

Furthermore, 14 CFR regulations and current system safety assessment policy and techniques do not address potential security vulnerabilities, which could be exploited by unauthorized access to airplane systems, data buses, and servers. Therefore, these special conditions are issued to ensure that the security (i.e., confidentiality, integrity, and availability) of airplane systems is not compromised by unauthorized wired or wireless electronic connections.

Applicability

As discussed above, these special conditions are applicable to the Model G280. Should GALP apply at a later date for a change to the type certificate to include another model incorporating the same novel or unusual design feature, the special conditions would apply to that model as well.

Conclusion

This action affects only certain novel or unusual design features on one model of airplane. It is not a rule of general applicability.

The substance of these special conditions has been subjected to the notice and comment period in several prior instances and has been derived without substantive change from those previously issued. It is unlikely that prior public comment would result in a significant change from the substance contained herein. Therefore, because a delay would significantly affect the certification of the airplane, which is imminent, the FAA has determined that prior public notice and comment are unnecessary and impracticable, and good cause exists for adopting these special conditions upon issuance. The FAA is requesting comments to allow interested persons to submit views that may not have been submitted in response to the prior opportunities for comment described above.

List of Subjects in 14 CFR Part 25

Aircraft, Aviation safety, Reporting and recordkeeping requirements.

The authority citation for these special conditions is as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701, 44702, 44704.

The Special Conditions

Accordingly, pursuant to the authority delegated to me by the Administrator, the following special conditions are issued as part of the type certification basis for Gulfstream Aerospace LP, Model Gulfstream G280 airplanes.

1. Airplane Electronic System Security Protection from Unauthorized

External Access. The applicant must ensure airplane electronic system security protection from access to or by unauthorized sources external to the airplane, including those possibly caused by maintenance activity.

2. The applicant must ensure that electronic system security threats are identified and assessed, and that effective electronic system security protection strategies are implemented to protect the airplane from all adverse impacts on safety, functionality, and continued airworthiness.

3. The applicant must establish appropriate procedures to allow the operator to ensure that continued airworthiness of the aircraft is maintained, including all post type certification modifications that may have an impact on the approved electronic system security safeguards.

Issued in Renton, Washington, on June 7, 2012.

Michael J. Kaszycki,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2012-14787 Filed 6-15-12; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2012-0250; Directorate Identifier 2011-CE-043-AD; Amendment 39-17063; AD 2012-10-14]

RIN 2120-AA64

Airworthiness Directives; SOCATA Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for SOCATA Model TBM 700 airplanes. This AD results from mandatory continuing airworthiness information (MCAI) issued by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as installation of an incorrect part number during overhaul of the nose landing gear. We are issuing this AD to require actions to address the unsafe condition on these products.

DATES: This AD is effective July 23, 2012.

The Director of the Federal Register approved the incorporation by reference

of certain publications listed in the AD as of July 23, 2012.

ADDRESSES: You may examine the AD docket on the Internet at <http://www.regulations.gov> or in person at Document Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

For service information identified in this AD, contact SOCATA—Direction des Services, 65921 Tarbes Cedex 9, France; telephone: +33 (0)5 62 41 73 00; fax: +33 (0)5 62 41 76 54; or in the United States contact SOCATA North America, Inc., North Perry Airport, 7501 South Airport Road, Pembroke Pines, Florida 33023; telephone: (954) 893-1400; fax: (954) 964-4141; email: mysocata@socata.daher.com; Internet: www.socatanorthamerica.com. You may review copies of the referenced service information at the FAA, Small Airplane Directorate, 901 Locust, Kansas City, Missouri 64106. For information on the availability of this material at the FAA, call (816) 329-4148.

FOR FURTHER INFORMATION CONTACT:

Albert Mercado, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4119; fax: (816) 329-4090; email: albert.mercado@faa.gov.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that would apply to the specified products. That NPRM was published in the **Federal Register** on March 9, 2012 (77 FR 14314). That NPRM proposed to correct an unsafe condition for the specified products. The MCAI states:

A TBM 700 operator reported a case of rupture of the bolt attaching the actuator hinge axle on the NLG of the aeroplane. The results of the technical investigations carried out by SOCATA revealed that this rupture could have been caused by the installation of a bolt bearing incorrect Part Number (P/N) during overhaul of the NLG. Furthermore, the investigations led to identify the NLG part numbers identified by S/N which are potentially affected after repair or overhaul.

This condition, if not detected and corrected, could lead to partial disengagement of the actuator hinge axle on the NLG of the aeroplane, resulting in nose landing gear collapse, possibly resulting in structural damage to the aeroplane.

To address this condition, SOCATA have developed Service Bulletin SB 70-194-32 which gives instructions for accomplishing repetitive checks of the bolt attaching actuator hinge axle on NLG and for replacing

the bolt attaching the actuator hinge axle with a correct bolt P/N.

