

Although § 25.562 was written with forward- and aft-facing seats in mind, the orientation of the seat does not change the relevant test conditions, and the rule applies to all seats regardless of orientation.

The dynamic test conditions included in § 25.562 are directly applicable to side-facing seats. However, for injury pass/fail criteria, the orientation of the seat may be significant. For forward-, aft-, and side-facing seats the injury criteria are currently limited to head, spine, and femur loads. The head and lumbar loads are critical but the femur load is not critical. For a side-facing seat, additional injury parameters may be identified and evaluation of those parameters would be necessary to provide an acceptable level of safety.

When evaluating side-facing seats the following should be taken into consideration:

1. The isolation of one occupant from another. Occupants should not rely on impact with other occupants to provide energy absorption; body-to-body impacts are unacceptable.

2. The restraint system and the retention of occupants in the seat. Addressing this concern may necessitate providing a means of restraint for the lower limbs as well as the torso. Failure to limit the forward (in the airplane's coordinate system) travel of the lower limbs may cause the occupant to come out of the restraint system or produce severe injuries due to the resulting position of the restraint system and/or twisting (torsional load) of the lower lumbar spinal column.

3. The load limit in the torso in the lateral direction. Human tolerance for side-facing seats differs from that for forward- or aft-facing seats.

The automotive industry has developed test procedures and occupant injury criteria appropriate for side impact conditions. The criteria include limiting lateral pelvic accelerations and using the "Thoracic Trauma Index," which is defined in 49 CFR 571.214. Use of the side impact dummy (SID) identified in 49 CFR part 572, subpart F, rather than the Hybrid II dummy identified in 49 CFR part 572, subpart B, is required to evaluate these parameters. The Hybrid II dummy is used in the current § 25.562 test. Testing with a SID is the best means available to assess the injury potential of a sideward impact condition. Such an evaluation is considered necessary to provide an acceptable level of safety for side-facing seats.

The side-facing seat proposed special conditions have been determined to result in a level of safety equivalent to that provided by the injury pass/fail

criteria in § 25.562 for forward- or aft-facing seats.

Applicability

As discussed above, this proposed special condition is applicable to the GVI. Should Gulfstream apply at a later date for a change to the type certificate to include another model incorporating the same novel or unusual design features, this proposed special condition would apply to that model as well.

Conclusion

This action affects only certain novel or unusual design features of the GVI. It is not a rule of general applicability.

List of Subjects in 14 CFR Part 25

Aircraft, Aviation safety, Reporting and recordkeeping requirements.

The authority citation for these special conditions is as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701, 44702, 44704.

The Proposed Special Conditions

Accordingly, the Federal Aviation Administration (FAA) proposes the following special conditions as part of the type certification basis for the GVI airplanes.

In addition to the airworthiness standards in §§ 25.562 and 25.785, the following proposed special conditions provide injury criteria and installation/testing guidelines that represent the minimum acceptable airworthiness standard for single-place side-facing seats:

A. The Proposed Injury Criteria

1. *Existing Criteria:* All injury protection criteria of § 25.562(c)(1) through (c)(6) apply to the occupant of a side-facing seat. Head injury criterion (HIC) assessments are only required for head contact with the seat and/or adjacent structures.

2. *Body-to-Wall/Furnishing Contact:* The seat must be installed aft of a structure such as an interior wall or furnishing that will support the pelvis, upper arm, chest, and head of an occupant seated next to the structure. A conservative representation of the structure and its stiffness must be included in the tests. It is recommended, but not required, that the contact surface of this structure be covered with at least two inches of energy absorbing protective padding (foam or equivalent), such as Ensolute.

3. *Thoracic Trauma:* Thoracic trauma index (TTI) injury criterion must be substantiated by dynamic test or by rational analysis based on previous test(s) of a similar seat installation. Testing must be conducted with a side

impact dummy (SID), as defined by Title 49, Code of Federal Regulations (49 CFR) part 572, Subpart F, or its equivalent. TTI must be less than 85, as defined in 49 CFR part 572, subpart F. SID TTI data must be processed as defined in Federal Motor Vehicle Safety Standard (FMVSS) part 571.214, section S6.13.5.

4. *Pelvis:* Pelvic lateral acceleration must be shown by dynamic test or by rational analysis based on previous test(s) of a similar seat installation not to exceed 130g. Pelvic acceleration data must be processed as defined in FMVSS part 571.214, section S6.13.5.

5. *Shoulder Strap Loads:* Where upper torso straps (shoulder straps) are used for occupants, tension loads in individual straps must not exceed 1,750 pounds. If dual straps are used for restraining the upper torso, the total strap tension loads must not exceed 2,000 pounds.

B. General Test Guidelines

1. One longitudinal test with the SID or its equivalent, undeformed floor, no yaw, and with all lateral structural supports (armrests/walls).

Pass/fail injury assessments: TTI and pelvic acceleration.

2. One longitudinal test with the Hybrid II anthropomorphic test dummy (ATD), deformed floor, with 10 degrees yaw, and with all lateral structural support (armrests/walls).

