

2016, for related information. The MCAI can be found in the AD docket on the Internet at: <https://www.regulations.gov/document?D=FAA-2016-7026-0002>.

(i) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless the AD specifies otherwise.

(i) PILATUS Aircraft Ltd. PC-7 Service Bulletin No: 53-013; dated February 25, 2016; and

(ii) PILATUS Aircraft Ltd. PC-7 Service Bulletin No: 53-014, dated February 25, 2016.

(3) For PILATUS Aircraft Ltd. service information identified in this AD, contact PILATUS Aircraft Ltd., Customer Technical Support (MCC), P.O. Box 992, CH-6371 Stans, Switzerland; phone: +41 (0)41 619 67 74; fax: +41 (0)41 619 67 73; email: techsupport@pilatus-aircraft.com; Internet: <http://www.pilatus-aircraft.com>.

(4) You may view this service information at the FAA, Small Airplane Directorate, 901 Locust, Kansas City, Missouri 64106. For information on the availability of this material at the FAA, call (816) 329-4148.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Kansas City, Missouri, on August 17, 2016.

Pat Mullen,

Acting Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2016-20074 Filed 8-25-16; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2016-8992; Directorate Identifier 2016-CE-021-AD; Amendment 39-18621; AD 2016-17-08]

RIN 2120-AA64

Airworthiness Directives; Textron Aviation, Inc. Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule; request for comments.

SUMMARY: We are superseding Airworthiness Directive (AD) 2016-07-24 for all Textron Aviation, Inc. Models 310 through 310R, E310H, E310J, T310P

through T310R, 310J-1, 320 through 320F, 320-1, 335, 340, 340A, 401 through 401B, 402 through 402C, 411, 411A, 414, 414A, and 421 through 421C airplanes (type certificates 3A10, 3A25, and A7CE previously held by Cessna Aircraft Company). AD 2016-07-24 required replacement and repetitive inspections of the hardware securing the elevator trim tab push-pull rod. This AD retains the actions for AD 2016-07-24 but revises the repetitive inspection intervals and allows for a longer bolt for the attachment of the elevator trim tab actuator rod end to the push-pull tube connection and/or for the elevator trim tab horn end to the push-pull tube connection. This AD was prompted by comments indicating difficulties with bolt installation and requesting a revision to repetitive inspection intervals to coincide with established inspection intervals. We are issuing this AD to prevent jamming of the elevator trim tab in a position outside the normal limits of travel due to the loss of the attachment hardware connecting the elevator trim tab actuator to the elevator trim tab push-pull rod, which could result in loss of control.

DATES: This AD is effective September 12, 2016.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in this AD as of September 12, 2016.

We must receive any comments on this AD by October 11, 2016.

ADDRESSES: You may send comments, using the procedures found in 14 CFR 11.43 and 11.45, by any of the following methods:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.

- *Fax:* 202-493-2251.

- *Mail:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

- *Hand Delivery:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this final rule, contact Textron Aviation Customer Service, P.O. Box 7706, Wichita, Kansas 67277; telephone: (316) 517-5800; fax: (316) 517-7271; email: customercare@txtav.com; Internet: <https://support.cessna.com/custsupt/csupport/newlogin.jsp>. You may review this referenced service information at the FAA, Small Airplane Directorate,

901 Locust, Kansas City, Missouri 64106. For information on the availability of this material at the FAA, call (816) 329-4148. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-8992.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-8992; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (phone: 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

FOR FURTHER INFORMATION CONTACT:

Adam Hein, Aerospace Engineer, Wichita Aircraft Certification Office, FAA, 1801 S. Airport Road, Room 100, Wichita, Kansas 67209; phone: (316) 946-4116; fax: (316) 946-4107; email: adam.hein@faa.gov.

SUPPLEMENTARY INFORMATION:

Discussion

On March 30, 2016, we issued AD 2016-07-24, Amendment 39-18469 (81 FR 21250, April 11, 2016), (“AD 2016-07-24”), for all Textron Aviation, Inc. Models 310 through 310R, E310H, E310J, T310P through T310R, 310J-1, 320 through 320F, 320-1, 335, 340, 340A, 401 through 401B, 402 through 402C, 411, 411A, 414, 414A, and 421 through 421C airplanes (type certificates 3A10, 3A25, and A7CE previously held by Cessna Aircraft Company). AD 2016-07-24 required replacing the hardware connecting the elevator trim tab push-pull rod to the elevator trim tab actuator and elevator trim tab. AD 2016-07-24 resulted from accident reports on Textron Aviation, Inc. Models T310Q, 310Q, and 402B airplanes; lessons learned in accident investigation support; and analysis of past accidents. The analysis of National Transportation Safety Board (NTSB) determination of probable cause indicated that following the loss of the attachment hardware connecting the elevator trim tab actuator to the elevator trim tab push-pull rod, the elevator trim tab may jam in a position outside the normal limits of travel. We issued AD 2016-07-24 to require replacement and repetitive inspections of the hardware securing the elevator trim tab push-pull rod.

