

intended to protect the airplane and its occupants if failure occurs.

Special conditions 3, 7, and 8 are self-explanatory.

Special condition no. 4 makes it clear that the flammable fluid fire protection requirements of § 25.863 apply to non-rechargeable lithium battery installations. Section 25.863 is applicable to areas of the airplane that could be exposed to flammable fluid leakage from airplane systems. Non-rechargeable lithium batteries contain an electrolyte that is a flammable fluid.

Special condition no. 5 requires that each non-rechargeable lithium battery installation not damage surrounding structure or adjacent systems, equipment, or electrical wiring from corrosive fluids or gases that may escape in such a way as to cause a major or more severe failure condition.

While special condition no. 5 addresses corrosive fluids and gases, special condition no. 6 addresses heat. Special condition no. 6 requires that each non-rechargeable lithium battery installation have provisions to prevent any hazardous effect on airplane structure or systems caused by the maximum amount of heat the battery installation can generate due to any failure of it or its individual cells. The means of meeting special conditions nos. 5 and 6 may be the same, but the requirements are independent and address different hazards.

These special conditions apply to all non-rechargeable lithium battery installations in lieu of § 25.1353(b)(1) through (4) at Amendment 25-123 or § 25.1353(c)(1) through (4) at earlier amendments. Those regulations remain in effect for other battery installations.

These special conditions contain the additional safety standards that the Administrator considers necessary to establish a level of safety equivalent to that established by the existing airworthiness standards.

Applicability

These special conditions are applicable to the 787-9 series airplanes. Should Boeing apply at a later date for a change to the type certificate to include another model incorporating the same novel or unusual design feature, these special conditions would apply to that model as well.

These special conditions are only applicable to design changes applied for after the effective date.

These special conditions are not applicable to changes to previously certified non-rechargeable lithium battery installations where the only change is either cosmetic or to relocate the installation to improve the safety of

the airplane and occupants. Previously certified non-rechargeable lithium battery installations, as used in this paragraph, are those installations approved for certification projects applied for on or before the effective date of these special conditions. A cosmetic change is a change in appearance only, and does not change any function or safety characteristic of the battery installation. These special conditions are also not applicable to unchanged, previously certified non-rechargeable lithium battery installations that are affected by a change in a manner that improves the safety of its installation. The FAA determined that these exclusions are in the public interest because the need to meet all of the special conditions might otherwise deter these design changes that improve safety.

Conclusion

This action affects only a certain novel or unusual design feature on one model of airplane. It is not a rule of general applicability.

The substance of these special conditions has been subjected to the notice and comment period in prior instances and has been derived without substantive change from those previously issued. It is unlikely that prior public comment would result in a significant change from the substance contained herein. Therefore, the FAA has determined that prior public notice and comment are unnecessary and impracticable, and good cause exists for adopting these special conditions upon publication in the **Federal Register**. The FAA is requesting comments to allow interested persons to submit views that may not have been submitted in response to the prior opportunities for comment described above.

List of Subjects in 14 CFR Part 25

Aircraft, Aviation safety, Reporting and recordkeeping requirements.

The authority citation for these special conditions is as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701, 44702, 44704.

The Special Conditions

Accordingly, pursuant to the authority delegated to me by the Administrator, the following special conditions are issued as part of the type certification basis for the Boeing Model 787-9 series airplanes.

Non-Rechargeable Lithium Battery Installations

In lieu of § 25.1353(b)(1) through (4) at Amendment 25-123 or § 25.1353(c)(1) through (4) at earlier amendments, each

non-rechargeable lithium battery installation must:

1. Be designed to maintain safe cell temperatures and pressures under all foreseeable operating conditions to prevent fire and explosion.

2. Be designed to prevent the occurrence of self-sustaining, uncontrollable increases in temperature or pressure.

3. Not emit explosive or toxic gases, either in normal operation or as a result of its failure, that may accumulate in hazardous quantities within the airplane.

4. Meet the requirements of § 25.863.

5. Not damage surrounding structure or adjacent systems, equipment, or electrical wiring from corrosive fluids or gases that may escape in such a way as to cause a major or more severe failure condition.

6. Have provisions to prevent any hazardous effect on airplane structure or systems caused by the maximum amount of heat it can generate due to any failure of it or its individual cells.

7. Have a failure sensing and warning system to alert the flightcrew if its failure affects safe operation of the airplane.

8. Have a means for the flightcrew or maintenance personnel to determine the battery charge state if the battery's function is required for safe operation of the airplane.

Note: A battery system consists of the battery and any protective, monitoring, and alerting circuitry or hardware inside or outside of the battery. It also includes vents (where necessary) and packaging. For the purpose of these special conditions, a "battery" and "battery system" are referred to as a battery.

Issued in Renton, Washington, on May 17, 2017.

