

been authorized by the Manager, Seattle ACO, to make those findings. For a repair method to be approved, the repair must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

(4) AMOCs approved previously in accordance with AD 2008-06-29, Amendment 39-15441 (73 FR 15397, March 24, 2008), are approved as AMOCs for the corresponding provisions of this AD.

(l) Related Information

For more information about this AD, contact Nancy Marsh, Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Seattle Aircraft Certification Office (ACO), 1601 Lind Avenue SW., Renton, WA 98057-3356; phone: (425) 917-6440; fax: (425) 917-6590; email: nancy.marsh@faa.gov.

(m) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless this AD specifies otherwise.

(i) Boeing Service Bulletin 737-57A1301, Revision 3, dated August 11, 2011.

(ii) Reserved.

(3) For service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P.O. Box 3707, MC 2H-65, Seattle, Washington 98124-2207; telephone 206-544-5000, extension 1; fax 206-766-5680; Internet <https://www.myboeingfleet.com>.

(4) You may view this service information at FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, Washington. For information on the availability of this material at the FAA, call 425-227-1221.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Renton, Washington, on July 21, 2013.

Stephen P. Boyd,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2013-19811 Filed 8-15-13; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2013-0362; Directorate Identifier 2013-NM-030-AD; Amendment 39-17531; AD 2013-15-15]

RIN 2120-AA64

Airworthiness Directives; The Boeing Company Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for certain The Boeing Company Model 727 airplanes. This AD was prompted by an evaluation by the design approval holder indicating that the frame-to-floor beam attachment is subject to widespread fatigue damage. This AD requires repetitive high frequency eddy current inspections for any crack of the frames at body station (STA) 188 through STA 344, and repair if necessary. We are issuing this AD to detect and correct fatigue cracking at the frame-to-floor beam attachment, on both the left- and right-sides, which could result in reduced structural integrity of the airplane, and decompression of the cabin.

DATES: This AD is effective September 20, 2013.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in the AD as of September 20, 2013.

ADDRESSES: For service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P.O. Box 3707, MC 2H-65, Seattle, WA 98124-2207; telephone 206-544-5000, extension 1; fax 206-766-5680; Internet <https://www.myboeingfleet.com>. You may review copies of the referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, Washington 98057-3356. For information on the availability of this material at the FAA, call 425-227-1221.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the

Docket Office (phone: 800-647-5527) is Document Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT:

Berhane Alazar, Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Seattle Aircraft Certification Office (ACO), 1601 Lind Avenue SW., Renton, Washington 98057-3356; phone: 425-917-6577; fax: 425-917-6590; email: berhane.alazar@faa.gov.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that would apply to the specified products. The NPRM published in the **Federal Register** on May 3, 2013 (78 FR 25905). The NPRM proposed to require repetitive high frequency eddy current inspections for any crack of the frames at body STA 188 through STA 344, and repair if necessary.

Comments

We gave the public the opportunity to participate in developing this AD. We have considered the comments received.

Boeing stated that it supports the NPRM (78 FR 25905, May 3, 2013).

Fedex stated that the NPRM (78 FR 25905, May 3, 2013) will be effective for twenty of its Model 727-200 airplanes, the inspection threshold and intervals will fit within its planned scheduled maintenance checks and therefore will be no impact to available lift, the number of man-hours and elapsed time to accomplish the inspections will not impact the overall span-time of its planned scheduled maintenance check, and the inspections do not require any special inspection techniques, training, or tooling.

Conclusion

We reviewed the relevant data, considered the comments received, and determined that air safety and the public interest require adopting this AD as proposed—except for minor editorial changes. We have determined that these minor changes:

- Are consistent with the intent that was proposed in the NPRM (78 FR 25905, May 3, 2013) for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the NPRM (78 FR 25905, May 3, 2013).

Costs of Compliance

We estimate that this AD affects 106 airplanes of U.S. registry.

We estimate the following costs to comply with this AD:

Action	Labor cost	Parts cost	Cost per product	Cost on U.S. operators
Inspection	118 work-hours × \$85 per hour = \$10,030 per inspection cycle.	\$0	\$10,030 per inspection cycle	\$1,063,180 per inspection cycle.