Actions Since AD 2016-07-24 Was Issued

Since we issued AD 2016-07-24, we have received reports of difficulties in installing the required bolt, part number (P/N) NAS464P3-8, because it was found in some cases to be too short to properly fasten with a cotter pin as required by the AD. It was also determined beneficial to revise the repetitive inspection intervals to better coincide with the standard established inspection intervals for these airplanes. We are issuing this AD to correct the unsafe condition on these products.

Related Service Information Under 1 CFR Part 51

We reviewed Textron Aviation, Inc. (Cessna Aircraft Company) Multi-engine Service Bulletin No. MEB-27-02, Revision 1, dated June 15, 2016. The service information describes procedures for replacing the hardware connecting the elevator trim tab push-pull rod to the elevator trim tab actuator and elevator trim tab. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the ADDRESSES section.

FAA's Determination

We are issuing this AD because we evaluated all the relevant information and determined the unsafe condition described previously is likely to exist or

develop in other products of the same type design.

AD Requirements

This AD requires replacement and repetitive inspection of the elevator trim tab push-pull rod connecting hardware.

Differences Between the AD and the Service Information

Due to the immediate safety of flight condition of this AD action, we are requiring replacement of the hardware within 90 days after the effective date of this AD rather than the potential of up to a year as allowed in the service information.

The kit referenced in the service bulletin contains only standard parts that may be procured from other sources.

FAA's Justification and Determination of the Effective Date

An unsafe condition exists that requires the immediate adoption of this AD. The FAA has found that the risk to the flying public justifies waiving notice and comment prior to adoption of this rule because the loss of the attachment hardware connecting the elevator trim tab actuator to the elevator trim tab push-pull rod may result in jamming of the elevator trim tab beyond normal limits, which could result in loss of ability to control the airplane. Therefore, we find that notice and opportunity for prior public comment

are impracticable and that good cause exists for making this amendment effective in less than 30 days.

Comments Invited

This AD is a final rule that involves requirements affecting flight safety, and we did not provide you with notice and an opportunity to provide your comments before it becomes effective. However, we invite you to send any written data, views, or arguments about this AD. Send your comments to an address listed under the ADDRESSES section. Include the docket number FAA-2016-8992 and directorate identifier 2016-CE-021-AD at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this AD. We will consider all comments received by the closing date and may amend this AD because of those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this AD.

Costs of Compliance

We estimate that this AD affects 5,066 airplanes of U.S. registry.

We estimate the following costs to comply with this AD:

ESTIMATED COSTS

| Action | Labor cost | Parts cost | Cost per product | Cost on U.S. operators |
|---|---|----------------------|------------------|------------------------|
| Elevator trim tab push-pull rod hardware replacement. | 5 work-hour × \$85 per hour = \$425 | \$42 | \$467 | \$2,365,822 |
| Repetitive Inspection | 1 work-hour × \$85 per hour = \$85 | Not applicable | 85 | 430,610 |

We estimate the following costs to do any necessary replacements that would be required based on the results of the

inspection. This is the same replacement that is initially required by this AD. We have no way of determining

the number of aircraft that might need this repetitive on-condition replacement:

ON-CONDITION COSTS

| Action | Labor cost | Parts cost | Cost per product |
|---|--|------------|------------------|
| Elevator trim tab push-pull rod hardware replacement. | 5 work-hours × \$85 per hour = \$425 | \$42 | \$467 |

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, Section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more

detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701, "General requirements." Under that section, Congress charges the FAA with

promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition

that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

This AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866,
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),
- (3) Will not affect intrastate aviation in Alaska, and
- (4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by removing Airworthiness Directive (AD) 2016-07-24, Amendment 39-18469 (81 FR 21250, April 11, 2016) and adding the following new AD:

2016-17-08 Textron Aviation, Inc.:

Amendment 39-18621; Docket No. FAA-2016-8992; Directorate Identifier 2016-CE-021-AD.

(a) Effective Date

This AD is effective September 12, 2016.

(b) Affected ADs

This AD replaces AD 2016-07-24, Amendment 39-18469 (81 FR 21250, April 11, 2016) ("AD 2016-07-24").

