

the Interim Rule will not have a significant economic impact on a substantial number of small business entities. The Final Rule merely extends the safe harbor of section 360.6(b) to securitizations issued before September 30, 2010 and does not represent a change in the law.

E. Small Business Regulatory Enforcement Fairness Act

The Office of Management and Budget has determined that the rule is not a "major rule" within the meaning of the relevant sections of the Small Business Regulatory Enforcement Fairness Act of 1996 (SBREFA) (5 U.S.C. 801 *et seq.*). As required by SBREFA, the FDIC will file the appropriate reports with Congress and the General Accounting Office so that the final rule may be reviewed.

F. Paperwork Reduction Act

No collection of information pursuant to section 3504(h) of the Paperwork Reduction Act (44 U.S.C. section 3501 *et seq.*) is contained in the final rule. Consequently, no information was submitted to the Office of Management and Budget for review.

List of Subjects in 12 CFR Part 360

Banks, Banking, Bank deposit insurance, Holding companies, National banks, Participations, Reporting and recordkeeping requirements, Savings associations, Securitizations.

■ For the reasons stated above, the Board of Directors of the Federal Deposit Insurance Corporation confirms as final, the interim rule amending chapter III of title 12 of the Code of Federal Regulations by amending Part 360 published on November 17, 2010 (74 FR 59066) with the following changes:

PART 360—RESOLUTION AND RECEIVERSHIP RULES

■ 1. The authority citation for part 360 continues to read as follows:

Authority: 12 U.S.C. 1821(d)(1), 1821(d)(10)(C), 1821(d)(11), 1821(e)(1), 1821(e)(8)(D)(i), 1823(c)(4), 1823(e)(2); Sec. 401(h), Pub.L. 101–73, 103 Stat. 357.

■ 2. Amend § 360.6 by revising paragraph (b)(2) to read as follows:

§ 360.6 Treatment by the Federal Deposit Insurance Corporation as conservator or receiver of financial assets transferred in connection with a securitization or participation.

* * * * *

(b) * * *

(2) With respect to any securitization for which transfers of financial assets were made, or for revolving trusts for which beneficial interests were issued,

on or before September 30, 2010, the FDIC as conservator or receiver shall not, in the exercise of its statutory authority to disaffirm or repudiate contracts, reclaim, recover, or recharacterize as property of the institution or the receivership any such transferred financial assets notwithstanding that such transfer does not satisfy all conditions for sale accounting treatment under generally accepted accounting principles as effective for reporting periods after November 15, 2009, provided that such transfer satisfied the conditions for sale accounting treatment set forth by generally accepted accounting principles in effect for reporting periods before November 15, 2009, except for the "legal isolation" condition that is addressed by this rule.

* * * * *

Dated at Washington, DC, this 10th day of March 2010.

By Order of the Board of Directors.
Federal Deposit Insurance Corporation.

Robert E. Feldman,

Executive Secretary.

[FR Doc. 2010–5707 Filed 3–18–10; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 25

[Docket No. NM425; Special Conditions No. 25–403–SC]

Special Conditions: Airbus Model A318, A319, A320, and A321 Series Airplanes; Seats With Non-Traditional, Large, Non-Metallic Panels

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final special conditions; request for comments.

SUMMARY: These special conditions are issued for the Airbus Model A318, A319, A320, and A321 series airplanes. These airplanes will have a novel or unusual design feature(s) associated with seats that include non-traditional, large, non-metallic panels that would affect survivability during a post-crash fire event. The applicable airworthiness regulations do not contain adequate or appropriate safety standards for this design feature. These special conditions contain the additional safety standards that the Administrator considers necessary to establish a level of safety equivalent to that established by the existing airworthiness standards.

DATES: The effective date of these special conditions is March 9, 2010. We must receive your comments by May 3, 2010.

