

(ii) Have not changed your policies and practices with regard to disclosing nonpublic personal information from the policies and practices that were disclosed to the customer under § 1016.6(a)(2) through (5) and (9) in the most recent privacy notice provided pursuant to this part.

(2) *Delivery of annual privacy notice after financial institution no longer meets requirements for exception.* If you have been excepted from delivering an annual privacy notice pursuant to paragraph (e)(1) of this section and change your policies or practices in such a way that you no longer meet the requirements for that exception, you must comply with paragraph (e)(2)(i) or (e)(2)(ii) of this section, as applicable.

(i) *Changes preceded by a revised privacy notice.* If you no longer meet the requirements of paragraph (e)(1) of this section because you change your policies or practices in such a way that § 1016.8 requires you to provide a revised privacy notice, you must provide an annual privacy notice in accordance with the timing requirements in paragraph (a) of this section, treating the revised privacy notice as an initial privacy notice.

(ii) *Changes not preceded by a revised privacy notice.* If you no longer meet the requirements of paragraph (e)(1) of this section because you change your policies or practices in such a way that § 1016.8 does not require you to provide a revised privacy notice, you must provide an annual privacy notice within 60 days of the change in your policies or practices that causes you to no longer meet the requirements of paragraph (e)(1).

(iii) *Example.* You change your policies and practices in such a way that you no longer meet the requirements of paragraph (e)(1) of this section effective April 1 of year 1. Assuming you define the 12-consecutive-month period pursuant to paragraph (a) of this section as a calendar year, if you were required to provide a revised privacy notice under § 1016.8 and you provided that notice on March 1 of year 1, you must provide an annual privacy notice by December 31 of year 2. If you were not required to provide a revised privacy notice under § 1016.8, you must provide an annual privacy notice by May 30 of year 1.

■ 4. Section 1016.9 is amended by revising paragraph (c) to read as follows:

§ 1016.9 Delivering privacy and opt out notices.

* * * * *

(c) *Annual notices only.* You may reasonably expect that a customer will

receive actual notice of your annual privacy notice if:

(1) The customer uses your Web site to access financial products and services electronically and agrees to receive notices at the Web site, and you post your current privacy notice continuously in a clear and conspicuous manner on the Web site; or

(2) The customer has requested that you refrain from sending any information regarding the customer relationship, and your current privacy notice remains available to the customer upon request.

* * * * *

Dated: June 29, 2016.
Richard Cordray,
Director, Bureau of Consumer Financial Protection.

[FR Doc. 2016-16132 Filed 7-8-16; 8:45 am]

BILLING CODE 4810-AM-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2015-3985; Directorate Identifier 2014-NM-182-AD]

RIN 2120-AA64

Airworthiness Directives; Airbus Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Supplemental notice of proposed rulemaking (NPRM); reopening of comment period.

SUMMARY: We are revising an earlier proposed airworthiness directive (AD) to supersede Airworthiness Directive (AD) 2010-04-03, for all Airbus Model A310 series airplanes. AD 2010-04-03 currently requires accomplishing repetitive detailed inspections for cracking around the fastener holes in certain wing top skin panels between the front and rear spars on the left- and right-hand sides of the fuselage, and repair if necessary. The NPRM proposed to continue to require the repetitive detailed inspections, and would also require supplemental repetitive ultrasonic inspections for cracking around the fastener holes in wing top skin panels 1 and 2 at rib 2, and repair if necessary. This action revises the NPRM by expanding the inspection area to include rib 3 due to widespread fatigue damage. We are proposing this supplemental NPRM (SNPRM) to detect and correct fatigue cracking around the fastener holes, which could result in

reduced structural integrity of the airplane. Since these actions impose an additional burden over those proposed in the NPRM, we are reopening the comment period to allow the public the chance to comment on these proposed changes.

DATES: We must receive comments on this SNPRM by August 25, 2016.

ADDRESSES: You may send comments, using the procedures found in 14 CFR 11.43 and 11.45, by any of the following methods:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.
- *Fax:* 202-493-2251.
- *Mail:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.
- *Hand Delivery:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this SNPRM, contact Airbus SAS, Airworthiness Office—EAW, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 44 51; email account.airworth-eas@airbus.com; Internet <http://www.airbus.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-3985; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (telephone: 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

FOR FURTHER INFORMATION CONTACT: Dan Rodina, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone 425-227-2125; fax 425-227-1149.

