

Alert Service Bulletin No. 427-09-29, REV A, dated November 17, 2009.

(2) Applicable to all SNs: As of April 14, 2011 (the effective date of this AD), you may not install replacement tail rotor driveshaft hanger bracket, P/N 427-044-223-101, unless the bracket has been inspected and found free of cracks and has been reworked following Bell Helicopter Alert Service Bulletin No. 427-09-29, REV A, dated November 17, 2009.

FAA AD Differences

Note: This AD differs from the MCAI and/or service information as follows: No differences.

Other FAA AD Provisions

(g) The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs):* The Manager, Standards Office, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. Send information to ATTN: Sharon Miles, Aerospace Engineer, FAA, Rotorcraft Directorate, 2601 Meacham Blvd., Fort Worth, Texas 76137; telephone: (817) 222-5122; fax: (817) 222-5961. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

(2) *Airworthy Product:* For any requirement in this AD to obtain corrective actions from a manufacturer or other source, use these actions if they are FAA-approved. Corrective actions are considered FAA-approved if they are approved by the State of Design Authority (or their delegated agent). You are required to assure the product is airworthy before it is returned to service.

(3) *Reporting Requirements:* For any reporting requirement in this AD, a federal agency may not conduct or sponsor, and a person is not required to respond to, nor shall a person be subject to a penalty for failure to comply with a collection of information subject to the requirements of the Paperwork Reduction Act unless that collection of information displays a current valid OMB Control Number. The OMB Control Number for this information collection is 2120-0056. Public reporting for this collection of information is estimated to be approximately 5 minutes per response, including the time for reviewing instructions, completing and reviewing the collection of information. All responses to this collection of information are mandatory. Comments concerning the accuracy of this burden and suggestions for reducing the burden should be directed to the FAA at: 800 Independence Ave., SW., Washington, DC 20591, Attn: Information Collection Clearance Officer, AES-200.

Related Information

(h) Refer to MCAI Transport Canada AD No. CF-2010-17, dated June 2, 2010; and Bell Helicopter Alert Service Bulletin No. 427-09-29, REV A, dated November 17, 2009, for related information.

Material Incorporated by Reference

(i) You must use Bell Helicopter Alert Service Bulletin No. 427-09-29, REV A, dated November 17, 2009, to do the actions required by this AD, unless the AD specifies otherwise.

(1) The Director of the Federal Register approved the incorporation by reference of this service information under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) For service information identified in this AD, contact Bell Helicopter Textron, Inc., P.O. Box 482, Fort Worth, TX 76101; telephone: (817) 280-2011; fax: (817) 280-2321; or at <http://www.bellhelicopter.com>.

(3) You may review copies of the referenced service information at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas 76137. For information on the availability of this material at the FAA, call 816-329-4148.

(4) You may also review copies of the service information incorporated by reference for this AD at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call (202) 741-6030, or go to: http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

Issued in Fort Worth, Texas, on January 13, 2011.

Kim Smith,

Manager, Rotorcraft Directorate, Aircraft Certification Service.

[FR Doc. 2011-4468 Filed 3-9-11; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2010-0891; Directorate Identifier 2009-SW-055-AD; Amendment 39-16585, AD 2011-03-02]

RIN 2120-AA64

Airworthiness Directives; EUROCOPTER FRANCE Model SA330F, SA330G, and SA330J Helicopters

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for the products listed above. This AD results from mandatory continuing airworthiness information (MCAI) issued by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as:

While adjusting the position of the pedal unit on a SA 330 helicopter, the copilot set the position beyond the end limit ("tall pilot" position). This resulted in the separation of the pedal adjustment system and the pedals rocking forward.

After investigation, it was determined that the Loctite bond on the "tall pilot" stop nut was damaged, most likely due to aging of the adhesive. The nut came loose and could no longer perform its stop function. The threaded rod of the adjustment system separated from the system.

The separation of the adjustment system, if not corrected, could result in the loss of control of the pedal units, causing the helicopter to begin rotating.

We are issuing this AD to require actions to correct the unsafe condition on these products.

DATES: This AD becomes effective April 14, 2011.

On April 14, 2011, the Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD.

FOR FURTHER INFORMATION CONTACT: Gary B. Roach, Aerospace Engineer, FAA, Rotorcraft Directorate, 2601 Meacham Blvd., Fort Worth, Texas 76137; telephone: (817) 222-5130; fax: (817) 222-5961.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that would apply to the specified products. That NPRM was published in the **Federal Register** on September 13, 2010 (75 FR 55492). That NPRM proposed to correct an unsafe condition for the specified products. The MCAI states:

While adjusting the position of the pedal unit on a SA 330 helicopter, the copilot set the position beyond the end limit ("tall pilot" position). This resulted in the separation of the pedal adjustment system and the pedals rocking forward.

After investigation, it was determined that the Loctite bond on the "tall pilot" stop nut was damaged, most likely due to aging of the adhesive. The nut came loose and could no longer perform its stop function. The threaded rod of the adjustment system separated from the system.

