

AIR IMMERSION DAC—Continued

Radio-nuclide	Half-life	( $\mu\text{Ci/mL}$ )	( $\text{Bq/m}^3$ )
Xe-120 .....	40.0 min	1E-05	4E+05
Xe-121 .....	40.1 min	2E-06	8E+04
Xe-122 .....	20.1 h .....	8E-05	3E+06
Xe-123 .....	2.14 h .....	6E-06	2E+05
Xe-125 .....	16.8 h .....	1E-05	6E+05
Xe-127 .....	36.406 d	1E-05	6E+05
Xe-129m ..	8.89 d .....	2E-04	7E+06
Xe-131m ...	11.84 d ...	5E-04	1E+07
Xe-133 .....	5.245 d ...	1E-04	5E+06
Xe-133m ..	2.19 d .....	1E-04	5E+06
Xe-135 .....	9.11 h .....	1E-05	6E+05
Xe-135m ..	15.36 min	1E-05	3E+05
Xe-138 .....	14.13 min	3E-06	1E+05

\* \* \* \* \*

[FR Doc. 2011-8836 Filed 4-12-11; 8:45 am]

BILLING CODE 6450-01-P

DEPARTMENT OF THE TREASURY

Office of Thrift Supervision

12 CFR Part 563e

Community Reinvestment

CFR Correction

In Title 12 of the Code of Federal Regulations, Parts 500 to 599, revised as of January 1, 2011, on page 278, in § 563e.12, the heading of paragraph (u) and paragraph (u)(1) are corrected to read as follows:

§ 563e.12 Definitions.

\* \* \* \* \*

(u) *Small savings association*—(1) *Definition.* *Small savings association* means a savings association that, as of December 31 of either of the prior two calendar years, had assets of less than \$1.122 billion. *Intermediate small savings association* means a small savings association with assets of at least \$280 million as of December 31 of both of the prior two calendar years and less than \$1.122 billion as of December 31 of either of the prior two calendar years.

\* \* \* \* \*

[FR Doc. 2011-8795 Filed 4-12-11; 8:45 am]

BILLING CODE 1505-01-D

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 27

[Docket No. SW026; Special Conditions No. 27-026-SC]

Special Conditions: Eurocopter France Model AS350B Series, AS350D, and EC130 Helicopters, Installation of a Hoh Aeronautics, Inc. Autopilot/Stabilization Augmentation System (AP/SAS)

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final special conditions; request for comments.

**SUMMARY:** These special conditions are issued for the modification of the Eurocopter France (Eurocopter) model AS350B series, AS350D, and EC130 helicopters. These model helicopters will have novel or unusual design features when modified by installing the Hoh Aeronautics, Inc. (Hoh) complex autopilot/stabilization augmentation system (AP/SAS) that has potential failure conditions with more severe adverse consequences than those envisioned by the existing applicable airworthiness regulations. These special conditions contain the added safety standards the Administrator considers necessary to ensure the failures and their effects are sufficiently analyzed and contained.

**DATES:** The effective date of these special conditions is March 31, 2011. We must receive your comments by June 13, 2011.

**ADDRESSES:** You may send your comments by e-mail to: [john.vanhoudt@faa.gov](mailto:john.vanhoudt@faa.gov); by mail to: Federal Aviation Administration, Rotorcraft Directorate, Attn: John VanHoudt (ASW-111), Special Conditions Docket No. SW026, 2601 Meacham Blvd., Fort Worth, Texas 76137; or by delivering your comments to the Rotorcraft Directorate at the indicated address. You must mark your comments: Docket No. SW026. You can inspect comments in the special conditions docket on weekdays, except Federal holidays, between 8:30 a.m. and 4 p.m., in the Rotorcraft Directorate.

**FOR FURTHER INFORMATION CONTACT:** John VanHoudt, Aviation Safety Engineer, FAA, Rotorcraft Directorate, Regulations and Policy Group (ASW-111), 2601 Meacham Blvd., Fort Worth, Texas 76137; telephone (817) 222-5167; facsimile (817) 222-5961; or e-mail to [john.vanhoudt@faa.gov](mailto:john.vanhoudt@faa.gov).

SUPPLEMENTARY INFORMATION:

Reason for No Prior Notice and Comment Before Adoption

The substance of these special conditions has been subjected to the notice and comment period previously and has been derived without substantive change from those previously issued. It is unlikely that prior public comment would result in a significant change from the substance contained herein. Further, a delay in the effective date of these special conditions would significantly delay issuance of the design approval and thus delivery of the helicopter, which is imminent. Therefore, the FAA has determined that prior public notice and comment are unnecessary, impracticable, and contrary to the public interest, and finds good cause exists for adopting these special conditions upon issuance. The FAA is requesting comments to allow interested persons to submit views that may not have been submitted in response to the prior opportunities for comment.

