

4. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

#### The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

#### PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

■ 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

**Airbus:** Docket No. FAA–2015–4204; Directorate Identifier 2015–NM–001–AD.

#### (a) Comments Due Date

We must receive comments by December 7, 2015.

#### (b) Affected ADs

None.

#### (c) Applicability

This AD applies to the Airbus airplanes specified in paragraphs (c)(1), (c)(2), (c)(3), and (c)(4) of this AD; certificated in any category; modified by Simmonds Precision Products, Inc., supplemental type certificate (STC) ST00092BO ([http://rgl.faa.gov/Regulatory\\_and\\_Guidance\\_Library/rqstc.nsf/0/D41C5AE8E46B4901862v574900069E004?OpenDocument&Highlight=st00092bo](http://rgl.faa.gov/Regulatory_and_Guidance_Library/rqstc.nsf/0/D41C5AE8E46B4901862v574900069E004?OpenDocument&Highlight=st00092bo)).

(1) Model A300 B4–601, B4–603, B4–620, and B4–622 airplanes.

(2) Model A300 B4–605R and B4–622R airplanes.

(3) Model A300 F4–605R and F4–622R airplanes.

(4) Model A300 C4–605R Variant F airplanes.

#### (d) Subject

Air Transport Association (ATA) of America Code 28, Fuel.

#### (e) Reason

This AD was prompted by a report of chafing found on the overflow sensor harness of the surge tank, and subsequent contact between the electrical wiring and fuel tank structure. We are issuing this AD to prevent chafing of the harness and subsequent contact between the electrical wiring and fuel tank structure, which could result in electrical arcing and a fuel tank explosion.

#### (f) Compliance

Comply with this AD within the compliance times specified, unless already done.

#### (g) One-Time Inspection and Repair

Within 12 months after the effective date of this AD: Do the actions required by paragraphs (g)(1), (g)(2), and (g)(3) of this AD, in accordance with the Accomplishment Instructions of UTC Aerospace Systems Service Bulletin 300723–28–03 (V–1577), Revision 01, dated July 20, 2015.

(1) Perform a one-time general visual inspection for damage of the outer tank sensor harness, and if any damage is found on the expando sleeving, before further flight, do a detailed inspection of the underlying wires for exposed conductor wires. If any exposed conductor wire is found, before further flight, replace the outer wing harness assembly.

(2) Install new brackets and re-route the surge tank overflow sensor harness.

(3) Modify the harness protection.

#### (h) Credit for Previous Actions

This paragraph provides credit for actions required by paragraph (g) of this AD, if those actions were performed before the effective date of this AD using UTC Aerospace Systems Service Bulletin 300723–28–03 (V–1577), dated October 10, 2014. This service information is not incorporated by reference in this AD.

#### (i) Alternative Methods of Compliance (AMOCs)

(1) The Manager, Boston Aircraft Certification Office (ACO) ANE–150, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the ACO, send it to the attention of the person identified in paragraph (j)(1) of this AD.

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

#### (j) Related Information

(1) For more information about this AD, contact Marc Ronell, Aerospace Engineer, Boston ACO, FAA, 12 New England Executive Park, Burlington, MA 01803; phone: 781–238–7776; fax: 781–238–7170; email: [marc.ronell@faa.gov](mailto:marc.ronell@faa.gov).

(2) For service information identified in this AD, contact Simmonds Precision Products, Inc., A UTC Aerospace Company, 100 Panton Road, Vergennes, Vermont 05491; phone 802–877–2911; fax 802–877–4444; Internet <http://www.utcaerospace.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425–227–1221.

Issued in Renton, Washington, on October 15, 2015.

**Jeffrey E. Duven,**

*Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 2015–26691 Filed 10–22–15; 8:45 am]

**BILLING CODE 4910–13–P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA–2015–3713; Directorate Identifier 2015–NE–23–AD]

**RIN 2120–AA64**

#### Airworthiness Directives; Engine Alliance Turbofan Engines

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** We propose to adopt a new airworthiness directive (AD) for certain Engine Alliance (EA) GP7270 turbofan engines. This proposed AD was prompted by reports of the installation of non-conforming honeycomb seals in the high-pressure compressor (HPC) adjacent to the HPC rotor stage 2 to 5 spool and stage 7 to 9 spool. This proposed AD would require removal and replacement of the affected HPC rotor stage 2 to 5 and stage 7 to 9 spools. We are proposing this AD to prevent failure of the HPC rotor stage 2 to 5 and stage 7 to 9 spools, which could lead to uncontained engine failure and damage to the airplane.

**DATES:** We must receive comments on this proposed AD by December 22, 2015.

**ADDRESSES:** You may send comments, using the procedures found in 14 CFR 11.43 and 11.45, by any of the following methods:

- **Federal eRulemaking Portal:** Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.

