

2. Will not affect intrastate aviation in Alaska, and

3. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

The FAA prepared an economic evaluation of the estimated costs to comply with this proposed AD and placed it in the AD docket.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

#### The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

#### PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

##### § 39.13 [Amended]

■ 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

**Airbus Helicopters:** Docket No. FAA-2019-0882; Product Identifier 2018-SW-113-AD.

##### (a) Applicability

This AD applies to Airbus Helicopters Model AS332C, AS332C1, AS332L, and AS332L1 helicopters, certificated in any category, delivered to the first owner or customer before September 1, 2018, and with attachment screws part number (P/N) 330A22013520 installed with main gearbox (MGB) right hand (RH) side rear attachment fitting P/N 330A22270207 and left hand (LH) side rear attachment fitting P/N 330A22270206 of the MGB suspension bars.

##### (b) Unsafe Condition

This AD defines the unsafe condition as elongation of the attachment screws and loss of tightening torque of the nut. This condition could result in structural failure of an MGB attachment fitting, detachment of an MGB suspension bar, and subsequent loss of control of the helicopter.

##### (c) Comments Due Date

The FAA must receive comments by January 7, 2020.

##### (d) Compliance

You are responsible for performing each action required by this AD within the specified compliance time unless it has already been accomplished prior to that time.

##### (e) Required Actions

Within 110 hours time-in-service, remove the sealing compound and inspect each

screw on the RH and LH rear attachment fitting by identifying the number of threads “F” that extend beyond the nut as shown in Detail “B” of Figure 2 of Airbus Helicopter Alert Service Bulletin No. AS332-53.02.04, Revision 0, dated November 21, 2018 (ASB AS332-53.02.04).

(1) If there are 2 or less threads on each of the four screws; or there are 3 or more threads on any screw with a thread height “H” less than 5 mm (0.196 in), before further flight, apply a sealing compound on the nuts, and convex and concave washers.

(2) If there are 3 or more threads on any screw with a thread height “H” of 5 mm (0.196 in) or more, before further flight, do the following, and for more than one screw, do one at a time while working in a cross pattern: Remove from service the nut; and remove the screw from the helicopter and measure the length “L” of the screw as shown in Detail “D” of Figure 2 of ASB AS332-53.02.04.

(i) If any washers are bent or corroded, before further flight, remove from service the washers.

(ii) If the length “L” measurement is less than or equal to 59.3 mm (2.334 in) for each screw removed as required by paragraph (e)(2) of this AD, visually inspect the screw for corrosion and cracks.

(A) For each screw with corrosion or a crack, before further flight, replace the screw with an airworthy screw.

(B) For any screw with no corrosion or cracks, before further flight, re-install the screw and washers. Install a new nut and apply sealant.

(iii) If the length “L” measurement is greater than 59.3 mm (2.334 in) for any screw removed as required by paragraph (e)(2) of this AD, before further flight, replace the rear attachment fitting that the screw was removed from and its set of four screws, washers, and nuts, and apply sealant as shown in Figures 2 and 3 of ASB AS332-53.02.04.

##### (f) Alternative Methods of Compliance (AMOCs)

(1) The Manager, Safety Management Section, Rotorcraft Standards Branch, FAA, may approve AMOCs for this AD. Send your proposal to: Matt Fuller, Senior Aviation Safety Engineer, Safety Management Section, Rotorcraft Standards Branch, FAA, 10101 Hillwood Pkwy., Fort Worth, TX 76177; telephone (817) 222-5110; email 9-ASW-FTW-AMOC-Requests@faa.gov.

(2) For operations conducted under a 14 CFR part 119 operating certificate or under 14 CFR part 91, subpart K, the FAA suggests that you notify your principal inspector, or lacking a principal inspector, the manager of the local flight standards district office or certificate holding district office before operating any aircraft complying with this AD through an AMOC.

##### (g) Additional Information

The subject of this AD is addressed in European Aviation Safety Agency (EASA) AD No. 2018-0282, dated December 19, 2018. You may view the EASA AD on the internet at <https://www.regulations.gov> in the AD Docket.

##### (h) Subject

Joint Aircraft Service Component (JASC)  
Code: 6320, Main Rotor Gearbox.

Issued in Fort Worth, Texas, on October 31, 2019.

