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This section of the FEDERAL REGISTER contains regulatory documents having general applicability and legal effect, most of which are keyed to and codified in the Code of Federal Regulations, which is published under 50 titles pursuant to 44 U.S.C. 1510.

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SMALL BUSINESS ADMINISTRATION

13 CFR Parts 107 and 121

SBA Reinvestor (“Fund-of-Funds”) Small Business Investment Company (SBIC) License Educational Public Webinar

AGENCY: U.S. Small Business Administration (SBA).

ACTION: Notification of public webinar.

SUMMARY: The SBA is holding a webinar to educate the public on the new Reinvestor SBIC License introduced to the market as part of the *SBIC Investment Diversification and Growth Final Rule* that went into effect on August 17, 2023.

DATES: The public webinar will be held on Friday, March 22, 2024, from 1 p.m. to 2 p.m. Eastern Time.

ADDRESSES: Information about applying for and managing a Reinvestor (“fund-of-funds”) SBIC License. The Webinar will be live streamed on Microsoft Teams for the public.

FOR FURTHER INFORMATION CONTACT: The meeting will be live streamed to the public, and anyone wishing to attend or needing accommodations because of a disability can contact Gretchen Kittel, SBA, Office of Investment & Innovation (OII), (202) 578-5502, investinnovate@sba.gov.

SUPPLEMENTARY INFORMATION:

I. Background

On August 17, 2023, the U.S. Small Business Administration (“SBA”) implemented new regulations for the Small Business Investment Company (“SBIC”) program as part of the SBIC Investment Diversification and Growth rulemaking. The new regulations significantly reduce barriers to program participation for new SBIC fund managers and funds investing in underserved communities and geographies, capital intensive investments, and technologies critical to national security and economic

development. The proposed rule introduced two additional types of SBIC Licensees, Reinvestor SBICs and Accrual SBICs, to increase program investment diversification and equity-oriented financing for American small businesses and innovative startups. Reinvestor SBICs expand SBA’s network of emerging fund managers, micro-funds, and funds addressing underserved communities and geographies and undercapitalized industries.

II. Questions

For the public webinar, OII strongly encourages questions be submitted in advance by March 20, 2024. Individuals may email investinnovate@sba.gov with subject line—“[Name/Organization] Question for 03/22/24 Public Webinar.” During the live event, attendees will be in listen-only mode and may submit additional questions via the Q&A Chat feature.

III. Information on Service for Individuals With Disabilities

For information on services for individuals with disabilities or to request special assistance, contact Gretchen Kittel at the telephone number or email address indicated under the **FOR FURTHER INFORMATION CONTACT** section of this notice.

Bailey DeVries,

Associate Administrator, Office of Investment & Innovation, U.S. Small Business Administration.

[FR Doc. 2024-05266 Filed 3-12-24; 8:45 am]

BILLING CODE 8026-09-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 25

[Docket No. FAA-2023-2435; Special Conditions No. 25-862-SC]

Special Conditions: Gulfstream Aerospace Corporation Model GVIII-G700 and GVIII-G800 Series Airplanes; Dynamic Test Requirements for Single- and Multiple-Occupant Side-Facing Seats With or Without Airbag Systems

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final special conditions.

SUMMARY: These special conditions are issued for the Gulfstream Aerospace Corporation (Gulfstream) Model GVIII-G700 and GVIII-G800 series airplanes. These airplanes will have a novel or unusual design feature when compared to the state of technology envisioned in the airworthiness standards for transport category airplanes. This design feature is side-facing seats oriented in the aircraft with the occupant facing 90 degrees to the direction of aircraft travel. The applicable airworthiness regulations do not contain adequate or appropriate safety standards for this design feature. These special conditions contain the additional safety standards that the Administrator considers necessary to establish a level of safety equivalent to that established by the existing airworthiness standards.

DATES: Effective March 13, 2024.

FOR FURTHER INFORMATION CONTACT: Myra Kuck, Cabin Safety, AIR-624, Technical Policy Branch, Policy and Standards Division, Aircraft Certification Service, Federal Aviation Administration, Aircraft Certification Policy and Standards, 3960 Paramount Blvd., Suite 100, Lakewood, CA 90712; telephone and fax 405-666-1059; email Myra.J.Kuck@faa.gov.

SUPPLEMENTARY INFORMATION:

Background

On December 31, 2019, Gulfstream applied for an amendment to Type Certificate No. T00015AT to include the new Model GVIII-G700 and GVIII-G800 series airplanes. These airplanes, which will be derivatives of the Model GVI currently approved under Type Certificate No. T00015AT, are twin-engine, transport-category airplanes, with seating for 19 passengers, and a maximum take-off weight of 107,600 pounds (GVIII-G700) and 105,600 pounds (GVIII-G800).

Type Certification Basis

Under the provisions of title 14, Code of Federal Regulations (14 CFR) 21.101, Gulfstream must show that the Model GVIII-G700 and GVIII-G800 series airplanes meet the applicable provisions of the regulations listed in Type Certificate No. T00015AT, or the applicable regulations in effect on the date of application for the change, except for earlier amendments as agreed upon by the FAA.

