

Proposed Rules

Federal Register

Vol. 68, No. 218

Wednesday, November 12, 2003

This section of the FEDERAL REGISTER contains notices to the public of the proposed issuance of rules and regulations. The purpose of these notices is to give interested persons an opportunity to participate in the rule making prior to the adoption of the final rules.

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2003-NE-46-AD]

RIN 2120-AA64

Airworthiness Directives; General Electric Company CF6-80C2 Series Turbofan Engines

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: The FAA proposes to adopt a new airworthiness directive (AD) for General Electric Company (GE) CF6-80C2 series turbofan engines. This proposed AD would require replacing certain high pressure turbine (HPT) stage 1 disks at or before reaching a new reduced life cycle limit. This proposed AD is prompted by an updated low-cycle-fatigue (LCF) analysis of the HPT stage 1 disk. We are proposing this AD to prevent LCF cracking and failure of the HPT stage 1 disk due to exceeding the life limit, which could result in an uncontained engine failure and damage to the airplane.

DATES: We must receive any comments on this proposed AD by January 12, 2004.

ADDRESSES: Use one of the following addresses to submit comments on this proposed AD:

- By mail: Federal Aviation Administration (FAA), New England Region, Office of the Regional Counsel, Attention: Rules Docket No. 2003-NE-46-AD, 12 New England Executive Park, Burlington, MA 01803-5299.
- By fax: (781) 238-7055.
- By e-mail: 9-ane-adcomment@faa.gov.

You may examine the AD docket, by appointment, at the FAA, New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA.

FOR FURTHER INFORMATION CONTACT: Karen Curtis, Aerospace Engineer, Aircraft Certification Office, FAA, Engine and Propeller Directorate, 12 New England Executive Office Park, Burlington, MA 01803; telephone (781) 238-7192; fax (781) 238-7199.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to submit any written relevant data, views, or arguments regarding this proposal. Send your comments to an address listed under **ADDRESSES**. Include "AD Docket No. 2003-NE-46-AD" in the subject line of your comments. If you want us to acknowledge receipt of your mailed comments, send us a self-addressed, stamped postcard with the docket number written on it; we will date-stamp your postcard and mail it back to you. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of the proposed AD. If a person contacts us verbally, and that contact relates to a substantive part of this proposed AD, we will summarize the contact and place the summary in the docket. We will consider all comments received by the closing date and may amend the proposed AD in light of those comments.

We are reviewing the writing style we currently use in regulatory documents. We are interested in your comments on whether the style of this document is clear, and your suggestions to improve the clarity of our communications that affect you. You can get more information about plain language at <http://www.faa.gov/language> and <http://www.plainlanguage.gov>.

Examining the AD Docket

You may examine the AD Docket (including any comments and service information), by appointment, between 8 a.m. and 4:30 p.m., Monday through Friday, except Federal holidays. See **ADDRESSES** for the location.

Discussion

GE has completed an updated LCF analysis for the CF6-80C2A5F, CF6-80C2B5F, CF6-80C2B7F, and CF6-80C2D1F HPT stage 1 disks, part numbers (P/Ns) 1531M84G10 and 1531M84G12, and has established a new reduced life cycle limit of 10,720 cycles-since-new (CSN) for these disks. In January 2003, the FAA became aware of

GE's in-process analysis and material testing of HPT stage 1 disks, P/Ns 1531M84G10 and 1531M84G12. The FAA approved temporary revisions (TRs) to Chapter 5, Life Limits, of the engine manual (EM), to incorporate revised life limits for these disks based on initial analytical results. The original life limit of 13,200 CSN for these disks was last published in EM GEK 92451, Revision 57, dated March 1, 2003. TRs 05-0093 and 05-0094, dated May 15, 2003, revised this life limit from 13,200 CSN to 9,000 CSN. The FAA chose to wait for final analytical results before taking action to mandate a lower life limit. This wait was possible due to the young age of the affected disks. The high-cycle disk has accumulated fewer than 7,500 CSN at this time, which is well below the interim limit of 9,000 CSN and the final mandated limit. The FAA now approves GE's final analytical results and the reduced life limit of 10,720 CSN. GE issued TRs 05-0096 and 05-0097 on June 19, 2003 to revise the life limits section of the EM for CF6-80C2A5F, CF6-80C2B5F, CF6-80C2B7F, and CF6-80C2D1F HPT stage 1 disks, P/Ns 1531M84G10 and 1531M84G12, to 10,720 CSN. Although interim publications of the EM showed lower life limits for this part, those limits were not mandated by an AD. Therefore, an AD is now required to mandate the approved 10,720 CSN life limit.

FAA's Determination and Requirements of the Proposed AD

We have evaluated all pertinent information and identified an unsafe condition that is likely to exist or develop on other products of this same type design. Therefore, we are proposing this AD, which would require replacing HPT stage 1 disks, P/Ns 1531M84G10 and 1531M84G12 at or before reaching a new reduced life cycle limit of 10,720 CSN.

