

## The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

### PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

■ 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

#### **Bell Helicopter Textron, Inc., Helicopters:**

Docket No. FAA-2008-0288; Directorate Identifier 2006-SW-25-AD.

#### (a) Applicability

This AD applies to Bell Helicopter Textron, Inc. (Bell), Model 214B, 214B-1, and 214ST helicopters, with pylon support spindle assembly (spindle), part number (P/N) 214-030-606-005 or -103, installed, certificated in any category.

#### (b) Unsafe Condition

This AD defines the unsafe condition as fatigue cracking of a spindle. This condition could result in failure of the spindle and subsequent loss of control of the helicopter.

#### (c) Compliance

You are responsible for performing each action required by this AD within the specified compliance time unless it has already been accomplished prior to that time.

#### (d) Required Actions

(1) Within 50 hours time-in-service (TIS):

(i) Create a component history card or equivalent record for each spindle, P/N 214-030-606-005 and 214-030-606-103, recording the spindle's P/N and serial number (S/N).

(ii) Review the helicopter records to determine the hours TIS of each spindle, if the hours TIS are not already recorded for your model helicopter. For each month for which the hours TIS is unknown, record 75 hours TIS.

(iii) Determine the total accumulated retirement index number (RIN) for each spindle. For the purpose of this AD, count 1 RIN for each takeoff and 2 RIN for each external load lift in which the helicopter achieves a vertical altitude difference of greater than 200 feet indicated altitude between the pick-up and drop-off point. For any time period for which the accumulated RIN cannot be determined while the spindle was installed on a helicopter, multiply the hours TIS by 30 to calculate the spindle's accumulated RIN.

(iv) Record the hours TIS and total accumulated RIN for each spindle on the component history card or equivalent record.

(2) Revise the Airworthiness Limitations section of the applicable maintenance manual or Instructions for Continued Airworthiness as follows:

(i) By establishing a new retirement life for the spindle, P/N 214-030-606-005, of 1,250 hours TIS or a total accumulated RIN of 20,000, whichever occurs first.

(ii) By reducing the retirement life for the spindle, P/N 214-030-606-103, from 5,000 hours TIS to 2,500 hours TIS or a total accumulated RIN of 50,000, whichever occurs first.

(3) Replace any spindle, P/N 214-030-606-005, that has been in service for 1,250 or more hours TIS, or a total accumulated RIN of 20,000 or more, whichever occurs first.

(4) Replace any spindle, P/N 214-030-606-103, that has been in service for 2,500 or more hours TIS, or a total accumulated RIN of 50,000 or more, whichever occurs first.

(5) Continue to count and record the accumulated RIN count and hours TIS for each spindle on its component history card or equivalent record.

#### (e) Alternative Methods of Compliance (AMOCs)

(1) The Manager, Rotorcraft Certification Office, FAA, may approve AMOCs for this AD. Send your proposal to: Martin Crane, Aviation Safety Engineer, Rotorcraft Certification Office, Rotorcraft Directorate, FAA, 2601 Meacham Blvd., Fort Worth, Texas 76137; telephone (817) 222-5056; email 7-AVS-ASW-170@faa.gov.

(2) For operations conducted under a 14 CFR part 119 operating certificate or under 14 CFR part 91, subpart K, we suggest that you notify your principal inspector, or lacking a principal inspector, the manager of the local flight standards district office or certificate holding district office before operating any aircraft complying with this AD through an AMOC.

#### (f) Subject

Joint Aircraft Service Component (JASC) Code: 6330, Transmission Mount.

Issued in Fort Worth, Texas, on May 17, 2013.

#### Kim Smith,

*Directorate Manager, Rotorcraft Directorate, Aircraft Certification Service.*

[FR Doc. 2013-12522 Filed 5-24-13; 8:45 am]

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## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA-2013-0454; Directorate Identifier 2009-SW-081-AD]

RIN 2120-AA64

#### **Airworthiness Directives; Sikorsky Aircraft Corporation-Manufactured (Sikorsky) Model Helicopters (type certificate currently held by Erickson Air-Crane Incorporated)**

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** We propose to supersede an existing airworthiness directive (AD) for Sikorsky Aircraft Corporation-manufactured Model S-64E helicopters (type certificate currently held by Erickson Air-Crane Incorporated (Erickson)). That AD currently requires inspecting and reworking the main gearbox (MGB) assembly second stage lower planetary plate (plate). This action would establish or reduce the life limits for certain flight-critical components, remove from service various parts, require repetitive inspections and other corrective actions, and require replacing any cracked part discovered during an inspection. This proposal is prompted by further analysis performed by the current type certificate holder and the service history of certain parts. The actions specified in the proposed AD are intended to prevent a crack in a flight critical component, failure of a critical part, and subsequent loss of control of the helicopter.

