

and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Operations office (phone: (800) 647-5527) is provided in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. The FAA amends § 39.13 by adding the following new AD:

2011-19-02 Dowty Propellers (formerly Dowty Aerospace; Dowty Rotol Limited; and Dowty Rotol): Amendment 39-16807. Docket No. FAA-2011-0735; Directorate Identifier 2011-NE-01-AD.

Effective Date

(a) This airworthiness directive (AD) becomes effective November 9, 2011.

Affected ADs

(b) None.

Applicability

(c) This AD applies to Dowty Propellers type R212/4-30-4/22 propeller assemblies with hub and driving center assembly part number (P/N) 601022105, 601022211, 601022294, 601021426, 601021858, or 601021859 installed, and type R251/4-30-4/49 propeller assemblies with hub and driving center assembly P/N 660207202 or P/N 660207203 installed.

Reason

(d) This AD results from mandatory continuing airworthiness information (MCAI) issued by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. We are issuing this AD to prevent propeller hub failure due to cracks in the hub, which could result in damage to the airplane.

Actions and Compliance

(e) Unless already done, do the following:

(1) Within 500 flight hours after the effective date of this AD, and thereafter at intervals not exceeding 500 flight hours, inspect the buttress threads in the propeller hub and driving center assembly for cracks.

(2) Use paragraphs 2.A.(1) through 2.A.(4)(a) of Accomplishment Instructions of Dowty Propellers Alert Service Bulletin No. 61-1043, Revision 7, dated March 1, 2011, and NDT Technique NDT 175U (Appendix A of Dowty Propellers Alert Service Bulletin No. 61-1043, Revision 7, dated March 1, 2011), to do the inspection.

(3) If a crack is found, remove the propeller assembly from service before further flight.

(4) After the effective date of this AD, do not install this propeller on any airplane unless the propeller hub and driving center has passed the inspections required by this AD.

FAA AD Differences

(f) This AD differs from the service information as follows:

(1) Although the service bulletin tells you to return the affected parts to the manufacturer, this AD does not require that action.

(2) Although the service bulletin tells you to submit information to the manufacturer, this AD does not require that action.

Alternative Methods of Compliance (AMOCs)

(g) The Manager, Boston Aircraft Certification Office, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19.

Related Information

(h) Refer to MCAI European Aviation Safety Agency AD 2011-0012, dated January 20, 2011, for related information.

(i) Contact Michael Schwetz, Aerospace Engineer, Boston Aircraft Certification Office, FAA, Engine & Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803; phone: 781-238-7761; fax: 781-238-7170, e-mail: michael.schwetz@faa.gov for more information about this AD.

Material Incorporated by Reference

(j) You must use Dowty Propellers Alert Service Bulletin No. 61-1043, Revision 7, dated March 1, 2011, to do the actions required by this AD, unless the AD specifies otherwise.

(k) The Director of the Federal Register approved the incorporation by reference of this service information under 5 U.S.C. 552(a) and 1 CFR part 51.

(l) For service information identified in this AD, contact Dowty Propellers, 114 Powers Court, Sterling, VA 20166, phone: 703-421-4434; fax: 703-450-0087.

(m) You may review copies at the FAA, New England Region, 12 New England Executive Park, Burlington, MA; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Burlington, Massachusetts, on September 7, 2011.

Peter A. White,

Manager, Engine & Propeller Directorate, Aircraft Certification Service.

[FR Doc. 2011-25653 Filed 10-4-11; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2011-0392; Directorate Identifier 2011-NE-12-AD; Amendment 39-16808; AD 2011-19-03]

RIN 2120-AA64

Airworthiness Directives; General Electric Company (GE) CT7-8, CT7-8A, CT7-8A1, CT7-8E, and CT7-8F5 Turboshaft Engines

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for the products listed above. This AD was prompted by four reports of unrecoverable engine stalls, during hover in a left-roll attitude. This AD requires the installation of an accessory gearbox (AGB) axis-A oil slinger nut to the axis-A shaft assembly. We are issuing this AD to prevent an unrecoverable engine stall, leading to a helicopter forced landing or accident.

DATES: This AD is effective November 9, 2011.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in the AD as of November 9, 2011.

ADDRESSES: For service information identified in this AD, contact GE-Aviation, M/D Rm. 285, One Neumann Way, Cincinnati, OH 45215; phone: 513-552-3272; e-mail: geaeaac@ge.com. You may review copies of the referenced service information at the FAA, Engine & Propeller Directorate, 12 New England Executive Park, Burlington, MA. For information on the availability of this material at the FAA, call 781-238-7125.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (phone: 800-647-5527) is Document Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT: Walter Meibaum, Aerospace Engineer,

Engine & Propeller Directorate, FAA, 12 New England Executive Park, Burlington, MA 01803; phone: 781-238-7119; fax: 781-238-7199; e-mail: walter.meibaum@faa.gov.