If the Administrator finds that the applicable airworthiness regulations do not contain adequate or appropriate safety standards for the Gulfstream Model GVIII-G700 and GVIII-G800 series airplanes because of a novel or unusual design feature, special conditions are prescribed under the provisions of § 21.16.

Special conditions are initially applicable to the model for which they are issued. Should the type certificate for that model be amended later to include any other model that incorporates the same novel or unusual design feature, or should any other model already included on the same type certificate be modified to incorporate the same novel or unusual design feature, these special conditions would also apply to the other model under § 21.101.

In addition to the applicable airworthiness regulations and special conditions, the Gulfstream Model GVIII-G700 and GVIII-G800 series airplanes must comply with the exhaust-emission requirements of 14 CFR part 34, and the noise-certification requirements of 14 CFR part 36.

The FAA issues special conditions, as defined in 14 CFR 11.19, in accordance with 14 CFR 11.38, and they become part of the type certification basis under § 21.101.

Novel or Unusual Design Feature

The Gulfstream Model GVIII-G700 and GVIII-G800 airplanes will incorporate the following novel or unusual design feature:

Side-facing seats, oriented in the aircraft with the occupant facing 90 degrees to the direction of aircraft travel, with or without incorporation of an airbag systems or inflatables.

Discussion

On June 16, 1988, 14 CFR part 25 was amended to revise the emergency landing conditions that must be considered in the design of transport category airplanes. This amendment (25-64) revised the static load conditions in § 25.561 and added a new § 25.562 that required dynamic testing for all seats approved for occupancy during takeoff and landing. The intent of Amendment 25-64 was to provide an improved level of safety for occupants on transport category airplanes; however, because most seating on transport category airplanes is forward-facing, the pass/fail criteria developed in Amendment 25-64 focused primarily on these seats.

Prior to 2012, the FAA granted exemptions¹ for the multiple-place side-facing-seat installations because the existing test methods and acceptance criteria did not produce a level of safety equivalent to the level of safety provided for forward-and aft-facing seats. These exemptions were subject to many conditions that reflected the injury-evaluation criteria and mitigation strategies available at the time of the exemption issuance. The FAA also issued special conditions to address single-place side-facing seats because it determined, at the time, that those conditions provided the same level of safety as for forward- and aft-facing seats.

Due to the novelty of side-facing seats in transport category airplanes, acceptable safety measures for § 25.562 were unknown. The FAA conducted research to develop an acceptable method of compliance with §§ 25.562 and 25.785(b) for side-facing seat installations. That research has identified injury considerations and evaluation criteria in addition to those previously used to approve side-facing seats (see published report DOT/FAA/AR-09/41, July 2011²). One particular concern that was identified during the FAA's research program but not addressed in special conditions prior to 2012 was the significant leg injuries that can occur to occupants of both single- and multiple-place side-facing seats. Because this type of injury does not occur on forward- and aft-facing seats, the FAA determined that to achieve the level of safety envisioned in Amendment 25-64, additional requirements would be needed as compared to previously issued special conditions. Nonetheless, the research has now allowed the development of a single set of special conditions that is applicable to all fully side-facing seats.

On November 5, 2012, the FAA released PS-ANM-25-03-R1, "Technical Criteria for Approving Side-Facing Seats," to update existing FAA certification policy on §§ 25.562 and 25.785(a) and (b) at Amendment 25-64 for single- and multiple-place side-facing seats. This policy addressed both the technical criteria for approving side-facing seats and the implementation of those criteria. The FAA methodology detailed in PS-ANM-25-03-R1 has been used to develop these special conditions. Some of the conditions issued for previous exemptions are still relevant and are included in these

special conditions; however, others have been replaced by different criteria that reflect current research findings described above, as well as design features from the Gulfstream GVII model side-facing seat design.

The special conditions contain the additional safety standards that the Administrator considers necessary to establish a level of safety equivalent to that established by the existing airworthiness standards.

Discussion of Comments

The FAA issued Notice of Proposed Special Conditions No. 25-23-07-SC for the Gulfstream Model GVIII-G700 and GVIII-G800 airplanes, which was published in the **Federal Register** on February 1, 2024 (89 FR 6443).

The FAA received responses from two anonymous commenters. One commenter stated that they support the special conditions as proposed. The second commenter requested the FAA make the following changes to the proposed special conditions:

1. The commenter requested the FAA replace the term "airbag" with "automatically deploying safety system." The commenter stated that it should be clear that the installation of a different kind of automatically deploying safety system would necessitate the issuance of a new special condition. The FAA acknowledges that the current automatically-deploying safety systems proposed by applicants are airbag systems. If in the future the technology proposed by applicants should change, then there may be need for another special condition. No changes were made to these special conditions as a result of this comment.

2. The commenter stated that the proposed special conditions paragraph 1.e(1)(b) may cause confusion, in that the word "bottom" is singular and the word "feet" is plural. This wording, according to the commenter, could lead to an interpretation that the force of about 20 pounds (lbs) may be applied to each foot, plus it is not specified that the force be applied uniformly. The commenter also suggested a specific revision to the text of this paragraph to address these concerns. The FAA disagrees that there is ambiguity or that a change is necessary. The 20 lbs of force is the total force applied to the bottom of the feet. The text of the special condition is similar to the language of FAA Policy Statement PS-ANM-25-03-R1 "Technical Criteria for Approving Side-Facing Seats." No changes were made to these special conditions as a result of this comment.

