

institution's loss on the exposure if the reference capital instrument were to have a value of zero.

(3) *Adjustments to reflect a short position.* In order to adjust the gross long position to recognize a short position in the same instrument under paragraph (h)(1) of this section, the following criteria must be met:

(i) The maturity of the short position must match the maturity of the long position, or the short position must have a residual maturity of at least one year (maturity requirement); or

(ii) For a position that is a trading asset or trading liability (whether on- or off-balance sheet) as reported on the FDIC-supervised institution's Call Report, if the FDIC-supervised institution has a contractual right or obligation to sell the long position at a specific point in time and the counterparty to the contract has an obligation to purchase the long position if the FDIC-supervised institution exercises its right to sell, this point in time may be treated as the maturity of the long position such that the maturity of the long position and short position are deemed to match for purposes of the maturity requirement, even if the maturity of the short position is less than one year; and

(iii) For an investment in an FDIC-supervised institution's own capital instrument under paragraph (c)(1) of this section, an investment in the capital of an unconsolidated financial institution under paragraphs (c)(4), (c)(5), and (d)(1)(iii) of this section, and an investment in a covered debt instrument under paragraphs (c)(4) and (c)(5) of this section:

(A) The FDIC-supervised institution may only net a short position against a long position in an investment in the FDIC-supervised institution's own capital instrument under paragraph (c)(1) of this section if the short position involves no counterparty credit risk;

(B) A gross long position in an investment in the FDIC-supervised institution's own capital instrument, an investment in the capital of an unconsolidated financial institution, or an investment in a covered debt instrument due to a position in an index may be netted against a short position in the same index;

(C) Long and short positions in the same index without maturity dates are considered to have matching maturities; and

(D) A short position in an index that is hedging a long cash or synthetic position in an investment in the FDIC-supervised institution's own capital instrument, an investment in the capital instrument of an unconsolidated

financial institution, or an investment in a covered debt instrument can be decomposed to provide recognition of the hedge. More specifically, the portion of the index that is composed of the same underlying instrument that is being hedged may be used to offset the long position if both the long position being hedged and the short position in the index are reported as a trading asset or trading liability (whether on- or off-balance sheet) on the FDIC-supervised institution's Call Report, and the hedge is deemed effective by the FDIC-supervised institution's internal control processes, which have not been found to be inadequate by the FDIC.

* * * * *

Dated: September 11, 2018.

Joseph M. Otting,

Comptroller of the Currency.

By order of the Board of Governors of the Federal Reserve System, March 22, 2019.

Ann E. Misback,

Secretary of the Board.

Dated at Washington, DC on September 19, 2018.

By order of the Board of Directors.
Federal Deposit Insurance Corporation.

Valerie J. Best,

Assistant Executive Secretary.

[FR Doc. 2019-06344 Filed 4-5-19; 8:45 am]

BILLING CODE 4810-33-P; 6210-01-P; 6714-01-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 25

[Docket No. FAA-2013-0772; Notice No. 25-19-01-SC]

Special Conditions: Embraer EMB-550 Airplane; Flight Envelope Protection: Normal Load Factor (g) Limiting

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed special conditions.

SUMMARY: This action proposes an amendment to special conditions for the Embraer EMB-550 airplane. This airplane will have novel or unusual design features when compared to the state of technology envisioned in the airworthiness standards for transport category airplanes. This design feature is associated with an electronic flight control system that prevents the pilot from inadvertently or intentionally exceeding the positive or negative airplane limit load factor. The applicable airworthiness regulations do not contain adequate or appropriate

safety standards for this design feature. These proposed special conditions contain the additional safety standards that the Administrator considers necessary to establish a level of safety equivalent to that established by the existing airworthiness standards.

DATES: Send comments on or before April 29, 2019.

ADDRESSES: Send comments identified by Docket No. FAA-2013-0772 using any of the following methods:

- *Federal eRegulations Portal:* Go to <http://www.regulations.gov/> and follow the online instructions for sending your comments electronically.

