

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) Modification

For Embraer S.A. Model ERJ 190 airplanes identified in Embraer Service Bulletin 190–21–0041, Revision 02, dated July 30, 2013, within 3 months after the effective date of this AD, replace the Hamilton Sundstrand air management system (AMS) controller operation program of the AMS controller processor boards, as specified in paragraph (g)(1) or (g)(2) of this AD.

(1) Replace with a new, improved program, in accordance with the Accomplishment Instructions of Embraer Service Bulletin 190–21–0041, Revision 02, dated July 30, 2013.

(2) Replace with a version of the Hamilton Sundstrand AMS controller operation program approved after August 31, 2012, using a method approved by the Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA; Agência Nacional de Aviação Civil (ANAC); or ANAC's authorized Designee.

(h) Valve Replacement

For Embraer S.A. Model ERJ 190 airplanes identified in Embraer Service Bulletin 190–21–0041, Revision 02, dated July 30, 2013, within 3 months after the effective date of this AD, and after accomplishment of the actions required by paragraph (g) of this AD: Replace the check valve and associated seals of the left-hand and right-hand engine bleed system with a check valve identified in paragraph (i) of this AD, and new seals, in accordance with the Accomplishment Instructions of Embraer Service Bulletin 190–36–0023, Revision 03, dated September 24, 2014.

(i) Allowed Valves

When complying with paragraph (h) of this AD, the low-stage bleed check valves having P/N 1001447–6, and associated seals, are replaced with new ones (zero-hour). Low-stage bleed check valves having P/N 1001447–6 that can be demonstrated with logged hours only on ERJ–170 aircraft and/or on ERJ–190 aircraft equipped with the AMS Controller Operational Program version Black Label 13, or a later version, can be used instead of new ones (zero-hour).

(j) Parts Installation Limitation

(1) For Model ERJ 170–100 STD, –100 LR, –100SU, –100SE, –200 STD, –200 LR, and –200 SU airplanes: No person may install on any airplane a low-stage bleed check valve having P/N 1001447–6 that was installed on any Model ERJ 190–100 STD, –100 LR, –100 IGW, –200 STD, –200 LR, or –200 IGW airplane, any serial number except 190–00587, 190–00589, and 190–00593 and subsequent, prior to accomplishment of paragraph (g) of this AD.

(2) For Model ERJ 190–100 STD, –100 LR, –100IGW, –200 STD, –200 LR, and –200 IGW airplanes: No person may install on any airplane on which the actions of paragraph (g) of this AD have been done, a low-stage bleed check valve having P/N 1001447–6 that was previously installed on any Model ERJ

190–100 STD, –100 LR, –100 IGW, –200 STD, –200 LR, or –200 IGW airplane, any serial number except 190–00587, 190–00589, 190–00593 and subsequent, prior to accomplishment of paragraph (g) of this AD.

(k) Credit for Previous Actions

(1) This paragraph provides credit for actions required by paragraph (g) of this AD, if those actions were performed before the effective date of this AD using the service information identified in paragraph (k)(1)(i) or (k)(1)(ii) of this AD. This service information is not incorporated by reference in this AD.

(i) Embraer Service Bulletin 190–21–0041, dated September 27, 2012.

(ii) Embraer Service Bulletin 190–21–0041, Revision 01, dated December 20, 2012.

(2) This paragraph provides credit for actions required by paragraph (h) of this AD, if those actions were performed before the effective date of this AD using the service information identified in paragraph (k)(2)(i), (k)(2)(ii), or (k)(2)(iii) of this AD. This service information is not incorporated by reference in this AD.

(i) Embraer Service Bulletin 190–36–0023, dated July 22, 2013.

(ii) Embraer Service Bulletin 190–36–0023, Revision 01, dated September 3, 2013.

(iii) Embraer Service Bulletin 190–36–0023, Revision 02, dated April 30, 2014.

(l) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs)*: The Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the International Branch, send it to ATTN: Ana Martinez Hueto, Aerospace Engineer, International Branch, ANM–116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057–3356; telephone 425–227–1622; fax 425–227–1149. Information may be emailed to: 9-ANM-116-AMOC-REQUESTS@faa.gov. Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office. The AMOC approval letter must specifically reference this AD.

(2) *Contacting the Manufacturer*: For any requirement in this AD to obtain corrective actions from a manufacturer, the action must be accomplished using a method approved by the Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA; or ANAC; or ANAC's authorized Designee. If approved by the ANAC Designee, the approval must include the Designee's authorized signature.

(m) Related Information

(1) Refer to Mandatory Continuing Airworthiness Information (MCAI) Brazilian Airworthiness Directive 2015–02–02, dated March 6, 2015, for related information. This

MCAI may be found in the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA–2015–6542.

(2) For service information identified in this AD, contact Embraer S.A., Technical Publications Section (PC 060), Av. Brigadeiro Faria Lima, 2170–Putim – 12227–901 São Jose dos Campos – SP – BRASIL; telephone +55 12 3927–5852 or +55 12 3309–0732; fax +55 12 3927–7546; email distrib@embraer.com.br; Internet <http://www.flyembraer.com>. You may view this service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425–227–1221.

Issued in Renton, Washington, on November 20, 2015.