3. The commenter suggested several changes to the formatting and technical

¹ See, generally, Exemption Nos. 7120C, 7878A, and 9900.

² Document available at <https://www.tc.faa.gov/its/worldpac/techrpt/ar09-41.pdf>.

content of paragraph 2.g. of the special conditions. The commenter suggested adding a colon at the end of paragraph g., and that subsequent special conditions should be numbered below that paragraph. The commenter also stated that paragraph 2.g. does not explicitly state when lap belt tension must be limited to 250 lbs, and that the rationale for the limit in the Civil Aerospace Medical Institute (CAMI) report does not have to be specified in the proposed special conditions. The commenter further stated that paragraph 2.g. has a typographical error in that data should be filtered at "CFC 600" as defined in SAE JS211-1 "Surface Vehicle Recommended Practice." The FAA agrees with most of the comments received on paragraph 2.g. The FAA has revised the formatting of paragraph 2.g. to reflect the requirements clearly. The FAA does not concur with the commenter that the data should be filtered at CFC 600 versus 60. Sixty is the correct value for belt loads in SAE JS211-1.

4. The commenter made two comments regarding paragraph 4.a. of these special conditions. The commenter suggested the FAA revise the phrase "that range of occupants" because it is missing explanation as to which range of occupants is being referred to. The FAA disagrees. The range of occupants is provided in Paragraph 3.b. Paragraph 3. states that "For all airbag systems in the shoulder harness and for leg flail the following apply" . . . Paragraph b. states that the means of protection must take into consideration a range of stature from a 2-year-old child to a 95th percentile male.

The commenter further stated that in paragraph 4.a., the situations that must be considered do not account for the possibility that the seat occupant is a child in a child restraint device or booster seat. The FAA disagrees. Booster seats are not allowed. If a child restraint device is installed, the installation must show it would not harm the occupant, otherwise an operating limitation would need to limit to no child restraint device per condition 4.b. No changes were made to these special conditions as a result of this comment.

Except as discussed above, the special conditions are adopted as proposed.

Applicability

As discussed above, these special conditions are applicable to the Gulfstream Model GVIII-G700 and GVIII-G800 series airplanes. Should Gulfstream apply at a later date for a change to the type certificate to include another model that incorporates the

same novel or unusual design feature, or should any other model already included on the same type certificate be modified to incorporate the same novel or unusual design feature, these special conditions would apply to the other model as well.

Under standard practice, the effective date of final special conditions would be 30 days after the date of publication in the **Federal Register**. However, as the certification date for the Gulfstream Model GVIII-G700 and GVIII-G800 series airplanes is imminent, the FAA finds that good cause exists to make these special conditions effective upon publication.

Conclusion

This action affects only a certain novel or unusual design feature on Gulfstream Model GVIII-G700 and GVIII-G800 series of airplanes. It is not a rule of general applicability.

List of Subjects in 14 CFR Part 25

Aircraft, Aviation safety, Reporting and recordkeeping requirements.

Authority Citation

The authority citation for these special conditions is as follows:

Authority: 49 U.S.C. 106(f), 106(g), 40113, 44701, 44702, 44704.

The Special Conditions

Accordingly, pursuant to the authority delegated to me by the Administrator, the following special conditions are issued as part of the type certification basis for Gulfstream Model GVIII-G700 and GVIII-G800 series airplanes.

In addition to the airworthiness standards in §§ 25.562 and 25.785, the FAA issues the following special conditions as part of the type certification basis for the Gulfstream Model GVIII-G700 and GVIII-G800 series aircraft. Items 1 through 3 are applicable to all side-facing seat installations on these airplanes. Item 4 imposes additional requirements applicable to side-facing seats equipped with an airbag system in the shoulder belt. Item 5 imposes additional requirements applicable to side-facing seats equipped with leg-flail airbag systems.

1. Additional requirements applicable to tests or rational analysis conducted to show compliance with §§ 25.562 and 25.785 for side-facing seats:

a. The longitudinal test(s) conducted in accordance with § 25.562(b)(2) to show compliance with the seat-strength requirements of § 25.562(c)(7) and (8), and these special conditions must have an ES-2re anthropomorphic test dummy

(ATD) (49 CFR part 572 subpart U) or equivalent, or a Hybrid-II ATD (49 CFR part 572, subpart B as specified in § 25.562) or equivalent, occupying each seat position and including all items contactable by the occupant (*e.g.*, armrest, interior wall, or furnishing) if those items are necessary to restrain the occupant. If included, the floor representation and contactable items must be located such that their relative position, with respect to the center of the nearest seat place, is the same at the start of the test as before floor misalignment is applied. For example, if floor misalignment rotates the centerline of the seat place nearest the contactable item 8 degrees clockwise about the aircraft x-axis, then the item and floor representations must be rotated by 8 degrees clockwise also to maintain the same relative position to the seat place, as shown in Figure 1. Each ATD's relative position to the seat after application of floor misalignment must be the same as before misalignment is applied. The ATD pelvis must remain supported by the seat pan, and the restraint system must remain on the pelvis and shoulder of the ATD until rebound begins. No injury-criteria evaluation is necessary for tests conducted only to assess seat-strength requirements.