Pass/fail injury assessments: HIC; and upper torso restraint load, restraint system retention and pelvic acceleration.

3. Vertical (14g) test is to be conducted with modified Hybrid II ATDs with existing pass/fail criteria.

Issued in Renton, Washington, on December 22, 2010.

Jeffrey E. Duven,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2006-24145; Directorate Identifier 2006-NE-06-AD]

RIN 2120-AA64

Airworthiness Directives; General Electric Company CF6-45 and CF6-50 Series Turbofan Engines

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: We propose to supersede an existing airworthiness directive (AD) that applies to General Electric Company (GE) CF6-45 and CF6-50 series turbofan engines. The existing AD requires replacing certain forward and aft centerbodies of the long fixed core exhaust nozzle (LFCEN) assembly. Since we issued that AD, we became aware that other forward and aft centerbodies are also affected. This proposed AD would add certain new centerbodies requiring replacement. This proposed AD is prompted by the discovery of more part numbers (P/Ns) of centerbodies requiring replacement. We are proposing this AD to prevent the forward and aft centerbody of the LFCEN assembly from separating from the engine, damage to the engine, and damage to the airplane.

DATES: We must receive comments on this proposed AD by February 18, 2011.

ADDRESSES: You may send comments by any of the following methods:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.

- *Fax:* 202-493-2251.

- *Mail:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

- *Hand Delivery:* Deliver to Mail address above between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this AD, contact General Electric Company, GE-Aviation, Room 285, 1 Neumann Way, Cincinnati, OH 45215, telephone 513-552-3272; fax 513-552-3329; e-mail: geae.aoc@ge.com. You may review copies of the referenced service information at the FAA, Engine & Propeller Directorate, 12 New England Executive Park, Burlington, MA. For information on the availability of this material at the FAA, call 781-238-7125.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (phone: 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

FOR FURTHER INFORMATION CONTACT: Tomasz Rakowski, Aerospace Engineer, Engine Certification Office, FAA, Engine & Propeller Directorate; phone: 781-238-7735; fax: 781-238-7199; e-mail: tomasz.rakowski@faa.gov.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposed AD. Send your comments to an address listed under the **ADDRESSES** section. Include "Docket No. FAA-2006-24145; Directorate Identifier 2006-NE-06-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD because of those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

Discussion

On February 12, 2009, we issued AD 2009-04-17, Amendment 39-15823 (74 FR 8735, February 26, 2009), for GE CF6-45 and CF6-50 series turbofan engines. That AD requires replacing LFCEN assembly forward centerbodies P/N 1313M55G01 or G02, P/N 9076M28G09 or G10, and aft centerbodies P/N 1313M56G01 or 9076M46G05 with modified centerbodies. That AD resulted from reports of separation of centerbodies from the engine due to high imbalance engine conditions caused by events including bird strikes. Separation of the centerbodies from the engine would cause engine damage and airplane damage. We issued that AD to prevent the forward and aft centerbody of the LFCEN assembly from separating from the engine, damage to the engine, and damage to the airplane.

Actions Since Existing AD Was Issued

Since we issued AD 2009-04-17, we identified seven additional centerbody P/Ns that should have been included in the AD. These centerbodies are of the same design and construction as those identified in the original AD and therefore, are subject to the same unsafe condition. We added forward centerbodies P/N 9076M28G05, G06, G08, P/N 9076M82G01, G03, and aft centerbodies P/N 9076M46G02, G04 to

the applicability of this proposed AD to remove them from service.

FAA's Determination

We are proposing this AD because we evaluated all the relevant information and determined the unsafe condition described previously is likely to exist or develop in other products of the same type design.

Proposed AD Requirements

This proposed AD would add forward centerbody P/Ns 9076M28G05, G06, and G08, P/Ns 9076M82G01, G03, and aft centerbody P/Ns 9076M46G02, G04, to those P/Ns in AD 2009-04-17 to be removed from service.

Costs of Compliance

We estimate that this proposed AD would affect 383 GE CF6-45 and CF6-50 series turbofan engines installed on airplanes of U.S. registry. We also estimate that it would take about 44 work hours per engine to perform the actions required by this AD, and that the average labor rate is \$85 per work-hour. Required parts would cost about \$11,000 per engine. Based on these figures, we estimate the total cost of this AD to U.S. operators to be \$5,645,420.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in subtitle VII, part A, subpart III, section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We have determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that the proposed regulation:

(1) Is not a "significant regulatory action" under Executive Order 12866,

(2) Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),

(3) Will not affect intrastate aviation in Alaska, and

(4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. The FAA amends § 39.13 by removing airworthiness directive (AD) 2009-04-17, Amendment 39-15823 (74 FR 8735; February 26, 2009), and adding the following new AD:

General Electric Company: Docket No. FAA-2006-24145; Directorate Identifier 2006-NE-06-AD.

Comments Due Date

(a) The FAA must receive comments on this AD action by February 18, 2011.

Affected ADs

(b) This AD supersedes AD 2009-04-17, Amendment 39-15823.

