

6. Occupancy of the lift must be limited to a single occupant secured in one of two possible configurations:

a. The occupant must be secured to a medical stretcher that is attached to the lift platform. The occupied stretcher must be designed to withstand the non-emergency load conditions defined in loads report SIE-327-301, revision D.

b. The occupant must be secured to a wheelchair that is attached to the lift platform.

7. Control panels must be located on both main and upper decks, connected with full duplex audio communications. On both operator control units, an emergency shut-off switch must be installed. In an emergency, this switch must immediately interrupt the main power supply to the motors. Lift operation must be stopped until the emergency shut-off switch is reset. As soon as one of the operators commands operation in a direction, the "Up" and "Down" option buttons must be disabled and the stop button enabled. Before one of the operators is able to change the lift-travel direction again, the lift must first be stopped.

8. Lift operation must require a trained operator at the main-deck control panel and a trained observer at the upper-deck control panel.

9. Sensors must be installed to detect the following conditions, and to prevent the start or continuation of lift travel if any conditions are not met:

a. Upper-deck seat, located on the left side of the aircraft and just forward of the master-bath bulkhead, is in its most forward, outboard position.

b. Upper-deck master-bedroom/lavatory port bulkhead is opened and secured.

c. Upper-deck shower door is closed and secured.

d. Upper-deck master-lavatory door is opened and secured.

e. Upper-deck floor panels are opened and configured to form the protective fencing.

f. Main-deck inboard doors are closed and secured. The doors must be lockable only from the outside of the lift. This ensures that the operator has control of this area and that nobody is located under the lift.

g. Aircraft seat-belt-fasten signs must not be illuminated.

10. Sensors must be installed to detect the following conditions during operation, and to prevent continued lift travel if any of these conditions occur:

a. Over-temperature of lift motors and/or power-frequency converter.

b. Presence of smoke at motors and in electrical-control cabinet.

c. Over-current at the lift motors.

d. Asynchronous operation of the spindles.

11. A built-in fire extinguisher must be installed in the motor and electrical-control cabinet. This fire extinguisher must be designed to discharge automatically upon the occurrence of a fire.

12. The lift must have the provision for manual operation in the event of a malfunction such as a loss of power to the lift and/or associated systems.

13. A separate battery backup system must provide lighting for the lift-control system, lift control/sensors, communication system, and lift lights for a minimum of 10 minutes in the event of loss of power to the lift and/or associated systems.

14. Lift placards must be installed near or adjacent the control panels identified in special condition 7. The placards must be stated as follows:

a. THIS LIFT IS APPROVED FOR MOVING ONLY A SINGLE OCCUPANT BETWEEN THE MAIN AND UPPER DECKS AND ONLY WHEN SECURED TO EITHER AN APPROVED MEDICAL STRETCHER OR WHEELCHAIR. NO OTHER USES OF THIS LIFT ARE APPROVED.

b. DO NOT OPERATE LIFT DURING TAXI, TAKEOFF, LANDING, OR TURBULENCE.

c. AN APPROVED MEDICAL STRETCHER OR WHEELCHAIR MUST BE PROPERLY SECURED TO THE LIFT PLATFORM BEFORE OPERATING THIS LIFT.

d. THE LIFT MUST BE STOWED FOR TAXI, TAKEOFF, AND LANDING. THE STOWED POSITION REQUIRES THE LIFT PLATFORM POSITIONED AT THE MAIN-DECK LEVEL WITH THE FLOOR PANELS CLOSED.

e. DURING MEDICAL-STRETCHER TRANSPORT, ALL PERSONNEL, MATERIEL, AND PATIENT EXTREMITIES MUST BE POSITIONED BETWEEN THE HEAD AND FOOT OF THE STRETCHER.

f. LIFT MAXIMUM CAPACITY: X LBS (X KG)

15. Lift operational-instruction placards must be installed near the control panels and must describe how to:

a. Configure the lift for operation, including ensuring that the bottom of the lift is clear of personnel and materiel before lowering the lift from the upper deck.

b. Operate the lift.

c. Stow the lift for non-operation such as during TTL and turbulence.

d. Operate the mechanical-override features in the event of a malfunction such as a loss of power to the lift and/or associated systems.

16. Training and related manuals must include:

a. Limitations and procedures for normal lift operation.

b. Backup and override procedure for evacuating the lift and returning it to TTL configuration.

17. Special conditions nos. 3, 4, and 14 must be documented in the Limitations section of the airplane flight manual.

Issued in Renton, Washington, on February 3, 2011.

**Ali Bahrami,**

*Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 2011-6618 Filed 3-21-11; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA-2010-1202; Directorate Identifier 2010-NM-167-AD; Amendment 39-16637; AD 2011-06-12]

RIN 2120-AA64

#### Airworthiness Directives; The Boeing Company Model MD-90-30 Airplanes

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Final rule.