**DATES:** We must receive comments on this proposed AD by July 29, 2013.

**ADDRESSES:** You may send comments by any of the following methods:

- *Federal eRulemaking Docket:* Go to <http://www.regulations.gov>. Follow the online instructions for sending your comments electronically.
- *Fax:* 202-493-2251.
- *Mail:* Send comments to the U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590-0001.

- *Hand Delivery:* Deliver to the "Mail" address between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

#### Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> or in person at the Docket Operations Office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the economic evaluation, any comments received and other information. The street address for the Docket Operations Office (telephone 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

For service information identified in this proposed AD, contact Erickson Air-Crane Incorporated, ATTN: Chris Erickson/Compliance Officer, 3100 Willow Springs Rd, PO Box 3247, Central Point, OR 97502, telephone

(541) 664-5544, fax (541) 664-2312, email address [cerickson@ericksonaircrane.com](mailto:cerickson@ericksonaircrane.com). You may review a copy of the service information at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas 76137.

**FOR FURTHER INFORMATION CONTACT:**

Michael Kohner, Aerospace Engineer, Rotorcraft Certification Office, Rotorcraft Directorate, FAA, 2601 Meacham Blvd., Fort Worth, TX 76137; telephone (817) 222-5170; email [7-avs-asw-170@faa.gov](mailto:7-avs-asw-170@faa.gov).

**SUPPLEMENTARY INFORMATION:**

**Comments Invited**

We invite you to participate in this rulemaking by submitting written comments, data, or views. We also invite comments relating to the economic, environmental, energy, or federalism impacts that might result from adopting the proposals in this document. The most helpful comments reference a specific portion of the proposal, explain the reason for any recommended change, and include supporting data. To ensure the docket does not contain duplicate comments, commenters should send only one copy of written comments, or if comments are filed electronically, commenters should submit only one time.

We will file in the docket all comments that we receive, as well as a report summarizing each substantive public contact with FAA personnel concerning this proposed rulemaking. Before acting on this proposal, we will consider all comments we receive on or before the closing date for comments. We will consider comments filed after the comment period has closed if it is possible to do so without incurring expense or delay. We may change this proposal in light of the comments we receive.

**Discussion**

On September 5, 1997, we issued AD 97-19-10, Amendment 39-10130 (62 FR 47933, September 12, 1997), to require, at 1,300 hours time-in-service (TIS), a fluorescent-magnetic particle inspection of the plate, part number (P/N) 6435-20229-102, for a crack, replacing the plate if a crack is found, and reworking the plate if there is no crack. That action also requires, at 1,500 hours TIS, and thereafter at intervals not to exceed 70 hours TIS, for a reworked plate, P/N 6435-20229-102, and for plate, P/N 6435-20229-104, inspecting for a crack and replacing the main gearbox assembly if a crack is found. Finally, AD 97-19-10 requires retiring

these part-numbered plates upon reaching 2,600 hours TIS. That action was prompted by reports that cracks were discovered in four plates, three of which had been reworked in accordance with previously superseded AD 77-20-01, Amendment 39-3045 (42 FR 51565, September 29, 1977) and Amendment 39-3064 (42 FR 56600, October 27, 1977). The requirements of AD 97-19-10 are intended to detect a crack in the plate to prevent failure of the plate, failure of the main gearbox, and subsequent loss of control of the helicopter.