**Helene T. Gandy,**

*Acting Director, Compliance & Airworthiness Division, Aircraft Certification Service.*

[FR Doc. 2019-24342 Filed 11-7-19; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA-2019-0859; Product Identifier 2019-NM-114-AD]

RIN 2120-AA64

#### Airworthiness Directives; The Boeing Company Airplanes

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** The FAA proposes to adopt a new airworthiness directive (AD) for certain The Boeing Company Model 747-100, 747-100B, 747-100B SUD, 747-200B, 747-200C, 747-200F, 747-300, 747-400, 747-400D, 747-400F, 747SR, and 747SP series airplanes. This proposed AD results from fuel system reviews conducted by the manufacturer. This proposed AD would require replacement of the bonding jumpers on the auxiliary power unit (APU) fuel pump. This proposed AD would also require, for certain airplanes, installation of a second bonding jumper; an inspection of the override/jettison fuel pumps and transfer/jettison fuel pumps to determine if the bonding jumper has a one-piece braid or two-piece braid and replacement of the bonding jumper if necessary; and replacement of the bonding jumper on the electrical scavenge fuel pump. The FAA is proposing this AD to address the unsafe condition on these products.

**DATES:** The FAA must receive comments on this proposed AD by December 23, 2019.

**ADDRESSES:** You may send comments, using the procedures found in 14 CFR 11.43 and 11.45, by any of the following methods:

- *Federal eRulemaking Portal:* Go to <https://www.regulations.gov>. Follow the instructions for submitting comments.
- *Fax:* 202-493-2251.
- *Mail:* U.S. Department of Transportation, Docket Operations, M-

30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue SE, Washington, DC 20590.

- **Hand Delivery:** Deliver to Mail address above between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this NPRM, contact Boeing Commercial Airplanes, Attention: Contractual & Data Services (C&DS), 2600 Westminister Blvd., MC 110–SK57, Seal Beach, CA 90740–5600; telephone 562–797–1717; internet <https://www.myboeingfleet.com>. You may view this service information at the FAA, Transport Standards Branch, 2200 South 216th St., Des Moines, WA. For information on the availability of this material at the FAA, call 206–231–3195. It is also available on the internet at <https://www.regulations.gov> by searching for and locating Docket No. FAA–2019–0859.

#### Examining the AD Docket

You may examine the AD docket on the internet at <https://www.regulations.gov> by searching for and locating Docket No. FAA–2019–0859; or in person at Docket Operations between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this NPRM, the regulatory evaluation, any comments received, and other information. The street address for Docket Operations is listed above. Comments will be available in the AD docket shortly after receipt.

#### FOR FURTHER INFORMATION CONTACT:

Jeffrey Rothman, Aerospace Engineer, Propulsion Section, FAA, Seattle ACO Branch, 2200 South 216th St., Des Moines, WA 98198; phone and fax: 206–231–3558; email: [jeffrey.rothman@faa.gov](mailto:jeffrey.rothman@faa.gov).

#### SUPPLEMENTARY INFORMATION:

##### Comments Invited

The FAA invites you to send any written relevant data, views, or arguments about this proposal. Send your comments to an address listed under the **ADDRESSES** section. Include “Docket No. FAA–2019–0859; Product Identifier 2019–NM–114–AD” at the beginning of your comments. The FAA specifically invites comments on the overall regulatory, economic, environmental, and energy aspects of this NPRM. The FAA will consider all comments received by the closing date and may amend this NPRM because of those comments.

The FAA will post all comments received, without change, to <https://www.regulations.gov>, including any

personal information you provide. The FAA will also post a report summarizing each substantive verbal contact received about this NPRM.

#### Discussion

The FAA has examined the underlying safety issues involved in fuel tank explosions on several large transport airplanes, including the adequacy of existing regulations, the service history of airplanes subject to those regulations, and existing maintenance practices for fuel tank systems. As a result of those findings, the FAA issued a final rule titled “Transport Airplane Fuel Tank System Design Review, Flammability Reduction, and Maintenance and Inspection Requirements” (66 FR 23086, May 7, 2001). In addition to new airworthiness standards for transport airplanes and new maintenance requirements, that rule included Amendment 21–78, which established Special Federal Aviation Regulation No. 88 (“SFAR 88”) at 14 CFR part 21. Subsequently, SFAR 88 was amended by Amendment 21–82 (67 FR 57490, September 10, 2002; corrected at 67 FR 70809, November 26, 2002) and Amendment 21–83 (67 FR 72830, December 9, 2002; corrected at 68 FR 37735, June 25, 2003, to change “21–82” to “21–83”).