b. The longitudinal test(s) conducted in accordance with § 25.562(b)(2), to show compliance with the injury assessments required by § 25.562(c) and these special conditions, may be conducted separately from the test(s) to show structural integrity. Structural-assessment tests must be conducted as specified in paragraph 1.a., above, and the injury-assessment test must be conducted without yaw or floor misalignment. Injury assessments may be accomplished by testing with ES-2re ATD (49 CFR part 572 subpart U) or equivalent at all places. Alternatively, these assessments may be accomplished by multiple tests that use an ES-2re at the seat place being evaluated, and a Hybrid-II ATD (49 CFR part 572, subpart B, as specified in § 25.562) or equivalent used in all seat places forward of the one being assessed, to evaluate occupant interaction. Seat places aft of the one being assessed may be unoccupied. If a seat installation includes adjacent items that are contactable by the occupant, the injury potential of that contact must be assessed. To make this assessment, tests may be conducted that include the actual item, located, and attached in a representative fashion. Alternatively, the injury potential may be assessed by a combination of tests with items having

the same geometry as the actual item, but having stiffness characteristics that would create the worst case for injury (injuries due to both contact with the item and lack of support from the item).

c. If a seat is installed aft of structure (e.g., an interior wall or furnishing) that does not have a homogeneous surface contactable by the occupant, additional analysis and/or test(s) may be required to demonstrate that the injury criteria are met for the area which an occupant could contact. For example, different yaw angles could result in different injury considerations and may require additional analysis or separate test(s) to evaluate.

d. To accommodate a range of occupant heights (5th percentile female to 95th percentile male), the surface of items contactable by the occupant must be homogenous 7.3 inches (185 mm) above and 7.9 inches (200 mm) below the point (center of area) that is contacted by the 50th percentile male size ATD's head during the longitudinal test(s) conducted in accordance with paragraphs a, b, and c, above.

Otherwise, additional head-injury criteria (HIC) assessment tests may be necessary. Any surface (inflatable or otherwise) that provides support for the occupant of any seat place must provide that support in a consistent manner regardless of occupant stature. For example, if an inflatable shoulder belt is used to mitigate injury risk, then it must be demonstrated by inspection to bear against the range of occupants in a similar manner before and after inflation. Likewise, the means of limiting lower-leg flail must be demonstrated by inspection to provide protection for the range of occupants in a similar manner.

e. For longitudinal test(s) conducted in accordance with § 25.562(b)(2) and these special conditions, the ATDs must be positioned, clothed, and have lateral instrumentation configured as follows:

(1) *ATD positioning:*

Lower the ATD vertically into the seat while simultaneously (see Figure 2 for illustration):

(a) Aligning the midsagittal plane (a vertical plane through the midline of the body; dividing the body into right and left halves) with approximately the middle of the seat place.

(b) Applying a horizontal x-axis direction (in the ATD coordinate system) force of about 20 pounds (lbs) (89 Newtons [N]) to the bottom of the feet of the ATD with the legs straight, to compress the seat back cushion.

(c) Keeping the legs nearly horizontal by supporting them just behind the ankles.

(d) Once all lifting devices have been removed from the ATD:

(i) Rock it slightly to settle it in the seat.

(ii) Gently lower the ankles of the ATD bending the legs at the knee joints. Do not allow the pelvis of the ATD to be moved when the lower legs are lowered. The seat back cushion must remain compressed. Separate the knees by about 4 inches (100 mm).

(iii) Set the ES-2re's head at approximately the midpoint of the available range of z-axis rotation (to align the head and torso midsagittal planes).

(iv) Position the ES-2re's arms at the joint's mechanical detent that puts them at approximately a 40-degree angle with respect to the torso. Position the Hybrid II ATD hands on top of its upper legs.

(v) Position the feet such that the centerlines of the lower legs are approximately parallel to a lateral vertical plane (in the aircraft coordinate system).

(2) *ATD clothing:* Clothe each ATD in form-fitting, mid-calf-length (minimum) pants and shoes (size 11E) weighing about 2.5 lb (1.1 Kg) total. The color of the clothing should be in contrast to the color of the restraint system. The ES-2re jacket is sufficient for torso clothing, although a form-fitting shirt may be used in addition if desired.

(3) *ES-2re ATD lateral instrumentation:* The rib-module linear slides are directional, i.e., deflection occurs in either a positive or negative ATD y-axis direction. The modules must be installed such that the moving end of the rib module is toward the front of the aircraft. The three abdominal-force sensors must be installed such that they are on the side of the ATD toward the front of the aircraft.

f. The combined horizontal/vertical test, required by § 25.562(b)(1) and these special conditions, must be conducted with a Hybrid II ATD (49 CFR part 572 subpart B as specified in § 25.562), or equivalent, occupying each seat position.

g. *Restraint systems:*

(1) If inflatable restraint systems are used, they must be active during all dynamic tests conducted to show compliance with § 25.562.

(2) The design and installation of seat-belt buckles must prevent unbuckling due to applied inertial forces or impact of the hands/arms of the occupant during an emergency landing.