The separation of the adjustment system, if not corrected, could result in the loss of control of the pedal units, causing the helicopter to begin rotating.

For the reasons described above, this Emergency AD requires a one-time functional test and modification (MOD 330A779820.00) of the pedal unit adjustment system.

Comments

We gave the public the opportunity to participate in developing this AD. We received no comments on the NPRM or on the determination of the cost to the public.

Conclusion

We reviewed the available data and determined that air safety and the public interest require adopting the AD as proposed.

Differences Between This AD and the MCAI or Service Information

We have reviewed the MCAI and related service information and, in general, agree with their substance. But we might have found it necessary to use different words from those in the MCAI to ensure the AD is clear for U.S. operators and is enforceable. In making these changes, we do not intend to differ substantively from the information provided in the MCAI and related service information.

We might also have required different actions in this AD from those in the MCAI in order to follow FAA policies. Any such differences are highlighted in a NOTE within the AD.

Costs of Compliance

We estimate that this AD will affect 6 products of U.S. registry. We also estimate that it will take about 3 work-hours per product to comply with the basic requirements of this AD. The average labor rate is \$85 per work-hour. Required parts will cost about \$100 per product.

Based on these figures, we estimate the cost of this AD to the U.S. operators to be \$2,130 or \$355 per product.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between

the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866;
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
- (3) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this AD and placed it in the AD Docket.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains the NPRM, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (telephone (800) 647-5527) is in the ADDRESSES section. Comments will be available in the AD docket shortly after receipt.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by adding the following new AD:

2011-03-02 EUROCOPTER FRANCE:
Amendment 39-16585; Docket No. FAA-2010-0891; Directorate Identifier 2009-SW-055-AD.

Effective Date

- (a) This airworthiness directive (AD) becomes effective April 14, 2011.

Affected ADs

- (b) None.

Applicability

(c) This AD applies to SA330F, SA330G, and SA330J helicopters, all serial numbers, certificated in any category, equipped with pedal position adjustment system modification (MOD 07.10.304).

Subject

(d) Air Transport Association of America (ATA) Code 67: Rotors Flight Control.

Reason

(e) The mandatory continuing airworthiness information (MCAI) states:

While adjusting the position of the pedal unit on a SA 330 helicopter, the copilot set the position beyond the end limit ("tall pilot" position). This resulted in the separation of the pedal adjustment system and the pedals rocking forward.

After investigation, it was determined that the Loctite bond on the "tall pilot" stop nut was damaged, most likely due to aging of the adhesive. The nut came loose and could no longer perform its stop function. The threaded rod of the adjustment system separated from the system.

The separation of the adjustment system, if not corrected, could result in the loss of control of the pedal units, causing the helicopter to begin rotating.

For the reasons described above, this Emergency AD requires a one-time functional test and modification (MOD 330A779820.00) of the pedal unit adjustment system.

Actions and Compliance

(f) Unless already done, do the following actions:

(1) Within the next 10 hours time-in-service after April 14, 2011 (the effective date of this AD), do a functional test of the pedal unit adjustment system following paragraph 2.B.1 of EUROCOPTER Emergency Alert Service Bulletin No. 67.18, dated August 3, 2009.

(2) If any non-conformity is found, before further flight, modify the pedal unit adjustment system following paragraphs 2.B.2, 2.B.3 or 2.B.4, and 2.B.5 of EUROCOPTER Emergency Alert Service Bulletin No. 67.18, dated August 3, 2009 (MOD 330A779820.00).

(3) If any non-conformity is not found, within 3 months after April 14, 2011 (the effective date of this AD), modify the pedal unit adjustment system following paragraphs 2.B.2, 2.B.3, and 2.B.5 of the EUROCOPTER Emergency Alert Service Bulletin No. 67.18, dated August 3, 2009 (MOD 330A779820.00).

(4) If half-bushings are not available when complying with paragraph (f)(2) or (f)(3) of this AD, flights are authorized without half-bushings for up to 12 months after April 14, 2011 (the effective date of this AD).

(5) After 3 months after April 14, 2011 (the effective date of this AD), do not install a pedal position adjustment system, unless it has been modified (MOD 330A779820.00) in accordance with the requirements of this AD.

FAA AD Differences

Note: This AD differs from the MCAI and/or service information as follows: No differences.

Other FAA AD Provisions

(g) The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs)*: The Manager, Standards Office, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. Send information to *Attn: Gary B. Roach, Aerospace Engineer, FAA, Rotorcraft Directorate, 2601 Meacham Blvd., Fort Worth, Texas 76137; telephone: (817) 222-5130; fax: (817) 222-5961*. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

(2) *Airworthy Product*: For any requirement in this AD to obtain corrective actions from a manufacturer or other source, use these actions if they are FAA-approved. Corrective actions are considered FAA-approved if they are approved by the State of Design Authority (or their delegated agent). You are required to assure the product is airworthy before it is returned to service.

