

**(d) Subject**

Air Transport Association of America (ATA) Code 28: Fuel.

**(e) Reason**

This AD was prompted by mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as a defective sealing of a tapped through bore hole at the inside of the fuel tank openings in combination with prolonged periods at maximum fuel level. We are issuing this AD to detect and correct a defective sealing of a tapped through bore hole at the inside of the fuel tank openings, which if not detected and corrected, could cause long-term structural degradation of the wing structure.

**(f) Actions and Compliance**

Unless already done, do the following actions, as specified in paragraphs (f)(1) through (f)(6), including subparagraphs, of this AD:

**Note 1 to paragraph (f) of this AD:** The service information referenced in this AD contains German to English translation. The MCAI cites the English translation. The following is the English to German translation of the service information entitled: AQUILA Aviation GmbH Vorgeschieden Technische Mitteilung SB-AT01-027, dated August 15, 2013 (English translation: AQUILA Aviation GmbH Mandatory Service Bulletin SB-AT01-027, Issue A.02, dated August 15, 2013). For paragraphs (f)(1) through (f)(6), the service information will be cited using the English translation.

(1) Within 100 hours time-in-service (TIS) after December 30, 2013 (the effective date of this AD) or 3 months after December 30, 2013 (the effective date of this AD), whichever occurs first, and repetitively thereafter at intervals not to exceed 12 months, visually inspect the left hand (LH) and right hand (RH) wing tank areas following paragraph (1) of the Actions section of AQUILA Aviation GmbH Mandatory Service Bulletin SB-AT01-027, Issue A.02, dated August 15, 2013.

(2) Concurrent with the initial inspection required in paragraph (f)(1) of this AD, seal the tapped through bore holes inside the LH and RH fuel tank openings following paragraph (2) of the Actions section of AQUILA Aviation GmbH Mandatory Service Bulletin SB-AT01-027, Issue A.02, dated August 15, 2013.

(3) If, during any subsequent inspection required in paragraph (f)(1) of this AD, a tapped through bore hole inside the LH or RH fuel tank opening is found to be improperly sealed, within the next 100 hours TIS after detecting the improper seal or 3 months after detecting the improper seal, whichever occurs first, renew the sealing of the affected bore hole following paragraph (2) of the Actions section of AQUILA Aviation GmbH Mandatory Service Bulletin SB-AT01-027, Issue A.02, dated August 15, 2013.

(4) If, during any inspection required in paragraph (f)(1) of this AD, the upper wing

shells show damaged finishing in the tank areas, before further flight, contact AQUILA Aviation GmbH following paragraph (3) of the Actions section of AQUILA Aviation GmbH Mandatory Service Bulletin SB-AT01-027, Issue A.02, dated August 15, 2013, at the address identified in paragraph (i)(3) of this AD for an approved repair scheme and, accomplish the repair scheme before further flight.

(5) Accomplishment of corrective actions required in paragraph (f)(3) or (f)(4) of this AD does not constitute terminating action for the repetitive inspections required by paragraph (f)(1) of this AD.

(6) After accomplishment of the required initial inspection and sealing in paragraphs (f)(1) and (f)(2) of this AD, compliance with the requirements of this AD can be demonstrated by:

(i) Revising the approved Aircraft Maintenance Program (AMP) and standard practices (Instructions for Continued Airworthiness) on the basis of which the operator or the owner ensures the continuing airworthiness of each airplane: Incorporate the repetitive 12 calendar month visual inspection of the LH and RH wing tank areas required in paragraph (f)(1) of this AD, Actions section of AQUILA Aviation GmbH Mandatory Service Bulletin SB-AT01-027, Issue A.02, dated August 15, 2013; and

(ii) Complying with the approved AMP described in paragraph (f)(6)(i) of this AD.

