

**DEPARTMENT OF TRANSPORTATION****Federal Aviation Administration****14 CFR Part 25**

[Docket No. NM442 Special Conditions No. 25-11-02-SC]

**Special Conditions: Gulfstream Model GVI Airplane; Interaction of Systems and Structures**

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Notice of proposed special conditions.

**SUMMARY:** This action proposes special conditions for the Gulfstream GVI airplane. This airplane will have novel or unusual design features when compared to the state of technology envisioned in the airworthiness standards for transport category airplanes. These design features include systems that affect the structural capability of the airplane. The applicable airworthiness regulations do not contain adequate or appropriate safety standards for these design features. The proposed special conditions contain the additional safety standards that the Administrator considers necessary to establish a level of safety equivalent to that established by the existing airworthiness standards.

**DATES:** We must receive your comments by March 31, 2011.

**ADDRESSES:** You must mail two copies of your comments to: Federal Aviation Administration, Transport Airplane Directorate, Attn: Rules Docket (ANM-113), Docket No. NM442, 1601 Lind Avenue, SW., Renton, Washington 98057-3356. You may deliver two copies to the Transport Airplane Directorate at the above address. You must mark your comments: Docket No. NM442. You can inspect comments in the Rules Docket weekdays, except Federal holidays, between 7:30 a.m. and 4 p.m.

**FOR FURTHER INFORMATION CONTACT:** Carl Niedermeyer, FAA, Airframe/Cabin Safety Branch, ANM-115, Transport Standards Staff, Transport Airplane Directorate, Aircraft Certification Service, 1601 Lind Avenue, SW., Renton, Washington 98057-3356; telephone (425) 227-2279; facsimile (425) 227-1320.

**SUPPLEMENTARY INFORMATION:**

**Comments Invited**

We invite interested people to take part in this rulemaking by sending written comments, data, or views. The most helpful comments reference a

specific portion of the special conditions, explain the reason for any recommended change, and include supporting data. We ask that you send us two copies of written comments.

We will file in the docket all comments we receive, as well as a report summarizing each substantive public contact with FAA personnel concerning these special conditions. You can inspect the docket before and after the comment closing date. If you wish to review the docket in person, go to the address in the **ADDRESSES** section of this preamble between 7:30 a.m. and 4 p.m., Monday through Friday, except Federal holidays.

We will consider all comments we receive on or before the closing date for comments. We will consider comments filed late if it is possible to do so without incurring expense or delay. We may change these special conditions based on the comments we receive.

If you want us to acknowledge receipt of your comments on this proposal, include with your comments a self-addressed, stamped postcard on which you have written the docket number. We will stamp the date on the postcard and mail it back to you.

**Background**

On March 29, 2005, Gulfstream Aerospace Corporation (hereafter referred to as "Gulfstream") applied for an FAA type certificate for its new Gulfstream Model GVI passenger airplane. Gulfstream later applied for, and was granted, an extension of time for the type certificate, which changed the effective application date to September 28, 2006. The Gulfstream Model GVI airplane will be an all-new, two-engine jet transport airplane with an executive cabin interior. The maximum takeoff weight will be 99,600 pounds, with a maximum passenger count of 19 passengers.

**Type Certification Basis**

Under provisions of Title 14, Code of Federal Regulations (14 CFR) 21.17, Gulfstream must show that the Gulfstream Model GVI airplane (hereafter referred to as "the GVI") meets the applicable provisions of 14 CFR part 25, as amended by Amendments 25-1 through 25-119, 25-122, and 25-124. If the Administrator finds that the applicable airworthiness regulations (*i.e.*, 14 CFR part 25) do not contain adequate or appropriate safety standards for the GVI because of a novel or unusual design feature, special conditions are prescribed under the provisions of § 21.16.

In addition to complying with the applicable airworthiness regulations

and special conditions, the GVI must comply with the fuel vent and exhaust emission requirements of 14 CFR part 34 and the noise certification requirements of 14 CFR part 36. The FAA must also issue a finding of regulatory adequacy pursuant to section 611 of Public Law 92-574, the "Noise Control Act of 1972."

The FAA issues special conditions, as defined in 14 CFR 11.19, in accordance with § 11.38, and they become part of the type certification basis under § 21.17(a)(2).

Special conditions are initially applicable to the model for which they are issued. Should the type certificate for that model be amended later to include any other model that incorporates the same novel or unusual design features, the special conditions would also apply to the other model under provisions of § 21.101.