Michael Kaszycki,

Assistant Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2017-10843 Filed 5-25-17; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2017-0186; Directorate Identifier 2017-NE-07-AD; Amendment 39-18899; AD 2017-10-25]

RIN 2120-AA64

Airworthiness Directives; Rolls-Royce Deutschland Ltd & Co KG Turbofan Engines

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule; request for comments.

SUMMARY: We are adopting a new airworthiness directive (AD) for all Rolls-Royce Deutschland Ltd & Co KG (RRD) model Spey 506–14A, Spey 555–15, Spey 555–15H, Spey 555–15N, and Spey 555–15P turbofan engines. This AD requires reducing the maximum approved life limits for certain high-pressure compressor (HPC) stage 12 rotor disks. We are issuing this AD to correct the unsafe condition on these products.

DATES: This AD becomes effective June 12, 2017.

We must receive comments on this AD by July 10, 2017.

ADDRESSES: You may send comments by any of the following methods:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.
- *Mail:* U.S. Department of Transportation, 1200 New Jersey Avenue SE., West Building Ground Floor, Room W12–140, Washington, DC 20590–0001.
- *Hand Delivery:* Deliver to Mail address above between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.
- *Fax:* 202–493–2251.

For service information identified in this AD, contact Rolls-Royce Deutschland Ltd & Co KG, Eschenweg 11–15827 Dahlewitz, Blankenfelde-Mahlow, Germany; phone: +49 0 33–7086–1944; fax: +49 0 33–7086–3276. You may view this service information at the FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA 01803. For information on the availability of this material at the FAA, call 781–238–7125.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA–2017–0186; or in person at the Docket Operations office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the mandatory continuing airworthiness information (MCAI), regulatory evaluation, any comments received, and other information. The address for the Docket Office (phone: 800–647–5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

FOR FURTHER INFORMATION CONTACT: Robert Green, Aerospace Engineer, Engine Certification Office, FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA 01803; phone: 781–238–7754; fax: 781–238–7199; email: robert.green@faa.gov.

SUPPLEMENTARY INFORMATION:

Comments Invited

This AD is a final rule that involves requirements affecting flight safety, and we did not precede it by notice and opportunity for public comment. We invite you to send any written relevant data, views, or arguments about this AD. Send your comments to an address listed under the **ADDRESSES** section. Include “Docket No. FAA–2017–0186; Directorate Identifier 2017–NE–07–AD” at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this AD. We will consider all comments received by the closing date and may amend this AD because of those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact with FAA personnel concerning this AD.

Discussion

The European Aviation Safety Agency (EASA), which is the Technical Agent for the Member States of the European Community, has issued EASA AD 2017–0014, dated January 30, 2017 (referred to hereinafter as “the MCAI”), to correct an unsafe condition for the specified products. The MCAI states:

Based on revised stress analysis and life calculation, Rolls-Royce Deutschland (RRD) determined new provisional life limits for high pressure compressor (HPC) stage 12 rotor disks, Part Number (P/N) EU25917, P/N EU56963, P/N JR10242 and P/N JR18449, reducing the maximum approved life limits currently defined in the RRD Spey 555–15 Engine Maintenance Manual (EMM), Chapter 5–10–1, currently at the revision dated July 2015 and the Engine Overhaul Manual (EOM), Chapter 5–10, revision dated November 2014. The Spey 506–14A EMM, Chapter 5–10–1 revision dated October 1993 as well as the Spey 506–14A EOM, Chapter 5–10 revision dated November 1992 already contain the applicable life limit. Failure to replace an affected HPC stage 12 rotor disk before exceeding these limits, could lead to an uncontained engine failure, possibly resulting in damage to, and/or reduced control of, the aeroplane. To address this potential unsafe condition, RRD issued Alert

Non-Modification Service Bulletin (NMSB) Sp72–A1071 to provide instructions to determine (re-calculate) the consumed and remaining service life for each part. For the reasons described above, this AD requires re-calculation of the service life (consumed and remaining) of the affected HPC stage 12 rotor disks and, depending on the results, implementation of the life limits. It is expected that the affected reduced life limits are introduced into a next revision of the RRD Spey 555–15 Engine EMM and EOM.

You may obtain further information by examining the MCAI in the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA–2017–0186.

Related Service Information

RRD has issued Alert Non-Modification Service Bulletin (NMSB) Sp72–A1071, Revision 1, dated January 27, 2017. The Alert NMSB provides instructions to re-calculate the consumed and remaining service life for HPC stage 12 rotor disks, part number (P/N) EU25917, P/N EU56963, P/N JR10242, and P/N JR18449. This service information is available by the means identified in the **ADDRESSES** section.