**Actions Since Existing AD Was Issued**

Since issuing AD 97-19-10, Amendment 39-10130 (62 FR 47933, September 12, 1997), Erickson has performed a configuration review and additional analyses of the safe life for various parts and released Erickson Service Bulletin (SB) No. 64B General-1, Revision 19, dated September 15, 2010 (SB 64B General-1). SB 64B General-1 specifies the retirement life for certain parts and assemblies as well as noting other maintenance actions. This and the previous revisions of SB 64B General-1 contain reduced or new life limits for certain parts, parts which should be removed from service, other maintenance actions, and various other provisions for certain parts. We have also reviewed Erickson SB No. 64B10-3, Revision D, dated October 15, 2007, which provides ultrasonic inspection procedures for the Main Rotor (M/R) hub horizontal hinge pins. Based on our review of this list of parts and assemblies, and an analysis of the service difficulties, we have determined that we need to propose to revise the life limit of certain critical parts, remove certain parts from service, and require additional inspections and other maintenance actions of certain parts. Failure to establish or revise a life limit or remove a part from service when there is repeated service difficulties with the part could result in failure of that part and subsequent loss of control of the helicopter.

**FAA's Determination**

We are proposing this AD because we evaluated all the relevant information and determined the unsafe condition described previously is likely to exist or develop in other helicopters of the same type design.

**Proposed AD Requirements**

The proposed AD would supersede AD 97-19-10 (62 FR 47933, September 12, 1997) to establish or revise the life limit for various parts, to remove various parts from service, to require

various inspections and other maintenance actions, and to revise the component history card or equivalent record and the airworthiness limitations section of the maintenance manual accordingly. Specifically, we propose, before further flight, to establish or to revise the retirement life for each of the following parts, and to remove from service those parts that have reached or exceeded the newly established or reduced life limit:

- M/R parts
  - Rod and bushing assembly, M/R, P/N 6410-21090-012, -013 or -014;
  - Lower plate, M/R hub, P/N 6410-23009-102;
  - Upper plate, M/R hub, P/N 6410-23011-102;
  - Swashplate, rotating, M/R, P/N 6410-24002-101;
  - Piston rod, P/N 6410-26005-104;
  - Cylinder, damper assembly, P/N 6410-26215-101;
  - M/R blade, P/N 6415-20201-045 or -047;
  - M/R blade, P/N 6415-20201-048, -049, -050, or -051;
  - M/R shaft assembly (includes shaft, P/N 6435-20078-104), P/N 6435-20078-014 or -015;
  - M/R shaft assembly (includes shaft, P/N 6435-20078-105), P/N 6435-20078-016;
  - Hub, M/R, P/N S1510-23001-005;
  - Spindle assembly, M/R, P/N S1510-23027-5;
  - Horn assembly, M/R, P/N S1510-23350-4, -6, or -8;
  - Sleeve, M/R, P/N S1510-23351-2;
  - Sleeve lockwasher, M/R, S1510-23458-0;
  - Cuff, M/R blade, P/N S1515-20320-0, -001 or -002;
  - Piston assembly, M/R tandem servo, P/N S1565-20443-0 or S1565-20443-301;
  - Fork assembly, M/R tandem servo, P/N S1565-20449 or P/N S1565-20449-301;
  - Tail Rotor (T/R) parts
    - T/R blade, P/N 65160-00001-042, -045, or -048;
    - T/R blade, P/N 65161-00001-042;
    - Bearing, T/R drive shaft, P/N SB1111-004 or -601;
    - Main Gearbox parts
      - Second stage planetary plate assembly, main gearbox assembly, P/N 6435-20231-012, -014, -015 or -016;
      - Oil cooler and support assembly, P/N 6435-60050-044; and
      - Other parts
        - Truss assembly, stabilizer, P/N 6420-66250-041.

In addition to establishing and revising the life limit for certain parts, we also propose to require, before further flight, removing from service the

following flight-critical parts due to previous service difficulties:

- Rod and bushing assembly, M/R, P/N 6410–21090–011;
- M/R blade, P/N 6415–20001–013, –014, or –015;
- Pylon stabilizer, P/N 6420–66201–010, –014, or –015;
- M/R shaft assembly, P/N 6435–20078–013;
- Oil cooler and support assembly, P/N 6435–60050–043;
- Pitch change link, rotary rudder, P/N 65113–07100–046; and
- Spindle, M/R blade, P/N S1510–23070–3.

Any part that is required to be removed from service is not eligible for installation on any helicopter.

