

same novel or unusual design feature, these special conditions would also apply to the other model under § 21.101.

In addition to the applicable airworthiness regulations and special conditions, the Beechcraft Model 400A airplanes must comply with the fuel-vent and exhaust-emission requirements of 14 CFR part 34, and the noise-certification requirements of 14 CFR part 36.

The FAA issues special conditions, as defined in 14 CFR 11.19, in accordance with § 11.38, and they become part of the type certification basis under § 21.101.

Novel or Unusual Design Features

The Beechcraft Model 400A airplanes, as modified by Garmin, will incorporate the following novel or unusual design feature:

Installation of the Garmin G5000 satellite-based navigation system into the airplanes.

Discussion

The Garmin G5000 satellite-based navigation-system design, installed in Beechcraft Model 400A airplanes, introduces the potential for unauthorized persons, accessing the passenger-services domain, to access the airplane-control domain and airplane information-services domain; and further may introduce security vulnerabilities related to the introduction of viruses, worms, user errors, and intentional sabotage of airplane networks, systems, and databases.

The operating systems for current airplane systems usually are proprietary. Therefore, they are not as susceptible to corruption from worms, viruses, and other malicious actions as are more widely used commercial operating systems, such as Microsoft Windows, because access to the design details of these proprietary operating systems is limited to the system developer and airplane integrator. Some systems installed on the Beechcraft Model 400A airplanes will use operating systems that are widely used and commercially available from third-party software suppliers. The security vulnerabilities of these operating systems may be more widely known than proprietary operating systems currently used by avionics manufacturers.

These special conditions contain the additional safety standards that the Administrator considers necessary to establish a level of safety equivalent to that established by the existing airworthiness standards.

Applicability

As discussed above, these special conditions are applicable to Beechcraft Model 400A airplanes modified by Garmin. Should Garmin apply at a later date for a supplemental type certificate to incorporate the same novel or unusual design feature for any other model included on the same type certificate, these special conditions would apply to that model as well.

Conclusion

This action affects only a certain novel or unusual design feature on one model of airplane. It is not a rule of general applicability and affects only the applicant who applied to the FAA for approval of this feature on the airplane.

List of Subjects in 14 CFR Part 25

Aircraft, Aviation safety, Reporting and recordkeeping requirements.

The authority citation for these special conditions is as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701, 44702, 44704.

The Special Conditions

Accordingly, pursuant to the authority delegated to me by the Administrator, the following special conditions are issued as part of the type certification basis for Beechcraft Model 400A airplanes modified by Garmin.

1. The applicant must ensure that the design provides isolation from, or airplane electronic-system security protection against, access by unauthorized sources internal to the airplane. The design must prevent inadvertent and malicious changes to, and all adverse impacts upon, airplane equipment, systems, networks, or other assets required for safe flight and operations.

2. The applicant must establish appropriate procedures to allow the operator to ensure that continued airworthiness of the airplane is maintained, including all post-type-certification modifications that may have an impact on the approved electronic-system security safeguards.

Issued in Renton, Washington, on July 31, 2017.

Victor Wicklund,

*Manager, Transport Standards Branch,
Aircraft Certification Service.*

[FR Doc. 2017-17071 Filed 8-11-17; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 25

[Docket No. FAA-2017-0316; Special Conditions No. 25-699-SC]

Special Conditions: Embraer S.A. Model ERJ 190-300 Airplane; Flight Envelope Protection: High Incidence Protection System

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final special conditions; request for comments.

SUMMARY: These special conditions are issued for the Embraer S.A. (Embraer) Model ERJ 190-300 airplane. This airplane will have a novel or unusual design feature when compared to the state of technology envisioned in the airworthiness standards for transport-category airplanes. This design feature is a high-incidence protection function that limits the angle of attack (AOA) at which the airplane can be flown during normal low-speed operation, and that cannot be overridden by the flightcrew. The applicable airworthiness regulations do not contain adequate or appropriate safety standards for this design feature. These special conditions contain the additional safety standards that the Administrator considers necessary to establish a level of safety equivalent to that established by the existing airworthiness standards.

DATES: This action is effective on Embraer S.A. on August 14, 2017. We must receive your comments by September 28, 2017.

ADDRESSES: Send comments identified by docket number FAA-2017-0316 using any of the following methods:

- **Federal eRegulations Portal:** Go to <http://www.regulations.gov/> and follow the online instructions for sending your comments electronically.

