

instrumentation specified in 49 CFR part 572, subpart E filtered in accordance with SAE International (SAE) recommended practice J211/1, "Instrumentation for Impact Test—Part 1—Electronic Instrumentation."

c. The occupant must not interact with the armrest or other seat components in any manner significantly different than would be expected for a forward-facing seat installation.

5. *Pelvis Criteria:*

Any part of the load-bearing portion of the bottom of the ATD pelvis must not translate beyond the edges of the seat bottom seat-cushion supporting structure.

6. *Femur Criteria:*

Axial rotation of the upper leg (about the z-axis of the femur per SAE Recommended Practice J211/1) must be limited to 35 degrees from the nominal seated position. Evaluation during rebound does not need to be considered.

7. *ATD and Test Conditions:*

Longitudinal tests conducted to measure the injury criteria above must be performed with the FAA Hybrid III ATD, as described in SAE 1999-01-1609, "A Lumbar Spine Modification to the Hybrid III ATD for Aircraft Seat Tests." The tests must be conducted with an undeformed floor, at the most-critical yaw cases for injury and with all lateral structural supports (e.g. armrests or walls) installed.

Note: Boeing must demonstrate that the installation of seats via plinths or pallets meets all applicable requirements. Compliance with the guidance contained in policy memorandum PS-ANM-100-2000-00123, "Guidance for Demonstrating Compliance with Seat Dynamic Testing for Plinths and Pallets," dated February 2, 2000, is acceptable to the FAA.

8. *Inflatable Airbag Restraint Systems Special Conditions:*

If inflatable airbag restraint systems are installed, the airbag systems must meet the requirements in one of the airbag (inflatable restraint) special conditions applicable to the Boeing Model 747-8 airplane.

Issued in Des Moines, Washington, on August 22, 2018.

Victor Wicklund,

Manager, Transport Standards Branch, Policy and Innovation Division, Aircraft Certification Service.

[FR Doc. 2018-19216 Filed 9-4-18; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 25

[Docket No. FAA-2018-0335; Special Conditions No. 25-725-SC]

Special Conditions: Bombardier Inc., Model BD-700-2A12 and BD-700-2A13 Series Airplanes; Flight Envelope Protection: High Incidence Protection System

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final special conditions; correction.

SUMMARY: The FAA is correcting an error that appeared in the **Federal Register** on May 1, 2018, for special conditions No. 25-725-SC, Docket No. FAA-2018-0335. As published, there was an error in the citation and the correct citation has been added.

DATES: Effective on Bombardier on September 5, 2018.

FOR FURTHER INFORMATION CONTACT: Joe Jacobsen, Airplane and Flight Crew Interface, AIR-671, Transport Standards Branch, Policy and Innovation Division, Aircraft Certification Service, Federal Aviation Administration, 2200 South 216th Street, Des Moines, Washington 98198; telephone and fax 206-231-3158; email Joe.Jacobsen@faa.gov.

SUPPLEMENTARY INFORMATION:

Background

On April 25, 2018, the FAA issued Special Conditions No. 25-725-SC, Docket No. FAA-2018-0335, which was published in the **Federal Register** on May 1, 2018 (83 FR 18934). Those special conditions pertain to the high incidence protection system that replaces the stall warning system during normal operating conditions, prohibits the airplane from stalling, limits the angle of attack at which the airplane can be flown during normal low speed operation, and cannot be overridden by the flight crew for Bombardier Model BD-700-2A12 and BD-700-2A13 series airplanes. As published, part II, paragraph 7 of the final special conditions cited § 25.143(j)(2)(i) instead of § 25.143(j)(1). There are no substantive changes to the document and it was apparent that § 25.143(j)(1) should have been referenced from the beginning.

Correction

In the final special conditions document FR Doc. 2018-09126 (Filed 4-30-2018; 8:45 a.m.), published on

May 1, 2018 (83 FR 18934), make the following correction:

On page 18938, column 2, under part II, paragraph 7, correct "§ 25.143(j)(2)(i)" to read "§ 25.143(j)(1)".

Issued in Des Moines, Washington, on August 27, 2018.

Victor Wicklund,

Manager, Transport Standards Branch, Policy and Innovation Division, Aircraft Certification Service.

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2018-0163; Product Identifier 2017-NM-168-AD; Amendment 39-19386; AD 2018-18-07]

RIN 2120-AA64

Airworthiness Directives; The Boeing Company Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for certain The Boeing Company Model 757 airplanes. This AD was prompted by an evaluation by the design approval holder (DAH) indicating that the longitudinal lap splices of the fuselage skin are subject to widespread fatigue damage (WFD). This AD requires repetitive inspections of the longitudinal lap splices of the fuselage skin for cracking and protruding fasteners, and applicable corrective actions. We are issuing this AD to address the unsafe condition on these products.

DATES: This AD is effective October 10, 2018.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in this AD as of October 10, 2018.

ADDRESSES: For service information identified in this final rule, contact Boeing Commercial Airplanes, Attention: Contractual & Data Services (C&DS), 2600 Westminister Blvd., MC 110-SK57, Seal Beach, CA 90740-5600; telephone 562-797-1717; internet <https://www.myboeingfleet.com>. You may view this service information at the FAA, Transport Standards Branch, 2200 South 216th St., Des Moines, WA. For information on the availability of this material at the FAA, call 206-231-3195.