FAA’s Determination and Requirements of This AD

This product has been approved by the aviation authority of Germany, and is approved for operation in the United States. Pursuant to our bilateral agreement with the European Community, EASA has notified us of the unsafe condition described in the MCAI. We are issuing this AD because we evaluated all information provided by EASA and determined the unsafe condition exists and is likely to exist or develop on other products of the same type design. This AD requires reducing the maximum approved life limits and re-calculating the consumed and remaining service life for HPC stage 12 rotor disks P/N EU25917, P/N EU56963, P/N JR10242, and P/N JR18449.

FAA’s Determination of the Effective Date

No domestic operators use this product. Therefore, we find that notice and opportunity for prior public comment are unnecessary and that good cause exists for making this amendment effective in less than 30 days.

Costs of Compliance

We estimate that this AD affects 0 engines installed on airplanes of U.S. registry. We estimate the following costs to comply with this AD:

ESTIMATED COSTS

Action	Labor cost	Parts cost	Cost per product	Cost on U.S. operators
Pro-rated lost life	1 work-hour × \$85 per hour = \$85	\$3,900	\$3,985	\$0

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA’s authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. “Subtitle VII: Aviation Programs,” describes in more detail the scope of the Agency’s authority.

We are issuing this rulemaking under the authority described in “Subtitle VII, Part A, Subpart III, Section 44701: General requirements.” Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this AD:

(1) Is not a “significant regulatory action” under Executive Order 12866,

(2) Is not a “significant rule” under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),

(3) Will not affect intrastate aviation in Alaska to the extent that it justifies making a regulatory distinction, and

(4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

2017–10–25 Rolls-Royce Deutschland Ltd & Co KG: Amendment 39–18899; Docket No. FAA–2017–0186; Directorate Identifier 2017–NE–07–AD.

(a) Effective Date

This AD is effective June 12, 2017.

(b) Affected ADs

None.

(c) Applicability

This AD applies to Rolls-Royce Deutschland Ltd & Co KG (RRD) Spey 506–14A, Spey 555–15, Spey 555–15H, Spey 555–15N, and Spey 555–15P turbofan engines with high-pressure compressor (HPC) stage 12 rotor disks, part number (P/N) EU25917, P/N EU56963, P/N JR10242, or P/N JR18449, installed.

(d) Subject

Joint Aircraft System Component (JASC) 7230, Turbine Engine Compressor Section.

(e) Reason

This AD was prompted by RRD re-calculating the life limits for HPC stage 12 rotor disks, P/N EU25917, P/N EU56963, P/N JR10242, and P/N JR18449. We are issuing this AD to prevent failure of the HPC stage 12 rotor disk, uncontained HPC stage 12 rotor disk release, damage to the engine, and damage to the airplane.

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) Required Actions

(1) Within 30 days after the effective date of this AD, determine if:

- (i) The affected part was ever operated in a Spey 555–15, Spey 555–15H, Spey 555–15N, or Spey 555–15P engine model, or
- (ii) the affected part was operated solely in a Spey 506–14A engine.

(2) If the affected part was operated solely in a Spey 506–14A engine with no history of operating in a Spey 555–15, Spey 555–15H, Spey 555–15N, or Spey 555–15P engine, no further action is required.

(3) If the affected part was operated in both Spey 506–14A and Spey 555–15, Spey 555–15H, Spey 555–15N, or Spey 555–15P engine models, or solely in Spey 555–15, Spey 555–15H, Spey 555–15N, or Spey 555–15P engines, re-calculate the consumed cyclic life (and remaining service life) using the Maximum Approved Life for each engine model and take-off monitoring procedure as defined in Figures 1 and 2 to paragraph (g) of this AD.

(4) After the effective date of this AD, the Maximum Approved Lives for the affected parts are as defined in Figure 2 to paragraph (g) of this AD. Calculate the consumed cyclic life (and remaining service life) using the Spey 555–15, Spey 555–15H, Spey 555–15N, or Spey 555–15P Maximum Approved Lives in Figure 2 to paragraph (g) of this AD.

(5) For Spey 506–14A engines with an affected part installed, that do not have an engine shop visit after the effective date of this AD before the re-calculated consumed cyclic life of the affected part exceeds 14,700 flight cycles (FC), remove the affected part from service before the re-calculated consumed cyclic life exceeds 14,700 FC, or 50 FC or 30 days after the effective date of this AD, whichever occurs later.

(6) For Spey 555–15, Spey 555–15H, Spey 555–15N, or Spey 555–15P engines with an affected part installed, that do not have an engine shop visit after the effective date of this AD before the re-calculated consumed cyclic life of the affected part exceeds the Maximum Approved Lives in Figure 2 to paragraph (g) of this AD, remove the affected part from service before the re-calculated consumed cyclic life exceeds the later of the following:

- (i) Maximum Approved Lives in Figure 2 to paragraph (g) of this AD, or
- (ii) 200 FC or 90 days after the effective date of this AD, or before exceeding the In-Service Replacement Limits defined in Figure 3 to paragraph (g) of this AD, whichever occurs first.

