

Dated: January 23, 2018.

**Dale L. Aultman,**

Secretary to the Board, Farm Credit System Insurance Corporation.

[FR Doc. 2018-01421 Filed 1-25-18; 8:45 am]

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## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA-2017-0716; Product Identifier 2016-NM-165-AD; Amendment 39-19165; AD 2018-02-12]

RIN 2120-AA64

#### Airworthiness Directives; Airbus Airplanes

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Final rule.

**SUMMARY:** We are superseding Airworthiness Directive (AD) 2016-02-01, which applied to certain Airbus Model A320-211, -212, and -231 airplanes. AD 2016-02-01 required repetitive inspections to detect cracks of the pressurized floor fittings at a certain frame, and renewal of the zone protective finish or replacement of fittings with new fittings if necessary. AD 2016-02-01 also provided an optional modification that was terminating action for the repetitive inspections. This new AD retains the requirements of AD 2016-02-01, and requires accomplishment of the modification that was optional in AD 2016-02-01. This AD was prompted by the results of an additional fatigue analysis of cracking of the pressurized floor fittings and a determination that the optional modification should become a required action. We are issuing this AD to address the unsafe condition on these products.

**DATES:** This AD is effective March 2, 2018.

The Director of the Federal Register approved the incorporation by reference of certain other publications listed in this AD as of March 3, 2016 (81 FR 4878, January 28, 2016).

**ADDRESSES:** For service information identified in this final rule, contact Airbus, Airworthiness Office—EIAS, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone: +33 5 61 93 36 96; fax: +33 5 61 93 44 51; email: [account.airworth-eas@airbus.com](mailto:account.airworth-eas@airbus.com); internet: <http://www.airbus.com>. You may view this referenced service information at the

FAA, Transport Standards Branch, 1601 Lind Avenue SW, Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221. It is also available on the internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2017-0716.

#### Examining the AD Docket

You may examine the AD docket on the internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2017-0716; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (telephone: 800-647-5527) is Docket Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE, Washington, DC 20590.

**FOR FURTHER INFORMATION CONTACT:** Sanjay Ralhan, Aerospace Engineer, International Section, Transport Standards Branch, FAA, 1601 Lind Avenue SW, Renton, WA 98057-3356; telephone: 425-227-1405; fax: 425-227-1149.

#### SUPPLEMENTARY INFORMATION:

##### Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to supersede AD 2016-02-01, Amendment 39-18380 (81 FR 4878, January 28, 2016) (“AD 2016-02-01”). AD 2016-02-01 applied to certain Airbus Model A320-211, -212, and -231 airplanes. The NPRM published in the *Federal Register* on August 15, 2017 (82 FR 38618).

The European Aviation Safety Agency (EASA), which is the Technical Agent for the Member States of the European Union, has issued EASA AD 2016-0181, dated September 13, 2016 (referred to after this as the Mandatory Continuing Airworthiness Information, or “the MCAI”), to correct an unsafe condition for certain Airbus Model A320-211, -212, and -231 airplanes. The MCAI states:

During centre fuselage certification full scale fatigue testing, damage was found on the pressurized floor fittings at Frame (FR) 36, below the lower surface panel.

This condition, if not detected and corrected, could affect the structural integrity of the aeroplane.

To prevent such damage, Airbus developed modification 21282, which was introduced in production from MSN [manufacturer serial

number] 0105, to reinforce the pressurized floor fitting lower surface by changing material. For affected in-service aeroplanes, Airbus issued Service Bulletin (SB) A320-57-1028, introducing repetitive inspections, and SB A320-57-1029, which provides modification instructions.

DGAC [Direction Générale de l'Aviation Civile] France issued AD 95-099-067 to require these repetitive inspections and, depending on findings, corrective action(s), while the modification was specified in that [French] AD as optional terminating action for these inspections.

Following new analysis in the frame of Extended Service Goal exercise, the inspection thresholds and intervals were revised to meet the original Design Service Goal. Consequently, EASA issued AD 2013-0226 [which corresponds to FAA AD 2016-02-01 (81 FR 4878, January 28, 2016)] to retain the requirements of DGAC France AD 95-099-067, which was superseded, but required those actions within reduced compliance times.

Since that [EASA] AD was issued, in the frame of Widespread Fatigue Damages analysis, the situation has been reassessed and it has been decided to reclassify the modification, still stated as ‘optional’ terminating action in EASA AD 2013-0226, to the status ‘mandatory’.

For the reasons described above, this [EASA] AD retains the requirements of EASA AD 2013-0226, which is superseded, but requires embodiment of the modification as specified in Airbus SB A320-57-1029.

You may examine the MCAI in the AD docket on the internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2017-0716.

#### Comments

We gave the public the opportunity to participate in developing this AD. The following presents the comment received on the NPRM and the FAA's response to that comment.

#### Request To Revise the Applicability

Allegiant Air noted that the applicability specified in paragraph (c) of the proposed AD included Airbus Model A320-214 airplanes. The commenter asked if Model A320-214 airplanes were included in the applicability in error. The commenter observed that neither the applicability of AD 2016-02-01 or EASA AD 2016-0181, nor the effectivity of Airbus Service Bulletin A320-57-1028, Revision 02, dated June 3, 2013, included Model A320-214 airplanes.