Comments Invited

While we did not precede this with a notice of proposed special conditions, we invite interested people to take part in this action by sending written comments, data, or views. The most helpful comments reference a specific portion of the special conditions, explain the reason for any recommended change, and include supporting data.

We will file in the special conditions docket all comments we receive, as well as a report summarizing each substantive public contact with FAA personnel about these special conditions. You can inspect the docket before and after the comment closing date. If you wish to review the docket in person, go to the address in the ADDRESSES section of this document between 8:30 a.m. and 4 p.m., Monday through Friday, except Federal holidays.

We will consider all comments we receive by the closing date for comments. We will consider comments filed late if it is possible to do so without incurring expense or delay. We may change these special conditions based on the comments we receive.

If you want us to let you know we received your mailed comments on these special conditions, send us a pre-addressed, stamped postcard on which the docket number appears. We will stamp the date on the postcard and mail it back to you.

Background

On February 5, 2010, Hoh submitted an application to the FAA's Los Angeles

Aircraft Certification Office (LA ACO) for a supplemental type certificate (STC) to install an AP/SAS on the Eurocopter model AS350B, AS350BA, AS350B1, AS350B2, AS350B3 (AS350B series), AS350D, and EC130 helicopters. The Eurocopter model AS350B series, AS350D, and EC130 helicopters are 14 CFR part 27 Normal category, single turbine engine, conventional helicopters designed for civil operation. These helicopter models are capable of carrying up to six passengers with one pilot, and have a maximum gross weight of approximately 5,290 pounds, depending on the model configuration. The major design features include a 3-blade, fully articulated main rotor, an anti-torque tail rotor system, a skid landing gear, and a visual flight rule (VFR) basic avionics configuration. Hoh proposes to modify these model helicopters by installing a two-axis AP/SAS.

#### Type Certification Basis

Under 14 CFR 21.115, Hoh must show that the Eurocopter model AS350B series, AS350D, and EC130 helicopters, as modified by the installed AP/SAS, continue to meet the 14 CFR 21.101 standards. The baseline of the certification basis for the unmodified Eurocopter model AS350B series, AS350D, and EC130 helicopters is listed in Type Certificate Number H9EU. Additionally, compliance must be shown to any applicable equivalent level of safety findings, exemptions, and special conditions, prescribed by the Administrator as part of the certification basis.

If the Administrator finds the applicable airworthiness regulations (that is, 14 CFR part 27), as they pertain to this STC, do not contain adequate or appropriate safety standards for the Eurocopter model AS350B series, AS350D, and EC130 helicopters because of a novel or unusual design feature, special conditions are prescribed under § 21.101(d).

In addition to the applicable airworthiness regulations and special conditions, Hoh must show compliance of the AP/SAS STC-altered Eurocopter model AS350B series, AS350D, and EC130 helicopters with the noise certification requirements of 14 CFR part 36.

The FAA issues special conditions, as defined in § 11.19, under § 11.38 and they become part of the type certification basis under § 21.101.

#### Novel or Unusual Design Features

The Hoh AP/SAS incorporates novel or unusual design features, for installation in a Eurocopter model

AS350B series, AS350D, and EC130 helicopter, Type Certificate Number H9EU. This AP/SAS performs non-critical control functions, since this model helicopter has been certificated to meet the applicable requirements independent of this system. However, the possible failure conditions for this system, and their effect on the continued safe flight and landing of the helicopters, are more severe than those envisioned by the present rules.

#### Discussion

The effect on safety is not adequately covered under § 27.1309 for the application of new technology and new application of standard technology. Specifically, the present provisions of § 27.1309(c) do not adequately address the safety requirements for systems whose failures could result in catastrophic or hazardous/severe-major failure conditions, or for complex systems whose failures could result in major failure conditions.

To comply with the provisions of the special conditions, we require that Hoh provide the FAA with a systems safety assessment (SSA) for the final AP/SAS installation configuration that will adequately address the safety objectives established by a functional hazard assessment (FHA) and a preliminary system safety assessment (PSSA), including the fault tree analysis (FTA). This will ensure that all failure conditions and their resulting effects are adequately addressed for the installed AP/SAS. The SSA process, FHA, PSSA, and FTA are all parts of the overall safety assessment (SA) process discussed in FAA Advisory Circular (AC) 27-1B (Certification of Normal Category Rotorcraft) and Society of Automotive Engineers (SAE) document Aerospace Recommended Practice (ARP) 4761 (Guidelines and Methods for Conducting the Safety Assessment Process on civil airborne Systems and Equipment).