- **Mail:** Send comments to Docket Operations, M-30, U.S. Department of Transportation (DOT), 1200 New Jersey Avenue SE., Room W12-140, West Building Ground Floor, Washington, DC 20590-0001.

- **Hand Delivery or Courier:** Take comments to Docket Operations in Room W12-140 of the West Building Ground Floor at 1200 New Jersey Avenue SE., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

- **Fax:** Fax comments to Docket Operations at 202-493-2251.

Privacy: The FAA will post all comments it receives, without change, to <http://www.regulations.gov/>,

including any personal information the commenter provides. Using the search function of the docket Web site, anyone can find and read the electronic form of all comments received into any FAA docket, including the name of the individual sending the comment (or signing the comment for an association, business, labor union, etc.). DOT's complete Privacy Act Statement can be found in the **Federal Register** published on April 11, 2000 (65 FR 19477–19478).

Docket: Background documents or comments received may be read at <http://www.regulations.gov/> at any time. Follow the online instructions for accessing the docket or go to Docket Operations in Room W12–140 of the West Building Ground Floor at 1200 New Jersey Avenue SE., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

FOR FURTHER INFORMATION CONTACT: Joe Jacobsen, FAA, Airplane and Flight Crew Interface, ANM–111, Transport Airplane Directorate, Aircraft Certification Service, 1601 Lind Avenue SW., Renton, Washington 98057–3356; telephone 425–227–2011; facsimile 425–227–1320.

SUPPLEMENTARY INFORMATION: The substance of these special conditions has been subject to the notice and comment period in several prior instances and has been derived without substantive change from those previously issued. It is unlikely that prior public comment would result in a significant change from the substance contained herein. Therefore, because a delay would significantly affect the certification of the airplane, the FAA has determined that prior public notice and comment are unnecessary and impracticable.

In addition, since the substance of these special conditions has been subject to the public comment process in several prior instances with no substantive comments received, the FAA finds it unnecessary to delay the effective date and finds that good cause exists for adopting these special conditions upon publication in the **Federal Register**.

The FAA is requesting comments to allow interested persons to submit views that may not have been submitted in response to the prior opportunities for comment described above.

Comments Invited

We invite interested people to take part in this rulemaking by sending written comments, data, or views. The most helpful comments reference a specific portion of the special conditions, explain the reason for any

recommended change, and include supporting data.

We will consider all comments we receive by the closing date for comments. We may change these special conditions based on the comments we receive.

Background

On September 13, 2013, Embraer applied for an amendment to Type Certificate No. A57NM to include the new Model ERJ 190–300 airplane. The Model ERJ 190–300 airplane, which is a derivative of the Embraer Model ERJ 190–100 STD airplane currently approved under Type Certificate No. A57NM, is a 97- to 114-passenger transport-category airplane, designed with a new wing with a high aspect ratio and raked wingtip, and a new electrical-distribution system. The maximum take-off weight is 124,340 lbs (56,400 kg).

Type Certification Basis

Under the provisions of title 14, Code of Federal Regulations (14 CFR) 21.101, Embraer must show that the Model ERJ 190–300 airplane meets the applicable provisions of the regulations listed in Type Certificate No. A57NM, or the applicable regulations in effect on the date of application for the change, except for earlier amendments as agreed upon by the FAA.

If the Administrator finds that the applicable airworthiness regulations (*i.e.*, 14 CFR part 25) do not contain adequate or appropriate safety standards for the Model ERJ 190–300 airplane because of a novel or unusual design feature, special conditions are prescribed under the provisions of § 21.16.

Special conditions are initially applicable to the model for which they are issued. Should the type certificate for that model be amended later to include any other model that incorporates the same novel or unusual design feature, or should any other model already included on the same type certificate be modified to incorporate the same novel or unusual design feature, these special conditions would also apply to the other model under § 21.101.

In addition to the applicable airworthiness regulations and special conditions, the Embraer Model ERJ 190–300 airplane must comply with the fuel-vent and exhaust-emission requirements of 14 CFR part 34, and the noise-certification requirements of 14 CFR part 36.

The FAA issues special conditions, as defined in 14 CFR 11.19, in accordance with § 11.38, and they become part of

the type certification basis under § 21.101.

Novel or Unusual Design Features

The Embraer Model ERJ 190–300 airplane will incorporate the following novel or unusual design feature:

A high-incidence protection function that limits the angle of attack (AOA) at which the airplane can be flown during normal low-speed operation, and that cannot be overridden by the flightcrew.