For the reasons described above, this AD requires accomplishment of repetitive checks of potentially affected NLGs and replacement of the bolt attaching the actuator hinge axle with a serviceable bolt. This AD also prohibits installation on any aeroplane of a potentially affected NLG, unless the bolt attaching the actuator hinge axle has been replaced with a serviceable bolt and the NLG has been marked with a green varnish line.

Following issuance of EASA AD 2011-0225-E, it has been determined that further NLG P/Ns and S/Ns are affected by this AD. SOCATA have developed an erratum to SB 70-194-32 amendment 2, which lists the new P/Ns and S/Ns as well affected by this AD.

For the above reason, this AD, which supersedes EASA AD 2011-0225-E, retaining its requirements, extends the list of NLG P/Ns and S/Ns affected by the AD requirements.

Comments

We gave the public the opportunity to participate in developing this AD. We received no comments on the NPRM (77 FR 14314, March 9, 2012) or on the determination of the cost to the public.

Conclusion

We reviewed the relevant data and determined that air safety and the public interest require adopting the AD as proposed except for minor editorial changes. We have determined that these minor changes:

- Are consistent with the intent that was proposed in the NPRM (77 FR 14314, March 9, 2012) for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the NPRM (77 FR 14314, March 9, 2012).

Costs of Compliance

We estimate that this AD will affect 448 products of U.S. registry. We also estimate that it would take about 1 work-hour per product to comply with the basic requirements of this AD. The average labor rate is \$85 per work-hour. Required parts would cost about \$35 per product.

Based on these figures, we estimate the cost of the AD on U.S. operators to be \$53,760, or \$120 per product.

According to the manufacturer, some of the costs of this AD may be covered under warranty, thereby reducing the cost impact on affected individuals. We do not control warranty coverage for affected individuals. As a result, we have included all costs in our cost estimate.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue

rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866,
- (2) Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),
- (3) Will not affect intrastate aviation in Alaska, and
- (4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains the NPRM, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (telephone (800) 647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by adding the following new AD:

2012-10-14 SOCATA: Amendment 39-17063; Docket No. FAA-2012-0250; Directorate Identifier 2011-CE-043-AD.

(a) Effective Date

This airworthiness directive (AD) becomes effective July 23, 2012.

(b) Affected ADs

None.

(c) Applicability

This AD applies to SOCATA Model TBM 700 airplanes, all serial numbers (S/N), certificated in any category, equipped with one of the following landing gears:

- (1) Part number (P/N) D23766000 or D23766000-X, serial numbers (S/N) B001 through B373; B375; AR1000 through AR1023; AR1025 through AR1031; AR1033 through AR1036; AAB00000A through AAB13766Z; AAB00000 through AAB13766Z; and EURXXX; or
- (2) P/N 21130-001-XY or 21130-000-XY, all S/N.

(d) Subject

Air Transport Association of America (ATA) Code 32: Landing Gear.

(e) Reason

This AD was prompted by mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as installation of an incorrect part number during overhaul of the nose landing gear. We are issuing this AD to detect and correct installation of incorrect P/N NLG bolts, which if not corrected could result in NLG collapse with consequent structural damage to the airplane.

(f) Actions and Compliance

Unless already done, do the following actions using the Accomplishment Instructions of DAHER-SOCATA TBM Aircraft Mandatory Service Bulletin SB 70-194-32, Amendment 2, dated November 2011, including Erratum, dated December 2011:

- (1) Although the European Aviation Safety Agency (EASA) MCAI allows the inspection of the NLG washer to be done by a pilot-owner, the U.S. regulatory system requires all actions of this AD to be done by a certified mechanic.

(2) Within 5 flight cycles (FC) after July 23, 2012 (the effective date of this AD), inspect the installed NLG to determine if it is one of the affected P/Ns and S/Ns as listed in paragraph (c) of this AD.

(i) If FC data is not available, the use of a one-to-one FC to flight hour conversion must be applied (example: 5 FC equal 5 hours time-in-service (TIS)).

(ii) For the purpose of this AD, when an NLG P/N reference is followed by -X or -XY, the X or XY can be any numerical digit, and when an NLG S/N reference is EURXXXX, the XXX can be any numerical digit.

(3) If during the inspection required in paragraph (f)(2) of this AD, you determine the NLG installed is one of the affected P/Ns and S/Ns listed in paragraph (c) of the AD, inspect for free rotation the washer of the NLG. Repetitively thereafter inspect the washer of the NLG for free rotation before every flight until the replacement and landing gear marking required in paragraphs (f)(4)(i) and (f)(4)(ii) or paragraphs (f)(5)(i) and (f)(5)(ii) of this AD are done.

(4) If, during any inspection required by paragraph (f)(3) of this AD, the washer of the NLG rotates freely, before further flight, do the following actions:

(i) Replace the bolt attaching the actuator hinge axle of the NLG with a serviceable bolt P/N 510130111.

(ii) Mark the landing gear with a green varnish line.