- *Mail:* Send comments to Docket Operations, M-30, U.S. Department of Transportation (DOT), 1200 New Jersey Avenue SE., Room W12-140, West Building Ground Floor, Washington, DC, 20590-0001.

- *Hand Delivery or Courier:* Take comments to Docket Operations in Room W12-140 of the West Building Ground Floor at 1200 New Jersey Avenue SE., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

- *Fax:* Fax comments to Docket Operations at 202-493-2251.

Privacy: The FAA will post all comments it receives, without change, to <http://www.regulations.gov/>, including any personal information the commenter provides. Using the search function of the docket website, anyone can find and read the electronic form of all comments received into any FAA docket, including the name of the individual sending the comment (or signing the comment for an association, business, labor union, etc.). DOT's complete Privacy Act Statement can be found in the **Federal Register** published on April 11, 2000 (65 FR 19477-19478).

Docket: Background documents or comments received may be read at <http://www.regulations.gov/> at any time. Follow the online instructions for accessing the docket or go to Docket Operations in Room W12-140 of the West Building Ground Floor at 1200 New Jersey Avenue SE., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

FOR FURTHER INFORMATION CONTACT: Joe Jacobsen, Airplane & Flight Crew Interface Section, AIR-671, Transport Standards Branch, Policy and Innovation Division, Aircraft Certification Service, Federal Aviation Administration, 2200 South 216th Street, Des Moines, Washington 98198; telephone and fax 206-231-3158; email joe.jacobsen@faa.gov.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite interested people to take part in this rulemaking by sending written comments, data, or views. The most helpful comments reference a specific portion of the special conditions, explain the reason for any recommended change, and include supporting data.

We will consider all comments we receive by the closing date for comments. We may change these special conditions based on the comments we receive.

Background

On August 9, 2016, Embraer applied for a change to Type Certificate No. TC00062IB to include additional flexibility to the normal load factor limit on the Model EMB-550 airplane, by amending the existing Model EMB-550 Special Conditions No. 25-520-SC as a result of harmonization efforts in the Flight Test Harmonization Working Group (FTHWG). The Embraer EMB-550 airplane, currently approved under Type Certificate No. TC00062IB, is a twin-engine, transport category airplane with a maximum takeoff weight of 42,857 pounds. The EMB-550 has a maximum seating capacity of 12 passengers.

The Model EMB-550 flight control system design incorporates normal load factor limiting on a full-time basis that will prevent the pilot from inadvertently or intentionally exceeding the positive or negative airplane limit load factor. This feature is considered novel and unusual in that the current regulations do not provide standards for maneuverability and controllability evaluations for such systems.

Type Certification Basis

Under the provisions of title 14, Code of Federal Regulations (14 CFR) 21.101, Embraer must show that the EMB-550 airplane, as changed, continues to meet the applicable provisions of the regulations listed in Type Certificate No. TC00062IB or the applicable regulations in effect on the date of application for the change, except for earlier amendments as agreed upon by the FAA.

If the Administrator finds that the applicable airworthiness regulations (*i.e.*, 14 CFR part 25) do not contain adequate or appropriate safety standards for the Embraer EMB-550 airplane because of a novel or unusual design feature, special conditions are prescribed under the provisions of § 21.16.

Special conditions are initially applicable to the model for which they

are issued. Should the type certificate for that model be amended later to include any other model that incorporates the same novel or unusual design feature, or should any other model already included on the same type certificate be modified to incorporate the same novel or unusual design feature, these special conditions would also apply to the other model under § 21.101.

In addition to the applicable airworthiness regulations and special conditions, the Embraer EMB-550 airplane must comply with the fuel vent and exhaust emission requirements of 14 CFR part 34 and the noise certification requirements of 14 CFR part 36.

The FAA issues special conditions, as defined in 14 CFR 11.19, in accordance with § 11.38, and they become part of the type certification basis under § 21.101.