Discussion

The Model ERJ 190–300 airplane design has a complex, full-digital flight-control system, referred to as fly-by-wire (FBW) architecture. This FBW architecture provides closed-loop flight-control laws and multiple protection functions.

The Model ERJ 190–300 airplane is equipped with a high-incidence protection function that limits the angle of attack (AOA) at which the airplane can be flown during normal low-speed operation, and cannot be overridden by the flightcrew. The application of this AOA limit impacts the stall-speed determination, the stall characteristics and stall-warning demonstration, and the longitudinal airplane-handling characteristics. The high-incidence protection function prevents the airplane from stalling at low speeds and, therefore, a stall warning system is backed up during normal flight conditions. If the high-incidence protection function has a failure that is not shown to be extremely improbable, stall warning must be provided in a conventional manner. Also, the flight characteristics at the AOA for $V_{CL\max}$ (AOA at the maximum lift co-efficient and associated speed) must be suitable in the traditional sense. Per 14 CFR 21.16, therefore, special conditions are needed to address the unique features of the high-incidence protection function installed on the Model ERJ 190–300 airplane.

These special conditions contain the additional safety standards that the Administrator considers necessary to establish a level of safety equivalent to that established by the existing airworthiness standards.

Applicability

As discussed above, these special conditions are applicable to the Model ERJ 190–300 airplane. Should Embraer apply at a later date for a change to the type certificate to include another model incorporating the same novel or unusual design feature, these special conditions would apply to that model as well.

Conclusion

This action affects only certain novel or unusual design features on one model of airplane. It is not a rule of general applicability.

List of Subjects in 14 CFR Part 25

Aircraft, Aviation safety, Reporting and recordkeeping requirements.

The authority citation for these special conditions is as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701, 44702, 44704.

The Special Conditions

Accordingly, pursuant to the authority delegated to me by the Administrator, the following special conditions are issued as part of the type certification basis for the Embraer Model ERJ 190–300 airplane.

Flight Envelope Protection: High Incidence Protection System

The current airworthiness standards do not contain adequate safety standards for the unique features of the high incidence protection system on the Embraer Model ERJ 190–300 airplane. Part I of the following special conditions are issued in lieu of the specified paragraphs of §§ 25.103, 25.145, 25.201, 25.203, 25.207, and 25.1323 of title 14, Code of Federal Regulations (CFR) or in addition to the requirements of § 25.21. Part II are in lieu of the specified paragraphs of §§ 25.103, 25.105, 25.107, 25.121, 25.123, 25.125, 25.143, and 25.207 of title 14 CFR.

Special Conditions Part I

Stall Protection and Scheduled Operating Speeds

The following special conditions are in lieu of §§ 25.103, 25.145(a), 25.145(b)(6), 25.201, 25.203, 25.207, and 25.1323(d) or in addition to the requirements of § 25.21.

Foreword

In the following paragraphs, “in icing conditions” means with the ice accretions (relative to the relevant flight phase) as defined in 14 CFR part 25, Amendment 121, appendix C.

1. Definitions

The following are terms relating to the novel or unusual design features these special conditions address:

- *High incidence protection system:* A system that operates directly and automatically on the airplane's flying controls to limit the maximum angle of attack that can be attained to a value below that at which an aerodynamic stall would occur.

- *Alpha-limit:* The maximum angle of attack at which the airplane stabilizes with the high incidence protection system operating, and the longitudinal control held on its aft stop.

- *V_{min}:* The minimum steady flight speed in the airplane configuration under consideration with the high incidence protection system operating. See Part I, section 3 of these special conditions.

- *V_{min}1g:* V_{min} corrected to 1g conditions. See Part I, section 3 of these special conditions. It is the minimum calibrated airspeed at which the airplane can develop a lift force normal to the flight path and equal to its weight when at an angle of attack not greater than that determined for V_{min}.

2. Capability and Reliability of the High Incidence Protection System

The capability and reliability of the high incidence protection system can be established by flight test, simulation, and analysis as appropriate. The capability and reliability required are as follows:

1. It must not be possible during pilot-induced maneuvers to encounter a stall, and handling characteristics must be acceptable, as required by section 5 of Part I of these special conditions.

2. The airplane must be protected against stalling due to the effects of wind-shears and gusts at low speeds as required by section 6 of Part I of these special conditions.

