

service fluorescent lamps, and to the measurement of lamp lumens, electrical characteristics for general service incandescent lamps and incandescent reflector lamps.

## 2. Definitions

2.1 To the extent that definitions in the referenced IESNA and CIE standards do not conflict with the DOE definitions, the definitions specified in § 1.2 of IESNA LM-9, § 3.0 of IESNA LM-20, § 1.2 and the Glossary of IESNA LM-45, § 2 of IESNA LM-58, and Appendix 1 of CIE Publication No. 13.3 shall be included.

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## 4. Test Methods and Measurements

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### 4.1 General Service Fluorescent Lamps

4.1.1 The measurement procedure shall be as described in IESNA LM-9, except that lamps shall be operated at the appropriate voltage and current conditions as described in ANSI C78.375 and in ANSI C78.81 or C78.901, and lamps shall be operated using the appropriate reference ballast at input voltage specified by the reference circuit as described in ANSI C82.3 (see 10 CFR 430.22). If, for a lamp, both low-frequency and high-frequency reference ballast settings are included in the ANSI standard, the lamp shall be operated using the low-frequency reference ballast.

4.1.2 For lamps not listed in ANSI C78.81 nor in C78.901, the lamp shall be operated using the following reference ballast settings:

4.1.2.1 4-Foot medium bi-pin lamps shall be operated using the following reference ballast settings: T10 or T12 lamps are to use 236 volts, 0.43 amps, and 439 ohms; T8 lamps are to use 300 volts, 0.265 amps, and 910 ohms.

4.1.2.2 2-Foot U-shaped lamps shall be operated using the following reference ballast settings: T12 lamps are to use 236 volts, 0.430 amps, and 439 ohms; T8 lamps are to use 300 volts, 0.265 amps, and 910 ohms.

4.1.2.3 8-Foot high output lamps shall be operated using the following reference ballast settings: T12 lamps are to use 400 volts, 0.800 amps, and 415 ohms; T8 lamps are to use 450 volts, 0.395 amps, and 595 ohms.

4.1.2.4 8-Foot slimline lamps shall be operated using the following reference ballast settings: T12 lamps are to use 625 volts, 0.425 amps, and 1280 ohms; T8 lamps are to use 625 volts, 0.260 amps, and 1960 ohms.

4.1.2.5 8-Foot very high output lamps shall be operated using the following reference ballast settings: T12 lamps are to use 400 volts, 1.500 amps, and 215 ohms.

4.1.2.6 Nominal 4-Foot T5 lamps shall be operated using the following reference ballast settings: Normal output lamps are to use 329 volts, 0.170 amps, and 950 ohms; high output lamps are to use 235 volts, 0.460 amps, and 255 ohms.

4.1.3 Lamp lumen output (lumens) and lamp electrical power input (watts), at the reference condition, shall be measured and recorded. Lamp efficacy shall be determined by computing the ratio of the measured lamp lumen output and lamp electrical power input at equilibrium for the reference condition.

### 4.2 General Service Incandescent Lamps

\* \* \* \* \*

4.2.2 The test procedure shall conform with sections 5 and 9 of IESNA LM-45 and the lumen output of the lamp shall be determined in accordance with section 9 of IESNA LM-45. Lamp electrical power input in watts shall be measured and recorded. Lamp efficacy shall be determined by computing the ratio of the measured lamp lumen output and lamp electrical power input at equilibrium for the reference condition. The test report shall conform to § 11 of IESNA LM-45 (see 10 CFR 430.22).

\* \* \* \* \*

### 4.4 Determination of Color Rendering Index and Correlated Color Temperature

4.4.1 The CRI shall be determined in accordance with the method specified in CIE Publication 13.3 for general service fluorescent lamps. The CCT shall be determined in accordance with the method specified in IESNA LM-9 for general service fluorescent lamps. The CCT shall be determined in accordance with the Journal of Optical Society of America, Vol. 58, pages 1528-1595 (1968) for incandescent lamps. The required spectroradiometric measurement and characterization shall be conducted in accordance with the methods set forth in IESNA LM-58 (see 10 CFR 430.22).