SUPPLEMENTARY INFORMATION:**Comments Invited**

We invite you to send any written relevant data, views, or arguments about this proposed AD. Send your comments to an address listed under the **ADDRESSES** section. Include “Docket No. FAA–2015–3985; Directorate Identifier 2014–NM–182–AD” at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD based on those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to supersede AD 2010–04–03, Amendment 39–16196 (75 FR 6852, February 12, 2010) (“AD 2010–04–03”). AD 2010–04–03 applied to all Airbus Model A310 series airplanes. The NPRM published in the **Federal Register** on October 13, 2015 (80 FR 61327) (“the NPRM”). The NPRM was prompted by development of an ultrasonic inspection program to allow for earlier crack detection and extend the repetitive inspection intervals. The NPRM proposed to retain the requirements of AD 2010–04–03, and proposed to require supplemental repetitive ultrasonic inspections for cracking around the fastener holes in wing top skin panels 1 and 2 at rib 2, and repair if necessary.

Since we issued the NPRM, a widespread fatigue damage analysis determined that the inspection area should be expanded to include cracking around the fastener holes in wing top skin panels 1 and 2 between the front and rear spar at rib 3.

The European Aviation Safety Agency (EASA), which is the Technical Agent for the Member States of the European Union, has issued EASA Airworthiness Directive 2016–0005, dated January 7, 2016 (referred to after this as the Mandatory Continuing Airworthiness Information, or “the MCAI”), to correct an unsafe condition on all Airbus Model A310 series airplanes. The MCAI states:

Following scheduled maintenance, cracks were found around the wing top skin panels fastener holes at Rib 2, between Stringer (STG) 2 and STG14.

This condition, if not detected and corrected, could affect the structural integrity of the aeroplane. The General Visual Inspection required by the existing applicable Airworthiness Limitation Items (ALI) tasks may not be adequate to detect these cracks.

To address this issue, Airbus developed an inspection programme based on repetitive detailed inspections (DET) to ensure that any visible cracks in the wing top skin panels 1 and 2 along Rib 2 are detected in time and repaired appropriately. EASA issued [EASA] AD 2008–0211 [<http://ad.easa.europa.eu/ad/2008-0211>], which corresponds to FAA AD 2010–04–03] to require implementation of this inspection programme.

After that [EASA] AD was issued, Airbus improved the inspection programme with an ultrasonic inspection to allow earlier crack detection, to subsequently reduce the scope of potential repair action, and to extend the intervals of the repetitive inspections.

Consequently, EASA issued AD 2014–0200 (later revised), superseding [EASA] AD 2008–0211, retaining its requirements, and to require supplementary repetitive ultrasonic inspections [for cracking] of the wing top skin panel 1 and 2 between STG2 and STG10 at Rib 2 [and repair if needed].

Since EASA AD 2014–0200R1 was issued, a widespread fatigue damage analysis concluded that the inspection programme has to be extended to include the wing top skin panels at Rib 3 attachments. For the reasons described above, this [EASA] AD retains the requirements of EASA AD 2014–0200R1, which is superseded, and extends the inspection area to include Rib 3.

You may examine the MCAI in the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA–2015–3985.

Related Service Information Under 14 CFR Part 51

Airbus has issued Service Bulletin A310–57–2096, Revision 03, dated June 30, 2015. This service information describes procedures for detailed and ultrasonic inspections for cracking around the fastener holes of wing top skin panels 1 and 2, at ribs 2 and 3, on the left- and right-hand sides of the fuselage. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the **ADDRESSES** section.

Comments

We gave the public the opportunity to participate in developing this proposed AD. We considered the comment received. The commenter, FedEx, supported the content of the NPRM and is currently complying with the requirements.

FAA’s Determination and Requirements of This SNPRM

This product has been approved by the aviation authority of another country, and is approved for operation in the United States. Pursuant to our bilateral agreement with the State of Design Authority, we have been notified of the unsafe condition described in the MCAI and service information referenced above. We are proposing this AD because we evaluated all pertinent information and determined an unsafe condition exists and is likely to exist or develop on other products of the same type design.

Certain changes described above expand the scope of the NPRM. As a result, we have determined that it is necessary to reopen the comment period to provide additional opportunity for the public to comment on this SNPRM.

Costs of Compliance

We estimate that this SNPRM affects 28 airplanes of U.S. registry.