3. The ability of the high incidence protection system to accommodate any reduction in stalling incidence must be verified in icing conditions.

4. The high incidence protection system must be provided in each abnormal configuration of the high lift devices that is likely to be used in flight following system failures.

5. The reliability of the system and the effects of failures must be acceptable in accordance with § 25.1309.

3. Minimum Steady Flight Speed and Reference Stall Speed

In lieu of § 25.103, the following requirements apply:

- (a) The minimum steady flight speed, V_{min}, is the final stabilized calibrated airspeed obtained when the airplane is decelerated until the longitudinal control is on its stop in such a way that the entry rate does not exceed 1 knot per second.

- (b) The minimum steady flight speed, V_{min}, must be determined if it is used to determine compliance with a required performance standard or other requirements demonstrations in icing or non-icing conditions with:

- (1) The high incidence protection system operating normally;

- (2) Idle thrust and automatic thrust system (if applicable) inhibited;

- (3) All combinations of flap settings and landing gear position for which V_{min} is required to be determined;

- (4) The weight used when reference stall speed, V_{SR}, is being used as a factor to determine compliance with a required performance standard;

- (5) The most unfavorable center of gravity allowable; and

- (6) The airplane trimmed for straight flight at a speed selected by the applicant, but not less than 1.13 V_{SR} and not greater than 1.3 V_{SR}.

- (c) The 1-g minimum steady flight speed, V_{min}1g, is the minimum calibrated airspeed at which the airplane can develop a lift force (normal to the flight path) equal to its weight, while at an angle of attack not greater than that at which the minimum steady flight speed of subparagraph (a) was determined. It must be determined if it is used to determine compliance with a required performance standard or other requirements demonstrations in icing or non-icing conditions.

- (d) The reference stall speed, V_{SR}, is a calibrated airspeed defined by the applicant. V_{SR} may not be less than a 1g stall speed. V_{SR} must be determined in non-icing conditions (and as an option, in icing conditions) and expressed as:

$$V_{SR} \geq \frac{V_{CL\ MAX}}{\sqrt{n_{zw}}}$$

where—

$V_{CL\ max}$ = Calibrated airspeed obtained when the load factor-corrected lift

coefficient ($\frac{n_{zw}W}{qS}$) is first a maximum during the maneuver

prescribed in sub-paragraph (e)(7) of this paragraph.

N_{zw} = Load factor normal to the flight path at $V_{CL\ max}$

W = Airplane gross weight;

S = Aerodynamic reference wing area; and

q = Dynamic pressure.

(e) $V_{CL\ max}$ is determined with:

(1) Engines idling, or, if that resultant thrust causes an appreciable decrease in stall speed, not more than zero thrust at the stall speed;

(2) The airplane in other respects (such as flaps, landing gear, and ice accretions) in the condition existing in the test or performance standard in which V_{SR} is being used;

(3) The weight used when V_{SR} is being used as a factor to determine compliance with a required performance standard;

(4) The center of gravity position that results in the highest value of reference stall speed;

(5) The airplane trimmed for straight flight at a speed achievable by the automatic trim system, but not less than 1.13 V_{SR} and not greater than 1.3 V_{SR} ; and

(6) The high incidence protection system adjusted or disabled, at the option of the applicant, to allow higher incidence than is possible with the normal production system.

(7) Starting from the stabilized trim condition, apply the longitudinal control to decelerate the airplane so that the speed reduction does not exceed 1 knot per second.

4. Stall Warning

In lieu of § 25.207, the following requirements apply:

4.1 Normal Operation

If the capability requirements of the high incidence protection system are successfully demonstrated, then the conditions of section 2, "Capability and Reliability of the High Incidence Protection System," are satisfied. These conditions provide safety equivalent to § 25.207, *Stall warning*, so the provision of an additional, unique warning device is not required.

4.2 High Incidence Protection System Failure

Following failures of the high incidence protection system, not shown to be extremely improbable, such that the capability of the system no longer satisfies items (1), (2), and (3) of section 2, "Capability and Reliability of the High Incidence Protection System," stall warning must be provided and must protect against encountering unacceptable stall characteristics and against encountering stall.

(a) Stall warning with the flaps and landing gear in any normal position must be clear and distinctive to the pilot and meet the requirements specified in paragraphs (d) and (e) below.