Among other actions, SFAR 88 requires certain type design (*i.e.*, type certificate (TC) and supplemental type certificate (STC)) holders to substantiate that their fuel tank systems can prevent ignition sources in the fuel tanks. This requirement applies to type design holders for large turbine-powered transport airplanes and for subsequent modifications to those airplanes. It requires them to perform design reviews and to develop design changes and maintenance procedures if their designs do not meet the new fuel tank safety standards. As explained in the preamble to the final rule published on May 7, 2001, the FAA intended to adopt airworthiness directives to mandate any changes found necessary to address unsafe conditions identified as a result of these reviews.

In evaluating these design reviews, the FAA has established four criteria intended to define the unsafe conditions associated with fuel tank systems that require corrective actions. The percentage of operating time during which fuel tanks are exposed to flammable conditions is one of these criteria. The other three criteria address the failure types under evaluation: Single failures, single failures in combination with another latent condition(s), and in-service failure

experience. For all four criteria, the evaluations included consideration of previous actions taken that may mitigate the need for further action.

The FAA has determined that the actions identified in this proposed AD are necessary to reduce the potential of ignition sources inside fuel tanks, which, in combination with flammable fuel vapors, could result in fuel tank explosions and consequent loss of the airplane.

The FAA has received data from the fuel tank inspection program indicating that the existing bond path design provides insufficient bond resistance margin between the fuel pump motor/impeller and structure. In the event of a fuel pump electrical fault, this condition might cause arcs at the existing fuel pump/tank interfaces and an ignition of fuel vapor in the wing fuel tank, which could result in a fuel tank explosion.

#### Related Service Information Under 1 CFR Part 51

The FAA reviewed Boeing Service Bulletin 747–28–2228, Revision 1, dated September 27, 2001. This service information describes procedures for a replacement of the bonding jumpers on the APU fuel pump; an inspection of the six override/jettison fuel pumps and of the two transfer/jettison fuel pumps to determine if the bonding jumper has a one-piece braid or two-piece braid, and replacement of the existing bonding jumper if the bonding jumper has a one-piece braid; installation of a second bonding jumper; and replacement of the bonding jumper on the electrical scavenge fuel pump.

This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the **ADDRESSES** section.

#### Other Relevant Rulemaking

Boeing Service Bulletin 747–28–2228, Revision 1, dated September 27, 2001, identifies “Boeing Service Bulletin 747–28–2033” as a concurrent requirement for certain airplanes. Boeing Alert Service Bulletin 747–28A2033, Revision 1, dated December 18, 2003, is the appropriate source of service information for accomplishing the installation required by AD 2005–01–07, Amendment 39–13931 (70 FR 1336, January 7, 2005) (“AD 2005–01–07”). The compliance time for accomplishing the installation required by AD 2005–01–07 has already passed; therefore, it is not necessary to include Boeing Alert Service Bulletin 747–28A2033 as a concurrent requirement in this proposed AD. The FAA issued AD 2005–01–07 to

ensure adequate electrical bonding between the housing of each fuel pump and airplane structure outside the fuel tanks. Inadequate electrical bonding, in the event of a lightning strike or fuel pump electrical fault, could cause electrical arcing and ignition of fuel vapor in the wing fuel tank, which could result in a fuel tank explosion.

**FAA’s Determination**

The FAA is proposing this AD because the agency evaluated all the relevant information and determined the unsafe condition described previously is likely to exist or develop in other products of the same type design.

**Proposed AD Requirements**

This proposed AD would require accomplishing the actions specified in the service information described previously.

**Costs of Compliance**

The FAA estimates that this proposed AD affects 74 airplanes of U.S. registry. The FAA estimates the following costs to comply with this proposed AD:

**ESTIMATED COSTS**

Action	Labor cost	Parts cost	Cost per product	Cost on U.S. operators
Replacement, Installation, and Inspection.	Up to 15 work-hours × \$85 per hour = Up to \$1,275.	Up to \$2,000 .....	Up to \$3,275 .....	Up to \$242,350.