**Novel or Unusual Design Features**

The Model GVI airplane will incorporate novel or unusual design features. These features are systems that may affect the airplane's structural performance, either directly or as a result of failure or malfunction. That is, the airplane's systems affect how it responds in maneuver and gust conditions, and thereby affect its structural capability. These systems may also affect the aeroelastic stability of the airplane. These systems include the GVI's flight control systems, autopilots, stability augmentation systems, load alleviation systems, and fuel management systems. Such systems represent a novel and unusual feature when compared to the technology envisioned in the current airworthiness standards.

**Discussion of Proposed Special Conditions**

Special conditions are needed to require consideration of the effects of systems on the structural capability and aeroelastic stability of the airplane, both in the normal and in the failed state, because these effects are not covered by current regulations.

These proposed special conditions are identical or nearly identical to those previously required for type certification of other transport airplane models. These proposed special conditions were derived initially from standardized requirements developed by the Aviation Rulemaking Advisory Committee (ARAC), comprised of representatives of the FAA, Europe's Joint Aviation Authorities (now replaced by the European Aviation Safety Agency), and industry.

These proposed special conditions require that the airplane meets the structural requirements of subparts C and D of 14 CFR part 25 when the airplane systems are fully operative. These proposed special conditions also require that the airplane meet these requirements considering failure conditions. In some cases, reduced margins are allowed for failure conditions based on system reliability.

These special conditions establish a level of safety that neither raises nor lowers the standard set forth in the applicable regulations.

In these proposed special conditions and in the current standards and regulations, the term “any” is used. Use of this term has traditionally been understood to require all items covered by the term are addressed, rather than addressing only a portion of the items. The use of the term “any” in these proposed special conditions continues this traditional understanding.

#### Applicability

As discussed above, these proposed special conditions are applicable to the GVI. Should Gulfstream apply at a later date for a change to the type certificate to include another model incorporating the same novel or unusual design features, these proposed special conditions would apply to that model as well.

#### Conclusion

This action affects only certain novel or unusual design features of the GVI. It is not a rule of general applicability.

#### List of Subjects in 14 CFR Part 25

Aircraft, Aviation safety, Reporting and recordkeeping requirements.

The authority citation for this special condition is as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701, 44702, 44704.

#### The Proposed Special Conditions

Accordingly, the Federal Aviation Administration (FAA) proposes the following special conditions as part of the type certification basis for the GVI airplanes.

##### A. General

The GVI is equipped with systems that affect structural performance, either directly or as a result of a failure or malfunction. The influence of these systems and their failure conditions on structural performance must be taken into account when showing compliance with the requirements of Subparts C and D of part 25 of Title 14, Code of Federal Regulations.

1. The following criteria must be used for showing compliance with this proposed special condition for airplanes equipped with flight control systems, autopilots, stability augmentation systems, load alleviation systems, fuel management systems, and other systems that either directly or as a result of failure or malfunction affect structural performance.

2. The criteria defined herein only address the direct structural consequences of the system responses and performance. They cannot be considered in isolation but should be included in the overall safety evaluation of the airplane. These criteria may in some instances duplicate standards already established for this evaluation. These criteria are only applicable to structure whose failure could prevent continued safe flight and landing. Specific criteria that define acceptable limits on handling characteristics or stability requirements when operating in the system degraded or inoperative mode are not provided in this proposed special condition.

3. Depending upon the specific characteristics of the airplane, additional studies may be required that go beyond the criteria provided in this proposed special condition in order to demonstrate the capability of the airplane to meet other realistic conditions such as alternative gust or maneuver descriptions for an airplane equipped with a load alleviation system.

4. The following definitions are applicable to this proposed special condition.

(a) Structural performance: Capability of the airplane to meet the structural requirements of 14 CFR part 25.

(b) Flight limitations: Limitations that can be applied to the airplane flight conditions following an in-flight occurrence and that are included in the flight manual (e.g., speed limitations, avoidance of severe weather conditions, etc.).

(c) Operational limitations: Limitations, including flight limitations, that can be applied to the airplane operating conditions before dispatch (e.g., fuel, payload and master minimum equipment list limitations).

(d) Probabilistic terms: The probabilistic terms (probable, improbable, extremely improbable) used in this proposed special condition are the same as those used in § 25.1309.

(e) Failure condition: The term failure condition is the same as that used in § 25.1309; however, this proposed special condition applies only to system failure conditions that affect the

structural performance of the airplane (e.g., system failure conditions that induce loads, change the response of the airplane to inputs such as gusts or pilot actions, or lower flutter margins).

##### B. Effects of Systems on Structures

1. *General.* The following criteria will be used in determining the influence of a system and its failure conditions on the airplane structure.

2. *System fully operative.* With the system fully operative, the following apply:

(a) Limit loads must be derived in all normal operating configurations of the system from all the limit conditions specified in Subpart C (or used in lieu of those specified in Subpart C), taking into account any special behavior of such a system or associated functions or any effect on the structural performance of the airplane that may occur up to the limit loads. In particular, any significant nonlinearity (rate of displacement of control surface, thresholds or any other system nonlinearities) must be accounted for in a realistic or conservative way when deriving limit loads from limit conditions.