4.4.2 The test report shall include a description of the test conditions, equipment, measured lamps, spectroradiometric measurement results, and CRI and CCT determinations.

[FR Doc. E8-4035 Filed 3-12-08; 8:45 am]

BILLING CODE 6450-01-P

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA-2008-0273; Directorate Identifier 2007-NM-369-AD]

RIN 2120-AA64

#### Airworthiness Directives; Boeing Model 747-400, 747-400D, and 747-400F Series Airplanes

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** The FAA proposes to supersede an existing airworthiness directive (AD) that applies to all Boeing Model 747-400, 747-400D, and 747-400F series airplanes. The existing AD currently requires reviewing airplane maintenance records, doing repetitive inspections for cracking of the yaw damper actuator portion of the upper and lower rudder power control

modules (PCMs), replacing the PCMs if necessary, and reporting all airplane maintenance records review and inspection results to the manufacturer. This proposed AD would limit the applicability, reduce the initial inspection threshold and repetitive interval, remove the reporting requirement, and require installation of a secondary retention device for the yaw damper modulating piston. Installation of the secondary retention device would terminate the repetitive inspection requirements. This proposed AD results from additional reports of failure or cracking of the PCM manifold in the area of the yaw damper cavity endcap at intervals well below the initial inspection threshold of the existing AD. We are proposing this AD to prevent an uncommanded left rudder hardover in the event of cracking in the yaw damper actuator portion of the upper or lower rudder PCMs, and subsequent failure of the PCM manifold, which could result in increased pilot workload, and possible runway departure upon landing.

**DATES:** We must receive comments on this proposed AD by April 14, 2008.

**ADDRESSES:** You may send comments by any of the following methods:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.

- *Fax:* 202-493-2251.

- *Mail:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

- *Hand Delivery:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this AD, contact Boeing Commercial Airplanes, P.O. Box 3707, Seattle, Washington 98124-2207.

#### Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (telephone 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

**FOR FURTHER INFORMATION CONTACT:**

Douglas Tsuji, Aerospace Engineer, Systems and Equipment Branch, ANM-130S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98057-3356; telephone (425) 917-6487; fax (425) 917-6590.

**SUPPLEMENTARY INFORMATION:****Comments Invited**

We invite you to send any written relevant data, views, or arguments about this proposed AD. Send your comments to an address listed under the

**ADDRESSES** section. Include "Docket No. FAA-2008-0273; Directorate Identifier 2007-NM-369-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD because of those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

**Discussion**

On August 30, 2006, we issued AD 2006-18-17, amendment 39-14756 (71 FR 52999, September 8, 2006), for all Boeing Model 747-400, 747-400D, and 747-400F series airplanes. That AD requires reviewing airplane maintenance records, doing repetitive inspections for cracking of the yaw damper actuator portion of the upper and lower rudder power control modules (PCMs), replacing the PCMs if necessary, and reporting all airplane maintenance records review and inspection results to the manufacturer. That AD resulted from manufacturer findings that the inspections required by the existing AD (AD 2003-23-01, amendment 39-13364, which AD 2006-18-17 superseded) must be performed at regular intervals. We issued AD 2006-18-17 to detect and correct cracking in the yaw damper actuator portion of the upper and lower rudder PCMs, which could result in an uncommanded left rudder hardover, consequent increased pilot workload, and possible runway departure upon landing.

**Actions Since Existing AD Was Issued**

Since the issuance of AD 2006-18-17, there have been additional reports of cracked rudder PCMs in the yaw

damper portion of the PCM manifold that failed well below the initial inspection threshold of the AD. As part of the National Transportation Safety Board (NTSB) and FAA investigation, it was determined that a permanent design modification must be made to the rudder PCM that prevents a left rudder hardover in the event of a failed rudder PCM manifold.

The preamble to AD 2006-18-17 specified that we considered the requirements "interim action." That AD explained that we might consider further rulemaking since the root cause of the cracking had not been determined. Because of the additional premature failures of the rudder PCM, the manufacturer has now developed a final action, which is installation of a secondary means of retention for the modulating piston assembly in the event of a rudder PCM manifold failure. Therefore, we have determined that further rulemaking is indeed necessary; this proposed AD follows from that determination.