We also estimate that it would take about 8 work-hours per product to comply with the basic requirements of this proposed AD. The average labor rate is \$85 per work-hour. Based on these figures, we estimate the cost of this proposed AD on U.S. operators to be \$19,040, or \$680 per product.

We estimate that it would take about 15 work-hours per product to do any necessary on-condition actions that would be required based on the results of the inspections. Required parts would cost about \$10,000 per product. We have no way of determining the number of aircraft that might need these actions.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA’s authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. “Subtitle VII: Aviation Programs,” describes in more detail the scope of the Agency’s authority.

We are issuing this rulemaking under the authority described in “Subtitle VII, Part A, Subpart III, Section 44701: General requirements.” Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this proposed regulation:

1. Is not a “significant regulatory action” under Executive Order 12866;
2. Is not a “significant rule” under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979);
3. Will not affect intrastate aviation in Alaska; and
4. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by removing Airworthiness Directive (AD) 2010–04–03, Amendment 39–16196 (75 FR 6852, February 12, 2010), and adding the following new AD:

Airbus: Docket No. FAA–2015–3985; Directorate Identifier 2014–NM–182–AD.

(a) Comments Due Date

We must receive comments by August 25, 2016.

(b) Affected ADs

This AD replaces AD 2010–04–03, Amendment 39–16196 (75 FR 6852, February 12, 2010) (“AD 2010–04–03”).

(c) Applicability

This AD applies to all Airbus Model A310–203, –204, –221, –222, –304, –322, –324, and –325 airplanes, certificated in any category, all manufacturer serial numbers.

(d) Subject

Air Transport Association (ATA) of America Code 57, Wings.

(e) Reason

This AD was prompted by the development of an ultrasonic inspection program to allow for earlier crack detection and extend the repetitive inspection intervals. We are issuing this AD to detect and correct fatigue cracking around the fastener holes in certain wing top skin panels between the front and rear spars on the left- and right-hand sides of the fuselage, which could result in reduced structural integrity of the airplane.

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) Repetitive Inspections

Except as required by paragraph (i) of this AD: Within the initial compliance time and thereafter at the repetitive intervals specified in paragraphs (h)(1) through (h)(3) of this AD, as applicable, accomplish the actions specified in paragraphs (g)(1) and (g)(2) of this AD concurrently and in sequence, in accordance with the Accomplishment Instructions of Airbus Service Bulletin A310–57–2096, Revision 03, dated June 30, 2015, except as provided by paragraph (j) of this AD.

(1) Accomplish a detailed inspection for cracking around the fastener holes in the wing top skin panels 1 and 2, along ribs 2 and 3, between the front and rear spars on the left- and right-hand sides of the fuselage.

(2) Accomplish an ultrasonic inspection for cracking around the fastener holes in the wing top skin panels 1 and 2, along ribs 2 and 3, between stringer (STG) 2 and STG10 on the left- and right-hand sides of the fuselage.

(h) Compliance Times for Airplanes Not Previously Inspected

(1) For Model A310–203, –204, –221, and –222 airplanes: Do the actions required by paragraphs (g)(1) and (g)(2) of this AD at the later of the times specified in paragraphs (h)(1)(i) and (h)(1)(ii) of this AD. Repeat the inspections specified in paragraphs (g)(1) and (g)(2) of this AD thereafter at intervals not to exceed 2,000 flight cycles or 4,100 flight hours, whichever occurs first.

(i) Prior to the accumulation of 18,700 flight cycles or 37,400 flight hours since first flight of the airplane, whichever occurs first.

(ii) Within 30 days after the effective date of this AD.

(2) For Model A310–304, –322, –324, and –325 airplanes having an average flight time (AFT) of less than 4 hours: Do the actions required by paragraphs (g)(1) and (g)(2) of this AD at the later of the times specified in paragraphs (h)(2)(i) and (h)(2)(ii) of this AD. Repeat the inspections specified in paragraphs (g)(1) and (g)(2) of this AD thereafter at intervals not to exceed 2,000 flight cycles or 5,600 flight hours, whichever occurs first.

(i) Prior to the accumulation of 17,300 flight cycles or 48,400 flight hours since first flight of the airplane, whichever occurs first.

(ii) Within 30 days after the effective date of this AD.

(3) For Model A310–304, –322, –324, and –325 airplanes having an AFT of equal to or

more than 4 hours: Do the actions required by paragraphs (g)(1) and (g)(2) of this AD at the later of the times specified in paragraphs (h)(3)(i) and (h)(3)(ii) of this AD. Repeat the inspections specified in paragraphs (g)(1) and (g)(2) of this AD thereafter at intervals not to exceed 1,500 flight cycles or 7,500 flight hours, whichever occurs first.