We have received no definitive data that would enable us to provide cost estimates for the on-condition actions specified in this AD.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA’s authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII: Aviation Programs, describes in more detail the scope of the Agency’s authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701: “General requirements.” Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

This AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a “significant regulatory action” under Executive Order 12866,
- (2) Is not a “significant rule” under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),
- (3) Will not affect intrastate aviation in Alaska, and
- (4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

2013–15–15 The Boeing Company:
Amendment 39–17531; Docket No. FAA–2013–0362; Directorate Identifier 2013–NM–030–AD.

(a) Effective Date

This AD is effective September 20, 2013.

(b) Affected ADs

None.

(c) Applicability

This AD applies to The Boeing Company Model 727, 727C, 727–100, 727–100C, 727–200, and 727–200F series airplanes, certificated in any category, as identified in Boeing Special Attention Service Bulletin 727–53–0234, dated January 17, 2013.

(d) Subject

Joint Aircraft System Component (JASC)/ Air Transport Association (ATA) of America Code 53, Fuselage.

(e) Unsafe Condition

This AD was prompted by an evaluation by the design approval holder indicating that the frame-to-floor beam attachment is subject to widespread fatigue damage. We are issuing this AD to detect and correct fatigue cracking at the frame-to-floor beam attachment, on both the left- and right-sides, which could result in reduced structural integrity of the airplane, and decompression of the cabin.

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) Inspection and Repair

Before the accumulation of 61,000 total flight cycles, or within 24 months after the effective date of this AD, whichever occurs later, do a high frequency eddy current inspection for cracking of the frames (for

certain stations), in the area of the floor beam attachments on both the left- and right-sides of the airplane, in accordance with the Accomplishment Instructions of Boeing Special Attention Service Bulletin 727–53–0234, dated January 17, 2013. Repeat this inspection thereafter at intervals not to exceed 20,000 flight cycles. If any crack is found during any inspection required by this AD, before further flight, repair the crack using a method approved in accordance with the procedures specified in paragraph (h) of this AD.

(h) Alternative Methods of Compliance (AMOCs)

(1) The Manager, Seattle Aircraft Certification Office (ACO), FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the ACO, send it to the attention of the person identified in the Related Information section of this AD. Information may be emailed to: 9-ANM-Seattle-ACO-AMOC-Requests@faa.gov.

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/ certificate holding district office.

(3) An AMOC that provides an acceptable level of safety may be used for any repair required by this AD if it is approved by the Boeing Commercial Airplanes Organization Designation Authorization (ODA) that has been authorized by the Manager, Seattle ACO, to make those findings. For a repair method to be approved, the repair must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

(i) Related Information

For more information about this AD, contact Berhane Alazar, Aerospace Engineer, Airframe Branch, ANM–120S, FAA, Seattle ACO, 1601 Lind Avenue SW., Renton, Washington 98057–3356; phone: 425–917–6577; fax: 425–917–6590; email: berhane.alazar@faa.gov.

(j) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless the AD specifies otherwise.

(i) Boeing Special Attention Service Bulletin 727–53–0234, dated January 17, 2013.

(ii) Reserved.

(3) For Boeing service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P.O. Box 3707, MC 2H-65, Seattle, WA 98124-2207; telephone 206-544-5000, extension 1; fax 206-766-5680; Internet <https://www.myboeingfleet.com>.

(4) You may view this service information at FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, Washington 98057-3356. For information on the availability of this material at the FAA, call 425-227-1221.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Renton, Washington, on July 21, 2013.

Stephen P. Boyd,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2013-18122 Filed 8-15-13; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2012-1321; Directorate Identifier 2011-NM-147-AD; Amendment 39-17528; AD 2013-15-12]

RIN 2120-AA64

Airworthiness Directives; Airbus Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are superseding airworthiness directive (AD) 2004-15-07, for certain Airbus Model A310 series airplanes. AD 2004-15-07 required repetitive inspections for fatigue cracking of the area around the fasteners of the landing plate of the aileron access doors of the bottom skin panel of the wings, and related corrective action. AD 2004-15-07 also provided for an optional terminating action to end the repetitive inspections. This new AD reduces the initial inspection compliance time and intervals, and provides additional terminating action options. This AD was prompted by a reassessment of a previous fatigue threshold and inspection interval, which resulted in a determination that reduced inspection thresholds and intervals for accomplishment of the

tasks are necessary. We are issuing this AD to detect and correct fatigue cracking of the area around the fasteners of the landing plate of the aileron access doors and the bottom skin panel of the wings, which could result in reduced structural integrity of the wings.