(b) Stall warning must also be provided in each abnormal configuration of the high lift devices that is likely to be used in flight following system failures.

(c) The warning may be furnished either through the inherent aerodynamic qualities of the airplane or by a device that will give clearly distinguishable indications under expected conditions of flight. However, a visual stall warning device that requires the attention of the crew within the cockpit is not acceptable by itself. If a warning device is used, it must provide a warning in each of the airplane configurations prescribed in paragraph (a) above and for the conditions prescribed in paragraphs (d) and (e) below.

(d) In non-icing conditions stall warning must provide sufficient margin to prevent encountering unacceptable stall characteristics and encountering stall in the following conditions:

(1) In power off (engine power or thrust at flight idle) straight deceleration not exceeding 1 knot per second to a speed 5 knots or 5 percent calibrated airspeed, whichever is greater, below the warning onset.

(2) In turning flight stall deceleration at entry rates up to 3 knots per second when recovery is initiated not less than 1 second after the warning onset.

(e) In icing conditions stall warning must provide sufficient margin to prevent encountering unacceptable characteristics and encountering stall, in power-off (engine power or thrust at flight idle) straight and turning flight decelerations not exceeding 1 knot per

second, when the pilot starts a recovery maneuver not less than three seconds after the onset of stall warning.

(f) An airplane is considered stalled when the behavior of the airplane gives the pilot a clear and distinctive indication of an acceptable nature that the airplane is stalled. Acceptable indications of a stall, occurring either individually or in combination are:

(1) A nose-down pitch that cannot be readily arrested;

(2) Buffeting, of a magnitude and severity that is strong and effective deterrent to further speed reduction; or

(3) The pitch control reaches the aft stop and no further increase in pitch attitude occurs when the control is held full aft for a short time before recovery is initiated.

(g) An aircraft exhibits unacceptable characteristics during straight or turning flight decelerations if it is not always possible to produce and to correct roll and yaw by unreversed use of aileron and rudder controls, or abnormal nose-up pitching occurs.

5. Handling Characteristics at High Incidence

In lieu of §§ 25.201 and 25.203, the following requirements apply:

5.1 High Incidence Handling Demonstration

In lieu of § 25.201:

(a) Maneuvers to the limit of the longitudinal control, in the nose-up pitch, must be demonstrated in straight flight and in 30° banked turns with:

(1) The high incidence protection system operating normally;

(2) Initial power conditions of:

i. Power off (engine power or thrust at flight idle); and

ii. The power necessary to maintain level flight at 1.5 V_{SR1} , where V_{SR1} is the reference stall speed with flaps in approach position, the landing gear retracted, and maximum landing weight;

(3) Flaps, landing gear, and deceleration devices in any likely combination of positions;

(4) Representative weights within the range for which certification is requested; and

(5) The airplane trimmed for straight flight at a speed selected by the applicant, but not less than 1.13 V_{SR} and not greater than 1.3 V_{SR} .

(b) The following procedures must be used to show compliance in non-icing and icing conditions:

(1) Starting at a speed sufficiently above the minimum steady flight speed to ensure that a steady rate of speed reduction can be established, apply the longitudinal control so that the speed

reduction does not exceed 1 knot per second until the control reaches the stop;

(2) The longitudinal control must be maintained at the stop until the airplane has reached a stabilized flight condition and must then be recovered by normal recovery techniques;

(3) Maneuvers with increased deceleration rates:

(i) In non-icing conditions, the requirements must also be met with increased rates of entry to the incidence limit, up to the maximum practical entry rate; and

(ii) In icing conditions, with the anti-icing system working normally, the requirements must also be met with increased rates of entry to the incidence limit, up to 3 knots per second; and

(4) Maneuver with ice accretion prior to operation of the normal anti-ice system. With the ice accretion prior to operation of the normal anti-ice system, the requirements must also be met in deceleration at 1 knot per second. The deceleration must be continued until one second after the activation of the tactile stall warning system or three seconds after reaching full back stick, whichever occurs first. A primary ice detection system must automatically activate the ice protection.

5.2 Characteristics in High Incidence Maneuvers

In lieu of § 25.203:

In icing and non-icing conditions:

(a) Throughout maneuvers with a rate of deceleration of not more than 1 knot per second, both in straight flight and in 30° banked turns, the airplane's characteristics must be as follows:

(1) There must not be any abnormal nose-up pitching.