SUPPLEMENTARY INFORMATION:

Discussion

We received four reports of GE CT7-8 series turboshaft helicopter engines experiencing unrecoverable engine stalls, during hover in a left-roll attitude. Investigation revealed that during a prolonged left roll, excessive return oil from the AGB may return to the A-sump and exceed the sump's scavenging capability. The sump then floods, leading to over-heated oil, which preheats the air entering the engine's compressor. This preheated air causes inlet thermal distortion. This condition, if not corrected, could result in an unrecoverable engine stall, leading to a helicopter forced landing or accident. We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that would apply to the specified products. That NPRM published in the **Federal Register** on May 2, 2011 (76 FR 24407). That NPRM proposed to require the installation of an AGB axis-A oil slinger nut to the axis-A shaft assembly.

Comments

We gave the public the opportunity to participate in developing this AD. We received no comments on the NPRM published in the **Federal Register** on May 2, 2011 (76 FR 24407).

Since we issued the NPRM published in the **Federal Register** on May 2, 2011 (76 FR 24407), GE issued a revision to the service bulletin we are incorporating by reference in this AD. The revision includes new information in the procedure required to torque the oil slinger nut. This AD incorporates by reference, GE Aircraft Engines CT7-8 Turboshaft Engine Service Bulletin No. CT7-8 S/B 72-0033, Revision 1, dated April 28, 2011.

Also since we issued the NPRM published in the **Federal Register** on May 2, 2011 (76 FR 24407), we discovered that in the applicability paragraph, we inadvertently omitted engine serial number 953071. We corrected that omission in paragraph (c) (4) by changing "CT7-8E, engine S/Ns 953068 and below, and S/Ns 953070 and 953072" to "CT7-8E, engine S/Ns 953068 and below, and S/Ns 953070 through 953072".

Conclusion

We reviewed the relevant data and determined that air safety and the public interest require adopting the AD

as proposed except for minor editorial changes. We have determined that these minor changes are consistent with the intent that was proposed in the NPRM published in the **Federal Register** on May 2, 2011 (76 FR 24407) for correcting the unsafe condition.

Costs of Compliance

We estimate that this AD will affect 80 engines installed on helicopters of U.S. registry. We also estimate that it will take about one work-hour per engine to perform the actions required by this AD, and that the average labor rate is \$85 per work-hour. Required parts will cost about \$700 per engine. Based on these figures, we estimate the total cost of the AD to U.S. operators to be \$62,800.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII: Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, part A, subpart III, Section 44701: "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

This AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866,
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),
- (3) Will not affect intrastate aviation in Alaska, and
- (4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities

under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

2011-19-03 General Electric Company:
Amendment 39-16808; Docket No. FAA-2011-0392; Directorate Identifier 2011-NE-12-AD.

Effective Date

- (a) This AD is effective November 9, 2011.

Affected ADs

- (b) None.

Applicability

(c) This AD applies to the following General Electric Company (GE) turboshaft engines:

- (1) CT7-8, all engine serial numbers (S/Ns).
- (2) CT7-8A, engine S/Ns 947565 and below.
- (3) CT7-8A1, engine S/Ns 530017 and below.
- (4) CT7-8E, engine S/Ns 953068 and below, and S/Ns 953070 through 953072.
- (5) CT7-8F5, engine S/Ns 731005 and below, and S/Ns 731007, 731008, 817021, and 817022.

Unsafe Condition

(d) This AD was prompted by four reports of unrecoverable engine stalls, during hover in a left-roll attitude. We are issuing this AD to prevent an unrecoverable engine stall, leading to a helicopter forced landing or accident.

Compliance

(e) Comply with this AD at the next engine shop visit, the next 1,500-hour helicopter inspection, or before operation after next engine installation, whichever occurs first.

Installation of Accessory Gearbox (AGB) Axis-A Oil Slinger Nut

(f) Install the AGB axis-A oil slinger nut to the axis-A shaft assembly. Use Accomplishment Instructions, paragraphs 3.A. through 3.C. of GE Aircraft Engines CT7-8 Turboshaft Engine Service Bulletin No. CT7-8 S/B 72-0033, Revision 1, dated April 28, 2011, to do the installation.

Previous Credit

(g) An oil slinger nut installation performed before the effective date of this AD using GE Aircraft Engines CT7–8 Turboshaft Engine Service Bulletin No. CT7–8 S/B 72–0033, dated February 11, 2011, satisfies the installation requirements of this AD.