Novel or Unusual Design Features

The Embraer EMB-550 airplane will incorporate the following novel or unusual design features:

The Model EMB-550 flight control system design incorporates normal load factor limiting on a full-time basis that will prevent the pilot from inadvertently or intentionally exceeding the positive or negative airplane limit load factor. This feature is considered novel and unusual in that the current regulations do not provide standards for maneuverability and controllability evaluations for such systems.

Discussion

The normal load factor limit on the Model EMB-550 is unique in that traditional airplanes with conventional flight control systems (mechanical linkages) are limited in the pitch axis only by the elevator surface area and deflection limit. The elevator control power is normally derived for adequate controllability and the maneuverability at the most critical longitudinal pitching moment. The result is that traditional airplanes have a significant portion of the flight envelope where maneuverability in excess of limit structural design values is possible. The Model EMB-550 because of the normal load factor limit does not have this excess maneuverability.

Title 14, Code of Federal Regulations (14 CFR) part 25 does not specify requirements for demonstrating maneuver control that impose any handling qualities requirements beyond the design limit structural loads. Nevertheless, some pilots are accustomed to the availability of this excess maneuver capacity in case of

extreme emergency such as upset recoveries or collision avoidance.

As a result of harmonization efforts with other civil aviation authorities through the Flight Test Harmonization Working Group (FTHWG), the FAA is including additional flexibility in maneuverability limits by amending the existing Model EMB-550 Special Conditions No. 25-520-SC. This additional flexibility allows for reduced maneuverability limits beyond V_{mo}/M_{mo}. The existing special conditions are otherwise unchanged.

The proposed special conditions contain the additional safety standards that the Administrator considers necessary to establish a level of safety equivalent to that established by the existing airworthiness standards.

Applicability

As discussed above, these special conditions are applicable to the Embraer EMB-550 airplane. Should Embraer apply at a later date for a change to the type certificate to include another model incorporating the same novel or unusual design feature, these special conditions would apply to that model as well.

Certification of the Embraer EMB-550 airplane is currently scheduled for April 2019. The substance of these special conditions has been subject to the notice and public comment procedure in several prior instances. Therefore, because a delay would significantly affect the applicant's installation of the system and the certification of the airplane, we are shortening the public comment period to 20 days.

Conclusion

This action affects only certain novel or unusual design features on one model of airplane. It is not a rule of general applicability.

List of Subjects in 14 CFR Part 25

Aircraft, Aviation safety, Reporting and recordkeeping requirements.

Authority Citation

The authority citation for these special conditions is as follows:

Authority: 49 U.S.C. 106(f), 106(g), 40113, 44701, 44702, 44704.

The Proposed Special Conditions

Accordingly, the Federal Aviation Administration (FAA) proposes the following special conditions as part of the type certification basis for Embraer EMB-550 airplanes.

Flight Envelope Protection: Normal Load Factor (g) Limiting

1. To meet the intent of adequate maneuverability and controllability required

by § 25.143(a), and in addition to the requirements of § 25.143(a) and in the absence of other limiting factors, the following special conditions are proposed based on § 25.333(b):

a. The positive limiting load factor must not be less than:

(1) 2.5g for the normal state of the electronic flight control system with the high lift devices retracted up to V_{mo}/M_{mo} . The positive limiting load factor may be gradually reduced down to 2.25g above V_{mo}/M_{mo} .

(2) 2.0g for the normal state of the electronic flight control system with the high lift devices extended.

b. The negative limiting load factor must be equal to or more negative than:

(1) Minus 1.0g for the normal state of the electronic flight control system with the high lift devices retracted.