(b) The airplane must meet the strength requirements of part 25 (static strength, residual strength), using the specified factors to derive ultimate loads from the limit loads defined above. The effect of nonlinearities must be investigated beyond limit conditions to ensure the behavior of the system presents no anomaly compared to the behavior below limit conditions. However, conditions beyond limit conditions need not be considered when it can be shown that the airplane has design features that will not allow it to exceed those limit conditions.

(c) The airplane must meet the aeroelastic stability requirements of § 25.629.

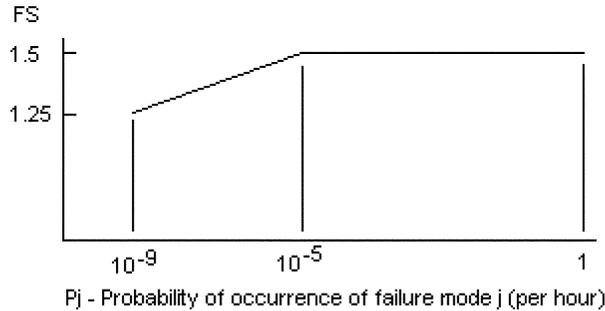
3. *System in the failure condition.* For any system failure condition not shown to be extremely improbable, the following apply:

(a) At the time of occurrence. Starting from 1-g level flight conditions, a realistic scenario, including pilot corrective actions, must be established to determine the loads occurring at the time of failure and immediately after the failure.

(1) For static strength substantiation, these loads multiplied by an appropriate factor of safety that is related to the probability of occurrence of the failure are ultimate loads to be considered for design. The factor of safety (FS) is defined in Figure 1.

Figure 1

Factor of safety at the time of occurrence



Pj - Probability of occurrence of failure mode j (per hour)

(2) For residual strength substantiation, the airplane must be able to withstand two thirds of the ultimate loads defined in subparagraph B.3(a)(1). For pressurized cabins, these loads must be combined with the normal operating differential pressure.

(3) Freedom from aeroelastic instability must be shown up to the speeds defined in § 25.629(b)(2). For failure conditions that result in speeds beyond V<sub>C</sub>/M<sub>C</sub>, freedom from aeroelastic instability must be shown to increased speeds, so that the margins intended by § 25.629(b)(2) are maintained.

(4) Failures of the system that result in forced structural vibrations (oscillatory failures) must not produce

loads that could result in detrimental deformation of primary structure.

(b) For the continuation of the flight. For the airplane in the system failed state, and considering any appropriate reconfiguration and flight limitations, the following apply:

(1) The loads derived from the following conditions (or used in lieu of the following conditions) at speeds up to V<sub>C</sub>/M<sub>C</sub> (or the speed limitation prescribed for the remainder of the flight) must be determined:

(i) The limit symmetrical maneuvering conditions specified in § 25.331 and in § 25.345.

(ii) The limit gust and turbulence conditions specified in § 25.341 and in § 25.345.

(iii) The limit rolling conditions specified in § 25.349 and the limit unsymmetrical conditions specified in § 25.367 and § 25.427(b) and (c).

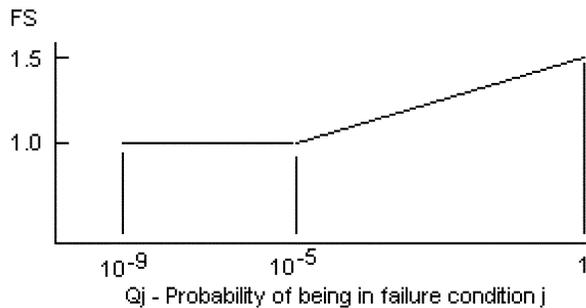
(iv) The limit yaw maneuvering conditions specified in § 25.351.

(v) The limit ground loading conditions specified in § 25.473 and § 25.491.

(2) For static strength substantiation, each part of the structure must be able to withstand the loads in paragraph B.3(b)(1) of this proposed special condition multiplied by a factor of safety depending on the probability of being in this failure state. The factor of safety is defined in Figure 2.

Figure 2

Factor of safety for continuation of flight



Qj - Probability of being in failure condition j

$Q_j = (T_j)(P_j)$

Where:

Q<sub>j</sub> = Probability of being in failure condition j

T<sub>j</sub> = Average time spent in failure condition j (in hours)

P<sub>j</sub> = Probability of occurrence of failure mode j (per hour)

**Note:** If P<sub>j</sub> is greater than 10<sup>-3</sup> per flight hour then a 1.5 factor of safety must be

applied to all limit load conditions specified in Subpart C.