This proposed AD also would also require the following inspections and other corrective actions:

- Within 20 hours TIS, and thereafter at intervals not to exceed 20 hours TIS, inspecting each M/R servo and control arm assembly, P/N S1565–20421–10, –11, –041, or –043, for any oil leaking from the M/R tandem servo housing assembly (servo housing), P/N S1565–20252–2. If there is any oil leaking from the servo housing, before further flight, replacing the M/R servo and control arm assembly.
  - Within 20 hours TIS or before reaching a total of 1,120 hours TIS, whichever occurs later, and thereafter at intervals not to exceed 200 hours TIS or 12 months, whichever occurs first, performing an ultrasonic inspection (UT) of each M/R hub horizontal hinge pin, P/N S1510–23099–1 or P/N S1510–23099–001, for a crack.
  - Performing a fluorescent-magnetic particle inspection (MPI) for a crack:
    - In each second stage planetary plate assembly, P/N 6435–20231–016, within 150 hours TIS, or before reaching 1,450 hours TIS, whichever occurs later;
    - In each M/R shaft, P/N 6435–20078–104, within 150 hours TIS or before reaching 1,450 hours TIS, whichever occurs later, and thereafter at intervals not to exceed 650 hours TIS;
    - In each M/R shaft, P/N 6435–20078–105, within 150 hours TIS or before reaching 1,450 hours TIS, whichever occurs later, and thereafter at intervals not to exceed 1,450 hours TIS; and
    - In each M/R shaft assembly, P/N 6435–20078–014, –015, or –016, at each overhaul of the main gearbox assembly, P/N 6435–20400–053, –054, –058, –060, –062, –063, –064, –065, or –066.
  - Within 150 hours TIS or before reaching 3,375 hours TIS, whichever occurs later, and thereafter at intervals not to exceed 3,375 hours TIS,

performing a fluorescent-penetrant inspection (FPI) of each housing lug on each M/R tandem servo housing assembly, P/N S1565–20252–2.

- If a there is a crack, before further flight, replacing the cracked part.
- At each overhaul of the damper assembly, P/N 6410–26200–042, replacing certain parts with parts that have zero (0) hours TIS.

#### Differences Between the Proposed AD and the Service Information

This proposed AD contains only those parts for the Model S–64E helicopters whose life limit has either been reduced or added for an existing P/N, whereas SB 64B General-1 also contains parts whose life limits have been extended. As a result, this proposed AD does not include all of the parts or P/Ns that are listed in SB 64B General-1.

#### Costs of Compliance

We estimate that this proposed AD would affect 13 helicopters of U.S. registry, and the proposed actions would take the following number of estimated work hours to accomplish:

- 26 work hours (2 work hours per helicopter) for the fleet to review the helicopter records or to remove a part to determine if an affected part is installed;
- 845 work hours (65 work hours per helicopter) for the fleet to replace the parts or assemblies on or before reaching the retirement lives stated in Table 1 of the proposed AD, assuming an annual usage of 600 hours TIS;
- 287 work hours per helicopter to replace all the parts or assemblies listed in Table 2 of the proposed AD;
- 130 work hours (10 work hours per helicopter) for the fleet to inspect the M/R servo housing assemblies for an oil leak, assuming each inspection would take approximately 0.25 work hour per helicopter and would be accomplished 40 times annually;
- Approximately 293 work hours (22.5 work hours per helicopter) for the fleet to UT inspect each M/R hub horizontal hinge pin, assuming that each inspection would take 7.5 work hours per helicopter and would be accomplished 3 times annually;
- 288 work hours (48 work hours per helicopter) to perform an MPI of each main gearbox second stage lower planetary plate and second stage planetary plate assembly assuming 6 helicopters would be inspected annually;
- 192 work hours (32 work hours per helicopter) to perform an MPI of each M/R shaft and M/R shaft assembly, assuming 6 helicopters would be inspected annually, and

- 96 work hours (32 work hours per helicopter) to perform an FPI of each M/R tandem servo housing assembly, assuming 3 helicopters would be inspected annually.

Therefore, we estimate that it would take approximately 2,157 work hours to accomplish the proposed actions at a cost of \$183,345, using an average labor rate \$85 per work hour. Replacement parts would cost approximately:

- \$5,363,449 (\$412,573 per helicopter) to replace the parts or assemblies on the entire fleet on or before reaching the proposed retirement lives, assuming parts for 13 helicopters would require replacement; and
- \$2,594,400 per helicopter to replace the parts or assemblies that are listed in Table 2 of the proposed AD.

Using these assumptions, the estimated total cost for the required parts would be approximately \$7,957,849. Based on these estimated amounts using these assumptions, we estimate the total cost impact of the proposed AD on the U.S. operators would be \$8,141,194.

#### Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on helicopters identified in this rulemaking action.

#### Regulatory Findings

We determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

*For the reasons discussed, I certify this proposed regulation:*

1. Is not a "significant regulatory action" under Executive Order 12866;

2. Is not a “significant rule” under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979);

3. Will not affect intrastate aviation in Alaska to the extent that it justifies making a regulatory distinction; and

4. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared an economic evaluation of the estimated costs to comply with this proposed AD and placed it in the AD docket.

**List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

**The Proposed Amendment**

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

**PART 39—AIRWORTHINESS DIRECTIVES**

■ 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

**§ 39.13 [Amended]**

■ 2. The FAA amends § 39.13 by removing Amendment 39–10130 (62 FR 47933, September 12, 1997), and adding the following new AD:

**Sikorsky Aircraft Corporation—Manufactured (Sikorsky) Model Helicopters (Type Certificate Currently Held By Erickson Air-Crane Incorporated):** Docket No. FAA–2013–0454; Directorate Identifier 2009–SW–81–AD.

**(a) Applicability**

This AD applies to Sikorsky Model CH–54A helicopters, now under the Erickson Air-Crane Incorporated (Erickson) Model S–64E type certificate, certificated in any category.

**(b) Unsafe Condition**

This AD defines the unsafe condition as fatigue cracking in a flight critical

component, failure of the component, and subsequent loss of control of the helicopter.

**(c) Affected ADs**

This AD supersedes AD 97–19–10, Amendment 39–10130 (62 FR 47933, September 12, 1997).

**(d) Compliance**

You are responsible for performing each action required by this AD within the specified compliance time unless it has already been accomplished prior to that time.

**(e) Required Actions**

(1) Before further flight, for each part listed in Table 1 to paragraph (e) of this AD:

(i) Remove any part that has reached or exceeded its newly established or revised retirement life.

(ii) Record the newly established or revised retirement life for each part on the component history card or equivalent record.

(iii) Make pen and ink changes or insert a copy of this AD into the Airworthiness Limitations section of the maintenance manual to establish or revise the retirement life for each part that is listed in Table 1 of this AD.

TABLE 1 TO PARAGRAPH (E) OF THIS AD—PARTS WITH NEW OR REVISED LIFE LIMITS

Part name	Part number (P/N)	Retirement life
Rod and bushing assembly, main rotor (M/R) .....	6410–21090–012 .....	5,700 hours time-in-service (TIS) or 60 months since the initial installation on any helicopter, whichever occurs first.
Rod and bushing assembly, M/R .....	6410–21090–013 or –014 .....	5,700 hours TIS.
Lower plate, M/R hub .....	6410–23009–102 .....	3,000 hours TIS.
Upper plate, M/R hub .....	6410–23011–102 .....	3,000 hours TIS.
Swashplate, rotating, M/R .....	6410–24002–101 .....	12,860 hours TIS.
Piston rod .....	6410–26005–104 .....	10,500 hours TIS.
Cylinder, damper assembly .....	6410–26215–101 .....	7,300 hours TIS.
M/R blade .....	6415–20201–045 or –047 .....	3,300 hours TIS.
M/R blade .....	6415–20201–048, –049, –050, or –051 ...	20,000 hours TIS.
Truss assembly, stabilizer .....	6420–66250–041 .....	4,720 hours TIS.
M/R shaft assembly (includes shaft, P/N 6435–20078–104).	6435–20078–014 or –015 .....	2,600 hours TIS.
M/R shaft assembly (includes shaft, P/N 6435–20078–105).	6435–20078–016 .....	5,000 hours TIS.
Second stage planetary plate assembly, main gearbox assembly.	6435–20231–012, –014, or –015 .....	1,300 hours TIS.
Second stage planetary plate assembly, main gearbox assembly.	6435–20231–016 .....	2,600 hours TIS.
Oil cooler and support assembly .....	6435–60050–044 .....	9,885 hours TIS.
Tail rotor (T/R) blade .....	65160–00001–042, –045, or –048 .....	23,300 hours TIS.
T/R blade .....	65161–00001–042 .....	23,300 hours TIS.
Hub, M/R .....	S1510–23001–005 .....	3,000 hours TIS.
Spindle assembly, M/R .....	S1510–23027–5 .....	5,675 hours TIS.
Horn assembly, M/R .....	S1510–23350–4, –6, or –8 .....	9,710 hours TIS.
Sleeve, M/R .....	S1510–23351–2 .....	12,930 hours TIS.
Sleeve lockwasher, M/R .....	S1510–23458–0 .....	2,700 hours TIS.
Cuff, M/R blade .....	S1515–20320–0 .....	6,410 hours TIS.
Cuff, M/R blade .....	S1515–20320–001 or –002 .....	12,930 hours TIS.
Piston assembly, M/R tandem servo .....	S1565–20443–0 or S1565–20443–301 ....	8,100 hours TIS.
Fork assembly, M/R tandem servo .....	S1565–20449 or S1565–20449–301 .....	8,100 hours TIS.
Bearing, T/R drive shaft .....	SB1111–004 or –601 .....	1,000 hours TIS or 12 months while installed on any helicopter, whichever occurs first.