FIGURE 1 TO PARAGRAPH (g)—SPEY 506–14A HIGH-PRESSURE COMPRESSOR (HPC) STAGE 12 ROTOR DISK MAXIMUM APPROVED LIFE

	Flight cycles
HPC stage 12 rotor disk, P/N EU25917, EU56963, and JR10242	14,700

FIGURE 2 TO PARAGRAPH (g)—SPEY 555–15, SPEY 555–15H, SPEY 555–15N, OR SPEY 555–15P HPC STAGE 12 ROTOR DISK, P/N EU25917, EU56963, JR10242, AND JR18449, MAXIMUM APPROVED LIFE

Take-off monitoring procedure	Maximum approved lives (flight cycles)
(A) With no high-pressure (HP) revolutions per minute (RPM) monitoring	11,500
HP RPM monitoring; stated RPM not exceeded on more than 15% of occasions:	
(B) 100% N2	13,600
(C) 99% N2	17,100
(D) 98% N2	19,300
(E) 97% N2	20,500
(F) No HP RPM monitoring required Datum (Average N2 at 99.5%)	16,800

FIGURE 3 TO PARAGRAPH (g)—SPEY 555–15, SPEY 555–15H, SPEY 555–15N, OR SPEY 555–15P HPC STAGE 12 ROTOR DISK, P/N EU25917, EU56963, JR10242, AND JR18449, IN-SERVICE REPLACEMENT LIMITS

Take-off monitoring procedure	In-service replacement limits (flight cycles)
(A) With no HP RPM monitoring	13,800
HP RPM monitoring; stated RPM not exceeded on more than 15% of occasions:	
(B) 100% N2	15,600
(C) 99% N2	17,600
(D) 98% N2	19,700
(E) 97% N2	22,100
(F) No HP RPM monitoring required Datum (Average N2 at 99.5%)	17,300

(h) Installation Prohibition

After the effective date of this AD, installation of a serviceable spare engine or release to service of an engine after any shop visit is allowed, provided the affected part has not exceeded the Maximum Approved Lives in Figures 1 or 2 to paragraph (g) of this AD.

(i) Definition

For the purpose of this AD, a shop visit is the induction of an engine into the shop for maintenance or overhaul. The separation of engine flanges solely for the purpose of transporting the engine without subsequent engine maintenance does not constitute an engine shop visit.

(j) Alternative Methods of Compliance (AMOCs)

The Manager, Engine Certification Office, FAA, may approve AMOCs for this AD. Use the procedures found in 14 CFR 39.19 to make your request. You may email your request to: *ANE-AD-AMOC@faa.gov*.

(k) Related Information

(1) For more information about this AD, contact Robert Green, Aerospace Engineer, Engine Certification Office, FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA 01803; phone: 781–238–7754; fax: 781–238–7199; email: *robert.green@faa.gov*.

(2) Refer to MCAI European Aviation Safety Agency (EASA), AD 2017–0014, dated January 30, 2017, for more information. You may examine the MCAI in the AD docket on the Internet at *http://www.regulations.gov* by searching for and locating it in Docket No. FAA–2017–0186.

(3) RRD Alert Non-Modification Service Bulletin Sp72–A1071, Revision 1, dated January 27, 2017, which is not incorporated by reference in this AD, can be obtained from RRD, using the contact information in paragraph (k)(4) of this AD.

(4) For RRD service information identified in this AD, contact Rolls-Royce Deutschland Ltd & Co KG, Eschenweg 11–15827 Dahlewitz, Blankenfelde-Mahlow, Germany; phone: +49 0 33–7086–1944; fax: +49 0 33–7086–3276.

(5) You may view this service information at FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA 01803. For information on the availability of this material at the FAA, call 781–238–7125.

Issued in Burlington, Massachusetts, on May 9, 2017.

Robert J. Ganley,
Acting Manager, Engine & Propeller Directorate, Aircraft Certification Service.
[FR Doc. 2017–10437 Filed 5–25–17; 8:45 am]

BILLING CODE 4910–13–P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA–2017–0451; Directorate Identifier 2017–CE–015–AD; Amendment 39–18885; AD 2017–10–11]

RIN 2120–AA64

Airworthiness Directives; Stemme AG Gliders

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule; request for comments.

SUMMARY: We are adopting a new airworthiness directive (AD) for Stemme AG Model S10–VT gliders (type certificate previously held by Stemme GmbH & Co. KG). This AD results from mandatory continuing airworthiness information (MCAI) issued by the aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as certain propeller front transmission gear wheels having insufficient material strength because of improper heat treatment