The inspection thresholds and intervals in AD 2006-18-17 correspond to those in Boeing Alert Service Bulletin 747-27A2397, Revision 2, dated September 1, 2005, which we referred to in AD 2006-18-17 as the appropriate source of service information for doing the initial and repetitive inspections. However, since we issued that AD, we have received additional reports of failure or cracking of the PCM manifold in the area of the yaw damper cavity endcap at intervals well below the initial inspection threshold of the existing AD. These incidents happened between 20,000 and 46,000 total flight hours. The initial flight-hour threshold in AD 2006-18-17 is 56,000 total flight hours, which corresponds to the threshold in Boeing Alert Service Bulletin 747-27A2397, Revision 2. Therefore, we are proposing that operators accomplish the ultrasonic inspections required by AD 2006-18-17 at reduced inspection threshold and intervals, until the final action is accomplished. Boeing has no objection to the reduced inspection thresholds and intervals.

**Relevant Service Information**

We have reviewed Boeing Alert Service Bulletin 747-27A2479, dated November 8, 2007. The service bulletin describes procedures for installing a new secondary retention device for the yaw damper piston assembly in both the upper and lower PCMs. The service bulletin specifies two options for

installing the new secondary retention device:

- Replacing the existing PCM with a new improved PCM.
- Modifying, testing, and re-identifying the existing PCM while installed on the airplane.

Accomplishing the actions specified in the service information is intended to adequately address the unsafe condition.

Boeing Alert Service Bulletin 747-27A2479 refers to Parker Service Bulletins 332700-27-312 and 333200-27-314, both dated September 13, 2007, as additional sources of service information for modifying the PCM.

**FAA's Determination and Requirements of the Proposed AD**

We have evaluated all pertinent information and identified an unsafe condition that is likely to develop on other airplanes of the same type design. For this reason, we are proposing this AD, which would supersede AD 2006-18-17, and retain all but the reporting requirement of that AD. This proposed AD would also limit the applicability, reduce the inspection threshold and repetitive interval, and require eventual installation of a secondary retention device for the yaw damper modulating piston. Installation of a secondary retention device would terminate the repetitive inspection requirements of the existing AD.

**Explanation of Reporting Requirement**

AD 2006-18-17 requires that operators report crack findings to Boeing and return cracked or broken PCMs to Parker Hannifin Corporation. This proposed AD would not include those actions. Boeing has now developed a final action, so the report and parts return are no longer necessary.

**Explanation of 30 Day Comment Period**

Operators should note that because of the critical need to prevent an uncommanded left rudder hardover, we have determined that a comment period of 30 days, rather than 45 days, is necessary in this case.

**Costs of Compliance**

There are about 655 airplanes of the affected design in the worldwide fleet. This proposed AD would affect about 86 airplanes of U.S. registry. The following table provides the estimated costs for U.S. operators to comply with this proposed AD. The average labor rate is \$80 per work hour.

## ESTIMATED COSTS

Action	Work hours	Parts	Cost per airplane	Fleet cost
Ultrasonic inspection (required by AD 2006–18–17) .....	4 .....	None .....	\$320, per inspection cycle.	\$27,520, per inspection cycle.
Option 1—replacement (new proposed action) .....	Up to 22 .....	Up to \$4,496 .....	Up to \$6,256 .....	Up to \$538,016.
Option 2—modification (new proposed action) .....	Up to 13 .....	Up to \$722 .....	Up to \$1,762 .....	Up to \$151,532.

**Authority for This Rulemaking**

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, Section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

**Regulatory Findings**

We have determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that the proposed regulation:

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
3. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this proposed AD and placed it in the AD docket. See the **ADDRESSES** section for a location to examine the regulatory evaluation.

**List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Safety.

**The Proposed Amendment**

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

**PART 39—AIRWORTHINESS DIRECTIVES**

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

**§ 39.13 [Amended]**

2. The Federal Aviation Administration (FAA) amends § 39.13 by removing amendment 39–14756 (71 FR 52999, September 8, 2006) and adding the following new airworthiness directive (AD):

**Boeing:** Docket No. FAA–2008–0273; Directorate Identifier 2007–NM–369–AD.

**Comments Due Date**

(a) The FAA must receive comments on this AD action by April 14, 2008.