These special conditions require that the AP/SAS installed on a Eurocopter model AS350B series, AS350D, or EC130 helicopter meet the requirements to adequately address the failure effects identified by the FHA, and subsequently verified by the SSA, within the defined design integrity requirements.

#### Applicability

These special conditions are applicable to the Hoh AP/SAS installed as an STC approval, in Eurocopter model AS350B, AS350BA, AS350B1, AS350B2, AS350B3, AS350D, and EC130 helicopters, Type Certificate Number H9EU.

#### Conclusion

This action affects only certain novel or unusual design features for a Hoh AP/SAS STC installed on the specified model series of helicopters. It is not a rule of general applicability and affects only the applicant who applied to the FAA for approval of these features on the model helicopters listed in the "Applicability" section.

#### List of Subjects in 14 CFR Part 27

Aircraft, Aviation safety.

The authority citation for these special conditions is as follows:

**Authority:** 42 U.S.C. 7572, 49 U.S.C. 106(g), 40105, 40113, 44701-44702, 44704, 44709, 44711, 44713, 44715, 45303.

#### The Special Conditions

Accordingly, pursuant to the authority delegated to me by the Administrator, the following special conditions are issued as part of the Hoh Aeronautics, Inc. (Hoh) supplemental type certificate basis for the installation of an autopilot/stabilization augmentation system (AP/SAS) on the Eurocopter model AS350B, AS350BA, AS350B1, AS350B2, AS350B3 (AS350B series), AS350D, and EC130 helicopters, Type Certificate Number H9EU.

The AP/SAS must be designed and installed so that the failure conditions identified in the functional hazard assessment (FHA) and verified by the system safety assessment (SSA), after design completion, are adequately addressed in accordance with the "failure condition categories" and "requirements" sections (including the system design integrity, system design environmental, and test and analysis requirements) of these special conditions.

##### I. Failure Condition Categories

Failure conditions are classified, according to the severity of their effects on the rotorcraft, into one of the following categories:

1. *No Effect*—Failure conditions that would have no effect on safety; for example, failure conditions that would not affect the operational capability of the rotorcraft or increase crew workload; however, could result in an inconvenience to the occupants, excluding the flight crew.

2. *Minor*—Failure conditions which would not significantly reduce rotorcraft safety, and which would involve crew actions that are well within their capabilities. Minor failure conditions would include, for example, a slight reduction in safety margins or functional capabilities, a slight increase in crew workload, such as, routine flight

plan changes, or result in some physical discomfort to occupants.

3. *Major*—Failure conditions which would reduce the capability of the rotorcraft or the ability of the crew to cope with adverse operating conditions to the extent that there would be, for example, a significant reduction in safety margins or functional capabilities, a significant increase in crew workload or result in impairing crew efficiency, physical distress to occupants, including injuries, or physical discomfort to the flight crew.

4. *Hazardous/Severe-Major*—Failure conditions which would reduce the capability of the rotorcraft or the ability of the crew to cope with adverse operating conditions to the extent that there would be:

- A large reduction in safety margins or functional capabilities;
- Physical distress or excessive workload that would impair the flight crew's ability to the extent that they could not be relied on to perform their tasks accurately or completely; or
- Possible serious or fatal injury to a passenger or a cabin crewmember, excluding the flight crew.

**Note 1:** "Hazardous/severe-major" failure conditions can include events that are manageable by the crew by the use of proper procedures, which, if not implemented correctly or in a timely manner, may result in a catastrophic event.

5. *Catastrophic*—Failure conditions which would result in multiple fatalities to occupants, fatalities or incapacitation to the flight crew, or result in loss of the rotorcraft.

The present §§ 27.1309 (b) and (c) regulations do not adequately address the safety requirements for systems whose failures could result in "catastrophic" or "hazardous/severe-major" failure conditions, or for complex systems whose failures could result in "major" failure conditions. The current regulations are inadequate because when §§ 27.1309(b) and (c) were promulgated, it was not envisioned that this type of rotorcraft would use systems that are complex or whose failure could result in "catastrophic" or "hazardous/severe-major" effects on the rotorcraft. This is particularly true with the application of new technology, new application of standard technology, or other applications not envisioned by the rule that affect safety.

Hoh must provide the FAA with a SSA for the final AP/SAS installation configuration that will adequately address the safety objectives established by the FHA and the preliminary system safety assessment (PSSA), including the

fault tree analysis (FTA). This will show that all failure conditions and their resulting effects are adequately addressed for the installed AP/SAS.