2. Additional performance measures applicable to tests and rational analysis conducted to show compliance with §§ 25.562 and 25.785 for side-facing seats:

a. *Body-to-body contact:* Contact between the head, pelvis, torso, or shoulder area of one ATD with the adjacent-seated ATD's head, pelvis, torso, or shoulder area is not allowed. Contact during rebound is allowed.

b. *Thoracic:* The deflection of any of the ES-2re ATD upper, middle, and lower ribs must not exceed 1.73 inches (44 mm). Data must be processed as defined in Federal Motor Vehicle Safety Standards (FMVSS) 571.214.

c. *Abdominal:* The sum of the measured ES-2re ATD front, middle, and rear abdominal forces must not exceed 562 lbs (2,500 N). Data must be processed as defined in FMVSS 571.214.

d. *Pelvic:* The pubic symphysis force measured by the ES-2re ATD must not exceed 1,350 lbs (6,000 N). Data must be processed as defined in FMVSS 571.214.

e. *Leg:* Axial rotation of the upper-leg (femur) must be limited to 35 degrees in either direction from the nominal seated position.

f. *Neck:* As measured by the ES-2re ATD and filtered at CFC 600 as defined in SAE J211:

(1) The upper-neck tension force at the occipital condyle (O.C.) location must be less than 405 lb (1,800 N).

(2) The upper-neck compression force at the O.C. location must be less than 405 lb (1,800 N).

(3) The upper-neck bending torque about the ATD x-axis at the O.C. location must be less than 1,018 in-lb (115 Nm).

(4) The upper-neck resultant shear force at the O.C. location must be less than 186 lb (825 N).

g. *Occupant (ES-2re ATD) retention:* The upper-torso restraint straps (if present) must remain on the ATD's shoulder during the impact. The pelvic restraint must remain on the ES-2re ATD's pelvis during the impact. The pelvic restraint must remain on the ES-2re ATD's pelvis during rebound unless the following criteria are met.

(1) A measurement of the belt loop load during the time when the belt moves above the pelvis (submarining) must not exceed 500 lbs (2,225 N) (*a 250 lb (1112.5 N) lap belt tension limit*). Data must be filtered at CFC 60 as defined in SAE J211. To evaluate the pelvic restraint performance using this criterion, three things are needed:

(a) A clear indication of when the belt moves above the pelvis. Loose clothing can make it difficult to determine where the top of the pelvis is, and in turn make it hard to discern exactly when the belt moved above it. This can be improved by marking the top of the pelvis clearly and by positioning the cameras so that

the position of the belt, relative to the top of the pelvis can be observed throughout the test (see Figure 3).

(b) A measurement of the belt tension during the time when the belt moves above the pelvis. Place the webbing transducer to measure the total tension in the forward lap belt segment. If a split (combined body-centered and conventional) leading belt is used, measure the tension in the common section so that it reflects the contribution of each segment. Since this placement typically produces contact between the ATD and the transducer, it is important to use a webbing transducer that is not sensitive to contact.

(c) Record useful video and belt load data until significant ATD rebound motion stops. Extra recording time is necessary because submarining usually occurs later in the test than other injury criteria maximums. To completely capture ATD rebound, the necessary time could exceed 500 ms.

h. *Occupant (ES-2re ATD) support:*

(1) *Pelvis excursion:* The load-bearing portion of the bottom of the ATD pelvis must not translate beyond the edges of its seat's bottom seat-cushion supporting structure.

(2) *Upper-torso support:* The lateral flexion of the ATD torso must not exceed 40 degrees from the normal upright position during the impact.

3. For all airbag systems in the shoulder harness and for leg flail, the following apply:

a. Show that the airbag system will deploy and provide protection under crash conditions where it is necessary to prevent serious injury.

b. The means of protection must take into consideration a range of stature from a 2-year-old child to a 95th percentile male.

c. The airbag system must provide adequate protection for each occupant regardless of the number of occupants of the seat assembly, considering that unoccupied seats may have an active airbag system.

d. It must be shown that the airbag system is not susceptible to inadvertent deployment as a result of wear and tear, or inertial loads resulting from in-flight or ground maneuvers (including gusts and hard landings), and other operating and environmental conditions (vibrations, moisture, etc.) likely to occur in service.

e. Deployment of the airbag system must not introduce injury mechanisms

to the seated occupant, or result in injuries that could impede rapid egress. This assessment should include an occupant whose seat belt is loosely fastened.

f. It must be shown that inadvertent deployment of the airbag system, during the most critical part of the flight, will either meet the requirement of § 25.1309(b) or not cause a hazard to the airplane or its occupants.

g. It must be shown that the airbag system will not impede rapid egress of occupants 10 seconds after airbag deployment.

h. The airbag system must be protected from lightning and high-intensity radiated fields (HIRF). The threats to the airplane specified in existing regulations regarding lightning, § 25.1316, and HIRF, § 25.1317, are adopted by reference for the purpose of measuring lightning and HIRF protection.

i. The airbag system must function properly after loss of normal aircraft electrical power, and after a transverse separation of the fuselage at the most critical location. A separation at the location of the airbag system does not have to be considered.

j. It must be shown that the airbag system will not release hazardous quantities of gas or particulate matter into the cabin.

k. The airbag system installation must be protected from the effects of fire such that no hazard to occupants will result.