**(g) Other FAA AD Provisions**

The following provisions also apply to this AD:

(1) Alternative Methods of Compliance (AMOCs): The Manager, Standards Office, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. Send information to ATTN: Doug Rudolph, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4059; fax: (816) 329-4090; email: [doug.rudolph@faa.gov](mailto:doug.rudolph@faa.gov). Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

(2) Airworthy Product: For any requirement in this AD to obtain corrective actions from a manufacturer or other source, use these actions if they are FAA-approved. Corrective actions are considered FAA-approved if they are approved by the State of Design Authority (or their delegated agent). You are required to assure the product is airworthy before it is returned to service.

**(h) Related Information**

Refer to MCAI European Aviation Safety Agency (EASA) AD No. 2013-0236, dated September 25, 2013, for related information. You may examine the MCAI on the Internet at <http://www.regulations.gov> by searching for and locating it in Docket No. FAA-2013-0963.

**(i) Material Incorporated by Reference**

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this

paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless the AD specifies otherwise.

(i) AQUILA Aviation GmbH Vorgeschieden Technische Mitteilung SB-AT01-027, dated August 15, 2013 (English translation: AQUILA Aviation GmbH Mandatory Service Bulletin SB-AT01-027, Issue A.02, dated August 15, 2013).

**Note 2 to paragraph (i)(2)(i) of this AD:** This service information contains German to English translation. EASA used the English translation in referencing the documents from AQUILA Aviation GmbH. For enforceability purposes, we will refer to the AQUILA Aviation GmbH service information as the titles appear on the documents.

(ii) Reserved.

(3) For AQUILA—Aviation by Excellence AG service information identified in this AD, contact AQUILA Aviation GmbH, OT Schoenhagen, Flugplatz, D-14959 Trebbin, Germany; phone: +49-(0) 33731-707-0; fax: +49 (0) 33731-707-11; Internet: <http://www.aquila-aviation.de/>; email: [maintenance@aquila-aviation.de](mailto:maintenance@aquila-aviation.de).

(4) You may view this service information at the FAA, Small Airplane Directorate, 901 Locust, Kansas City, Missouri 64106. For information on the availability of this material at the FAA, call (816) 329-4148.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Kansas City, Missouri, on November 5, 2013.

**Earl Lawrence,**

*Manager, Small Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 2013-27914 Filed 11-22-13; 8:45 am]

**BILLING CODE 4910-13-P**

**DEPARTMENT OF TRANSPORTATION****Federal Aviation Administration****14 CFR Part 39**

**[Docket No. FAA-2013-0354; Directorate Identifier 2011-SW-072-AD; Amendment 39-17665; AD 2013-23-10]**

**RIN 2120-AA64**

**Airworthiness Directives; Eurocopter France Helicopters**

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Final rule.

**SUMMARY:** We are superseding Airworthiness Directive (AD) 2010-21-01 for Eurocopter France (Eurocopter) Model AS350B, BA, B1, B2, B3, D, AS355E, F, F1, F2, and N

helicopters. AD 2010–21–01 required an inspection to determine whether a cross-member is installed at station X 2165 and doublers at X 2325 and Y 269, and installing them if they are missing. This new AD retains the requirements of AD 2010–21–01 but clarifies the inspection procedures and limits the applicability to only those helicopters with collective-to-yaw control coupling. This AD is prompted by a crack discovered in the area of the center cross-member at station X 2325, at the attachment point of the yaw channel ball-type control sheath stop, of a Model AS355N helicopter fitted with the collective-to-yaw control coupling. The actions of this AD are intended to prevent reduced yaw control and subsequent loss of helicopter control.

**DATES:** This AD is effective December 30, 2013.

The Director of the Federal Register approved the incorporation by reference of certain documents listed in this AD as of December 30, 2013.

**ADDRESSES:** For service information identified in this AD, contact American Eurocopter Corporation, 2701 N. Forum Drive, Grand Prairie, TX 75052; telephone (972) 641–0000 or (800) 232–0323; fax (972) 641–3775; or at <http://www.eurocopter.com/techpub>. You may review the referenced service information at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas 76137.

#### Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> or in person at the Docket Operations Office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the European Aviation Safety Agency (EASA) AD, any incorporated-by-reference service information, the economic evaluation, any comments received, and other information. The street address for the Docket Operations Office (phone: 800–647–5527) is U.S. Department of Transportation, Docket Operations Office, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue SE., Washington, DC 20590.

**FOR FURTHER INFORMATION CONTACT:** Gary Roach, Aviation Safety Engineer, Regulations and Policy Group, Rotorcraft Directorate, FAA, 2601 Meacham Blvd., Fort Worth, TX 76137; telephone (817) 222–5110; email [gary.b.roach@faa.gov](mailto:gary.b.roach@faa.gov).

#### SUPPLEMENTARY INFORMATION:

#### Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to supersede AD 2010–21–01, Amendment 39–16461 (75 FR 63050, October 14, 2010), which applied to Eurocopter Model AS350B, BA, B1, B2, B3, D, AS355E, F, F1, F2, and N helicopters. The NPRM was published in the **Federal Register** on July 3, 2013 (78 FR 40045). AD No. 2010–21–01 required within 10 hours time-in-service (TIS) or 1 month, inspecting the helicopters to determine whether a cross-member is installed at station X 2165 and doublers at X 2325 and Y 269. If the cross-member and doublers are not installed, AD No. 2010–21–01 required inspecting for a crack in the center cross-member, and replacing the center cross-member if there is a crack before further flight. If a crack does not exist, AD No. 2010–21–01 required inspecting the tail rotor control rigging before further flight. Lastly, if needed, AD 2010–21–01 required installing a cross-member and two doublers within 55 hours TIS. AD 2010–21–01 was prompted by AD No. 2007–0139–E, dated May 15, 2007 (corrected May 23, 2007), issued by EASA, which is the Technical Agent for the Member States of the European Union. EASA advised that a crack was discovered in the area of the center cross-member at station X 2325, at the attachment point of the yaw channel ball-type control sheath stop, of a Model AS355N helicopter fitted with the collective-to-yaw control coupling.

Since we issued AD No. 2010–21–01, we discovered that we included all helicopters in the AD applicability rather than limiting it to only those helicopters with collective-to-yaw control coupling. Therefore, this new AD retains the requirements in AD No. 2010–21–01 with some revisions for the inspection of the tail rotor control rigging to clarify those procedures. This AD also reduces the applicability to only those Model AS350 and AS355 helicopters with collective-to-yaw control coupling installed.

#### Comments

We gave the public the opportunity to participate in developing this AD, but we received no comments on the NPRM (78 FR 40045, July 3, 2013).

#### FAA's Determination

These helicopters have been approved by the aviation authority of France and are approved for operation in the United States. Pursuant to our bilateral agreement with France, EASA, its technical representative, has notified us of the unsafe condition described in the

EASA AD. We are issuing this AD because we evaluated all information provided by EASA and determined the unsafe condition exists and is likely to exist or develop on other helicopters of these same type designs and that air safety and the public interest require adopting the AD requirements as proposed.

#### Differences Between This AD and the EASA AD

This AD differs from the EASA AD as follows:

- We require the installation of the cross-member at station X 2165 and the two doublers at stations X 2325 and Y 269 within 55 hours time-in-service. The EASA AD requires that this action be accomplished within 12 months.
- We do not require repetitive inspections if no crack exists in the center cross-member, whereas the EASA AD does.
- We do not include military model helicopters in the applicability.

#### Related Service Information

We reviewed Eurocopter Emergency Alert Service Bulletin (EASB), Revision 0, dated April 11, 2007, that contains three different numbers (Nos. 53.00.37, 53.00.11, and 53.00.23) for Eurocopter Model 350, 355, 550, and 555 helicopters. EASB No. 53.00.37 relates to two Model 350 (350 BB and 350 L1) helicopters that are not type certificated in the United States. EASB No. 53.00.11 relates to four Model 550 and six Model 555 military helicopters that are not type-certificated in the United States. The EASB describes procedures for checking the conformity for the cross member at X 2325 under the cabin floor. The actions in the EASA AD are intended to correct the same unsafe condition as that identified in the service bulletin.