We infer that the commenter is requesting that Model A320-214 airplanes be removed from the applicability of the proposed AD. We agree, for the reasons provided by the commenter. This final rule is not applicable to Model A320-214 airplanes; therefore, we have revised the

applicability specified in paragraph (c) of this AD by removing Model A320–214 airplanes.

**Conclusion**

We reviewed the available data, including the comment received, and determined that air safety and the public interest require adopting this AD with the changes described previously and minor editorial changes. We have determined that these changes:

- Are consistent with the intent that was proposed in the NPRM for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the NPRM.

We also determined that these changes will not increase the economic burden on any operator or increase the scope of this AD.

**Related Service Information Under 1 CFR Part 51**

Airbus has issued Service Bulletin A320–57–1028, Revision 02, dated June 3, 2013. The service information describes procedures for an inspection to detect cracks of the pressurized floor fittings at FR 36, renewal of the zone protective finish, and replacement of fittings with new fittings.

Airbus has also issued Service Bulletin A320–57–1029, Revision 02,

dated June 16, 1999. The service information describes procedures for modification of the pressurized floor fittings at FR 36.

This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the **ADDRESSES** section.

**Costs of Compliance**

We estimate that this AD affects 13 airplanes of U.S. registry. We estimate the following costs to comply with this AD:

**ESTIMATED COSTS**

Action	Labor cost	Parts cost	Cost per product	Cost on U.S. operators
Inspection .....	11 work-hours × \$85 per hour = \$935 per inspection cycle.	\$0	\$935 per inspection cycle.	\$12,155 per inspection cycle.
Modification .....	85 work-hours × \$85 per hour = \$7,225 .....	5,320	\$12,545 .....	\$163,085.

We have received no definitive data that will enable us to provide cost estimates for the on-condition actions specified in this AD.

**Authority for This Rulemaking**

Title 49 of the United States Code specifies the FAA’s authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. “Subtitle VII: Aviation Programs,” describes in more detail the scope of the Agency’s authority.

We are issuing this rulemaking under the authority described in “Subtitle VII, Part A, Subpart III, Section 44701: General requirements.” Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

This AD is issued in accordance with authority delegated by the Executive Director, Aircraft Certification Service, as authorized by FAA Order 8000.51C. In accordance with that order, issuance of ADs is normally a function of the Compliance and Airworthiness Division, but during this transition period, the Executive Director has delegated the authority to issue ADs applicable to transport category

airplanes to the Director of the System Oversight Division.

**Regulatory Findings**

We determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

1. Is not a “significant regulatory action” under Executive Order 12866;
2. Is not a “significant rule” under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979);
3. Will not affect intrastate aviation in Alaska; and
4. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

**List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

**Adoption of the Amendment**

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

**PART 39—AIRWORTHINESS DIRECTIVES**

- 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

**§ 39.13 [Amended]**

- 2. The FAA amends § 39.13 by removing Airworthiness Directive (AD) 2016–02–01, Amendment 39–18380 (81 FR 4878, January 28, 2016), and adding the following new AD:

**2018–02–12 Airbus:** Amendment 39–19165; Docket No. FAA–2017–0716; Product Identifier 2016–NM–165–AD.

**(a) Effective Date**

This AD is effective March 2, 2018.

**(b) Affected ADs**

This AD replaces AD 2016–02–01, Amendment 39–18380 (81 FR 4878, January 28, 2016) (“AD 2016–02–01”).

**(c) Applicability**

This AD applies to Airbus Model A320–211, –212, and –231 airplanes, certificated in any category, manufacturer serial numbers up through 0104 inclusive.

**(d) Subject**

Air Transport Association (ATA) of America Code 57, Wings.

**(e) Reason**

This AD is intended to complete certain mandated programs intended to support the airplane reaching its limit of validity (LOV) of the engineering data that support the established structural maintenance program. We are issuing this AD to prevent fatigue cracking in the pressurized floor fittings at frame (FR) 36, which could result in the reduced structural integrity of the floor

fittings and subsequent depressurization of the fuselage.

#### (f) Compliance

Comply with this AD within the compliance times specified, unless already done.

#### (g) Inspection

(1) At the latest of the times specified in paragraphs (g)(1)(i), (g)(1)(ii), and (g)(1)(iii) of this AD: Do a detailed inspection of the pressurized floor fittings at FR 36, in accordance with the Accomplishment Instructions of Airbus Service Bulletin A320-57-1028, Revision 02, dated June 3, 2013. Repeat the inspection thereafter at intervals not to exceed 9,300 flight cycles or 18,600 flight hours, whichever occurs first.

(i) Before exceeding 20,900 flight cycles or 41,800 flight hours, whichever occurs first since first flight of the airplane.

(ii) Within 9,300 flight cycles or 18,600 flight cycles since the most recent inspection accomplished in accordance with the Accomplishment Instructions of Airbus Service Bulletin A320-57-1028, Revision 02, dated June 3, 2013.