ADDRESSES: You must mail two copies of your comments to: Federal Aviation Administration, Transport Airplane Directorate, Attn: Rules Docket (ANM–113), Docket No. NM425, 1601 Lind Avenue SW., Renton, Washington 98057–3356. You may deliver two copies to the Transport Airplane Directorate at the above address. You must mark your comments: Docket No. NM425. You can inspect comments in the Rules Docket weekdays, except Federal holidays, between 7:30 a.m. and 4 p.m.

FOR FURTHER INFORMATION CONTACT:

Alan Sinclair, FAA, Airframe/Cabin Safety Branch, ANM–115, Transport Airplane Directorate, Aircraft Certification Service, 1601 Lind Avenue SW., Renton, Washington 98057–3356; telephone (425) 227–2785; facsimile (425) 227–2195; electronic mail alan.sinclair@faa.gov.

SUPPLEMENTARY INFORMATION: The FAA has determined that notice and opportunity for prior public comment hereon are impracticable because these procedures would significantly delay issuance of the design approval and thus delivery of the affected aircraft. In addition, the substance of these special conditions has been subject to the public comment process in several prior instances with no substantive comments received. The FAA therefore finds that good cause exists for making these special conditions effective upon issuance.

Comments Invited

We invite interested people to take part in this rulemaking by sending written comments, data, or views. The most helpful comments reference a specific portion of the special conditions, explain the reason for any recommended change, and include supporting data. We ask that you send us two copies of written comments.

We will file in the docket all comments we receive, as well as a report summarizing each substantive public contact with FAA personnel about these special conditions. You can inspect the docket before and after the comment closing date. If you wish to review the docket in person, go to the address in the **ADDRESSES** section of this preamble between 7:30 a.m. and 4 p.m., Monday through Friday, except Federal holidays.

We will consider all comments we receive by the closing date for comments. We will consider comments

filed late if it is possible to do so without incurring expense or delay. We may change these special conditions based on the comments we receive.

If you want us to let you know we received your comments on these special conditions, send us a pre-addressed, stamped postcard on which the docket number appears. We will stamp the date on the postcard and mail it back to you.

Background

On January 15, 2010, Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac, Cedex, France, applied for a design change to Type Certificate No. A28NM for installation of seats that include non-traditional, large, non-metallic panels in Airbus Model A318, A319, A320, and A321 series airplanes. These airplanes, currently approved under Type Certificate No. A28NM, are swept-wing, conventional-tail, twin-engine, turbofan-powered, single aisle, medium sized transport category airplanes.

The applicable regulations to airplanes currently approved under Type Certificate No. A28NM do not require seats to meet the more stringent flammability standards required of large, non-metallic panels in the cabin interior. At the time the applicable rules were written, seats were designed with a metal frame covered by fabric, not with large, non-metallic panels. Seats also met the then recently adopted standards for flammability of seat cushions. With the seat design being mostly fabric and metal, the contribution to a fire in the cabin had been minimized and was not considered a threat. For these reasons, seats did not need to be tested to heat release and smoke emission requirements.

Seat designs have now evolved to occasionally include non-traditional, large, non-metallic panels. Taken in total, the surface area of these panels is on the same order as the sidewall and overhead stowage bin interior panels. To provide the level of passenger protection intended by the airworthiness standards, these non-traditional, large, non-metallic panels in the cabin must meet the standards of Title 14, Code of Federal Regulations (14 CFR), part 25, Appendix F, parts IV and V, heat release and smoke emission requirements.

Type Certification Basis

Under the provisions of 14 CFR 21.101 Airbus must show that the Model A318, A319, A320, and A321 series airplanes, as changed, continue to meet the applicable provisions of the regulations incorporated by reference in

Type Certificate No. A28NM, or the applicable regulations in effect on the date of application for the change. The regulations incorporated by reference in the type certificate are commonly referred to as the "original type certification basis." The regulations incorporated by reference in Type Certificate No. A28NM are as follows: 14 CFR part 25, effective February 1, 1965, including Amendments 25-1 through 25-56; SFAR 27, effective February 1, 1974, including Amendments 27-1 through 27-5; and 14 CFR part 36 effective December 1, 1969, including Amendments 36-1 through 36-12.