#### Costs of Compliance

We estimate that this AD affects 72 helicopters of U.S. Registry and that labor costs average \$85 a work-hour. It takes about one work-hour to perform the inspections, and if needed, to install the cross-member, two doublers and an airworthy center-cross member. Required parts cost about \$161 per helicopter. Based on these figures, we estimate the cost of the AD to be \$246 per helicopter and \$17,712 for the fleet if all repairs are needed.

#### Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII:

Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701: "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on helicopters identified in this rulemaking action.

### Regulatory Findings

This AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866;
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979);
- (3) Will not affect intrastate aviation in Alaska to the extent that it justifies making a regulatory distinction; and
- (4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared an economic evaluation of the estimated costs to comply with this AD and placed it in the AD docket.

### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

### Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

### PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

- 2. The FAA amends § 39.13 by removing Airworthiness Directive (AD)

2010–21–01, Amendment 39–16461 (75 FR 63050, October 14, 2010), and adding the following new AD:

**2013–23–10 Eurocopter France:**  
Amendment 39–17665; Docket No. FAA–2013–0354; Directorate Identifier 2011–SW–072–AD.

#### (a) Applicability

This AD applies to Eurocopter France Model AS350B, BA, B1, B2, B3, D, AS355E, F, F1, F2, and N helicopters with collective-to-yaw control coupling, part number 350A27–2178–04, 350A27–2178–06, or 350A27–2178–0601, certificated in any category.

#### (b) Unsafe Condition

This AD defines the unsafe condition as reduced yaw control travel, which could result in loss of control of the helicopter.

#### (c) Affected ADs

This AD supersedes AD 2010–21–01, Amendment 39–16461 (75 FR 63050, October 14, 2010).

#### (d) Effective Date

This AD becomes effective December 30, 2013.

#### (e) Compliance

You are responsible for performing each action required by this AD within the specified compliance time unless it has already been accomplished prior to that time.

#### (f) Required Actions

(1) Within 10 hours time-in-service (TIS) or within one month, whichever occurs first, determine whether the cross-member (numbered "1") at station X 2165 and the two doublers (numbered "2" and "3") at stations X 2325 and Y 269 are installed as shown in Figure 1 of Eurocopter Emergency Alert Service Bulletin (EASB) No. 53.00.37, Revision 0, dated April 11, 2007 (EASB 53.00.37), for Model AS350 helicopters and EASB No. 53.00.23, Revision 0, dated April 11, 2007 (EASB 53.00.23), for Model AS355 helicopters.

(2) If the cross-member (numbered "1") and doublers (numbered "2" and "3") are not installed, before further flight, inspect for a crack in the center cross-member (numbered "4") in the area around the attachment point of the tail rotor directional ball-type control as shown in Figure 1 of EASB 53.00.37 for Model AS350 helicopters or EASB 53.00.23 for Model AS355 helicopters.

(i) If a crack exists, before further flight, replace the unairworthy center cross-member (Numbered "4") with an airworthy center cross-member as described in paragraph (f)(3) of this AD.

(ii) If a crack does not exist, before further flight, inspect the tail rotor control rigging to determine whether it meets conformity limits.

(A) If all items of the tail rotor control rigging are found within conformity limits, install the cross-member and doublers as described in paragraph (f)(3) of this AD.

(B) For any items of the tail rotor control rigging found outside of conformity limits, perform appropriate corrective action in

accordance with FAA-accepted procedures, and install the cross-member and doublers as described in paragraph (f)(3) of this AD.