(iii) Within 1,250 flight cycles or 2,500 flight hours after March 3, 2016 (the effective date of AD 2016-02-01), without exceeding 12,000 flight cycles since the most recent inspection accomplished in accordance with the Accomplishment Instructions of Airbus Service Bulletin A320-57-1028, Revision 02, dated June 3, 2013.

(2) If any crack is found during any inspection required by paragraph (g)(1) of this AD: Before further flight, repair using a method approved by the Manager, International Section, Transport Standards Branch, FAA; or the European Aviation Safety Agency (EASA); or Airbus's EASA Design Organization Approval (DOA). If approved by the DOA, the approval must include the DOA-authorized signature.

#### (h) Modification

Before exceeding 48,000 total flight cycles or 96,000 total flight hours, whichever occurs first since first flight of the airplane: Modify (replace aluminum fittings with titanium fittings) the pressurized floor fittings at FR 36, in accordance with the Accomplishment Instructions of Airbus Service Bulletin A320-57-1029, Revision 02, dated June 16, 1999. Accomplishment of this modification is terminating action for the repetitive inspections required by paragraph (g) of this AD for the modified airplane only.

#### (i) Credit for Previous Actions

(1) This paragraph provides credit for the inspection required by paragraph (g) of this AD, if that inspection was performed before the effective date of this AD using Airbus Service Bulletin A320-57-1028, dated August 12, 1991; or Revision 01, dated April 19, 1996.

(2) This paragraph provides credit for the modification required by paragraph (h) of this AD, if that modification was performed before the effective date of this AD using Airbus Service Bulletin A320-57-1029, dated August 12, 1991; or Revision 01, dated November 10, 1992.

#### (j) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs)*: The Manager, International Section, Transport Standards Branch, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the International Section, send it to the attention of the person identified in paragraph (k)(2) of this AD. Information may be emailed to [9-ANM-116-AMOC-REQUESTS@faa.gov](mailto:9-ANM-116-AMOC-REQUESTS@faa.gov). Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(2) *Contacting the Manufacturer*: As of the effective date of this AD, for any requirement in this AD to obtain corrective actions from a manufacturer, the action must be accomplished using a method approved by the Manager, International Section, Transport Standards Branch, FAA; or EASA; or Airbus's EASA DOA. If approved by the DOA, the approval must include the DOA-authorized signature.

#### (k) Related Information

(1) Refer to Mandatory Continuing Airworthiness Information (MCAI) EASA AD 2016-0181, dated September 13, 2016, for related information. This MCAI may be found in the AD docket on the internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2017-0716.

(2) For more information about this AD, contact Sanjay Ralhan, Aerospace Engineer, International Section, Transport Standards Branch, FAA, 1601 Lind Avenue SW, Renton, WA 98057-3356; telephone: 425-227-1405; fax: 425-227-1149.

(3) Service information identified in this AD that is not incorporated by reference is available at the addresses specified in paragraphs (l)(4) and (l)(5) of this AD.

#### (l) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless this AD specifies otherwise.

(3) The following service information was approved for IBR on March 3, 2016 (81 FR 4878, January 28, 2016).

(i) Airbus Service Bulletin A320-57-1028, Revision 02, dated June 3, 2013.

(ii) Airbus Service Bulletin A320-57-1029, Revision 02, dated June 16, 1999.

(4) For service information identified in this AD, contact Airbus, Airworthiness Office—EIAS, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone: +33 5 61 93 36 96; fax: +33 5 61 93 44 51; email: [account.airworth-eas@airbus.com](mailto:account.airworth-eas@airbus.com); internet: <http://www.airbus.com>.

(5) You may view this service information at the FAA, Transport Standards Branch,

1601 Lind Avenue SW, Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

(6) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Renton, Washington, on January 10, 2018.

**John P. Piccola, Jr.,**

Acting Director, System Oversight Division, Aircraft Certification Service.

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**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA-2017-1201; Product Identifier 2017-SW-068-AD; Amendment 39-19155; AD 2018-02-02]

RIN 2120-AA64

#### Airworthiness Directives; Airbus Helicopters

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Final rule; request for comments.

**SUMMARY:** We are adopting a new airworthiness directive (AD) for Airbus Helicopters Model AS350B, AS350BA, AS350B1, AS350B2, AS350B3, AS350D, AS355E, AS355F, AS355F1, AS355F2, AS355N, AS355NP, EC130B4, and EC130T2 helicopters. This AD requires inspecting the main rotor (M/R) mast jet oil lubrication hose (oil hose). This AD is prompted by a report of a blocked oil hose. The actions of this AD are intended to prevent an unsafe condition on these helicopters.

**DATES:** This AD becomes effective February 12, 2018.

We must receive comments on this AD by March 27, 2018.

**ADDRESSES:** You may send comments by any of the following methods:

- *Federal eRulemaking Docket:* Go to <http://www.regulations.gov>. Follow the online instructions for sending your comments electronically.

- *Fax:* 202-493-2251.

- *Mail:* Send comments to the U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE, Washington, DC 20590-0001.