DATES: This AD becomes effective September 20, 2013.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD as of September 20, 2013.

The Director of the Federal Register approved the incorporation by reference of certain other publications listed in this AD as of August 31, 2004 (69 FR 44592, July 27, 2004).

ADDRESSES: You may examine the AD docket on the Internet at <http://www.regulations.gov> or in person at the U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC.

FOR FURTHER INFORMATION CONTACT: Dan Rodina, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone (425) 227-2125; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that would apply to the specified products. The NPRM was published in the **Federal Register** on February 5, 2013 (78 FR 8054), and proposed to supersede AD 2004-15-07, Amendment 39-13741 (69 FR 44592, July 27, 2004). The NPRM proposed to correct an unsafe condition for the specified products. The European Aviation Safety Agency (EASA), which is the Technical Agent for the Member States of the European Community, has issued EASA Airworthiness Directive 2011-0125, dated June 30, 2011 (referred to after this as "the MCAI"), to correct an unsafe condition for the specified products. The MCAI states:

DGAC [Direction Générale de l'Aviation Civile] France issued AD 2003-242(B) [which corresponds to FAA AD 2004-15-07, Amendment 39-13741 (69 FR 44592, July 27, 2004)] to require an inspection programme for aeroplanes with pre- and post-Airbus modification 05106 configurations (Airbus SB A310-57-2004) in order to detect any crack located on the trailing edge of the wing bottom skin No. 2 panel of the all-speed-aileron servo control bay. A crack at this location, if not detected and corrected, would propagate towards the wing rear spar and

ultimately into the wing fuel tank area. Undetected cracks would affect the structural integrity of the [left hand] LH and/or [right hand] RH wing.

Since issuance of DGAC France AD 2003-242(B) [which corresponds to FAA AD 2004-15-07, Amendment 39-13741 (69 FR 44592, July 27, 2004)], a reassessment of the previous fatigue threshold and inspection interval has been completed. As a result of the reassessment, the inspection thresholds and intervals for accomplishment of the tasks as defined in Airbus SB A310-57-2082 have been adjusted and reduced. Airbus SB A310-57-2082 Revision 03 has been published, in which the compliance time periods for these inspection thresholds and intervals have been amended.

For the reasons stated above, this [EASA] AD retains the requirements of the DGAC France AD 2003-242(B) [which corresponds to FAA AD 2004-15-07, Amendment 39-13741 (69 FR 44592, July 27, 2004)], which is superseded, and requires implementation of the amended inspection programme.

Corrective action includes doing a permanent repair (installing a repair plate and new landing plates), a temporary repair (crack-stop drilling and application of a protective coating) followed by repetitive inspections until a permanent repair is done, and a repair approved by the FAA or EASA (or its delegated agent). This AD also adds optional permanent repairs.

The initial inspection compliance times are dependent on the configuration (modification status, repair status, and crack length), and type of use (short range, long range, and normal). For airplanes without temporary repairs, the initial inspection compliance time ranges between 2,000 total flight cycles or 10,200 total flight hours, whichever occurs first; and 12,000 total flight cycles or 24,000 total flight hours, whichever occurs first. If the total flight cycles or total flight hours compliance time has been exceeded, the initial inspection compliance time (grace period) ranges between 200 flight cycles or 1,000 flight hours, to within 1,000 flight cycles or 2,800 flight hours, whichever occurs first.

For airplanes with temporary repairs, the initial inspection compliance time is dependent on crack length and ranges between 7 flight cycles or 35 flight hours, whichever occurs first, since the repair; to within 100 flight cycles or 200 flight hours, whichever occurs first, since the repair.

For airplanes with a temporary repair, the compliance time for completing the permanent repair ranges between 35 flight cycles or 175 flight hours, whichever occurs first, after completing the temporary repair; to within 500 flight cycles or 1,000 flight hours,