(2) There must not be any uncommanded nose-down pitching, which would be indicative of stall. However, reasonable attitude changes associated with stabilizing the incidence at Alpha limit as the longitudinal control reaches the stop would be acceptable.

(3) There must not be any uncommanded lateral or directional motion and the pilot must retain good lateral and directional control, by conventional use of the controls, throughout the maneuver.

(4) The airplane must not exhibit buffeting of a magnitude and severity that would act as a deterrent from completing the maneuver specified in paragraph 5.1(a).

(b) In maneuvers with increased rates of deceleration, some degradation of characteristics is acceptable, associated with a transient excursion beyond the stabilized Alpha limit. However, the

airplane must not exhibit dangerous characteristics or characteristics that would deter the pilot from holding the longitudinal control on the stop for a period of time appropriate to the maneuver.

(c) It must always be possible to reduce incidence by conventional use of the controls.

(d) The rate at which the airplane can be maneuvered from trim speeds associated with scheduled operating speeds such as V_2 and V_{REF} up to Alpha limit must not be unduly damped or be significantly slower than can be achieved on conventionally controlled transport airplanes.

5.3 Characteristics Up to Maximum Lift Angle of Attack

Also in lieu of § 25.201:

(a) In non-icing conditions:

Maneuvers with a rate of deceleration of not more than 1 knot per second up to the angle of attack at which V_{CLmax} was obtained as defined in section 3, "Minimum Steady Flight Speed and Reference Stall Speed," must be demonstrated in straight flight and in 30° banked turns in the following configurations:

(1) The high incidence protection deactivated or adjusted, at the option of the applicant, to allow higher incidence than is possible with the normal production system;

(2) Automatic thrust increase system inhibited (if applicable);

(3) Engines idling;

(4) Flaps and landing gear in any likely combination of positions; and

(5) The airplane trimmed for straight flight at a speed selected by the applicant, but not less than 1.13 V_{SR} and not greater than 1.3 V_{SR} .

(b) In icing conditions:

Maneuvers with a rate of deceleration of not more than 1 knot per second up to the maximum angle of attack reached during maneuvers from paragraph 5.1(b)(3)(ii) must be demonstrated in straight flight with:

(1) The high incidence protection deactivated or adjusted, at the option of the applicant, to allow higher incidence than is possible with the normal production system;

(2) Automatic thrust increase system inhibited (if applicable);

(3) Engines idling;

(4) Flaps and landing gear in any likely combination of positions, and

(5) The airplane trimmed for straight flight at a speed achievable by the automatic trim system.

(c) During the maneuvers used to show compliance with paragraphs (a) and (b) above, the airplane must not exhibit dangerous characteristics, and it

must always be possible to reduce angle of attack by conventional use of the controls. The pilot must retain good lateral and directional control, by conventional use of the controls, throughout the maneuver.

6. Atmospheric Disturbances

Operation of the high incidence protection system must not adversely affect aircraft control during expected levels of atmospheric disturbances, nor impede the application of recovery procedures in case of wind-shear. This must be demonstrated in non-icing and icing conditions.

7. Proof of Compliance

In addition to the requirements of § 25.21, the following requirement applies:

(b) The flying qualities must be evaluated at the most unfavorable center-of-gravity position.

8. Sections 25.145(a), 25.145(b)(6), and 25.1323(d)

The following requirements apply:

- For § 25.145(a), add “ V_{min} ” in lieu of “stall identification.”

- For § 25.145(b)(6), and “ V_{min} ” in lieu of “ V_{sw} .”

- For § 25.1323(d), add “From 1.23 V_{SR} to $V_{min} . . .$ ” in lieu of, “1.23 V_{SR} to the speed at which stall warning begins . . .,” and, “. . . speeds below $V_{min} . . .$ ” in lieu of, “. . . speeds below stall warning.”

Special Conditions Part II

Credit for Robust Envelope Protection in Icing Conditions

The following special conditions are in lieu of the specified paragraphs of §§ 25.103, 25.105, 25.107, 25.121, 25.123, 25.125, 25.143, and 25.207.