(c) Applicability

This AD applies to Textron Aviation, Inc. Models 310 through 310R, E310H, E310J, T310P through T310R, 310J-1, 320 through

320F, 320-1, 335, 340, 340A, 401 through 401B, 402 through 402C, 411, 411A, 414, 414A, and 421 through 421C airplanes (type certificates 3A10, 3A25, and A7CE previously held by Cessna Aircraft Company), all serial numbers, certificated in any category.

(d) Subject

Joint Aircraft System Component (JASC)/ Air Transport Association (ATA) of America Code 2731, Elevator Tab Control System.

(e) Unsafe Condition

This AD supersedes AD 2016-07-24, which required replacement and repetitive inspections of the hardware securing the elevator trim tab push-pull rod. This AD retains the actions for AD 2016-07-24 but revises the repetitive inspection intervals and allows for a longer bolt for the attachment of the elevator trim tab actuator rod end to the push-pull tube connection and/or for the elevator trim tab horn end to the push-pull tube connection. This AD was prompted by comments indicating difficulties with bolt installation and requesting a revision to repetitive inspection intervals to coincide with established inspection intervals. We are issuing this AD to prevent jamming of the elevator trim tab in a position outside the normal limits of travel due to the loss of the attachment hardware connecting the elevator trim tab actuator to the elevator trim tab push-pull rod, which could result in loss of control.

(f) Actions and Compliance

Do the actions in paragraphs (f)(1) through (3) of this AD. If paragraph (f)(1) of this AD has already been done before September 12, 2016 (the effective date of this AD) following either Textron Aviation, Inc. (Cessna) Multi-engine Service Bulletin (SB) No. MEB-27-02, dated February 29, 2016 (see paragraph (g) of this AD) or Textron Aviation, Inc. (Cessna) Multi-engine Service Bulletin (SB) No. MEB-27-02, Revision 1, dated June 15, 2016, then credit can be taken and the only required actions are the repetitive actions of paragraphs (f)(2) and (3) of this AD.

(1) Within the next 90 days after September 12, 2016 (the effective date of this AD), replace the elevator trim tab push-pull rod attachment hardware on the elevator trim tab actuator and the trim tab ends of the push-pull rod following steps 3 through 6 of the accomplishment instructions in Textron Aviation, Inc. (Cessna) Multi-engine Service Bulletin (SB) No. MEB-27-02, Revision 1, dated June 15, 2016.

(2) Following the replacement required in paragraph (f)(1) of this AD or the replacement or previous repetitive general visual inspection done per AD 2016-07-24, whichever occurs later, repetitively conduct general visual inspections of the elevator trim tab push-pull rod attachment hardware on the elevator trim tab actuator and the trim tab ends of the push-pull rod at intervals not to exceed 110 hours TIS or 12 months, whichever occurs first. Before further flight, replace the hardware if necessary following the Compliance NOTE on page 1 of Textron Aviation, Inc. (Cessna) Multi-engine Service Bulletin (SB) No. MEB-27-02, Revision 1, dated June 15, 2016.

Note 1 to paragraph (f)(2) of this AD: The intent is to require these repetitive inspections during your regular maintenance schedule.

(3) After September 12, 2016 (the effective date of this AD), any time the elevator trim tab push-pull rod attachment hardware on the elevator trim tab actuator and/or trim tab ends of the push-pull rod is removed for any reason, discard the old hardware (bolt, nut, washer and cotter pin) and replace with new hardware following steps 4 and/or step 6 of Textron Aviation, Inc. (Cessna) Multi-engine Service Bulletin (SB) No. MEB-27-02, Revision 1, dated June 15, 2016.

(g) Credit for Actions Accomplished in Accordance With Previous Service Information

This AD allows credit for the actions required in paragraphs (f)(1) of this AD if done before September 12, 2016 (the effective date of this AD) following the instructions of Textron Aviation, Inc. (Cessna) Multi-engine Service Bulletin (SB) No. MEB-27-02, dated February 29, 2016.

(h) Special Flight Permit

Special flight permits are allowed for this AD per 14 CFR 39.23 with the following limitation: Before flight a pre-flight visual inspection is required of the attachment hardware connecting the elevator trim tab actuator to the elevator trim tab push-pull rod. Confirmation of the presence of a castellated nut and cotter pin is required.

(i) Alternative Methods of Compliance (AMOCs)

(1) The Manager, Wichita ACO, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the ACO, send it to the attention of the person identified in paragraph (i) of this AD.

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(3) AMOCs approved previously for AD 2016-07-24 are valid as AMOCs for this AD.