- **Fax:** 202–493–2251.

- **Mail:** U.S. Department of Transportation, Docket Operations, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue SE., Washington, DC 20590.

- **Hand Delivery:** Deliver to Mail address above between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this proposed AD, contact Engine Alliance, 400 Main St., East Hartford, CT 06108, M/S 169–10, phone: 800–565–0140; email: [help24@pw.utc.com](mailto:help24@pw.utc.com);

**Web site:**

[www.engineallianceportal.com](http://www.engineallianceportal.com). You may view this service information at the FAA, Engine & Propeller Directorate, 12 New England Executive Park, Burlington, MA. For information on the availability of this material at the FAA, call 781-238-7125.

**Examining the AD Docket**

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-3713; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (phone: 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

**FOR FURTHER INFORMATION CONTACT:**

Martin Adler, Aerospace Engineer, Engine & Propeller Directorate, FAA, 12 New England Executive Park, Burlington, MA 01803; phone: 781-238-7157; fax: 781-238-7199; email: [martin.adler@faa.gov](mailto:martin.adler@faa.gov).

**SUPPLEMENTARY INFORMATION:****Comments Invited**

We invite you to send any written relevant data, views, or arguments about this proposal. Send your comments to an address listed under the **ADDRESSES** section. Include "Docket No. FAA-2015-3713; Directorate Identifier 2015-NE-23-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD because of those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

**Discussion**

We learned from the manufacturer that non-conforming honeycomb seals were installed in the affected HPCs adjacent to the HPC rotor stage 2 to 5 spools and stage 7 to 9 spools. The honeycomb seals in the HPC were machined to an incorrect radial height which resulted in reduced clearances between the honeycomb and the

rotating spools. This error could lead to cracks on the spools prior to reaching their full life. This condition, if not corrected, could result in failure of the HPC rotor stage 2 to 5 and stage 7 to 9 spools, which could lead to uncontained engine failure, and damage to the airplane.

**Relevant Service Information Under 1 CFR Part 51**

Engine Alliance has issued EA Service Bulletin (SB) No. EAGP7-72-327, dated July 21, 2015; and SB No. EAGP7-72-328, dated July 21, 2015. The SBs describe procedures for removal and replacement of HPC rotor stage 2 to 5 spools and HPC rotor stage 7 to 9 spools, respectively. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the **ADDRESSES** section of this document.

**FAA's Determination**

We are proposing this AD because we evaluated all the relevant information and determined the unsafe condition described previously is likely to exist or develop in other products of the same type design.

**Proposed AD Requirements**

This proposed AD would require removal and replacement of the affected HPC rotor stage 2 to 5 and stage 7 to 9 spools.

**Costs of Compliance**

We estimate that this proposed AD affects zero engines installed on airplanes of U.S. registry. The average labor rate is \$85 per hour. Based on these figures, we estimate the cost of this proposed AD on U.S. operators to be \$0.

**Authority for This Rulemaking**

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII: Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701: "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority

because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

**Regulatory Findings**

We determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this proposed regulation:

(1) Is not a "significant regulatory action" under Executive Order 12866,

(2) Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),

(3) Will not affect intrastate aviation in Alaska to the extent that it justifies making a regulatory distinction, and

(4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

**List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

**The Proposed Amendment**

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

**PART 39—AIRWORTHINESS DIRECTIVES**

■ 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

**§ 39.13 [Amended]**

■ 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

**Engine Alliance:** Docket No. FAA-2015-3713; Directorate Identifier 2015-NE-23-AD.

**(a) Comments Due Date**

We must receive comments by December 22, 2015.

**(b) Affected ADs**

None.

**(c) Applicability**

This AD applies to Engine Alliance (EA) GP7270 turbofan engines with one or both of the following installed:

(1) A high-pressure compressor (HPC) rotor stage 2 to 5 spool, part number (P/N) 382-104-807-0, with a serial number (S/N) listed in EA Service Bulletin (SB) No. EAGP7-72-327, dated July 21, 2015; or

(2) an HPC rotor stage 7 to 9 spool, P/N 2031M90G04, 2031M90G05, or 2031M90G07, with an S/N listed in EA SB No. EAGP7-72-328, dated July 21, 2015.

#### (d) Unsafe Condition

This AD was prompted by reports of the installation of non-conforming honeycomb seals in the HPC adjacent to the HPC rotor stage 2 to 5 spool and stage 7 to 9 spool. We are issuing this AD to prevent failure of the HPC rotor stage 2 to 5 spools and stage 7 to 9 spools, which could lead to uncontained engine failure and damage to the airplane.