In addition, the certification basis includes other regulations and special conditions that are not pertinent to these special conditions.

If the Administrator finds that the applicable airworthiness regulations (i.e., 14 CFR part 25) do not contain adequate or appropriate safety standards for the Model A318, A319, A320, and A321 series airplanes because of a novel or unusual design feature, special conditions are prescribed under the provisions of § 21.16.

In addition to the applicable airworthiness regulations and special conditions, the Model A318, A319, A320, and A321 series airplanes must comply with the fuel vent and exhaust emission requirements of 14 CFR part 34 and the noise certification requirements of 14 CFR part 36.

The FAA issues special conditions, as defined in § 11.19, under § 11.38 and they become part of the type certification basis under § 21.101.

Special conditions are initially applicable to the model for which they are issued. Should the type certificate for that model be amended later to include any other model that incorporates the same novel or unusual design feature, or should any other model already included on the same type certificate be modified to incorporate the same novel or unusual design feature, the special conditions would also apply to the other model.

Novel or Unusual Design Features

The Model A318, A319, A320, and A321 series airplanes will incorporate the following novel or unusual design features: These models offer interior arrangements that include passenger seats that incorporate non-traditional, large, non-metallic panels in lieu of the traditional metal frame covered by fabric. The flammability properties of these panels have been shown to significantly affect the survivability of occupants of the cabin in the case of fire. These seats are considered a novel

design for transport category airplanes that include Amendment 25-61 and Amendment 25-66 in the certification basis, and were not considered when those airworthiness standards were established.

The existing regulations do not provide adequate or appropriate safety standards for seat designs that incorporate non-traditional, large, non-metallic panels. In order to provide a level of safety that is equivalent to that provided by the balance of the cabin, additional airworthiness standards, in the form of special conditions, are necessary. These special conditions supplement § 25.853. The requirements contained in these special conditions consist of applying the identical test conditions required of all other large panels in the cabin, to seats with non-traditional, large, non-metallic panels.

Definition of "Non-Traditional, Large, Non-Metallic Panel"

A non-traditional, large, non-metallic panel, in this case, is defined as a panel with exposed-surface areas greater than 1.5 square feet installed per seat place. The panel may consist of either a single component or multiple components in a concentrated area. Examples of parts of the seat where these non-traditional panels are installed include, but are not limited to: seat backs, bottoms and leg/foot rests, kick panels, back shells, credenzas and associated furniture. Examples of traditional exempted parts of the seat include: Arm caps, armrest close-outs such as end bays and armrest-styled center consoles, food trays, video monitors and shrouds.

Clarification of "Exposed"

"Exposed" is considered to include those panels directly exposed to the passenger cabin in the traditional sense, plus those panels enveloped such as by a dress cover. Traditional fabrics or leathers currently used on seats are excluded from these special conditions. These materials must still comply with § 25.853(a) and § 25.853(c) if used as a covering for a seat cushion, or § 25.853(a) if installed elsewhere on the seat. Non-traditional, large, non-metallic panels covered with traditional fabrics or leathers will be tested without their coverings or covering attachments.

Discussion

In the early 1980s the FAA conducted extensive research on the effects of post-crash flammability in the passenger cabin. As a result of this research and service experience, we adopted new standards for interior surfaces associated with large surface area parts. Specifically, the rules require

measurement of heat release and smoke emission (part 25, Appendix F, parts IV and V) for the affected parts. Heat release has been shown to have a direct correlation with post-crash fire survival time. Materials that comply with the standards (i.e., § 25.853 entitled "Compartment interiors" as amended by Amendment 25-61 and Amendment 25-66) extend survival time by approximately 2 minutes, over materials that do not comply.