(3) Within 55 hours TIS, if the cross member (Numbered "1") is not installed, install the cross-member at station X 2165 and the 2 doublers (Numbered "2" and "3") at stations X 2325 and Y 269 by following the Appendix, the referenced figures 2 and 3 of EASB 53.00.37 for Model AS350 helicopters or EASB 53.00.23 for Model AS355 helicopters.

#### (g) Alternative Methods of Compliance (AMOCs)

(1) The Manager, Safety Management Group, FAA, may approve AMOCs for this AD. Send your proposal to: Gary Roach, Aviation Safety Engineer, Regulations and Policy Group, Rotorcraft Directorate, FAA, 2601 Meacham Blvd., Fort Worth, TX 76137; telephone (817) 222–5110; email [gary.b.roach@faa.gov](mailto:gary.b.roach@faa.gov).

(2) For operations conducted under a 14 CFR part 119 operating certificate or under 14 CFR part 91, subpart K, we suggest that you notify your principal inspector, or lacking a principal inspector, the manager of the local flight standards district office or certificate holding district office, before operating any aircraft complying with this AD through an AMOC.

#### (h) Additional Information

The subject of this AD is addressed in the European Aviation Safety Agency (EASA) AD No. 2007–0139–E, dated May 15, 2007 (corrected May 23, 2007). You may view the EASA AD at <http://www.regulations.gov> by searching for and locating it in Docket No. FAA–2013–0354.

#### (i) Subject

Joint Aircraft Service Component (JASC) Code: 5320, Fuselage Miscellaneous Structure.

#### (j) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless the AD specifies otherwise.

(i) Eurocopter Emergency Alert Service Bulletin No. 53.00.37, Revision 0, dated April 11, 2007.

(ii) Eurocopter Emergency Alert Service Bulletin No. 53.00.23, Revision 0, dated April 11, 2007.

**Note 1 to paragraph (j)(2):** Eurocopter Emergency Alert Service Bulletin (EASB) No. 53.00.37, Revision 0, dated April 11, 2007, and Eurocopter EASB No. 53.00.23, Revision 0, dated April 11, 2007, are co-published as one document along with Eurocopter EASB No. 53.00.11, Revision 0, dated April 11, 2007, which is not incorporated by reference in this AD.

(3) For Eurocopter service information identified in this AD, contact American Eurocopter Corporation, 2701 N. Forum Drive, Grand Prairie, TX 75052; telephone (972) 641–0000 or (800) 232–0323; fax (972)

641–3775; or at <http://www.eurocopter.com/techpub>.

(4) You may view this service information at FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas 76137. For information on the availability of this material at the FAA, call (817) 222–5110.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call (202) 741–6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Fort Worth, Texas, on November 5, 2013.

**Lance T. Gant,**

*Acting Directorate Manager, Rotorcraft Directorate, Aircraft Certification Service.*

[FR Doc. 2013–27637 Filed 11–22–13; 8:45 am]

**BILLING CODE 4910–13–P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA–2013–0487; Directorate Identifier 2010–SW–056–AD; Amendment 39–17666; AD 2013–23–11]

RIN 2120–AA64

#### Airworthiness Directives; Eurocopter France Helicopters

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Final rule.

**SUMMARY:** We are adopting a new airworthiness directive (AD) for Eurocopter France (Eurocopter) Model AS332L2 and EC225LP helicopters. This AD requires inspecting the torque value of the bolts that secure the front and rear main gearbox (MGB) suspension bar attaching fittings, and re-torquing the bolts to the proper value if the torque value is out of tolerance. This AD also requires, if the torque value is out of tolerance by more than 20 percent, inspecting the bolts, frames, and related equipment for a crack and repairing or replacing them if cracked. This AD was prompted by reports of cracks on Frame 5295 of Model AS332L2 helicopters. The actions of this AD are intended to detect the torque loss of the bolts that secure the MGB bar attaching fittings and to prevent cracks that could lead to failure of the MGB supporting structure, detachment of the MGB, and loss of helicopter control.