l. A means must be available for a crewmember to verify the integrity of the airbag system prior to each flight, or it must be demonstrated to reliably operate between inspection intervals. The FAA considers that the loss of the airbag-system deployment function alone (*i.e.*, independent of the conditional event that requires the airbag-system deployment) is a major-failure condition.

m. The inflatable material may not have an average burn rate of greater than 2.5 inches/minute when tested using the horizontal flammability test defined in part 25, appendix F, part I, paragraph (b)(5).

n. The airbag system, once deployed, must not adversely affect the emergency-lighting system (*i.e.*, block floor proximity lights to the extent that the lights no longer meet their intended function).

o. The airbag system must perform its intended function after impact from

other proximate assemblies (*e.g.*, life raft) that may become detached under the loads specified in §§ 25.561 and 25.562.

4. For seats with an airbag system in the shoulder belts, the following apply:

a. The airbag system in the shoulder belt must provide a consistent approach to energy absorption throughout that range of occupants. The airbag system must be included in each of the certification tests as it would be installed in the airplane. In addition, the following situations must be considered:

(1) The seat occupant is holding an infant.

(2) The seat occupant is a pregnant woman.

b. The design must prevent the airbag system in the shoulder belt from being either incorrectly buckled or incorrectly installed, such that the airbag system in the shoulder belt would not properly deploy. Alternatively, it must be shown that such deployment is not hazardous to the occupant, and will provide the required injury protection.

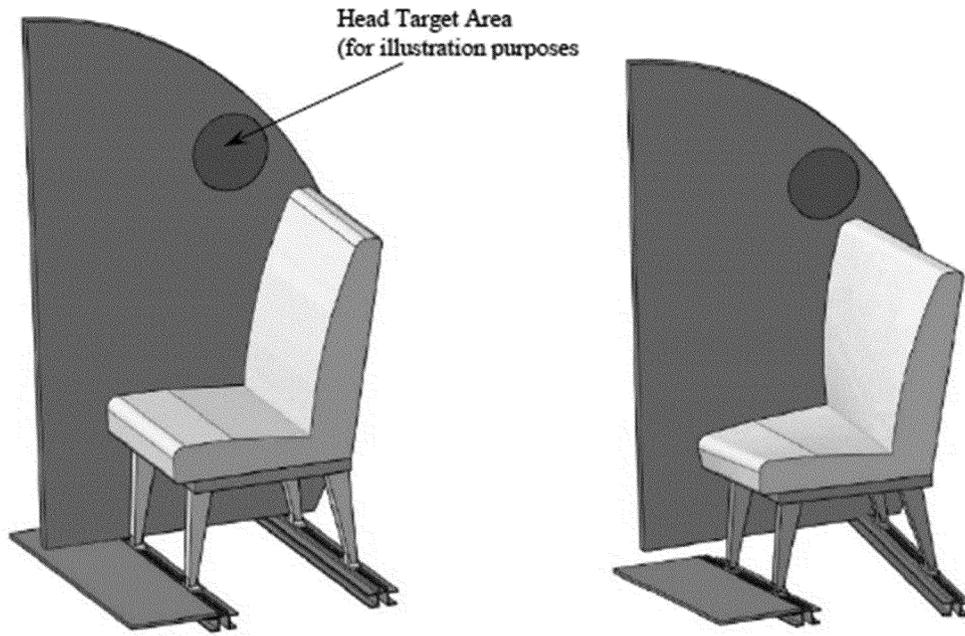
5. For seats using an airbag system to meet the leg-flail conditions of 2.e. the following apply:

a. At some buttock popliteal length and effective seat bottom depth the lower legs will not be able to make a 90-degree angle with the upper leg; at this point the lower leg flail would not occur. The leg flail airbag system must provide a consistent approach to prevention of leg flail throughout that range of occupants whose lower legs can make a 90-degree angle with the upper legs when seated upright in the seat. Items that need to be considered include, but are not limited to the range of occupants' popliteal height, the range of occupants' buttock popliteal length, the design of the seat effective height above the floor, and the effective depth of the seat bottom cushion.

b. For all g-levels, if the design of the leg flail limited device does absorb some of the impact energy and returns only a portion to the legs (a qualitative assessment), then a rebound leg flail of greater than 35 degrees is acceptable.

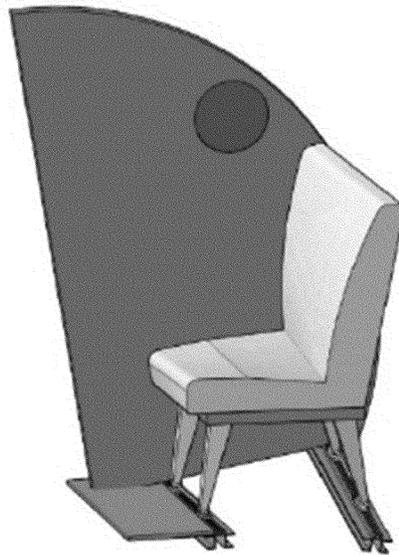
c. Threshold test severity must be shown to be non-injurious (less than the post-mortem human subject (PMHS) low-g research testing) for g-levels up to the point where the leg flail airbag is designed to deploy.