**Note to Table 1 to paragraph (e) of this AD:** The list of parts in Table 1 to paragraph (e) of this AD contains only a portion of the

life-limited parts for this model helicopter and is not an all-inclusive list.

(2) Before further flight, remove from service any part with a P/N listed in Table 2 to Paragraph (e) of this AD, regardless of

the part's TIS. The part numbers listed in Table 2 to paragraph (e)(2) of this AD are not eligible for installation on any helicopter.

**TABLE 2 TO PARAGRAPH (e) OF THIS AD—PARTS TO BE REMOVED FROM SERVICE**

Part name	P/N
Rod and bushing assembly, M/R.	6410-21090-011
M/R blade .....	6415-20001-013, -014, or -015
Pylon stabilizer .....	6420-66201-010, -014, or -015
M/R shaft assembly .....	6435-20078-013
Oil cooler and support assembly.	6435-60050-043
Pitch change link, rotary rudder.	65113-07100-046
Spindle, M/R blade .....	S1510-23070-3

(3) Within 20 hours TIS, and thereafter at intervals not to exceed 20 hours TIS, visually inspect each M/R servo and control arm assembly, P/N S1565-20421-10, -11, -041, or -043, and determine if there is any oil leaking from the M/R tandem servo housing assembly (servo housing), P/N S1565-20252-2. If there is any oil leaking from the servo housing, before further flight, replace the M/R servo and control arm assembly.

(4) Within 20 hours TIS or before reaching 1,120 hours TIS, whichever occurs later, and thereafter at intervals not to exceed 200 hours TIS or 12 months, whichever occurs first, ultrasonic (UT) inspect each M/R hub horizontal hinge pin (hinge pin), P/N S1510-23099-1 or P/N S1510-23099-001, for a crack in accordance with the Accomplishment Instructions, paragraphs 2.A through 2.C, of Erickson Service Bulletin (SB) No. 64B10-3, Revision D, dated October 15, 2007, except you are not required to contact Erickson nor send hinge pins to them. A non-destructive testing (NDT) UT Level I Special, Level II, or Level III inspector who is qualified under the guidelines established by ASNT SNT-TC-1A, ISO 9712, or an FAA-accepted equivalent qualification standard for NDT inspection and evaluation, must perform the UT inspection.

(5) Within 150 hours TIS or before reaching 1,450 hours TIS, whichever occurs later, perform a fluorescent-magnetic particle inspection (MPI) of each second stage planetary plate assembly, P/N 6435-20231-016, for a crack.

(6) Within 150 hours TIS or before reaching 1,450 hours TIS, whichever occurs later, and thereafter at intervals not to exceed 650 hours TIS, perform an MPI of each M/R shaft, P/N 6435-20078-104, for a crack, paying particular attention to the lower spline area.

(7) Within 150 hours TIS or before reaching 1,450 hours TIS, whichever occurs later, and thereafter at intervals not to exceed 1,450 hours TIS, perform an MPI of each M/R shaft, P/N 6435-20078-105, for a crack, paying particular attention to the lower spline area.

(8) Within 150 hours TIS or before reaching 3,375 hours TIS, whichever occurs later, and

thereafter at intervals not to exceed 3,375 hours TIS, perform a fluorescent penetrant inspection of each housing lug on each servo housing, P/N S1565-20252-2, for a crack.

(9) At each overhaul of the main gearbox assembly, P/N 6435-20400-053, -054, -058, -060, -062, -063, -064, -065, or -066, perform an MPI of the entire shaft of each M/R shaft assembly, P/N 6435-20078-014, -015, or -016, for a crack, paying particular attention to the rotating swashplate spherical bearing ball travel area, which is located approximately ten inches above the upper roller bearing journal shoulder.