1. Define the stall speed as provided in these special conditions, Part I, in lieu of § 25.103.

2. In lieu of § 25.105(a)(2)(i), the following requirement applies:

(i) The V_2 speed scheduled in non-icing conditions does not provide the maneuvering capability specified in § 25.143(h) for the takeoff configuration, or

3. In lieu of § 25.107(c) and (g), the following requirements apply, with additional sections (c') and (g'):

Takeoff speeds:

(c) In non-icing conditions V_2 , in terms of calibrated airspeed, must be selected by the applicant to provide at least the gradient of climb required by § 25.121(b) but may not be less than—

(1) V_{2MIN} ;

(2) V_R plus the speed increment attained (in accordance with § 25.111(c)(2)) before reaching a height of 35 feet above the takeoff surface; and

(3) A speed that provides the maneuvering capability specified in § 25.143(h).

(c') In icing conditions with the “takeoff ice” accretion defined in part 25, appendix C, V_2 may not be less than—

(1) The V_2 speed determined in non-icing conditions; and

(2) A speed that provides the maneuvering capability specified in § 25.143(h).

(g) In non-icing conditions, V_{FTO} , in terms of calibrated airspeed, must be selected by the applicant to provide at least the gradient of climb required by § 25.121(c), but may not be less than—

(1) 1.18 V_{SR} ; and

(2) A speed that provides the maneuvering capability specified in § 25.143(h).

(g') In icing conditions with the “final takeoff ice” accretion defined in part 25, appendix C, V_{FTO} , may not be less than—

(1) The V_{FTO} speed determined in non-icing conditions.

(2) A speed that provides the maneuvering capability specified in § 25.143(h).

4. In lieu of §§ 25.121(b)(2)(ii)(A), 25.121(c)(2)(ii)(A), and 25.121(d)(2)(ii), the following requirements apply:

In lieu of § 25.121(b)(2)(ii)(A):

(A) The V_2 speed scheduled in non-icing conditions does not provide the maneuvering capability specified in § 25.143(h) for the takeoff configuration; or

In lieu of § 25.121(c)(2)(ii)(A):

(A) The V_{FTO} speed scheduled in non-icing conditions does not provide the maneuvering capability specified in § 25.143(h) for the en-route configuration; or

In lieu of § 25.121(d)(2)(ii):

(d)(2) The requirements of subparagraph (d)(1) of this paragraph must be met: (ii) In icing conditions with the approach ice accretion defined in appendix C, in a configuration corresponding to the normal all-engines-operating procedure in which V_{min1g} for this configuration does not exceed 110% of the V_{min1g} for the related all-engines-operating landing configuration in icing, with a climb speed established with normal landing procedures, but not more than 1.4 V_{SR} (V_{SR} determined in non-icing conditions).

5. In lieu of § 25.123(b)(2)(i), the following requirements apply:

(i) The minimum en-route speed scheduled in non-icing conditions does not provide the maneuvering capability specified in § 25.143(h) for the en-route configuration, or

6. In lieu of §§ 25.125(b)(2)(ii)(B) and 25.125(b)(2)(ii)(C), the following requirements apply:

(B) A speed that provides the maneuvering capability specified in § 25.143(h) with the landing ice accretion defined in part 25, appendix C.

(C) 1.17 V_{min1g} .

7. In lieu of § 25.143(j)(1), the following requirement applies:

(1) The airplane is controllable in a pull-up maneuver up to 1.5 g load factor or lower if limited by angle of attack protection; and

8. In lieu of § 25.207, *Stall warning*, to read as the requirements defined in these special conditions Part I, Section 4.

Issued in Renton, Washington, on July 31, 2017.

Victor Wicklund,

*Manager, Transport Standards Branch,
Aircraft Certification Service.*

[FR Doc. 2017-17072 Filed 8-11-17; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 25

[Docket No. FAA-2017-0484; Special Conditions No. 25-700-SC]

Special Conditions: Textron Aviation Inc. Model 700 Airplanes; Use of Automatic Power Reserve for Go-Around Performance Credit

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final special conditions; request for comments.

SUMMARY: These special conditions are issued for the Textron Aviation Inc. (Textron) Model 700 airplane. This airplane will have a novel or unusual design feature when compared to the state of technology envisioned in the airworthiness standards for transport-category airplanes. This design feature is an Automatic Takeoff Thrust Control System (ATTCS), referred to as an Automatic Power Reserve (APR), to set the performance level for approach-climb operation after an engine failure. The applicable airworthiness regulations do not contain adequate or appropriate safety standards for this design feature. These special conditions contain the additional safety standards that the Administrator considers necessary to establish a level of safety equivalent to that established by the existing airworthiness standards.

DATES: This action is effective on Textron on August 14, 2017. Send your comments by September 28, 2017.