Changes to 14 CFR Part 39—Effect on the Proposed AD

On July 10, 2002, we issued a new version of 14 CFR part 39 (67 FR 47998, July 22, 2002), which governs the FAA's AD system. This regulation now includes material that relates to altered products, special flight permits, and alternative methods of compliance. This material previously was included in each individual AD. Since this material

is included in 14 CFR part 39, we will not include it in future AD actions.

Costs of Compliance

There are about 526 CF6–80C2A5F, CF6–80C2B5F, CF6–80C2B7F, and CF6–80C2D1F turbofan engines of the affected design in the worldwide fleet. We estimate that 208 engines installed on airplanes of U.S. registry would be affected by this proposed AD. The proposed action does not impose any additional labor costs. The prorated cost of a new HPT stage 1 disk would cost approximately \$43,306 per engine. Based on these figures, and on the prorating for the usage of the HPT stage 1 disks, the cost of the proposed AD on U.S. operators is estimated to be \$9,007,648.

Regulatory Findings

We have determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that the proposed regulation:

1. Is not a “significant regulatory action” under Executive Order 12866;
2. Is not a “significant rule” under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
3. Would not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a summary of the costs to comply with this proposal and placed it in the AD Docket. You may get a copy of this summary by sending a request to us at the address listed under

ADDRESSES. Include “AD Docket No. 2003–NE–46–AD” in your request.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. The FAA amends § 39.13 by adding the following new airworthiness directive:

General Electric Company: Docket No. 2003–NE–46–AD.

Comments Due Date

(a) The Federal Aviation Administration (FAA) must receive comments on this airworthiness directive (AD) action by January 12, 2004.

Affected ADs

(b) None.

Applicability

(c) This AD applies to General Electric Company (GE) CF6–80C2A5F, CF6–80C2B5F, CF6–80C2B7F, and CF6–80C2D1F turbofan engines with high pressure turbine (HPT) stage 1 disks, part numbers (P/Ns) 1531M84G10 or 1531M84G12 installed. These engines are installed on, but not limited to, Airbus Industrie A300 and A330 series, Boeing 747 and 767 series, and McDonnell Douglas MD–11 airplanes.

Unsafe Condition

(d) This AD was prompted by an updated low-cycle-fatigue (LCF) analysis of the HPT stage 1 disk. We are issuing this AD to prevent LCF cracking and failure of the HPT stage 1 disk due to exceeding the life limit, which could result in an uncontained engine failure and damage to the airplane.

Compliance

(e) You are responsible for having the actions required by this AD performed within the compliance times specified unless the actions have already been done.

(f) Replace HPT stage 1 disks, P/Ns 1531M84G10 and 1531M84G12, at or before the disk accumulates 10,720 cycles-since-new (CSN).

(g) After the effective date of this AD, do not install any HPT stage 1 disk, P/N 1531M84G10 or 1531M84G12, that exceeds 10,720 CSN.

Alternative Methods of Compliance

(h) The Manager, Engine Certification Office, has the authority to approve alternative methods of compliance for this AD if requested using the procedures found in 14 CFR 39.19.

Material Incorporated by Reference

(i) None.

Related Information

(j) None.

Issued in Burlington, Massachusetts, on November 4, 2003.

Peter A. White,

Acting Manager, Engine and Propeller Directorate, Aircraft Certification Service.

[FR Doc. 03–28323 Filed 11–10–03; 8:45 am]

BILLING CODE 4910–13–P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2000–NM–65–AD]

RIN 2120–AA64

Airworthiness Directives; Cessna Model 500, 501, 550, and 551 Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Supplemental notice of proposed rulemaking; reopening of comment period.

SUMMARY: This document revises an earlier proposed airworthiness directive (AD); applicable to certain Cessna Model 500, 501, 550, and 551 airplanes; that would have required inspection of the piston housing for an “SB” impression stamp; a one-time inspection of the brake assembly to detect cracked or broken brake stator disks; and replacement of the brake assembly with a new or serviceable assembly, if necessary. This new action revises the proposed rule by eliminating the inspection of the brake assembly to determine if the letters “SB” have been impression-stamped on the piston housing, and, instead, requiring a one-time inspection of the brake stator disks to determine to what change level they have been modified (if any), and follow-on actions if necessary. This new proposed AD would also require that the existing markings on the piston housing of certain brake assemblies be eliminated. The actions specified by this new proposed AD are intended to prevent wheel lockups that may be caused by cracked or broken brake stator disks becoming jammed in the brake assembly and preventing rotation. Such jamming of the brake assembly may result in reduced directional control or braking performance during landing. This action is intended to address the identified unsafe condition.

DATES: Comments must be received by December 8, 2003.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM–114, Attention: Rules Docket No. 2000–NM–65–AD, 1601 Lind Avenue, SW., Renton, Washington 98055–4056. Comments may be inspected at this location between 9 a.m. and 3 p.m., Monday through Friday, except Federal holidays. Comments may be submitted via fax to (425) 227–1232. Comments may also be sent via the Internet using the following address: 9-anm-