At the time these standards were written, the potential application of the requirements of heat release and smoke emission to seats was explored. The seat frame itself was not a concern because it was primarily made of aluminum and there were only small amounts of non-metallic materials. It was determined that the overall effect on survivability was negligible, whether or not the food trays met the heat release and smoke requirements. The requirements, therefore, did not address seats. The preambles to both the Notice of Proposed Rule Making (NPRM), Notice No. 85-10 (50 FR 15038, April 16, 1985), and the Final Rule at Amendment 25-61 (51 FR 26206, July 21, 1986), specifically note that seats were excluded "because the recently-adopted standards for flammability of seat cushions will greatly inhibit involvement of the seats."

Subsequently, the Final Rule at Amendment 25-83 (60 FR 6615, March 6, 1995) clarified the definition of minimum panel size: "It is not possible to cite a specific size that will apply in all installations; however, as a general rule, components with exposed-surface areas of one square foot or less may be considered small enough that they do not have to meet the new standards. Components with exposed-surface areas greater than two square feet may be considered large enough that they do have to meet the new standards. Those with exposed-surface areas greater than one square foot, but less than two square feet, must be considered in conjunction with the areas of the cabin in which they are installed before a determination could be made."

In the late 1990s, the FAA issued Policy Memorandum 97-112-39, "Guidance for Flammability Testing of Seat/Console Installations," October 17, 1997 (<http://rgl.faa.gov>). That memo was issued when it became clear that seat designs were evolving to include large non-metallic panels with surface areas that would impact survivability during a cabin fire event, comparable to partitions or galleys. The memo noted that large surface area panels must comply with heat release and smoke emission requirements, even if they

were attached to a seat. If the FAA had not issued such policy, seat designs could have been viewed as a loophole to the airworthiness standards that would result in an unacceptable decrease in survivability during a cabin fire event.

In October of 2004, an issue was raised regarding the appropriate flammability standards for passenger seats that incorporated non-traditional, large, non-metallic panels in lieu of the traditional metal covered by fabric. The Seattle Aircraft Certification Office and Transport Standards Staff reviewed this design and determined that it represented the kind and quantity of material that should be required to pass the heat release and smoke emissions requirements. We have determined that special conditions would be promulgated to apply the standards defined in § 25.853(d) to seats with large non-metallic panels in their design.

Applicability

As discussed above, these special conditions are applicable to Airbus Model A318, A319, A320, and A321 series airplanes. Although the heat release and smoke testing requirements of § 25.853 per Appendix F, parts IV and V, are not part of the part 25 certification basis for the Airbus Model A318, A319, A320, and A321 series airplanes, these special conditions are applicable if the airplanes are in 14 CFR part 121 service. Part 121 requires applicable interior panels to comply with § 25.853, Appendix F, parts IV and V, regardless of the certification basis. It is not our intent to require seats with large non-metallic panels to meet § 25.853, Appendix F, parts IV and V, if they are installed in cabins of airplanes that otherwise are not required to meet these standards. Should Airbus apply at a later date for a change to the type certificate to include another model incorporating the same novel or unusual design feature, the special conditions would apply to that model as well.

Conclusion

This action affects only certain novel or unusual design features on Airbus Model A318, A319, A320, and A321 series airplanes. It is not a rule of general applicability.

The substance of these special conditions has been subjected to the notice and comment period in several prior instances and has been derived without substantive change from those previously issued. It is unlikely that prior public comment would result in a significant change from the substance contained herein. Therefore, the FAA has determined that prior public notice

and comment are unnecessary and good cause exists for adopting these special conditions upon issuance. The FAA is requesting comments to allow interested persons to submit views that may not have been submitted in response to the prior opportunities for comment described above.

List of Subjects in 14 CFR Part 25

Aircraft, Aviation safety, Reporting and recordkeeping requirements.

■ The authority citation for these special conditions is as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701, 44702, 44704.