**SUMMARY:** We are adopting a new airworthiness directive (AD) for the products listed above. This AD requires repetitive inspections for cracking of the left and right upper center skin panels of the horizontal stabilizer, and corrective action if necessary. This AD was prompted by a report of a crack found in the upper skin panel at the aft inboard corner of a right horizontal stabilizer. We are issuing this AD to detect and correct cracks in the upper center skin panels of the horizontal stabilizer. Uncorrected cracks might ultimately lead to the loss of overall structural integrity of the horizontal stabilizer.

**DATES:** This AD is effective April 26, 2011.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in the AD as of April 26, 2011.

**ADDRESSES:** For service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, 3855 Lakewood Boulevard, MC D800-0019, Long Beach, California 90846-0001; phone: 206-544-5000, extension 2; fax: 206-766-5683; e-mail: [dse.boecom@boeing.com](mailto:dse.boecom@boeing.com); Internet: <https://www.myboeingfleet.com>. You

may review copies of the referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington. For information on the availability of this material at the FAA, call 425-227-1221.

**Examining the AD Docket**

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (phone: 800-647-5527) is Document Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

**FOR FURTHER INFORMATION CONTACT:** Roger Durbin, Aerospace Engineer, Los Angeles ACO, Airframe Branch, ANM-120L, FAA, 3960 Paramount Boulevard,

Lakewood, CA 90712-4137; phone: 562-627-5233; fax: 562-627-5210; e-mail: [Roger.Durbin@faa.gov](mailto:Roger.Durbin@faa.gov).

**SUPPLEMENTARY INFORMATION:**

**Discussion**

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an airworthiness directive (AD) that would apply to the specified products. That NPRM published in the **Federal Register** on December 23, 2010 (75 FR 80742). That NPRM proposed to require repetitive inspections for cracking of the left and right upper center skin panels of the horizontal stabilizer, and corrective action if necessary.

**Comments**

We gave the public the opportunity to participate in developing this AD. We received no comments on the NPRM or on the determination of the cost to the public.

**Explanation of Change to Applicability**

We have revised the applicability of the existing AD to identify model

designations as published in the most recent type certificate data sheet for the affected models.

**Conclusion**

We reviewed the relevant data, and determined that air safety and the public interest require adopting the AD with the change described previously. We have determined that these minor changes:

- Are consistent with the intent that was proposed in the NPRM for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the NPRM.

We also determined that these changes will not increase the economic burden on any operator or increase the scope of the AD.

**Costs of Compliance**

We estimate that this AD affects 19 airplanes of U.S. registry.

We estimate the following costs to comply with this AD:

**ESTIMATED COSTS**

| Action           | Labor cost   | Parts cost | Cost per product                 | Cost on U.S. operators        |
|------------------|--|------------|----------------------------------|-------------------------------|
| Inspection ..... | 4 work-hours × \$85 per hour = \$340 per inspection cycle. | \$0        | \$340 per inspection cycle ..... | \$6,460 per inspection cycle. |

We estimate the following costs to do any necessary repairs that would be

required based on the results of the inspection. We have no way of

determining the number of aircraft that might need these repairs.

**ON-CONDITION COSTS**

| Action                       | Labor cost                                      | Parts cost | Cost per product |
|------------------------------|---|------------|------------------|
| Skin panel replacement ..... | 648 work-hours × \$85 per hour = \$55,080 ..... | \$55,608   | \$110,688        |

**Authority for This Rulemaking**

Title 49 of the United States Code specifies the FAA’s authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII: Aviation Programs, describes in more detail the scope of the Agency’s authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701: “General requirements.” Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority

because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

**Regulatory Findings**

This AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

*For the reasons discussed above, I certify that this AD:*

- (1) Is not a “significant regulatory action” under Executive Order 12866,

(2) Is not a “significant rule” under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),

(3) Will not affect intrastate aviation in Alaska, and

(4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

**List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

**Adoption of the Amendment**

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

## PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

### § 39.13 [Amended]

■ 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

#### 2011-06-12 The Boeing Company:

Amendment 39-16637; Docket No. FAA-2010-1202; Directorate Identifier 2010-NM-167-AD.

#### Effective Date

(a) This AD is effective April 26, 2011.

#### Affected ADs

(b) None.

#### Applicability

(c) This AD applies to all The Boeing Company Model MD-90-30 airplanes, certificated in any category.

#### Subject

(d) Joint Aircraft System Component (JASC)/Air Transport Association (ATA) of America Code 55, Stabilizers.

