

## Applicability

As discussed above, these special conditions are applicable to the Models BD-500-1A10 and BD-500-1A11 series airplanes. Should Bombardier Aerospace apply at a later date for a change to the type certificate to include another model incorporating the same novel or unusual design feature, the special conditions would apply to that model as well.

## Conclusion

This action affects only certain novel or unusual design features on two model series of airplanes. It is not a rule of general applicability.

## List of Subjects in 14 CFR Part 25

Aircraft, Aviation safety, Reporting and recordkeeping requirements.

The authority citation for these special conditions is as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701, 44702, 44704.

## The Proposed Special Conditions

Accordingly, the Federal Aviation Administration (FAA) proposes the following special conditions as part of the type certification basis for Bombardier Aerospace Models BD-500-1A10 and BD-500-1A11 series airplanes.

### Composite Wing and Fuel Tank Post-Crash Fire Survivability

1. The wing fuel tank structure must withstand an external fuel-fed pool fire for a minimum of 5 minutes.

2. The integrity of the wing fuel tank structure must be demonstrated at:

- Minimum fuel load, not less than reserve fuel level;
- Maximum fuel load equal to the maximum range fuel quantity; and
- Any other critical fuel loads.

3. The demonstration must consider fuel tank flammability, burn-through resistance, wing structural strength retention properties, and auto-ignition threats from localized heating of composite structure, fasteners, or any other feature that may produce an ignition source during a ground fire event for the required time duration.

Issued in Renton, Washington, on June 19, 2014.

**Michael Kaszycki,**

*Acting Manager, Transport Airplane Directorate.*

[FR Doc. 2014-16645 Filed 7-15-14; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA-2014-0448; Directorate Identifier 2013-NM-055-AD]

RIN 2120-AA64

### Airworthiness Directives; Airbus Airplanes

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** We propose to adopt a new airworthiness directive (AD) for all Airbus Model A300 series airplanes; Airbus Model A300 B4-600, B4-600R, and F4-600R series airplanes, and Model A300 C4-605R Variant F airplanes (collectively called Model A300-600 series airplanes); and Airbus Model A310 series airplanes. This proposed AD was prompted by a report of early ruptures on the levers of the nose landing gear (NLG) sequence valve. This proposed AD would require a one-time inspection for damage of the landing gear sequence valve levers and pin shearing indicating areas on the NLG and the main landing gears (MLGs); and depending on findings, replacing the sequence valve and lever, or doing a one-time inspection to detect interference between control rods and sequence valves and corrective actions if necessary. We are proposing this AD to detect and correct interference between a landing gear leg and door, which could result in failure of that landing gear to extend and could damage the airplane and injure occupants.

**DATES:** We must receive comments on this proposed AD by September 2, 2014.

**ADDRESSES:** You may send comments by any of the following methods:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.
- *Fax:* (202) 493-2251.
- *Mail:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.
- *Hand Delivery:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this proposed AD, contact Airbus SAS,

Airworthiness Office—EAW, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 44 51; email [account.airworth-eas@airbus.com](mailto:account.airworth-eas@airbus.com); Internet <http://www.airbus.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

### Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2014-0448; or in person at the Docket Operations office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Operations office (telephone (800) 647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

**FOR FURTHER INFORMATION CONTACT:** Dan Rodina, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone 425-227-2125; fax 425-227-1149.

### SUPPLEMENTARY INFORMATION:

#### Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposed AD. Send your comments to an address listed under the **ADDRESSES** section. Include “Docket No. FAA-2014-0448; Directorate Identifier 2013-NM-055-AD” at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD based on those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

### Discussion

The European Aviation Safety Agency (EASA), which is the Technical Agent for the Member States of the European Community, has issued EASA Airworthiness Directive 2013-0058, dated March 11, 2013 (referred to after

this as the Mandatory Continuing Airworthiness Information, or “the MCAI”), to correct an unsafe condition for all Airbus Model A300 series airplanes; Airbus Model A300 B4–600, B4–600R, and F4–600R series airplanes, and Model A300 C4–605R Variant F airplanes (collectively called Model A300–600 series airplanes); and Airbus Model A310 series airplanes. The MCAI states:

Operators have reported five cases of early ruptures on levers of the nose landing gear (NLG) sequence valve.

Analysis showed that these fatigue ruptures were due to an incorrect adjustment of the mechanical links. As the design of the main landing gear (MLG) sequence valve lever is similar, there is sufficient reason to assume that these parts are similarly affected by fatigue.

This condition, if not detected and corrected, could lead to interference between landing gear leg and door and consequent failure of the landing gear to extend, possibly resulting in damage to the aeroplane and injury to occupants.