#### (e) Compliance

Comply with this AD within the compliance times specified, unless already done. Within 30 days after the effective date of this AD or before accumulating 2,100 engine cycles since the last disassembly of the compressor module of the engine, whichever occurs later:

(1) For engines with an HPC rotor stage 2 to 5 spool, P/N 382-104-807-0, installed with a S/N listed in EA SB No. EAGP7-72-327, dated July 21, 2015, do the following:

(i) Remove the HPC rotor stage 2 to 5 spool from service and replace with a part eligible for installation.

(ii) Remove and replace the honeycomb seals on the HPC stage 5 vanes.

(2) For engines with an HPC rotor stage 7 to 9 spool, P/N 2031M90G04, 2031M90G05, or 2031M90G07 installed with an S/N listed in EA SB No. EAGP7-72-328, dated July 21, 2015, do the following:

(i) Remove the HPC rotor stage 7 to 9 spool from service and replace with a part eligible for installation.

(ii) Remove and replace the honeycomb seals on the HPC stage 6, stage 7, and stage 8 vanes.

#### (f) Alternative Methods of Compliance (AMOCs)

The Manager, Engine Certification Office, may approve AMOCs for this AD. Use the procedures found in 14 CFR 39.19 to make your request. You may email your request to: [ANE-AD-AMOC@faa.gov](mailto:ANE-AD-AMOC@faa.gov).

#### (g) Related Information

(1) For more information about this AD, contact Martin Adler, Aerospace Engineer, Engine & Propeller Directorate, FAA, 12 New England Executive Park, Burlington, MA 01803; phone: 781-238-7157; fax: 781-238-7199; email: [martin.adler@faa.gov](mailto:martin.adler@faa.gov).

(2) EA SBs No. EAGP7-72-327, dated July 21, 2015; and No. EAGP7-72-328, dated July 21, 2015 can be obtained from EA using the contact information in paragraph (g)(3) of this proposed AD.

(3) For service information identified in this AD, contact Engine Alliance, 400 Main St., East Hartford, CT 06108, M/S 169-10, phone: 800-565-0140; email: [help24@pw.utc.com](mailto:help24@pw.utc.com); Web site: [www.engineallianceportal.com](http://www.engineallianceportal.com).

(4) You may view this service information at the FAA, Engine & Propeller Directorate,

12 New England Executive Park, Burlington, MA. For information on the availability of this material at the FAA, call 781-238-7125.

Issued in Burlington, Massachusetts, on October 16, 2015.

**Colleen M. D'Alessandro,**

*Directorate Manager, Engine & Propeller Directorate, Aircraft Certification Service.*

[FR Doc. 2015-26755 Filed 10-22-15; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA-2015-4202; Directorate Identifier 2014-NM-016-AD]

**RIN 2120-AA64**

#### Airworthiness Directives; Airbus Airplanes

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** We propose to supersede Airworthiness Directive (AD) 2012-18-12, for certain Airbus Model A318, A319, and A320 series airplanes. AD 2012-18-12 currently requires modifying the off-wing escape slide (OWS) enclosures on the left-hand (LH) side and right-hand (RH) side of the fuselage. Since we issued AD 2012-18-12, we have received reports that additional OWS part numbers have been affected. This proposed AD would retain the requirements of AD 2012-18-12 and expand the applicability to all Airbus Model A318, A319, and A320 series airplanes. We are proposing this AD to prevent off-wing exits on the LH and RH sides of the fuselage from becoming inoperative, which, during an emergency, could impair the safe evacuation of occupants, possibly resulting in personal injuries.

**DATES:** We must receive comments on this proposed AD by December 7, 2015.

**ADDRESSES:** You may send comments by any of the following methods:

- Federal eRulemaking Portal: Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.
- Fax: 202-493-2251.
- Mail: U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.
- Hand Delivery: U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE.,

Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For Airbus service information identified in this proposed AD, contact Airbus, Airworthiness Office—EIAS, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 44 51; email [account.airworth-eas@airbus.com](mailto:account.airworth-eas@airbus.com); Internet <http://www.airbus.com>. For Air Cruisers service information identified in this proposed AD, contact Air Cruisers Company, Cage Code 70167, 1747 State Route 34, Wall Township, NJ 07727-3935; telephone 732-681-3527; fax 732-681-9163; Internet <http://www.zodiac aerospace.com/en/our-activities/aerosafety/zodiac-evacuation-systems>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

#### Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-4202; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Operations office (telephone 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

**FOR FURTHER INFORMATION CONTACT:** Sanjay Ralhan, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone 425-227-1405; fax 425-227-1149.

#### SUPPLEMENTARY INFORMATION:

##### Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposed AD. Send your comments to an address listed under the **ADDRESSES** section. Include "Docket No. FAA-2015-4202; Directorate Identifier 2014-NM-016-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD based on those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>.