**Note 2:** The SSA process, FHA, PSSA, and FTA are all parts of the overall safety assessment (SA) process discussed in FAA Advisory Circular (AC) 27-1B (Certification of Normal Category Rotorcraft) and Society of Automotive Engineers (SAE) document Aerospace Recommended Practice (ARP) 4761 (Guidelines and Methods for Conducting the Safety Assessment Process on civil airborne Systems and Equipment).

## II. Requirements

Hoh must comply with the existing requirements of § 27.1309 for all applicable design and operational aspects of the AP/SAS with the failure condition categories of "no effect," and "minor," and for non-complex systems whose failure condition category is classified as "major." Hoh must comply with the requirements of these special conditions for all applicable design and operational aspects of the AP/SAS with the failure condition categories of "catastrophic" and "hazardous/severe/major," and for complex systems whose failure condition category is classified as "major." A complex system is a system whose operations, failure conditions, or failure effects are difficult to comprehend without the aid of analytical methods (for example, FTA, Failure Modes and Effect Analysis, FHA).

### System Design Integrity Requirements

Each of the failure condition categories defined in these special conditions relate to the corresponding aircraft system integrity requirements. The system design integrity requirements, for the Hoh AP/SAS, as they relate to the allowed probability of occurrence for each failure condition category, and the proposed software design assurance level, are as follows:

- "Major"—For systems with "major" failure conditions, failures resulting in these major effects must be shown to be remote, a probability of occurrence on the order of between  $1 \times 10^{-5}$  to  $1 \times 10^{-7}$  failures/hour, and associated software must be developed to the RTCA/DO-178B (Software Considerations in Airborne Systems And Equipment Certification) Level C software design assurance level.

- "Hazardous/Severe-Major"—For systems with "hazardous/severe-major" failure conditions, failures resulting in these hazardous/severe-major effects must be shown to be extremely remote, a probability of occurrence on the order of between  $1 \times 10^{-7}$  to  $1 \times 10^{-9}$  failures/hour, and associated software

must be developed to the RTCA/DO-178B (Software Considerations in Airborne Systems And Equipment Certification) Level B software assurance level.

- "Catastrophic"—For systems with "catastrophic" failure conditions, failures resulting in these catastrophic effects must be shown to be extremely improbable, a probability of occurrence on the order of  $1 \times 10^{-9}$  failures/hour or less, and associated software must be developed to the RTCA/DO-178B (Software Considerations in Airborne Systems And Equipment Certification) Level A design assurance level.

### System Design Environmental Requirements

The AP/SAS system equipment must be qualified to the appropriate environmental level per RTCA document DO-160F (Environmental Conditions and Test Procedures for Airborne Equipment), for all relevant aspects. This is to show that the AP/SAS system performs its intended function under any foreseeable operating condition, which includes the expected environment in which the AP/SAS is intended to operate. Some of the main considerations for environmental concerns are installation locations and the resulting exposure to environmental conditions for the AP/SAS system equipment, including considerations for other equipment that may be affected environmentally by the AP/SAS equipment installation. The level of environmental qualification must be related to the severity of the considered failure conditions and effects on the rotorcraft.

### Test and Analysis Requirements

Compliance with the requirements of these special conditions may be shown by a variety of methods, which typically consist of analysis, flight tests, ground tests, and simulation, as a minimum. Compliance methodology is related to the associated failure condition category. If the AP/SAS is a complex system, compliance with the requirements for failure conditions classified as "major" may be shown by analysis, in combination with appropriate testing to validate the analysis. Compliance with the requirements for failure conditions classified as "hazardous/severe-major" may be shown by flight-testing in combination with analysis and simulation, and the appropriate testing to validate the analysis. Flight tests may be limited for "hazardous/severe-major" failure conditions and effects due to safety considerations. Compliance with the requirements for failure conditions

classified as “catastrophic” may be shown by analysis, and appropriate testing in combination with simulation to validate the analysis. Very limited flight tests in combination with simulation are used as a part of a showing of compliance for “catastrophic” failure conditions. Flight tests are performed only in circumstances that use operational variations, or extrapolations from other flight performance aspects to address flight safety.

These special conditions require that the Hoh AP/SAS system installed on a Eurocopter model AS350B, AS350BA, AS350B1, AS350B2, AS350B3, AS350D, and EC130 helicopter, Type Certificate Number H9EU, meet these requirements to adequately address the failure effects identified by the FHA, and subsequently verified by the SSA, within the defined design system integrity requirements.

Issued in Fort Worth, Texas, on March 31, 2011.