For the reasons described above, this [EASA] AD requires a one-time inspection of the sequence valve control lever [for damage, which could include cracking or deformation], of the adjustment of the control rod between doors and landing gear sequence valve and depending on inspections results, accomplishment of applicable corrective actions.

The corrective actions include adjusting the control rod between the door and the sequence valves; adjusting mechanical linkages; and replacing/ installing a serviceable valve and lever. You may examine the MCAI in the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA–2014–0448.

#### Relevant Service Information

Airbus has issued Service Bulletins:

- A300–32–0464, dated July 17, 2012 (for Model A300 airplanes);
- A300–32–6110, dated July 17, 2012 (for Model A300–600 airplanes); and
- A310–32–2146, dated July 17, 2012 (for Model A310 airplanes).

The actions described in this service information are intended to correct the unsafe condition identified in the MCAI.

#### FAA’s Determination and Requirements of This Proposed AD

This product has been approved by the aviation authority of another country, and is approved for operation in the United States. Pursuant to our bilateral agreement with the State of Design Authority, we have been notified of the unsafe condition described in the MCAI and service information

referenced above. We are proposing this AD because we evaluated all pertinent information and determined an unsafe condition exists and is likely to exist or develop on other products of the same type design.

#### “Contacting the Manufacturer” Paragraph in This Proposed AD

Since late 2006, we have included a standard paragraph titled “Airworthy Product” in all MCAI ADs in which the FAA develops an AD based on a foreign authority’s AD.

The MCAI or referenced service information in an FAA AD often directs the owner/operator to contact the manufacturer for corrective actions, such as a repair. Briefly, the Airworthy Product paragraph allowed owners/operators to use corrective actions provided by the manufacturer if those actions were FAA-approved. In addition, the paragraph stated that any actions approved by the State of Design Authority (or its delegated agent) are considered to be FAA-approved.

In another NPRM, Directorate Identifier 2012–NM–101–AD (78 FR 78285, December 26, 2013), we proposed to prevent the use of repairs that were not specifically developed to correct the unsafe condition, by requiring that the repair approval provided by the State of Design Authority or its delegated agent specifically refer to the FAA AD. This change was intended to clarify the method of compliance and to provide operators with better visibility of repairs that are specifically developed and approved to correct the unsafe condition. In addition, we proposed to change the phrase “its delegated agent” to include a design approval holder (DAH) with State of Design Authority design organization approval (DOA), as applicable, to refer to a DAH authorized to approve required repairs for the proposed AD.

One commenter to the other NPRM, Directorate Identifier 2012–NM–101–AD (78 FR 78285, December 26, 2013), stated the following: “The proposed wording, being specific to repairs, eliminates the interpretation that Airbus messages are acceptable for approving minor deviations (corrective actions) needed during accomplishment of an AD mandated Airbus service bulletin.”

This comment has made the FAA aware that some operators have misunderstood or misinterpreted the Airworthy Product paragraph to allow the owner/operator to use messages provided by the manufacturer as approval of deviations during the accomplishment of an AD-mandated action. The Airworthy Product

paragraph does not approve messages or other information provided by the manufacturer for deviations to the requirements of the AD-mandated actions. The Airworthy Product paragraph only addresses the requirement to contact the manufacturer for corrective actions for the identified unsafe condition and does not cover deviations from other AD requirements. However, deviations to AD-required actions are addressed in 14 CFR 39.17, and anyone may request the approval for an alternative method of compliance to the AD-required actions using the procedures found in 14 CFR 39.19.

To address this misunderstanding and misinterpretation of the Airworthy Product paragraph, we have changed that paragraph and retitled it “Contacting the Manufacturer.” This paragraph now clarifies that for any requirement in this proposed AD to obtain corrective actions from a manufacturer, the action must be accomplished using a method approved by the FAA, the European Aviation Safety Agency (EASA), or Airbus’s EASA DOA.

The Contacting the Manufacturer paragraph also clarifies that, if approved by the DOA, the approval must include the DOA-authorized signature. The DOA signature indicates that the data and information contained in the document are EASA-approved, which is also FAA-approved. Messages and other information provided by the manufacturer that do not contain the DOA-authorized signature approval are not EASA-approved, unless EASA directly approves the manufacturer’s message or other information.

This clarification does not remove flexibility previously afforded by the Airworthy Product paragraph. Consistent with long-standing FAA policy, such flexibility was never intended for required actions. This is also consistent with the recommendation of the Airworthiness Directive Implementation Aviation Rulemaking Committee to increase flexibility in complying with ADs by identifying those actions in manufacturers’ service instructions that are “Required for Compliance” with ADs. We continue to work with manufacturers to implement this recommendation. But once we determine that an action is required, any deviation from the requirement must be approved as an alternative method of compliance.