(5) For the NLG P/Ns and S/Ns as listed in paragraph (c) of this AD, within 10 months after July 23, 2012 (the effective date of this AD), unless already done following a discrepancy identified during any inspection as required by paragraph (f)(3) of this AD, do the following actions:

(i) Replace the bolt attaching the actuator hinge axle of the NLG with a serviceable bolt P/N 510130111 and;

(ii) Mark the landing gear with a green varnish line.

(6) Replacing of the bolt attaching the actuator hinge axle of the NLG with a serviceable bolt P/N 510130111 and marking the landing gear with a green varnish line terminates the repetitive inspections required by paragraph (f)(3) of this AD.

(7) After July 23, 2012 (the effective date of this AD), do not install an NLG with P/N and S/N as listed in paragraph (c) of this AD, unless the bolt attaching the actuator hinge axle of the NLG has been replaced and the NLG has been marked with a green varnish line following the requirements of this AD.

(g) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs)*: The Manager, Standards Office, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. Send information to ATTN: Albert Mercado, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4119; fax: (816) 329-4090; email: albert.mercado@faa.gov. Before using any approved AMOC on any airplane

to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

(2) *Airworthy Product*: For any requirement in this AD to obtain corrective actions from a manufacturer or other source, use these actions if they are FAA-approved. Corrective actions are considered FAA-approved if they are approved by the State of Design Authority (or their delegated agent). You are required to assure the product is airworthy before it is returned to service.

(3) *Reporting Requirements*: For any reporting requirement in this AD, a federal agency may not conduct or sponsor, and a person is not required to respond to, nor shall a person be subject to a penalty for failure to comply with a collection of information subject to the requirements of the Paperwork Reduction Act unless that collection of information displays a current valid OMB Control Number. The OMB Control Number for this information collection is 2120-0056. Public reporting for this collection of information is estimated to be approximately 5 minutes per response, including the time for reviewing instructions, completing and reviewing the collection of information. All responses to this collection of information are mandatory. Comments concerning the accuracy of this burden and suggestions for reducing the burden should be directed to the FAA at: 800 Independence Ave. SW., Washington, DC 20591, Attn: Information Collection Clearance Officer, AES-200.

(h) Related Information

Refer to MCAI EASA AD No.: 2011-0235-E, dated December 13, 2011; DAHER-SOCATA TBM Aircraft Mandatory Service Bulletin SB 70-194-32, Amendment 2, dated November 2011; and Erratum to DAHER-SOCATA TBM Aircraft Mandatory Service Bulletin SB 70 194-32, Amendment 2, dated December 2011, for related information.

(i) Material Incorporated by Reference

(1) You must use DAHER-SOCATA TBM Aircraft Mandatory Service Bulletin SB 70-194-32, Amendment 2, dated November 2011, including Erratum, dated December 2011, to do the actions required by this AD, unless the AD specifies otherwise. The Director of the Federal Register approved the incorporation by reference (IBR) under 5 U.S.C. 552(a) and 1 CFR part 51:

(2) For service information identified in this AD, contact SOCATA—Direction des Services, 65921 Tarbes Cedex 9, France; telephone: +33 (0)5 62 41 73 00; fax: +33 (0)5 62 41 7654; or in the United States contact SOCATA North America, Inc., North Perry Airport, 7501 South Airport Road, Pembroke Pines, Florida 33023; telephone: (954) 893-1400; fax: (954) 964-4141; email: mysocata@socata.daher.com; Internet: www.socatanorthamerica.com.

(3) You may review copies of the service information at the FAA, Small Airplane Directorate, 901 Locust, Kansas City, Missouri 64106. For information on the availability of this material at the FAA, call (816) 329-4148.

(4) You may also review copies of the service information that is incorporated by

reference at the National Archives and Records Administration (NARA). For information on the availability of this material at an NARA facility, call 202-741-6030, or go to http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

Issued in Kansas City, Missouri, on May 17, 2012.

Earl Lawrence,

Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2012-12649 Filed 6-15-12; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2012-0188; Directorate Identifier 2011-NM-120-AD; Amendment 39-17079; AD 2012-11-15]

RIN 2120-AA64

Airworthiness Directives; BAE SYSTEMS (Operations) Limited Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for all BAE SYSTEMS (Operations) Limited Model 4101 airplanes. This AD was prompted by reports of cracking found in the wing rear spar. This AD requires a one-time detailed inspection for cracks, corrosion, and other defects of the rear face of the wing rear spar, and repair if necessary. We are issuing this AD to detect and correct cracking in the rear spar, which could propagate to a critical length, possibly affecting the structural integrity of the area and resulting in a fuel tank rupture, with consequent damage to the airplane and possible injury to its occupants.

DATES: This AD becomes effective July 23, 2012.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD as of July 23, 2012.

ADDRESSES: You may examine the AD docket on the Internet at <http://www.regulations.gov> or in person at the U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC.

FOR FURTHER INFORMATION CONTACT: Todd Thompson, Aerospace Engineer,