(3) *Reporting Requirements*: For any reporting requirement in this AD, under the provisions of the Paperwork Reduction Act (44 U.S.C. 3501 *et seq.*), the Office of Management and Budget (OMB) has approved the information collection requirements and has assigned OMB Control Number 2120-0056.

Related Information

(h) Refer to MCAI European Aviation Safety Agency (EASA) Emergency AD No.: 2009-0172-E, dated August 5, 2009; and, for related information.

Material Incorporated by Reference

(i) You must use EUROCOPTER Emergency Alert Service Bulletin No. 67.18, dated August 3, 2009, to do the actions required by this AD, unless the AD specifies otherwise.

(1) The Director of the Federal Register approved the incorporation by reference of this service information under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) For service information identified in this AD, contact American Eurocopter Corporation, 2701 Forum Drive, Grand Prairie, TX 75053-4005; *telephone: (800) 232-0323; fax: (972) 641-3710; or Internet: <http://www.eurocopter.com>*.

(3) You may review copies of the referenced service information at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas 76137.

(4) You may also review copies of the service information incorporated by reference for this AD at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call (202) 741-6030, or go to: http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

Issued in Fort Worth, Texas, on January 10, 2011.

Kim Smith,

Manager, Rotorcraft Directorate, Aircraft Certificate Service.

[FR Doc. 2011-4466 Filed 3-9-11; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION**Federal Aviation Administration****14 CFR Part 39**

[Docket No. FAA-2010-0781; Directorate Identifier 2007-SW-49-AD; Amendment 39-16590; AD 2011-03-06]

RIN 2120-AA64

Airworthiness Directives; Eurocopter France Model AS-365N2, AS 365 N3, and SA-365N1 Helicopters

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD) for the specified Eurocopter France (Eurocopter) model helicopters. This AD requires replacing the aluminum tail rotor (T/R) blade pitch control shaft with a steel T/R blade pitch control shaft. This AD is prompted by an incident involving a Eurocopter Model AS-365N2 helicopter on which there was a loss of control of the T/R due to a broken shaft. The actions specified by this AD are intended to prevent failure of the T/R blade pitch control shaft, loss of T/R control, and subsequent loss of control of the helicopter.

DATES: Effective April 14, 2011.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of April 14, 2011.

ADDRESSES: You may get the service information identified in this AD from American Eurocopter Corporation, 2701 Forum Drive, Grand Prairie, Texas 75053-4005, telephone (800) 232-0323, fax (972) 641-3710, or at <http://www.eurocopter.com>.

Examining the Docket: You may examine the docket that contains this AD, any comments, and other information on the Internet at <http://www.regulations.gov> or at the Docket Operations office, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC.

FOR FURTHER INFORMATION CONTACT: Jim Grigg, Aviation Safety Engineer, Safety Management Group, Rotorcraft

Directorate, FAA, 2601 Meacham Blvd., Fort Worth, Texas 76137, telephone (817) 222-5126, fax (817) 222-5961.

SUPPLEMENTARY INFORMATION:**Discussion**

On August 2, 2010 we issued a Notice of Proposed Rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that would apply to the Eurocopter Model AS-365N2, AS 365 N3, and SA-365N1 helicopters, all serial numbers, with an aluminum T/R blade pitch control shaft, part number (P/N) 365A33.6161.20 or P/N 365A33.6161.21. That NPRM was published in the **Federal Register** on August 11, 2010 (75 FR 48618) and proposed to require replacing the aluminum T/R blade pitch control shaft with a steel T/R blade pitch control shaft. The actions specified by the NPRM are intended to prevent failure of the T/R blade pitch control shaft, loss of T/R control, and subsequent loss of control of the helicopter.

The European Aviation Safety Agency (EASA), which is the Technical Agent for the Member States of the European Community, has issued EASA AD No. 2007-0220, dated August 13, 2007, to correct an unsafe condition for the Eurocopter Model AS-365N2, AS 365 N3, and SA-365N1 helicopters, all serial numbers, equipped with an aluminum T/R blade pitch control shaft, P/N 365A33.6161.20 or P/N 365A33.6161.21. EASA advises of an incident in which the pilot of a Model AS 365 N2 helicopter encountered a loss of control of the T/R, but executed an uneventful run-on landing. A subsequent investigation revealed that the T/R blade pitch control shaft, P/N 365A33.6161.21, had broken in the main section of the shaft sliding area, which appeared to be damaged by peening. The origin of the crack, which developed under fatigue loading, could not be determined. However, accidental damage (i.e., shock impact), is believed to have caused the initiation of a crack.

Related Service Information

Eurocopter has issued Alert Service Bulletin No. 01.00.59, dated June 21, 2007 (ASB), which specifies removing any T/R blade pitch control shaft, P/N 365A33.6161.20 or P/N 365A33.6161.21, and replacing it with a steel T/R blade pitch control shaft, P/N 365A33.6214.20. EASA classified this ASB as mandatory and issued EASA AD No. 2007-0220, dated August 13, 2007, to ensure the continued airworthiness of these helicopters.