(3) For residual strength substantiation, the airplane must be able to withstand two thirds of the ultimate loads defined in paragraph B.3(b)(2) of this proposed special condition. For pressurized cabins, these loads must be

combined with the normal operating differential pressure.

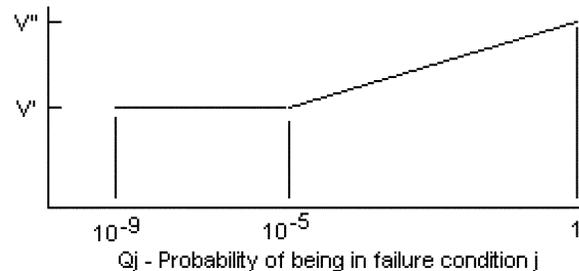
(4) If the loads induced by the failure condition have a significant effect on fatigue or damage tolerance then their effects must be taken into account.

(5) Freedom from aeroelastic instability must be shown up to a speed determined from Figure 3. Flutter clearance speeds V' and V'' may be

based on the speed limitation specified for the remainder of the flight using the margins defined by § 25.629(b).

Figure 3

## Clearance speed



$V'$  = Clearance speed as defined by § 25.629(b)(2).

$V''$  = Clearance speed as defined by § 25.629(b)(1).

$Q_j = (T_j)(P_j)$  where:

$Q_j$  = Probability of being in failure condition  $j$

$T_j$  = Average time spent in failure condition  $j$  (in hours)

$P_j$  = Probability of occurrence of failure mode  $j$  (per hour)

**Note:** If  $P_j$  is greater than  $10^{-3}$  per flight hour, then the flutter clearance speed must not be less than  $V''$ .

(6) Freedom from aeroelastic instability must also be shown up to  $V'$  in Figure 3 above, for any probable system failure condition combined with any damage required or selected for investigation by § 25.571(b).

(c) Consideration of certain failure conditions may be required by other sections of 14 CFR part 25 regardless of calculated system reliability. Where analysis shows the probability of these failure conditions to be less than  $10^{-9}$ , criteria other than those specified in this paragraph may be used for structural substantiation to show continued safe flight and landing.

4. *Failure indications.* For system failure detection and indication, the following apply:

(a) The system must be checked for failure conditions, not extremely improbable, that degrade the structural capability below the level required by part 25 or significantly reduce the reliability of the remaining system. As far as reasonably practicable, the flight crew must be made aware of these failures before flight. Certain elements of the control system, such as mechanical and hydraulic components, may use special periodic inspections, and electronic components may use daily checks, in lieu of detection and

indication systems to achieve the objective of this requirement. These certification maintenance requirements must be limited to components that are not readily detectable by normal detection and indication systems, and where service history shows that inspections will provide an adequate level of safety.

(b) The existence of any failure condition, not extremely improbable, during flight that could significantly affect the structural capability of the airplane and for which the associated reduction in airworthiness can be minimized by suitable flight limitations, must be signaled to the flight crew. For example, failure conditions that result in a factor of safety between the airplane strength and the loads of Subpart C below 1.25, or flutter margins below  $V''$ , must be signaled to the crew during flight.

5. *Dispatch with known failure conditions.* If the airplane is to be dispatched in a known system failure condition that affects structural performance, or that affects the reliability of the remaining system to maintain structural performance, then the provisions of this proposed special condition must be met, including the provisions of paragraph B.2 for the dispatched condition and paragraph B.3 for subsequent failures. Expected operational limitations may be taken into account in establishing  $P_j$  as the probability of failure occurrence for determining the safety margin in Figure 1. Flight limitations and expected operational limitations may be taken into account in establishing  $Q_j$  as the combined probability of being in the dispatched failure condition and the subsequent failure condition for the safety margins in Figures 2 and 3. These

limitations must be such that the probability of being in this combined failure state and then subsequently encountering limit load conditions is extremely improbable. No reduction in these safety margins is allowed if the subsequent system failure rate is greater than  $1E-3$  per hour.

Issued in Renton, Washington, on February 3, 2011.

**Jeffrey E. Duven,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 2011-3215 Filed 2-11-11; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 25

[Docket No. NM441 Special Conditions No. 25-11-01-SC]

#### Special Conditions: Gulfstream Model GVI Airplane; Design Roll Maneuver Requirement for Electronic Flight Controls

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Notice of proposed special conditions.

**SUMMARY:** This action proposes special conditions for the Gulfstream GVI airplane. This airplane will have novel or unusual design features when compared to the state of technology envisioned in the airworthiness standards for transport category airplanes. These design features include an electronic flight control system that provides roll control of the airplane through pilot inputs to the flight