**Affected ADs**

(b) This AD supersedes AD 2006–18–17.

**Applicability**

(c) This AD applies to Boeing Model 747–400, 747–400D, and 747–400F series airplanes, certificated in any category, as identified in Boeing Alert Service Bulletin 747–27A2479, dated November 8, 2007.

**Unsafe Condition**

(d) This AD results from additional reports of failure or cracking of the power control module (PCM) manifold in the area of the yaw damper cavity endcap at intervals well below the initial inspection threshold of the existing AD. We are issuing this AD to prevent an uncommanded left rudder hardover in the event of cracking in the yaw damper actuator portion of the upper or lower rudder PCM, and subsequent failure of the PCM manifold, which could result in increased pilot workload, and possible runway departure upon landing.

**Compliance**

(e) You are responsible for having the actions required by this AD performed within the compliance times specified, unless the actions have already been done.

**Verification of Rudder PCM/Main Manifold Time-in-Service**

(f) For any affected airplane, if it can be positively verified that any rudder PCM or PCM main manifold installed on that

airplane has accumulated a different total of flight hours or flight cycles than the totals accumulated by that airplane, the flight cycles or flight hours accumulated by the rudder PCM or PCM main manifold will be acceptable as valid starting points for meeting the compliance times required by this AD.

**Ultrasonic Inspections**

(g) Do an ultrasonic inspection for cracking of the yaw damper actuator portion of the upper and lower rudder PCMs at the applicable times specified in paragraph (g)(1) or (g)(2) of this AD in accordance with the Accomplishment Instructions of Boeing Service Bulletin 747–27A2397, Revision 2, dated September 1, 2005. Doing the installation required by paragraph (j) of this AD ends the inspection requirements of this paragraph for that PCM.

(1) For airplanes that have been inspected before the effective date of this AD in accordance with Boeing Alert Service Bulletin 747–27A2397, dated July 24, 2003; Revision 1, dated March 31, 2005; or Revision 2, dated September 1, 2005: Do the ultrasonic inspection at the later of the times specified in paragraph (g)(1)(i) and (g)(1)(ii) of this AD. Repeat the inspection thereafter at intervals not to exceed 7,000 flight hours or 1,125 flight cycles, whichever occurs first, until the action required by paragraph (j) of this AD is accomplished.

(i) Within 7,000 flight hours or 1,125 flight cycles after the prior inspection, whichever occurs first.

(ii) Within 6 months after the effective date of this AD.

(2) For airplanes that have not been inspected before the effective date of this AD in accordance with Boeing Alert Service Bulletin 747–27A2397, dated July 24, 2003; Revision 1, dated March 31, 2005; or Revision 2, dated September 1, 2005: Do the ultrasonic inspection at the later of the times specified in paragraph (g)(2)(i) and (g)(2)(ii) of this AD. Repeat the inspection thereafter at intervals not to exceed 7,000 flight hours or 1,125 flight cycles, whichever occurs first, until the action required by paragraph (j) of this AD is accomplished.

(i) Prior to the accumulation of 14,000 total flight hours or 2,250 total flight cycles, whichever occurs first.

(ii) Within 6 months after the effective date of this AD.

**Action if No Cracking Is Found**

(h) If no cracking is found during any inspection required by paragraph (g) of this AD: Before further flight, apply sealant and a torque stripe and install a lockwire on the rudder PCM in accordance with the Accomplishment Instructions, and Figure 1

or Figure 2, as applicable, of Boeing Service Bulletin 747-27A2397, Revision 2, dated September 1, 2005.

#### Action if Cracking Is Found

(i) If any cracking is found during any inspection required by paragraph (g) of this AD: Before further flight, do the action in paragraph (i)(1) or (i)(2) of this AD.

(1) Replace the affected PCM with a new or serviceable PCM in accordance with the Accomplishment Instructions of Boeing Service Bulletin 747-27A2397, Revision 2, dated September 1, 2005.

(2) Replace the PCM with a PCM that has the new secondary retention device installed as specified in paragraph (j) of this AD.