#### Costs of Compliance

We estimate that this proposed AD affects 128 airplanes of U.S. registry.

We also estimate that it would take about 4 work-hours per product to comply with the basic requirements of this proposed AD. The average labor rate is \$85 per work-hour. Based on these figures, we estimate the cost of this proposed AD on U.S. operators to be \$43,520, or \$340 per product.

In addition, we estimate that any necessary follow-on actions would take up to 9 work-hours and require parts costing up to \$42,000, for a cost of \$42,765 per product. We have no way of determining the number of aircraft that might need these actions.

#### Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

#### Regulatory Findings

We determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

*For the reasons discussed above, I certify this proposed regulation:*

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979);
3. Will not affect intrastate aviation in Alaska; and
4. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

#### The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

#### PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

- 2. Amend § 39.13 by adding the following new airworthiness directive (AD):

**Airbus:** Docket No. FAA-2014-0448; Directorate Identifier 2013-NM-055-AD.

#### (a) Comments Due Date

We must receive comments by September 2, 2014.

#### (b) Affected ADs

None.

#### (c) Applicability

This AD applies to all Airbus airplanes identified in paragraphs (c)(1), (c)(2), and (c)(3) of this AD, certificated in any category.

- (1) Airbus Model A300 B2-1A, B2-1C, B2K-3C, B2-203, B4-2C, B4-103, and B4-203 airplanes.
- (2) Airbus Model A300 B4-601, B4-603, B4-620, B4-622, B4-605R, B4-622R, F4-605R, F4-622R, and C4-605R Variant F airplanes.
- (3) Airbus Model A310-203, -204, -221, -222, -304, -322, -324, and -325 airplanes.

#### (d) Subject

Air Transport Association (ATA) of America Code 32, Landing Gear.

#### (e) Reason

This AD was prompted by a report of early ruptures on the levers of the nose landing gear (NLG) sequence valve. We are issuing this AD to detect and correct interference between a landing gear leg and door, which could result in failure of that landing gear to extend, and could damage the airplane and injure occupants.

#### (f) Compliance

Comply with this AD within the compliance times specified, unless already done.

#### (g) Service Information

Do the actions required by paragraph (h) of this AD in accordance with the applicable service information identified in paragraphs (g)(1) through (g)(3) of this AD.

- (1) For Model A300 airplanes: Airbus Service Bulletin A300-32-0464, dated July 17, 2012.

(2) For Model A300-600 airplanes: Airbus Service Bulletin A300-32-6110, dated July 17, 2012.

(3) For Model A310 airplanes: Airbus Service Bulletin A310-32-2146, dated July 17, 2012.

#### (h) Inspections and Corrective Actions

Within 4,000 flight cycles, 6,000 flight hours, or 30 months after the effective date of this AD, whichever occurs first: Do a detailed inspection of each sequence valve lever and pin shearing indicating area on the nose landing gear and main landing gears for any damage, in accordance with the Accomplishment Instructions of the applicable service information identified in paragraph (g) of this AD. Do the actions required by paragraphs (h)(1) and (h)(2) of this AD in accordance with the Accomplishment Instructions of the applicable service information identified in paragraph (g) of this AD.

(1) If damage is found, before further flight, replace the affected sequence valve and its lever with a serviceable sequence valve and lever. No further action is required by paragraph (h) of this AD for that replaced valve and lever.

(2) If no damage is found, within the compliance time required by paragraph (h) of this AD, do a detailed inspection to detect interference between the landing gear door control rod and the landing gear sequence valve, and do all applicable corrective actions. Do all applicable corrective actions before further flight. No further action is required by paragraph (h) of this AD.

(3) For the purposes of this AD, a detailed inspection is: An intensive examination of a specific item, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at an intensity deemed appropriate. Inspection aids such as mirror, magnifying lenses, etc., may be necessary. Surface cleaning and elaborate procedures may be required.

#### (i) Parts Installation Limitation

As of the effective date of this AD, no person may install on any airplane a landing gear sequence valve, unless that valve has been inspected and corrected, as applicable, in accordance with the requirements of paragraph (h) of this AD.

#### (j) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) Alternative Methods of Compliance (AMOCs): The Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the International Branch, send it to ATTN: Dan Rodina, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone 425-227-2125; fax 425-227-1149. Information may be emailed to: 9-ANM-116-

AMOC-REQUESTS@faa.gov. Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office. The AMOC approval letter must specifically reference this AD.