#### Unsafe Condition

(e) This AD was prompted by a report of a crack found in the upper center skin panel at the aft inboard corner of a right horizontal stabilizer. We are issuing this AD to detect and correct cracks in the upper center skin panel of the horizontal stabilizer. Uncorrected cracks might ultimately lead to the loss of overall structural integrity of the horizontal stabilizer.

#### Compliance

(f) Comply with this AD within the compliance times specified, unless already done.

#### Inspections

(g) Before the accumulation of 20,000 total flight cycles, or within 3,778 flight cycles after the effective date of this AD, whichever occurs later, do eddy current inspections to detect cracking of the left and right upper center skin panels of the horizontal stabilizer, in accordance with the Accomplishment Instructions of Boeing Alert Service Bulletin MD90-55A015, dated July 16, 2010.

(1) If no crack is found during any inspection required by paragraph (g) of this AD, repeat the applicable inspections thereafter at the applicable times specified in paragraph 1.E., "Compliance," of Boeing Alert Service Bulletin MD90-55A015, dated July 16, 2010.

(2) If any crack is found during any inspection required by paragraph (g) of this AD, before further flight, replace the skin panel with a serviceable skin panel, in accordance with the Accomplishment Instructions of Boeing Alert Service Bulletin MD90-55A015, dated July 16, 2010. Within 20,000 flight cycles after the replacement, do eddy current inspections as required by paragraph (g) of this AD.

#### Alternative Methods of Compliance (AMOCs)

(h)(1) The Manager, Los Angeles Aircraft Certification Office (ACO), FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the ACO, send it to the attention of the person identified in the Related Information section of this AD.

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(3) An AMOC that provides an acceptable level of safety may be used for any repair required by this AD if it is approved by the Boeing Commercial Airplanes Organization Designation Authorization (ODA) that has been authorized by the Manager, Los Angeles ACO, to make those findings. For a repair method to be approved, the repair must meet the certification basis of the airplane, and 14 CFR 25.571, Amendment 45, and the approval must specifically refer to this AD.

#### Related Information

(i) For more information about this AD, contact Roger Durbin, Aerospace Engineer, Los Angeles ACO, Airframe Branch, ANM-120L, FAA, 3960 Paramount Boulevard, Lakewood, CA 90712-4137; *phone:* 562-627-5233; *fax:* 562-627-5210; *e-mail:* [Roger.Durbin@faa.gov](mailto:Roger.Durbin@faa.gov).

#### Material Incorporated by Reference

(j) You must use Boeing Alert Service Bulletin MD90-55A015, dated July 16, 2010, to do the actions required by this AD, unless the AD specifies otherwise.

(1) The Director of the Federal Register approved the incorporation by reference of Boeing Alert Service Bulletin MD90-55A015, dated July 16, 2010, under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) For service information identified in this AD, contact Boeing Commercial Airplanes, *Attention:* Data & Services Management, 3855 Lakewood Boulevard, MC D800-0019, Long Beach, California 90846-0001; *phone:* 206-544-5000, extension 2; *fax:* 206-766-5683; *e-mail:* [dse.boecom@boeing.com](mailto:dse.boecom@boeing.com); *Internet:* <https://www.myboeingfleet.com>.

(3) You may review copies of the service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington. For information on the availability of this material at the FAA, call 425-227-1221.

(4) You may also review copies of the service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at an NARA facility, call 202-741-6030, or go to [http://www.archives.gov/federal\\_register/code\\_of\\_federal\\_regulations/ibr\\_locations.html](http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html).

Issued in Renton, Washington, on March 9, 2011.

**Kalene C. Yanamura,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 2011-6249 Filed 3-21-11; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA-2011-0212; Directorate Identifier 2010-SW-055-AD; Amendment 39-16632; AD 2011-06-07]

RIN 2120-AA64

#### Airworthiness Directives; Eurocopter France (Eurocopter) Model EC130 B4 Helicopters

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule; request for comments.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD) for Eurocopter Model EC130 B4 helicopters. This action requires identifying and inspecting a certain emergency flotation gear unit "1G" (1G unit). This action also requires modification of certain affected 1G units. This amendment is prompted by an uncommanded in-flight deployment of the emergency flotation gear when it was not armed by the crew. The actions specified in this AD are intended to prevent an uncommanded in-flight deployment of the emergency flotation gear, unexpected deceleration and pitch down movement of the helicopter, and subsequent loss of control of the helicopter.

**DATES:** Effective April 6, 2011.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of April 6, 2011.

Comments for inclusion in the Rules Docket must be received on or before May 23, 2011.

**ADDRESSES:** Use one of the following addresses to submit comments on this AD:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.

- *Fax:* 202-493-2251.

- *Mail:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

- *Hand Delivery:* U.S. Department of Transportation, Docket Operations, M-