#### Terminating Action

(j) Within 24 months or 8,400 flight hours after the effective date of this AD, whichever occurs earlier: Install a new secondary retention device for the yaw damper piston assembly in both the upper and lower PCMs by either replacing the existing PCM with a new improved PCM that already has the new secondary retention device, or by modifying, testing, and re-identifying the existing PCM. Do the installation in accordance with the Accomplishment Instructions of Boeing Alert Service Bulletin 747-27A2479, dated November 8, 2007. Doing the installation terminates the inspection requirements of this AD.

**Note 1:** Boeing Alert Service Bulletin 747-27A2479 refers to Parker Service Bulletins 332700-27-312 and 333200-27-314, both dated September 13, 2007, as additional sources of service information for modifying the PCM.

#### Prior Accomplishment of Requirements

(k) Actions accomplished before October 13, 2006 (the effective date of AD 2006-18-17), in accordance with Boeing Alert Service Bulletin 747-27A2397, dated July 24, 2003; or Revision 1, dated March 31, 2005; are considered acceptable for compliance with the corresponding requirements of this AD.

#### Parts Installation

(l) As of October 13, 2006 no person may install on any airplane a rudder PCM having a top assembly part number (P/N) 332700-1003, -1005, or -1007; or P/N 333200-1003, -1005, or -1007; unless the PCM has been ultrasonically inspected and found to be without cracks; in accordance with the Accomplishment Instructions of Boeing Service Bulletin 747-27A2397, Revision 2, dated September 1, 2005, as specified by paragraph (g) of this AD.

#### Alternative Methods of Compliance (AMOCs)

(m)(1) The Manager, Seattle Aircraft Certification Office, FAA, has the authority to approve AMOCs for this AD, if requested in accordance with the procedures found in 14 CFR 39.19.

(2) To request a different method of compliance or a different compliance time for this AD, follow the procedures in 14 CFR 39.19. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector

(PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

(3) AMOCs approved previously in accordance with AD 2006-18-17 are approved as AMOCs for the corresponding provisions of paragraphs (g), (h), and (i) of this AD.

Issued in Renton, Washington, on March 4, 2008.

**Ali Bahrami,**

*Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. E8-5013 Filed 3-12-08; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA-2008-0274; Directorate Identifier 2008-NM-038-AD]

RIN 2120-AA64

#### Airworthiness Directives; Boeing Model 757 Airplanes, Model 767 Airplanes, and Model 777-200 and -300 Series Airplanes

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** We propose to adopt a new airworthiness directive (AD) for certain Boeing Model 757 airplanes, Model 767 airplanes, and Model 777-200 and -300 series airplanes. This proposed AD would require repetitive inspections for damage of the electrical terminal at the left and right flightdeck window # 1, and corrective actions if necessary. This proposed AD would also allow for replacing the flightdeck window # 1 with a new improved flightdeck window equipped with electrical connections, which would end the need for the repetitive inspections for that flightdeck window # 1. This proposed AD results from several reports of electrical arcs at the terminal blocks of the electrically heated flightdeck window # 1. In more than one of the incidents, the arcs resulted in open flames. We are proposing this AD to prevent smoke and fire in the cockpit, which could lead to loss of visibility, and injuries to or incapacitation of the flightcrew.

**DATES:** We must receive comments on this proposed AD by April 28, 2008.

**ADDRESSES:** You may send comments by any of the following methods:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.

- *Fax:* 202-493-2251.

- *Mail:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

- *Hand Delivery:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this AD, contact Boeing Commercial Airplanes, P.O. Box 3707, Seattle, Washington 98124-2207.

#### Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (telephone 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

#### FOR FURTHER INFORMATION CONTACT:

Louis Natsiopolous, Aerospace Engineer, Systems and Equipment Branch, ANM-130S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98057-3356; telephone (425) 917-6478; fax (425) 917-6590.

#### SUPPLEMENTARY INFORMATION:

##### Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposed AD. Send your comments to an address listed under the **ADDRESSES** section. Include "Docket No. FAA-2008-0274; Directorate Identifier 2008-NM-038-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD because of those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.