The FAA estimates the following costs to do any necessary replacements that would be required based on the

results of the proposed inspection. The FAA has no way of determining the

number of aircraft that might need these replacements:

**ON-CONDITION COSTS**

Action	Labor cost	Parts cost	Cost per product
Replacement .....	Up to 6 work-hours × \$85 per hour = Up to \$510.	Up to \$950 .....	Up to \$1,460.

**Authority for This Rulemaking**

Title 49 of the United States Code specifies the FAA’s authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII: Aviation Programs, describes in more detail the scope of the Agency’s authority.

The FAA is issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701: “General requirements.” Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

This proposed AD is issued in accordance with authority delegated by the Executive Director, Aircraft Certification Service, as authorized by FAA Order 8000.51C. In accordance with that order, issuance of ADs is normally a function of the Compliance and Airworthiness Division, but during this transition period, the Executive Director has delegated the authority to issue ADs applicable to transport category airplanes and associated

appliances to the Director of the System Oversight Division.

**Regulatory Findings**

The FAA determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this proposed regulation:

- (1) Is not a “significant regulatory action” under Executive Order 12866,
- (2) Will not affect intrastate aviation in Alaska, and
- (3) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

**List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

**The Proposed Amendment**

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

**PART 39—AIRWORTHINESS DIRECTIVES**

- 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

**§ 39.13 [Amended]**

- 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

**The Boeing Company:** Docket No. FAA–2019–0859; Product Identifier 2019–NM–114–AD.

**(a) Comments Due Date**

The FAA must receive comments by December 23, 2019.

**(b) Affected ADs**

None.

**(c) Applicability**

This AD applies to The Boeing Company Model 747–100, 747–100B, 747–100B SUD, 747–200B, 747–200C, 747–200F, 747–300, 747–400, 747–400D, 747–400F, 747SR, and 747SP series airplanes, certificated in any category, line numbers (L/Ns) 1 through 1229 inclusive.

**(d) Subject**

Air Transport Association (ATA) of America Code 28, Fuel.

**(e) Unsafe Condition**

This AD was prompted by fuel system reviews conducted by the manufacturer indicating that the existing bond path design

provides insufficient bond resistance margin between the fuel pump motor/impeller and structure. The FAA is issuing this AD to address insufficient bond resistance margin between the fuel pump motor/impeller and structure. In the event of a fuel pump electrical fault, this condition might cause arcs at the existing fuel pump/tank interfaces and an ignition of fuel vapor in the wing fuel tank, which could result in a fuel tank explosion and consequent loss of the airplane.

#### (f) Compliance

Comply with this AD within the compliance times specified, unless already done.

#### (g) Definitions

For the purposes of this AD, the definitions specified in paragraphs (g)(1) through (4) of this AD apply.

(1) Group 1 airplanes: L/Ns 1 through 167 inclusive.

(2) Group 2 airplanes: L/Ns 168 through 971 inclusive.

(3) Group 3 airplanes: L/Ns 972 through 1161 inclusive.

(4) Group 4 airplanes: L/Ns 1162 through 1229 inclusive.

#### (h) Replacement, Installation, and Inspection

Within 60 months after the effective date of this AD, do the applicable actions specified in paragraphs (h)(1) through (4) of this AD, in accordance with the Accomplishment Instructions of Boeing Service Bulletin 747-28-2228, Revision 1, dated September 27, 2001.

(1) For Groups 1, 2, and 3 airplanes: Do the actions specified in paragraphs (h)(1)(i) and (ii) of this AD.

(i) Do a general visual inspection of the six override/jettison fuel pumps to determine if the bonding jumper has a one-piece braid or two-piece braid. If the bonding jumper has a one-piece braid, within 60 months after the effective date of this AD, replace the existing bonding jumper.

(ii) Install a second bonding jumper.

(2) For Groups 1, 2 and 3 airplanes with horizontal stabilizer fuel tanks: Do the actions specified in paragraphs (h)(2)(i) and (ii) of this AD.

(i) Do a general visual inspection of the two transfer/jettison fuel pumps to determine if the bonding jumper has a one-piece braid or a two-piece braid. If the bonding jumper has a one-piece braid, within 60 months after the effective date of this AD, replace the existing bonding jumper.

(ii) Install a second bonding jumper.

(3) For all airplanes: Replace the bonding jumpers on the auxiliary power unit (APU) fuel pump.