**DATES:** This AD is effective December 30, 2013.

The Director of the Federal Register approved the incorporation by reference

of certain documents listed in this AD as of December 30, 2013.

**ADDRESSES:** For service information identified in this AD, contact American Eurocopter Corporation, 2701 N. Forum Drive, Grand Prairie, TX 75052; telephone (972) 641–0000 or (800) 232–0323; fax (972) 641–3775; or at <http://www.eurocopter.com/techpub>. You may review the referenced service information at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas 76137.

#### Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> or in person at the Docket Operations Office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the foreign authority's AD, any incorporated-by-reference service information, the economic evaluation, any comments received, and other information. The street address for the Docket Operations Office (phone: 800–647–5527) is U.S. Department of Transportation, Docket Operations Office, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue SE., Washington, DC 20590.

**FOR FURTHER INFORMATION CONTACT:** Gary Roach, Aviation Safety Engineer, Regulations and Policy Group, Rotorcraft Directorate, FAA, 2601 Meacham Blvd., Fort Worth, Texas 76137; telephone (817) 222–5110; email [gary.b.roach@faa.gov](mailto:gary.b.roach@faa.gov).

#### SUPPLEMENTARY INFORMATION:

##### Discussion

On June 7, 2013, at 78 FR 34288, the **Federal Register** published our notice of proposed rulemaking (NPRM), which proposed to amend 14 CFR part 39 to include an AD that would apply to Eurocopter Model AS332L2 and EC225LP helicopters. The NPRM proposed to require inspecting the torque value of the bolts that secure the front and rear MGB suspension bar attaching fittings, and re-torquing the bolts to the proper value if the torque value is out of tolerance. The NPRM also proposed to require that if the torque value is out of tolerance by more than 20 percent, inspecting the bolts, frames, and related equipment for a crack and repairing or replacing them if cracked. The proposed requirements were intended to detect the torque loss of the bolts that secure the MGB bar attaching fittings and to prevent cracks that could lead to failure of the MGB

supporting structure, detachment of the MGB, and loss of helicopter control.

The NPRM was prompted by AD No. 2006–0163 R1, dated December 13, 2007 (AD No. 2006–0163R1), issued by the European Aviation Safety Agency (EASA), which is the Technical Agent for the Member States of the European Community, to correct an unsafe condition in Model AS 332 L2 and Model EC 225 LP helicopters. According to EASA, analysis of tightening torques revealed some cases of tightening torque loss, which can lead to the formation of a crack at the MGB bar attaching fittings. As a result, EASA AD No. 2006–0163R1 requires checking the bolts securing the front and rear of the MGB bar attaching fittings for tightening torque loss and, if the loss is equal to or greater than 20 percent, readjusting the torque and checking the four bolts securing the MGB bar attaching fitting mounting plate, as well as the frame 3855, for a crack. If there is a crack in at least one of the bolts, AD No. 2006–0163R1 requires replacing all four bolts. If there is a crack in frame 3855, AD No. 2006–0163R1 requires suspending all flights and contacting the manufacturer for corrective action.

#### Comments

We gave the public the opportunity to participate in developing this AD, but we received no comments on the NPRM (78 FR 34288, June 7, 2013).

#### FAA's Determination

These helicopters have been approved by the aviation authority of France and are approved for operation in the United States. Pursuant to our bilateral agreement with France, EASA, its technical representative, has notified us of the unsafe condition described in the EASA AD. We are issuing this AD because we evaluated all information provided by EASA and determined the unsafe condition exists and is likely to exist or develop on other helicopters of these same type designs and that air safety and the public interest require adopting the AD requirements as proposed.

#### Differences Between This AD and the EASA AD

This AD differs from the EASA AD in that we use the word “inspect” to describe actions required by a mechanic versus the word “check,” which is how we describe actions allowed by a pilot. We also require that if you find a crack in a frame or fitting, you repair or replace the cracked part instead of contacting the manufacturer. Also, we have different compliance times for the initial inspection for the tightening