(2) 0.0g for the normal state of the electronic flight control system with high lift devices extended.

c. Maximum reachable positive load factor wings level may be limited by the characteristics of the electronic flight control system or flight envelope protections (other than load factor protection) provided that:

(1) The required values are readily achievable in turns, and

(2) Wings level pitch up responsiveness is satisfactory.

d. Maximum achievable negative load factor may be limited by the characteristics of the electronic flight control system or flight envelope protections (other than load factor protection) provided that:

(1) Pitch down responsiveness is satisfactory, and

(2) From level flight, 0g is readily achievable or alternatively, a satisfactory trajectory change is readily achievable at operational speeds. For the FAA to consider a trajectory change as satisfactory, the applicant should propose and justify a pitch rate that provides sufficient maneuvering capability in the most critical scenarios.

e. Compliance demonstration with the above requirements may be performed without ice accretion on the airframe.

f. These special conditions do not impose an upper bound for the normal load factor limit, nor does it require that the limiter exist. If the limit is set at a value beyond the structural design limit maneuvering load factor “n” of §§ 25.333(b), 25.337(b) and 25.337(c), there should be a very obvious positive tactile feel built into the controller so that it serves as a deterrent to inadvertently exceeding the structural limit.

Issued in Des Moines, Washington.

Suzanne Masterson,

Acting Manager, Transport Standards Branch, Policy and Innovation Division, Aircraft Certification Service.

[FR Doc. 2019-06647 Filed 4-5-19; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2019-0188; Product Identifier 2018-NM-174-AD]

RIN 2120-AA64

Airworthiness Directives; The Boeing Company Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: We propose to adopt a new airworthiness directive (AD) for all The Boeing Company Model 747-200B, 747-200C, 747-200F, 747-300, 747-400, 747-400D, 747-400F, 747-8F, and 747-8 series airplanes. This AD was prompted by reports of uncommanded movement of the Captain’s and First Officer’s seats. This proposed AD would require, for the Captain’s and First Officer’s seats, repetitive horizontal actuator identifications, repetitive checks of the horizontal movement system (HMS), a detailed inspection of the HMS for certain airplanes, and applicable on-condition actions. This proposed AD would also require an inspection to determine the part number and, if applicable, the serial number of the Captain’s and First Officer’s seats and applicable on-condition actions. This proposed AD would also provide an optional terminating action for the repetitive checks of the HMS for certain airplanes. We are proposing this AD to address the unsafe condition on these products.

DATES: We must receive comments on this proposed AD by May 23, 2019.

ADDRESSES: You may send comments, using the procedures found in 14 CFR 11.43 and 11.45, by any of the following methods:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.
- *Fax:* 202-493-2251.
- *Mail:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE, Washington, DC 20590.
- *Hand Delivery:* Deliver to Mail address above between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this NPRM, contact Boeing Commercial Airplanes, Attention: Contractual & Data Services (C&DS), 2600 Westminister Blvd., MC 110-SK57, Seal Beach, CA

90740-5600; telephone 562-797-1717; internet <https://www.myboeingfleet.com>. You may view this service information at the FAA, Transport Standards Branch, 2200 South 216th St., Des Moines, WA. For information on the availability of this material at the FAA, call 206-231-3195. It is also available on the internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2019-0188.

Examining the AD Docket

You may examine the AD docket on the internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2019-0188; or in person at Docket Operations between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this NPRM, the regulatory evaluation, any comments received, and other information. The street address for Docket Operations (phone: 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

FOR FURTHER INFORMATION CONTACT: Brandon Lucero, Aerospace Engineer, Cabin Safety and Environmental Systems Section, FAA, Seattle ACO Branch, 2200 South 216th St., Des Moines, WA 98198; phone and fax: 206-231-3569; email: Brandon.Lucero@faa.gov.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposal. Send your comments to an address listed under the **ADDRESSES** section. Include “Docket No. FAA-2019-0188; Product Identifier 2018-NM-174-AD” at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this NPRM. We will consider all comments received by the closing date and may amend this NPRM because of those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

Discussion

We have received reports of uncommanded movement of the Captain’s and First Officer’s seats. An operator reported that during a takeoff, the First Officer’s seat unlocked from its