(j) Related Information

For more information about this AD, contact Adam Hein, Aerospace Engineer, Wichita Aircraft Certification Office, FAA, 1801 S. Airport Road, Room 100, Wichita, Kansas 67209; phone: (316) 946-4116; fax: (316) 946-4107; email: adam.hein@faa.gov.

(k) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless the AD specifies otherwise.

(3) The following service information was approved for IBR on September 12, 2016 (the effective date of this AD).

(i). Textron Aviation, Inc. (Cessna) Multi-engine Service Bulletin (SB) No. MEB-27-02, Revision 1, dated June 15, 2016.

(ii) Reserved.

(4) For Textron Aviation, Inc. (Cessna) service information identified in this AD, contact Textron Aviation Customer Service, P.O. Box 7706, Wichita, Kansas 67277; telephone: (316) 517-5800; fax: (316) 517-7271; email: customercare@cessna.textron.com; Internet: <https://support.cessna.com/custsupt/csupt/newlogin.jsp>.

(5) You may view this service information at the FAA, Small Airplane Directorate, 901 Locust, Kansas City, Missouri 64106. For information on the availability of this material at the FAA, call (816) 329-4148. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-8992.

(6) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Kansas City, Missouri, on August 17, 2016.

Pat Mullen,

Acting Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2016-20073 Filed 8-25-16; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2016-4221; Directorate Identifier 2015-NM-167-AD; Amendment 39-18619; AD 2016-17-06]

RIN 2120-AA64

Airworthiness Directives; The Boeing Company Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for all The Boeing Company Model 767-200 and -300 series airplanes. This AD was prompted by an evaluation by the design approval holder (DAH) indicating that the aft pressure bulkhead web to pressure chord joint is subject to widespread fatigue damage (WFD). This AD requires repetitive high frequency eddy current (HFEC) inspections of the aft pressure bulkhead web at fasteners common to the bulkhead web and pressure chord, around the entire circumference of the pressure chord, for any crack, and repair of cracks. We are

issuing this AD to detect and correct cracks in the aft pressure bulkhead web. Such cracking could result in the loss of structural integrity of the airplane.

DATES: This AD is effective September 30, 2016.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in this AD as of September 30, 2016.

ADDRESSES: For service information identified in this final rule, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P.O. Box 3707, MC 2H-65, Seattle, WA 98124-2207; telephone 206-544-5000, extension 1; fax 206-766-5680; Internet <https://www.myboeingfleet.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-4221.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-4221; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (phone: 800-647-5527) is Docket Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT: Wayne Lockett, Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Seattle Aircraft Certification Office (ACO), 1601 Lind Avenue SW., Renton, WA 98057-3356; phone: 425-917-6447; fax: 425-917-6590; email: wayne.lockett@faa.gov.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 by adding an AD that would apply to all The Boeing Company Model 767-200 and -300 series airplanes. The NPRM published in the **Federal Register** on March 8, 2016 (81 FR 12047) (“the NPRM”). The NPRM was prompted by an evaluation by the DAH

indicating that the aft pressure bulkhead web to pressure chord joint is subject to WFD. The NPRM proposed to require repetitive HFEC inspections of the aft pressure bulkhead web at fasteners common to the bulkhead web and pressure chord, around the entire circumference of the pressure chord, for any crack, and repair of cracks. We are issuing this AD to detect and correct cracks in the aft pressure bulkhead web. Such cracking could result in the loss of structural integrity of the airplane.

Comments

We gave the public the opportunity to participate in developing this AD. The following presents the comments received on the NPRM and the FAA’s response. Boeing and United Airlines supported the NPRM.

Effect of Winglets on Accomplishment of the Proposed Actions

Aviation Partners Boeing stated that accomplishing supplemental type certificate (STC) ST01920SE does not affect the actions specified in the proposed AD.

We concur with the commenter. We have redesignated paragraph (c) of the proposed AD as paragraph (c)(1), and added new paragraph (c)(2) to this AD to state that installation of STC ST01920SE does not affect the ability to accomplish the actions required by this AD. Therefore, for airplanes on which STC ST01920SE is installed, a “change in product” alternative method of compliance (AMOC) approval request is not necessary to comply with the requirements of 14 CFR 39.17.

Conclusion

We reviewed the relevant data, considered the comments received, and determined that air safety and the public interest require adopting this AD with the changes described previously and minor editorial changes. We have determined that these minor changes:

- Are consistent with the intent that was proposed in the NPRM for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the NPRM.

We also determined that these changes will not increase the economic burden on any operator or increase the scope of this AD.

Related Service Information Under 14 CFR Part 51

We reviewed Boeing Alert Service Bulletin 767-53A0268, dated April 1, 2015. The service information describes procedures for HFEC inspections of the aft pressure bulkhead web at fasteners