(i) Prior to the accumulation of 12,800 flight cycles or 64,300 flight hours since first flight of the airplane, whichever occurs first.

(ii) Within 30 days after the effective date of this AD.

(i) Compliance Times for Airplanes Previously Inspected

For airplanes previously inspected before the effective date of this AD using Airbus Service Bulletin A310–57–2096, dated May 6, 2008; Airbus Service Bulletin A310–57–2096, Revision 01, dated August 5, 2010; or Airbus Service Bulletin A310–57–2096, Revision 02, dated March 5, 2014: At the applicable compliance times specified in paragraphs (i)(1), (i)(2), and (i)(3) of this AD, accomplish the actions specified in paragraphs (g)(1) and (g)(2) concurrently and in sequence, in accordance with the Accomplishment Instructions of Airbus Service Bulletin A310–57–2096, Revision 03, dated June 30, 2015. Repeat the inspections specified in paragraphs (g)(1) and (g)(2) of this AD thereafter at the repetitive intervals specified in paragraphs (h)(1), (h)(2), and (h)(3) of this AD, as applicable.

(1) For Model A310–203, –204, –221, and –222 airplanes: Do the actions required by paragraphs (g)(1) and (g)(2) of this AD within 3,500 flight hours or 1,700 flight cycles, whichever occurs first since the most recent inspection.

(2) For Model A310–304, –322, –324, and –325 airplanes having an AFT of less than 4 hours: Do the actions required by paragraphs (g)(1) and (g)(2) of this AD within 4,600 flight hours or 1,600 flight cycles, whichever occurs first since the most recent inspection.

(3) For Model A310–304, –322, –324, and –325 airplanes having an AFT of equal to or more than 4 hours: Do the actions required by paragraphs (g)(1) and (g)(2) of this AD within 6,100 flight hours or 1,200 flight cycles, whichever occurs first since the most recent inspection.

(j) Compliance Times if No Ultrasonic Equipment Is Available

If no ultrasonic equipment is available for the initial or second inspection required by paragraph (g) or (h) of this AD, accomplish the detailed inspection specified in paragraph (g)(1) of this AD within the applicable compliance times specified in paragraphs (j)(1) and (j)(2) of this AD. After accomplishing the detailed inspection, do the inspections specified in paragraphs (g)(1) and (g)(2) of this AD at the applicable compliance times specified by paragraphs (i)(1), (i)(2), and (i)(3) of this AD. Subsequently, repeat the inspections specified in paragraphs (g)(1) and (g)(2) of this AD thereafter at the applicable repetitive intervals specified in paragraphs (h)(1), (h)(2), and (h)(3) of this AD.

(1) For airplanes not previously inspected before the effective date of this AD using the

service information identified in paragraph (j)(2)(i), (j)(2)(ii), or (j)(2)(iii) of this AD: Do the actions required by paragraph (g)(1) of this AD within the initial compliance time specified by paragraphs (h)(1), (h)(2), and (h)(3) of this AD, as applicable.

(2) For airplanes previously inspected before the effective date of this AD using the service information identified in paragraph (j)(2)(i), (j)(2)(ii), or (j)(2)(iii) of this AD: Do the actions required by paragraph (g)(1) of this AD within the applicable compliance times specified in paragraphs (i)(1), (i)(2), and (i)(3) of this AD.

(i) Airbus Service Bulletin A310-57-2096, dated May 6, 2008.

(ii) Airbus Service Bulletin A310-57-2096, Revision 01, dated August 5, 2010.

(iii) Airbus Service Bulletin A310-57-2096, Revision 02, dated March 5, 2014.

(k) Repair of Cracking

If any cracking is found during any inspection required by paragraph (g), (h), (i), or (j) of this AD, before further flight, repair the cracking using a method approved by the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA; or the European Aviation Safety Agency (EASA); or Airbus's EASA Design Organization Approval (DOA). Accomplishing the repair specified in this paragraph terminates the repetitive inspections required by paragraph (h), (i), or (j) of this AD, as applicable, for the repaired area only.

(l) Definition of Average Flight Time (AFT)

For the purposes of this AD, the AFT should be established as specified in paragraphs (l)(1), (l)(2), and (l)(3) of this AD for the determination of the compliance times.