Applicability

(c) This AD applies to the following engines with a long fixed core exhaust nozzle (LFCEN) assembly forward centerbody, part number (P/N) 1313M55G01 or G02, P/N 9076M28G05, G06, G08, G09, or G10, P/N 9076M82G01 or G03, and aft centerbody P/N 1313M56G01, or P/N 9076M46G02, G04, or G05, installed in:

(1) General Electric Company (GE) CF6-45A, CF6-45A2, CF6-50A, CF6-50C, CF6-50CA, CF6-50C1, CF6-50C2, CF6-50C2B, CF6-50C2D, CF6-50E, CF6-50E1, CF6-50E2, and CF6-50E2B turbofan engines; including engines marked on the engine data plate as CF6-50C2-F and CF6-50C2-R.

(2) These engines are installed on, but not limited to, Airbus A300 series, Boeing 747 series, McDonnell Douglas DC-10 series, and DC-10-30F (KDC-10) airplanes.

Unsafe Condition

(d) This AD was prompted by the discovery of more P/Ns of centerbodies

affected, requiring replacement. We are issuing this AD to prevent the forward and aft centerbody of the LFCEN assembly from separating from the engine, damage to the engine, and damage to the airplane.

Compliance

(e) Comply with this AD within the compliance times specified, unless already done.

(1) Within 18 months after the effective date of this AD, replace forward centerbody, P/N 1313M55G01 and G02, P/N 9076M28G05, G06, G08, G09, and G10, P/N 9076M82G01 and G03, and aft centerbody P/N 1313M56G01, P/N 9076M46G02, G04, and G05 with a forward and aft centerbody that has been modified using the Accomplishment Instructions, Section 3, of GE Service Bulletin (SB) No. CF6-50 S/B 78-0244, Revision 1, dated March 13, 2008, CF6-50 S/B 78-0244, dated July 30, 2007, or CF6-50 S/B 78-0242, dated September 26, 2005.

Centerbody Installation Prohibition

(2) After 18 months from the effective date of this AD, do not install any engine with forward centerbody, P/N 1313M55G01 or G02, P/N 9076M28G05, G06, G08, G09, or G10, P/N 9076M82G01 or G03, or aft centerbody P/N 1313M56G01, P/N 9076M46G02, G04, or G05 on any airplane.

Alternative Methods of Compliance (AMOCs)

(f) The Manager, Engine Certification Office, FAA, may approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19.

Related Information

(g) For more information about this AD, contact Tomasz Rakowski, Aerospace Engineer, Engine Certification Office, FAA, Engine & Propeller Directorate; phone: 781-238-7735; fax: 781-238-7199; e-mail: tomasz.rakowski@faa.gov.

(h) For service information identified in this AD, contact General Electric Company, GE-Aviation, Room 285, 1 Neumann Way, Cincinnati, OH 45215, telephone 513-552-3272; fax 513-552-3329; e-mail: gae.aoc@ge.com. You may review copies of the referenced service information at the FAA, Engine & Propeller Directorate. For information on the availability of this material at the FAA, call 781-238-7125.

Issued in Burlington, Massachusetts, on December 28, 2010.

Peter A. White,

Assistant Manager, Engine & Propeller Directorate, Aircraft Certification Service.

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DEPARTMENT OF COMMERCE

National Oceanic and Atmospheric Administration

15 CFR Part 922

[0908041219-0073-02]

RIN 0648-AX79

Amendments to National Marine Sanctuary Regulations Regarding Low Overflights in Designated Zones

AGENCY: Office of National Marine Sanctuaries (ONMS), National Oceanic and Atmospheric Administration (NOAA), Department of Commerce (DOC).

ACTION: Extension of public comment period.

SUMMARY: On December 8, 2010, NOAA published a proposed rule in the *Federal Register* to amend the low overflight regulations of the Channel Islands, Monterey Bay, Gulf of the Farallones, and Olympic Coast national marine sanctuaries. Specifically, NOAA proposes to amend the regulations requiring that motorized aircraft maintain certain minimum altitudes above specified locations within the boundaries of the listed sanctuaries; and state that failure to comply with these altitude limits is presumed to disturb marine mammals or seabirds and is a violation of the sanctuary regulations.

DATES: The public comment period on the proposed rule published at 75 FR 76319, December 8, 2010, will be extended an additional 30 days from the original due date of January 7, 2011. Comments will be accepted through February 7, 2011.

ADDRESSES: You may submit comments, identified by RIN 0648-AX79 by any one of the following methods:

- **Electronic Submissions:** Submit all electronic public comments via the Federal eRulemaking Portal <http://www.regulations.gov>.

- **Mail:** Debra Malek, Office of National Marine Sanctuaries, 1305 East-West Highway, 11th floor, Silver Spring, MD 20910.

Instructions: No comments will be posted for public viewing until after the comment period has closed. All comments received are a part of the public record and will be posted to <http://www.regulations.gov> without change. All Personal Identifying Information (for example, name, address, etc.) voluntarily submitted by the commenter may be publicly accessible. Do not submit confidential business information or otherwise sensitive or protected information.