Alternative Methods of Compliance (AMOCs)

(h) The Manager, Engine Certification Office, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19.

Related Information

(i) For more information about this AD, contact Walter Meibaum, Aerospace Engineer, Engine & Propeller Directorate, FAA, 12 New England Executive Park, Burlington, MA 01803; *phone*: 781–238–7119; *fax*: 781–238–7199; *e-mail*: walter.meibaum@faa.gov.

Material Incorporated by Reference

(j) You must use the following service information to do the actions required by this AD, unless the AD specifies otherwise. The Director of the Federal Register approved the incorporation by reference (IBR) under 5 U.S.C. 552(a) and 1 CFR part 51 of the following service information on the date specified:

(1) GE Aircraft Engines CT7–8 Turboshaft Engine Service Bulletin No. CT7–8 S/B 72–0033, Revision 1, dated April 28, 2011, approved for IBR November 9, 2011.

(2) For service information identified in this AD, contact GE–Aviation, M/D Rm. 285, One Neumann Way, Cincinnati, OH 45215; *phone*: 513–552–3272; *e-mail*: geaeaac@ge.com.

(3) You may review copies of the service information at the FAA, Engine & Propeller Directorate, 12 New England Executive Park, Burlington, MA. For information on the availability of this material at the FAA, call 781–238–7125.

(4) You may also review copies of the service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at an NARA facility, call 202–741–6030, or go to http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

Issued in Burlington, Massachusetts, on September 8, 2011.

Peter A. White,

Manager, Engine & Propeller Directorate, Aircraft Certification Service.

[FR Doc. 2011–25654 Filed 10–4–11; 8:45 am]

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DEPARTMENT OF TRANSPORTATION**Federal Aviation Administration****14 CFR Part 39**

[Docket No. FAA–2010–1313; Directorate Identifier 2010–NM–158–AD; Amendment 39–16823; AD 2011–20–10]

RIN 2120–AA64

Airworthiness Directives; The Boeing Company Model 737–600, –700, –700C, –800, –900, and –900ER Series Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for the products listed above. This AD requires inspecting to determine the clearance and any wire bundle damage between wire bundle W443 and the left forward rudder quadrant, followed by adjusting the minimum clearance between the wire bundle and the left forward rudder quadrant, and repairing any wire bundle damage. This AD was prompted by reports of contact between wire bundle W443 and the left forward rudder quadrant. We are issuing this AD to detect and correct contact between the wire bundle and the left forward rudder quadrant. Damage to the wire bundle from contact between the wire bundle and the left forward rudder quadrant could result in uncommanded stabilizer trim and autopilot disconnects due to shorted wires, potentially affecting the capability of the flightcrew during high work load and consequently reducing control of the airplane. Restricted movement of the rudder quadrant at full right rudder travel would reduce controllability of the airplane.

DATES: This AD is effective November 9, 2011.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in the AD as of November 9, 2011.

ADDRESSES: For service information identified in this AD, contact Boeing Commercial Airplanes, *Attention:* Data & Services Management, P.O. Box 3707, MC 2H–65, Seattle, Washington 98124–2207; telephone 206–544–5000, extension 1; fax 206–766–5680; e-mail me.boecom@boeing.com; Internet <https://www.myboeingfleet.com>. You may review copies of the referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington. For information on the availability of

this material at the FAA, call 425–227–1221.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (*phone*: 800–647–5527) is Document Management Facility, U.S. Department of Transportation, Docket Operations, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT: Dean Thompson, Aerospace Engineer, Systems and Equipment Branch, ANM–130S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98057–3356; telephone (425) 917–6409; fax (425) 917–6590.

SUPPLEMENTARY INFORMATION:**Discussion**

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an airworthiness directive (AD) that would apply to the specified products. That NPRM was published in the **Federal Register** on January 18, 2011 (76 FR 2840). That NPRM proposed to require inspecting to determine the clearance and any wire bundle damage between wire bundle W443 and the left forward rudder quadrant, followed by adjusting the minimum clearance between the wire bundle and the left forward rudder quadrant, and repairing any wire bundle damage.

Comments

We gave the public the opportunity to participate in developing this AD. We have considered the comments received. The following presents the comments received on the proposal and the FAA's response to each comment.

Support for the NPRM

American Airlines, Delta Air Lines, and Continental Airlines (CAL) support the NPRM (76 FR 2840, January 18, 2011), and stated that they have been inspecting the affected airplanes in accordance with the original issue and revision 1 of the service information cited in the NPRM.

Request for Boeing To Add Instructions for Continued Airworthiness (ICAs)

CAL requested that the FAA ask Boeing to add the appropriate ICAs to