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A. Prior to Test Setup

B. Inboard Seat Tracks Twisted



C. Partition Rotated to maintain

Figure 1

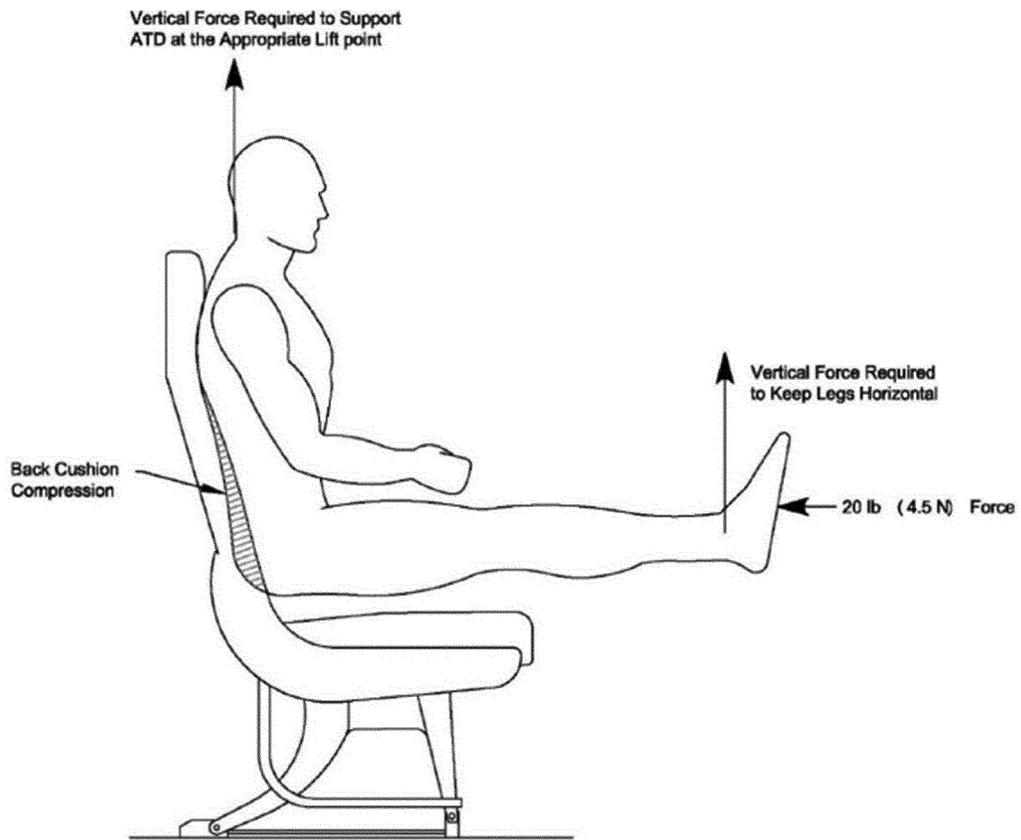


Figure 2

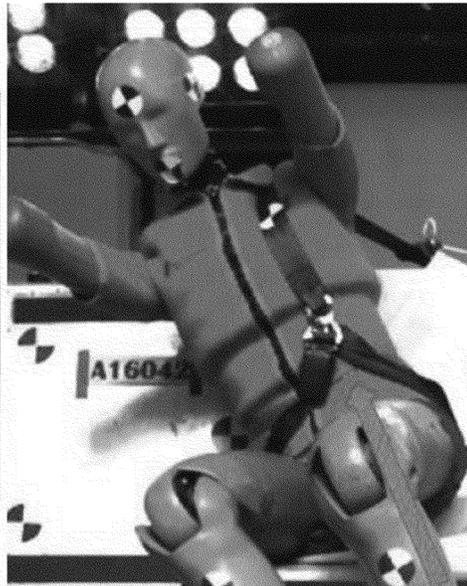
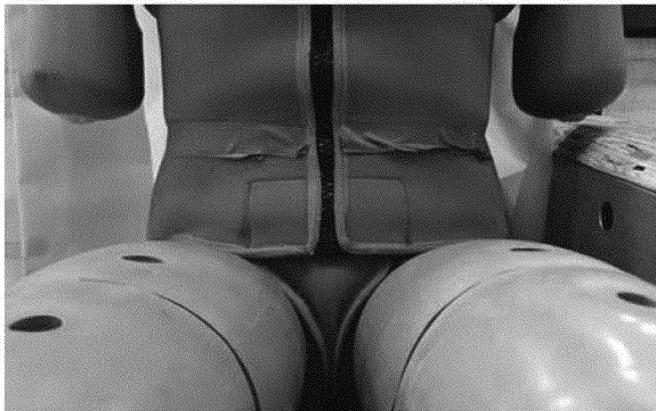


Figure 3

Issued in Kansas City, Missouri, on March 6, 2024.

James David Foltz,

Manager, Technical Policy Branch, Policy and Standards Division, Aircraft Certification Service.