(10) If there is a crack in any part, before further flight, replace the cracked part.

(11) At each overhaul of the damper assembly, P/N 6410-26200-042, replace the following parts with airworthy parts that have zero (0) hours TIS:

- (i) All Air Force-Navy Aeronautical Standard (AN), Aerospace Standard (AS), Military Standard (MS), and National Aerospace Standard (NAS) nuts, bolts, washers, and packings, except packing, P/N MS28775-011, installed on stud, P/N SHF111-11SN-12A;
- (ii) Lock washer, P/N SS5073-2;
- (iii) Nut, P/N SS5081-05;
- (iv) Felt seal, P/N S1510-26017;
- (v) Retaining ring, P/N UR106L; and
- (vi) Nut, P/N 6410-26214-101.

**(f) Alternative Methods of Compliance (AMOCs)**

(1) The Manager, Rotorcraft Certification Office, FAA, may approve AMOCs for this AD. Send your proposal to: Michael Kohner, Aerospace Engineer, Rotorcraft Certification Office, Rotorcraft Directorate, FAA, 2601 Meacham Blvd., Fort Worth, TX 76137; telephone (817) 222-5170; email [7-avs-asw-170@faa.gov](mailto:7-avs-asw-170@faa.gov).

(2) For operations conducted under a 14 CFR part 119 operating certificate or under 14 CFR part 91, subpart K, we suggest that you notify your principal inspector, or lacking a principal inspector, the manager of the local flight standards district office or certificate holding district office before operating any aircraft complying with this AD through an AMOC.

**(g) Additional Information**

Erickson Service Bulletin (SB) No. 64B General-1, Revision 19, dated September 15, 2010, which is not incorporated by reference, contains additional information about the subject of this AD. For service information identified in this AD, contact Erickson Air-Crane Incorporated, ATTN: Chris Erickson/ Compliance Officer, 3100 Willow Springs Rd, PO Box 3247, Central Point, OR 97502, telephone (541) 664-5544, fax (541) 664-2312, email address [cerickson@ericksonaircrane.com](mailto:cerickson@ericksonaircrane.com). You may review a copy of this information at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas 76137.

**(h) Subject**

Joint Aircraft Service Component (JASC) Code: 6200: Main Rotor System; 6300: Main Rotor Drive System; 6410: Tail Rotor Blades; 6500: Tail Rotor Drive System.

Issued in Fort Worth, Texas, on May 17, 2013.

**Kim Smith,**

*Directorate Manager, Rotorcraft Directorate, Aircraft Certification Service.*

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**BILLING CODE 4910-13-P**

**DEPARTMENT OF TRANSPORTATION**

**Federal Aviation Administration**

**14 CFR Part 39**

**[Docket No. FAA-2013-0425; Directorate Identifier 2012-NM-224-AD]**

**RIN 2120-AA64**

**Airworthiness Directives; The Boeing Company Airplanes**

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** We propose to adopt a new airworthiness directive (AD) for certain The Boeing Company Model 747 airplanes. This proposed AD was prompted by reports of cracking in the forward and aft inner chord of the body station (BS) 2598 bulkhead near the upper corners of the cutout for the horizontal stabilizer rear spar, and cracking in the bulkhead upper and lower web panels near the inner chord to shear deck connection. This proposed AD would require doing repetitive inspections for cracking in the bulkhead splice fitting, frame supports, forward and aft inner chords, and floor support; doing an inspection for cracking in the bulkhead upper web, doubler, and bulkhead lower web; and corrective actions if necessary; for certain airplanes, inspections for cracking in the repaired area of the bulkhead, and corrective actions if necessary; for certain airplanes, support frame modification and support frame inspections, and related investigative and corrective actions, if necessary; for certain airplanes, repetitive support frame post-modification inspections and inspections for cracking in the hinge support, and related investigative and corrective actions if necessary; for certain airplanes, a one-time inspection of the frame web and upper shear deck (floor support) chord aft side for fasteners; and a one-time inspection of the upper forward inner chord, frame support fitting and splice fitting, for the installation of certain fasteners; and related investigative and corrective actions if necessary; for certain airplanes, a one-time inspection of the upper forward inner chord, frame