(1) The inspection threshold is defined as the total flight hours accumulated (counted from take-off to touch-down), divided by the total number of flight cycles accumulated at the effective date of this AD.

(2) The initial inspection interval is defined as the total flight hours accumulated divided by the total number of flight cycles accumulated at the time of the initial inspection threshold.

(3) The second inspection interval is defined as the total flight hours accumulated divided by the total number of flight cycles accumulated between the initial and second inspection threshold.

(m) Credit for Previous Actions

This paragraph provides credit for the actions required by paragraph (g)(1) of this AD, if those actions were performed before the effective date of this AD using the applicable service information identified in paragraph (m)(1), (m)(2), or (m)(3) of this AD.

(1) Airbus Service Bulletin A310-57-2096, dated May 6, 2008, which was incorporated by reference in AD 2010-04-03.

(2) Airbus Service Bulletin A310-57-2096, Revision 01, dated August 5, 2010, which is not incorporated by reference in this AD.

(3) Airbus Service Bulletin A310-57-2096, Revision 02, dated March 5, 2014, which is not incorporated by reference in this AD.

(n) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) Alternative Methods of Compliance (AMOCs): The Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the International Branch, send it to ATTN: Dan Rodina, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone 425-227-2125; fax 425-227-1149. Information may be emailed to: 9-ANM-116-AMOC-REQUESTS@faa.gov. Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office. The AMOC approval letter must specifically reference this AD.

(2) Contacting the Manufacturer: As of the effective date of this AD, for any requirement in this AD to obtain corrective actions from a manufacturer, the action must be accomplished using a method approved by the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA; or the EASA; or Airbus's EASA DOA. If approved by the DOA, the approval must include the DOA-authorized signature.

(3) Required for Compliance (RC): Except as required by paragraph (k) of this AD, if any service information contains procedures or tests that are identified as RC, those procedures and tests must be done to comply with this AD; any procedures or tests that are not identified as RC are recommended. Those procedures and tests that are not identified as RC may be deviated from using accepted methods in accordance with the operator's maintenance or inspection program without obtaining approval of an AMOC, provided the procedures and tests identified as RC can be done and the airplane can be put back in an airworthy condition. Any substitutions or changes to procedures or tests identified as RC require approval of an AMOC.

(o) Related Information

(1) Refer to Mandatory Continuing Airworthiness Information (MCAI) EASA Airworthiness Directive 2016-0005, dated January 7, 2016, for related information. This MCAI may be found in the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-3985.

(2) For service information identified in this AD, contact Airbus SAS, Airworthiness Office—EAW, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 44 51; email account.airworth-eas@airbus.com; Internet <http://www.airbus.com>. You may view this service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

Issued in Renton, Washington, on July 1, 2016.

Phillip Forde,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2016-16210 Filed 7-8-16; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF HOMELAND SECURITY

Coast Guard

33 CFR Part 100

[Docket Number USCG-2016-0012]

RIN 1625-AA08

Special Local Regulation; Bucksport/Lake Murray Drag Boat Fall Nationals, Atlantic Intracoastal Waterway; Bucksport, SC

AGENCY: Coast Guard, DHS.

ACTION: Notice of proposed rulemaking.

SUMMARY: The Coast Guard proposes to establish a special local regulation on the Atlantic Intracoastal Waterway in Bucksport, South Carolina during the Bucksport/Lake Murray Drag Boat Fall Nationals, on September 10 and September 11, 2016. This special local regulation is necessary to ensure the safety of participants, spectators, and the general public during the event. This proposed rulemaking would prohibit persons and vessels from being in the regulated area unless authorized by the Captain of the Port Charleston or a designated representative. We invite your comments on this proposed rulemaking.

DATES: Comments and related material must be received by the Coast Guard on or before August 10, 2016.

ADDRESSES: You may submit comments identified by docket number USCG-2016-0012 using the Federal eRulemaking Portal at <http://www.regulations.gov>. See the "Public Participation and Request for Comments" portion of the **SUPPLEMENTARY INFORMATION** section for further instructions on submitting comments.

FOR FURTHER INFORMATION CONTACT: If you have questions about this proposed rulemaking, call or email Lieutenant John Downing, Sector Charleston Office of Waterways Management, Coast Guard; telephone (843) 740-3184, email John.Z.Downing@uscg.mil.

SUPPLEMENTARY INFORMATION:

I. Table of Abbreviations

CFR Code of Federal Regulations