(2) Contacting the Manufacturer: For any requirement in this AD to obtain corrective actions from a manufacturer, the action must be accomplished using a method approved by the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA; or the European Aviation Safety Agency (EASA); or Airbus's EASA Design Organization Approval (DOA). If approved by the DOA, the approval must include the DOA-authorized signature.

#### (k) Related Information

(1) Refer to Mandatory Continuing Airworthiness Information (MCAI) European Aviation Safety Agency (EASA) Airworthiness Directive 2013-0058, dated March 11, 2013, for related information. This MCAI may be found in the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating it in Docket No. FAA-2014-0448.

(2) For service information identified in this AD, contact Airbus SAS, Airworthiness Office—EAW, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 44 51; email [account.airworth-eas@airbus.com](mailto:account.airworth-eas@airbus.com); Internet <http://www.airbus.com>. You may view this service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

Issued in Renton, Washington, on July 3, 2014.

**Dionne Palermo,**

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2014-16690 Filed 7-15-14; 8:45 am]

BILLING CODE 4910-13-P

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA-2014-0472; Directorate Identifier 2013-SW-040-AD]

RIN 2120-AA64

#### Airworthiness Directives; Agusta S.p.A Helicopters (Type Certificate Currently Held by AgustaWestland S.p.A.) (Agusta)

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** We propose to adopt a new airworthiness directive (AD) for certain Agusta Model A109E, A109K2, A119,

and AW119 MKII helicopters. This proposed AD was prompted by a report of a crack that was found on a Gleason crown. This proposed AD would require repetitively performing a magnetic particle inspection of the Gleason crown for a crack. We are proposing this AD to detect a crack, which could cause damage to or loss of the main rotor drive and subsequent loss of control of the helicopter.

**DATES:** We must receive comments on this proposed AD by September 2, 2014.

**ADDRESSES:** You may send comments, using the procedures found in 14 CFR 11.43 and 11.45, by any of the following methods:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.
- *Fax:* 202-493-2251.
- *Mail:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

- *Hand Delivery:* Deliver to Mail address above between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this proposed AD, contact AgustaWestland, Product Support Engineering, Via del Gregge, 100, 21015 Lonate Pozzolo (VA) Italy, ATTN: Maurizio D'Angelo; telephone 39-0331-664757; fax 39-0331-664680; or at <http://www.agustawestland.com/technical-bulletins>. You may review the referenced service information at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas 76137.

#### Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the European Aviation Safety Agency (EASA) AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (phone: 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

**FOR FURTHER INFORMATION CONTACT:** Rao Edupuganti, Aviation Safety Engineer, Regulations and Policy Group, Rotorcraft Directorate, FAA, 2601 Meacham Blvd., Fort Worth, Texas 76137; telephone (817) 222-5110; email [rao.edupuganti@faa.gov](mailto:rao.edupuganti@faa.gov).

#### SUPPLEMENTARY INFORMATION:

##### Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposal. Send your comments to an address listed under the **ADDRESSES** section. Include "Docket No. FAA-2014-0472; Directorate Identifier 2013-SW-040-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD because of those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

##### Discussion

EASA, which is the Technical Agent for the Member States of the European Union, has issued EASA AD No. 2013-0118, dated June 3, 2013, to correct an unsafe condition for Agusta Model A109K2, A109E, A119, and AW119MKII helicopters. EASA advises that during an overhaul of an A119 main transmission, part number (P/N) 109-0400-05-103, a crack on the Gleason crown, P/N 109-0403-07-103, was found. EASA further states that an investigation by Agusta revealed that the crack originated from the bottom of one of the 40 threaded holes in the Gleason crown, and that this part-numbered Gleason crown is also installed on Model A109 helicopters. EASA states that this condition, if not corrected, could cause damage to or loss of the main rotor drive and loss of control of the helicopter. To correct this unsafe condition, EASA AD No. 2013-0118 requires repetitive magnetic particle inspections of the Gleason crown and, if there is a crack, replacing the Gleason crown with a different part-numbered Gleason crown. EASA AD No. 2013-0118 also prohibits installing a Gleason crown, P/N 109-0403-07-103, or a Gleason crown assembly, P/N 109-0401-27-101 or P/N 109-0401-27-109, on any helicopter, as Gleason crown, P/N 109-0403-07-103, is a component of these assemblies.

##### Relevant Service Information

We reviewed Agusta Bollettino Tecnico (BT) No. 109EP-128 for Model A109E helicopters, Agusta BT No. 109K-57 for Model A109K2 helicopters, and Agusta BT No. 119-058 for Model