**Scott A. Horn,**

*Acting Manager, Rotorcraft Directorate, Aircraft Certification Service.*

[FR Doc. 2011–8294 Filed 4–12–11; 8:45 am]

**BILLING CODE 4910–13–P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA–2011–0262; Directorate Identifier 2010–NM–215–AD; Amendment 39–16649; AD 2011–07–12]

RIN 2120–AA64

#### **Airworthiness Directives; Fokker Services B.V. Model F.27 Mark 050 Airplanes**

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Final rule; request for comments.

**SUMMARY:** We are adopting a new airworthiness directive (AD) for the products listed above. This AD results from mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as:

\* \* \* [T]he Federal Aviation Administration (FAA) has published Special Federal Aviation Regulation (SFAR) 88, and the Joint Aviation Authorities (JAA) has published Interim Policy INT/POL/25/12. The review conducted by Fokker Services on

the Fokker 50 and Fokker 60 type design, in response to these regulations, revealed that the clearance between parts of the main landing gear (MLG) and the fuel pipes may be insufficient.

This condition, if not detected and corrected, could lead to chafing, possibly resulting in fuel leakage and, in combination with other factors, a fuel fire.

This AD requires actions that are intended to address the unsafe condition described in the MCAI.

**DATES:** This AD becomes effective April 28, 2011.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in the AD as of April 28, 2011.

We must receive comments on this AD by May 31, 2011.

**ADDRESSES:** You may send comments by any of the following methods:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.

- *Fax:* 202–493–2251.

- *Mail:* U.S. Department of Transportation, Docket Operations, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

- *Hand Delivery:* U.S. Department of Transportation, Docket Operations, M–30, West Building Ground Floor, Room W12–40, 1200 New Jersey Avenue, SE., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

#### **Examining the AD Docket**

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Operations office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Operations office (phone: 800–647–5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

**FOR FURTHER INFORMATION CONTACT:** Tom Rodriguez, Aerospace Engineer, International Branch, ANM–116, Transport Airplane Directorate, FAA, 1601 Lind Avenue, SW., Renton, Washington 98057–3356; phone: 425–227–1137; fax: 425–227–1149.

#### **SUPPLEMENTARY INFORMATION:**

##### **Discussion**

The European Aviation Safety Agency (EASA), which is the Technical Agent for the Member States of the European Community, has issued EASA Airworthiness Directive 2010–0197, dated October 1, 2010 (referred to after

this as “the MCAI”), to correct an unsafe condition for the specified products.

The MCAI states:

\* \* \* [T]he Federal Aviation Administration (FAA) has published Special Federal Aviation Regulation (SFAR) 88, and the Joint Aviation Authorities (JAA) has published Interim Policy INT/POL/25/12. The review conducted by Fokker Services on the Fokker 50 and Fokker 60 type design, in response to these regulations, revealed that the clearance between parts of the main landing gear (MLG) and the fuel pipes may be insufficient.

This condition, if not detected and corrected, could lead to chafing, possibly resulting in fuel leakage and, in combination with other factors, a fuel fire.

EASA issued AD 2010–0182 to require actions to ensure that a minimum clearance is maintained between the parts of the MLG and the fuel pipes in both nacelles.

Since that AD was issued, it was discovered that aeroplane serial numbers 20133 through 20142 were erroneously omitted in the original Fokker Service Bulletins (SB) and consequently the AD did not apply to those aeroplanes. The two SB’s (some typographical errors in part numbers were also found) have now been revised to correct this omission.

For the reasons described above, this new AD retains the requirements of AD 2010–0182, which is superseded, and expands the Applicability to add the 10 missing serial numbers.

The required actions include an inspection to determine fuel pipe part numbers, a general visual inspection to determine the clearance between certain fuel pipes and parts of the main landing gear, and replacement of certain pipes with insufficient main landing gear clearance. The required actions also include revising the maintenance program to incorporate a fuel limitation and a critical design configuration control limitation (CDCCL). You may obtain further information by examining the MCAI in the AD docket.

The FAA has examined the underlying safety issues involved in fuel tank explosions on several large transport airplanes, including the adequacy of existing regulations, the service history of airplanes subject to those regulations, and existing maintenance practices for fuel tank systems. As a result of those findings, we issued a regulation titled “Transport Airplane Fuel Tank System Design Review, Flammability Reduction and Maintenance and Inspection Requirements” (66 FR 23086, May 7, 2001). In addition to new airworthiness standards for transport airplanes and new maintenance requirements, this rule included Special Federal Aviation Regulation No. 88 (“SFAR 88,” Amendment 21–78, and subsequent Amendments 21–82 and 21–83).