Jeffrey E. Duven,

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2015–30224 Filed 11–27–15; 8:45 am]

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DEPARTMENT OF TRANSPORTATION**Federal Aviation Administration****14 CFR Part 39**

[Docket No. FAA–2015–1014; Directorate Identifier 2015–NE–14–AD]

RIN 2120–AA64

Airworthiness Directives; Rolls-Royce Deutschland Ltd & Co KG Turbofan Engines

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Proposed rule; withdrawal.

SUMMARY: The FAA is withdrawing a notice of proposed rulemaking (NPRM). The NPRM proposed a new airworthiness directive (AD) that had applied to all Rolls-Royce Deutschland Ltd & Co KG (RRD) Tay 650–15 and Tay 651–54 turbofan engines. The proposed action would have required reducing the cyclic life limits for certain high-pressure turbine (HPT) disks. Accordingly, we withdraw the proposed rule.

DATES: The proposed rule published in the **Federal Register** on June 8, 2015 (80 FR 32315, June 8, 2015) is withdrawn as of November 30, 2015.

FOR FURTHER INFORMATION CONTACT: Philip Haberlen, Aerospace Engineer, Engine Certification Office, FAA, Engine & Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803; phone: 781–238–7770; fax: 781–238–7199; email: philip.haberlen@faa.gov.

SUPPLEMENTARY INFORMATION: The FAA proposed to amend 14 CFR part 39 with

a proposed AD (80 FR 32315, June 8, 2015). The proposed AD had applied to Rolls-Royce Deutschland Ltd & Co KG (RRD) Tay 650–15 and Tay 651–54 turbofan engines. The NPRM proposed to reduce the cyclic life limits for certain HPT disks. The proposed action was prompted by an analysis that showed the need to reduce the cyclic life limits for certain HPT disks. The proposed actions intended to prevent failure of the HPT disk, which could result in uncontained disk release, damage to the engine, and damage to the airplane.

Since we issued the NPRM (80 FR 32315, June 8, 2015), additional information became available after the public comment period closed on August 7, 2015.

Upon further consideration, we hereby withdraw the proposed rule because we will propose to supersede AD 2006–18–14 (71 FR 52988, September 8, 2006).

Withdrawal of the NPRM (80 FR 32315, June 8, 2015) constitutes only such action, and does not preclude the agency from issuing another notice in the future, nor does it commit the agency to any course of action in the future.

Since this action only withdraws a notice of proposed rulemaking, it is neither a proposed nor a final rule. Therefore, Executive Order 12866, the Regulatory Flexibility Act, or DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979) do not cover this withdrawal.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

The Withdrawal

Accordingly, the notice of proposed rulemaking, Docket No. FAA–2015–1014; Directorate Identifier 2015–NE–14–AD, published in the **Federal Register** on June 8, 2015 (80 FR 32315), is withdrawn.

Issued in Burlington, Massachusetts, on November 20, 2015.

Colleen M. D'Alessandro,

Directorate Manager, Engine & Propeller Directorate, Aircraft Certification Service.

[FR Doc. 2015–30010 Filed 11–27–15; 8:45 am]

BILLING CODE 4910–13–P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA–2015–6539; Directorate Identifier 2015–NM–036–AD]

RIN 2120–AA64

Airworthiness Directives; Airbus Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: We propose to adopt a new airworthiness directive (AD) for all Airbus Model A318–111 and –112 airplanes; Model A319–111, –112, –113, –114, and –115 airplanes; Model A320–211, –212, and –214 airplanes; and Model A321–111, –112, –211, –212, and –213 airplanes. This proposed AD was prompted by an evaluation by the design approval holder (DAH) indicating that the forward engine mounts are subject to widespread fatigue damage (WFD). This proposed AD would require repetitive detailed inspections of the right and left forward engine mounts, and corrective action if necessary. These inspections are required by AD 2015–05–02. This proposed AD would reduce the compliance times for those inspections. We are proposing this AD to detect and correct fatigue cracking in the forward engine mounts, which could result in reduced structural integrity of the airplane and could lead to in-flight loss of an engine, possibly resulting in reduced controllability of the airplane.

DATES: We must receive comments on this proposed AD by January 14, 2016.

ADDRESSES: You may send comments, using the procedures found in 14 CFR 11.43 and 11.45, by any of the following methods:

- Federal eRulemaking Portal: Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.
- Fax: 202–493–2251.
- Mail: U.S. Department of Transportation, Docket Operations, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue SE., Washington, DC 20590.
- Hand Delivery: U.S. Department of Transportation, Docket Operations, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue SE., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this proposed AD, contact Airbus,

Airworthiness Office—EIAS, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 44 51; email account.airworth-eas@airbus.com; Internet <http://www.airbus.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425–227–1221.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA–2015–6539; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Operations office (telephone 800–647–5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

FOR FURTHER INFORMATION CONTACT: Sanjay Ralhan, Aerospace Engineer, International Branch, ANM–116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057–3356; telephone 425–227–1405; fax 425–227–1149.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposed AD. Send your comments to an address listed under the **ADDRESSES** section. Include “Docket No. FAA–2015–6539; Directorate Identifier 2015–NM–036–AD” at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD based on those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

Discussion

Structural fatigue damage is progressive. It begins as minute cracks, and those cracks grow under the action of repeated stresses. This can happen because of normal operational conditions and design attributes, or