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2023-2149; Project Identifier MCAI-2023-00136-E; Amendment 39-22675; AD 2024-03-05]

RIN 2120-AA64

Airworthiness Directives; GE Aviation Czech s.r.o. (Type Certificate Previously Held by WALTER Engines a.s., Walter a.s., and MOTORLET a.s.) Engines

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: The FAA is superseding Airworthiness Directive (AD) 2022-13-16 for all GE Aviation Czech s.r.o. (GEAC) (type certificate previously held by WALTER Engines a.s., Walter a.s., and MOTORLET a.s.) Model M601D-11 engines; and AD 2022-14-12, for certain GEAC Model M601D-11, M601E-11, M601E-11A, M601E-11AS, M601E-11S, and M601F engines. AD 2022-13-16 required revising the airworthiness limitations section (ALS) of the existing engine maintenance manual (EMM) to incorporate a visual inspection of the centrifugal compressor case for cracks. AD 2022-14-12 required replacing the propeller shaft for Model M601F engines. AD 2022-14-12 also required calculating the accumulated life of the propeller shaft and replacing the propeller shaft, if necessary, for model M601D-11, M601E-11, M601E-11A, M601E-11AS, and M601E-11S engines. Since the FAA issued AD 2022-13-16 and AD 2022-14-12, the manufacturer revised the ALS of the existing EMM to introduce new and more restrictive tasks and limitations, expand the applicability to all Model M601 engines, and incorporate certain requirements addressed by AD 2021-13-07 and AD 2023-01-10, which prompted this AD. This AD requires revising the ALS of the existing EMM and the operator's existing approved engine maintenance or inspection program, as applicable, to incorporate new and more restrictive tasks and limitations, as specified in a European Union Aviation Safety Agency

(EASA) AD, which is incorporated by reference. The FAA is issuing this AD to address the unsafe condition on these products.

DATES: This AD is effective April 17, 2024.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in this AD as of April 17, 2024.

ADDRESSES:

AD Docket: You may examine the AD docket at *regulations.gov* under Docket No. FAA-2023-2149; or in person at Docket Operations between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this final rule, the mandatory continuing airworthiness information (MCAI), any comments received, and other information. The address for Docket Operations is U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE, Washington, DC 20590.

Material Incorporated by Reference:

- For service information identified in this final rule, contact EASA, Konrad-Adenauer-Ufer 3, 50668 Cologne, Germany; phone: +49 221 8999 000; email: ADs@easa.europa.eu; website: easa.europa.eu. You may find this material on the EASA website at ad.easa.europa.eu.

- You may view this service information at the FAA, Airworthiness Products Section, Operational Safety Branch, 1200 District Avenue, Burlington, MA 01803. For information on the availability of this material at the FAA, call (817) 222-5110. It is also available at *regulations.gov* under Docket No. FAA-2023-2149.

FOR FURTHER INFORMATION CONTACT:

Barbara Caufield, Aviation Safety Engineer, FAA, 1600 Stewart Avenue, Suite 410, Westbury, NY 11590; phone: (781) 238-7146; email: barbara.caufield@faa.gov.

SUPPLEMENTARY INFORMATION:

Background

The FAA issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to supersede AD 2022-13-16, Amendment 39-22102 (87 FR 37986, June 27, 2022) (AD 2022-13-16); and AD 2022-14-12, Amendment 39-22117 (87 FR 42066, July 14, 2022) (AD 2022-14-12).

AD 2022-13-16 applied to all GEAC Model M601D-11 engines and required revising the ALS of the existing EMM to incorporate a visual inspection of the centrifugal compressor case. The FAA issued AD 2022-13-16 to prevent

failure of the centrifugal compressor case.

AD 2022-14-12 applied to certain GEAC Model M601D-11, M601E-11, M601E-11A, M601E-11AS, M601E-11S, and M601F engines. For Model M601F engines, AD 2022-14-12 required replacement of the propeller shaft. For Model M601D-11, M601E-11, M601E-11A, M601E-11AS, and M601E-11S engines, AD 2022-14-12 required calculating the accumulated life of the propeller shaft and replacing the propeller shaft if necessary.

The NPRM published in the **Federal Register** on November 14, 2023 (88 FR 77918). The NPRM was prompted by EASA AD 2023-0020, dated January 23, 2023 (EASA AD 2023-0020) (also referred to as the MCAI), issued by EASA, which is the Technical Agent for the Member States of the European Union. The MCAI states that the manufacturer revised the ALS to incorporate new and more restrictive tasks and limitations, expand the applicability to all model M601 series engines, and include certain requirements that were previously addressed by EASA Emergency AD 2021-0125-E and EASA AD 2021-0264. The MCAI also states that the manufacturer published service information that specifies instructions to determine the accumulated life of certain propeller shafts.

You may examine the MCAI in the AD docket at *regulations.gov* under Docket No. FAA-2023-2149.

In the NPRM, the FAA proposed to require revising the ALS of the existing EMM and the operator's existing approved engine maintenance or inspection program, as applicable, to incorporate new and more restrictive tasks and limitations.

Discussion of Final Airworthiness Directive

Comments

The FAA received no comments on the NPRM or on the determination of the costs.

Conclusion

These products have been approved by the aviation authority of another country and are approved for operation in the United States. Pursuant to the FAA's bilateral agreement with this State of Design Authority, it has notified the FAA of the unsafe condition described in the MCAI referenced above. The FAA reviewed the relevant data and determined that air safety requires adopting this AD as proposed. Accordingly, the FAA is issuing this AD to address the unsafe condition on these