(4) For Groups 1 and 2 airplanes: Replace the bonding jumper on the electrical scavenge fuel pump.

#### (i) Alternative Methods of Compliance (AMOCs)

(1) The Manager, Seattle ACO Branch, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your

principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the certification office, send it to the attention of the person identified in paragraph (j)(1) of this AD. Information may be emailed to: [9-ANM-Seattle-ACO-AMOC-Requests@faa.gov](mailto:9-ANM-Seattle-ACO-AMOC-Requests@faa.gov).

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(3) An AMOC that provides an acceptable level of safety may be used for any repair, modification, or alteration required by this AD if it is approved by The Boeing Company Organization Designation Authorization (ODA) that has been authorized by the Manager, Seattle ACO Branch, FAA, to make those findings. To be approved, the repair method, modification deviation, or alteration deviation must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

#### (j) Related Information

(1) For more information about this AD, contact Jeffrey Rothman, Aerospace Engineer, Propulsion Section, FAA, Seattle ACO Branch, 2200 South 216th St., Des Moines, WA 98198; phone and fax: 206-231-3558; email: [jeffrey.rothman@faa.gov](mailto:jeffrey.rothman@faa.gov).

(2) For service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Contractual & Data Services (C&DS), 2600 Westminister Blvd., MC 110-SK57, Seal Beach, CA 90740-5600; telephone 562-797-1717; internet <https://www.myboeingfleet.com>. You may view this service information at the FAA, Transport Standards Branch, 2200 South 216th St., Des Moines, WA. For information on the availability of this material at the FAA, call 206-231-3195.

Issued in Des Moines, Washington, on October 29, 2019.

#### Dionne Palermo,

Acting Director, System Oversight Division, Aircraft Certification Service.

[FR Doc. 2019-24329 Filed 11-7-19; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 71

[Docket No. FAA-2019-0785; Airspace Docket No. 19-AEA-14]

#### Proposed Revocation of Class E Airspace; Grundy, VA

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** This action proposes to remove Class E airspace at Grundy, VA, as Grundy Municipal Airport has been abandoned, and controlled airspace is

no longer required. This action would enhance the safety and management of controlled airspace within the national airspace system.

**DATES:** Comments must be received on or before December 23, 2019.

**ADDRESSES:** Send comments on this rule to: U.S. Department of Transportation, Docket Operations, 1200 New Jersey Avenue SE, West Building Ground Floor, Room W12-140, Washington, DC 20590; Telephone: (800) 647-5527, or (202) 366-9826. You must identify the Docket No. FAA-2019-0785; Airspace Docket No. 19-AEA-14, at the beginning of your comments. You may also submit comments through the internet at <https://www.regulations.gov>. You may review the public docket containing the proposal, any comments received, and any final disposition in person in the Dockets Office between 9:00 a.m. and 5:00 p.m., Monday through Friday, except federal holidays.

FAA Order 7400.11D, Airspace Designations and Reporting Points, and subsequent amendments can be viewed on line at [https://www.faa.gov/air\\_traffic/publications/](https://www.faa.gov/air_traffic/publications/). For further information, you can contact the Airspace Policy Group, Federal Aviation Administration, 800 Independence Avenue, SW, Washington, DC, 20591; telephone: 202-267-8783. The Order is also available for inspection at the National Archives and Records Administration (NARA). For information on the availability of FAA Order 7400.11D at NARA, email [fedreg.legal@nara.gov](mailto:fedreg.legal@nara.gov) or go to <https://www.archives.gov/federal-register/cfr/ibr-locations.html>.

**FOR FURTHER INFORMATION CONTACT:** John Fornito, Operations Support Group, Eastern Service Center, Federal Aviation Administration, 1701 Columbia Ave, College Park, GA 30337; telephone (404) 305-6364.

#### SUPPLEMENTARY INFORMATION:

##### Authority for This Rulemaking

The FAA's authority to issue rules regarding aviation safety is found in Title 49 of the United States Code. Subtitle I, Section 106 describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the agency's authority. This proposed rulemaking is promulgated under the authority described in Subtitle VII, part A, Subpart I, Section 40103. Under that section, the FAA is charged with prescribing regulations to assign the use of airspace necessary to ensure the safety of aircraft and the efficient use of airspace. This regulation is within the scope of that authority as it would