The Special Conditions

■ Accordingly, pursuant to the authority delegated to me by the Administrator, the following special conditions are issued as part of the type certification basis for Airbus Model A318, A319, A320, and A321 series airplanes.

1. Except as provided in paragraph 3 of these special conditions, compliance with heat release and smoke emission testing requirements per 14 CFR part 25, § 25.853 Appendix F, parts IV and V, is required for seats that incorporate non-traditional, large, non-metallic panels that may either be a single component or multiple components in a concentrated area in their design.

2. The applicant may designate up to and including 1.5 square feet of non-traditional, non-metallic panel material per seat place that does not have to comply with special condition Number 1, above. A triple seat assembly may have a total of 4.5 square feet excluded on any portion of the assembly (e.g., outboard seat place 1 square foot, middle 1 square foot, and inboard 2.5 square feet).

3. Seats do not have to meet the test requirements of 14 CFR part 25, Appendix F, parts IV and V, when installed in compartments that are not otherwise required to meet these requirements. Examples include:

a. Airplanes with passenger capacities of 19 or less,

b. Airplanes that do not have § 25.853, Amendment 25-61 or later, in their certification basis and do not need to comply with the requirements of 14 CFR 121.312, and

c. Airplanes exempted from § 25.853, Amendment 25-61 or later.

4. Only airplanes associated with new seat certification programs approved after the effective date of these special conditions will be affected by the requirements in these special conditions. Previously certificated interiors on the existing airplane fleet and follow-on deliveries of airplanes

with previously certificated interiors are not affected.

Issued in Renton, Washington, on March 9, 2010.

Jeffrey E. Duven,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2010-5871 Filed 3-17-10; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2009-0883; Directorate Identifier 97-ANE-08; Amendment 39-16237; AD 97-17-04R1]

RIN 2120-AA64

Airworthiness Directives; Pratt & Whitney JT8D-209, -217, -217C, and -219 Turbofan Engines

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: The FAA is revising an existing airworthiness directive (AD) for Pratt & Whitney JT8D-209, -217, -217C, and -219 turbofan engines with front compressor front hub (fan hub), part number (P/N) 5000501-01 installed. That AD currently requires cleaning the front compressor front hubs (fan hubs), initial and repetitive eddy current (ECI) and fluorescent penetrant inspections (FPI) of tierod and counterweight holes for cracks, removal of bushings, cleaning and ECI and FPI of bushed holes for cracks and, if necessary, replacement with serviceable parts. In addition, that AD currently requires reporting the findings of cracked fan hubs and monthly reports of the number of inspections completed. This AD requires the same actions, except for the monthly reporting of the number of completed inspections. This AD results from the FAA determining that it has collected a sufficient amount of data since issuing AD 97-17-04 and that therefore, it no longer needs the monthly reporting of the number of completed inspections. We are issuing this AD to prevent fan hub failure due to tierod, counterweight, or bushed hole cracking, which could result in an uncontained engine failure and damage to the airplane.

DATES: This AD becomes effective April 22, 2010. The Director of the Federal Register previously approved the incorporation by reference of the

publications listed in the regulations as of March 5, 1997 (62 FR 4902).

ADDRESSES: You can get the service information identified in this AD from Pratt & Whitney, 400 Main St., East Hartford, CT 06108; telephone (860) 565-8770; fax (860) 565-4503.

The Docket Operations office is located at Docket Management Facility, U.S. Department of Transportation, 1200 New Jersey Avenue, SE., West Building Ground Floor, Room W12-140, Washington, DC 20590-0001.

FOR FURTHER INFORMATION CONTACT:

Kevin Dickert, Aerospace Engineer, Engine Certification Office, FAA, Engine & Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803; e-mail: kevin.dickert@faa.gov; telephone (781) 238-7117; fax (781) 238-7199.

SUPPLEMENTARY INFORMATION: The FAA proposed to amend 14 CFR part 39 with a proposed AD. The proposed AD applies to Pratt & Whitney JT8D-209, -217, -217C, and -219 turbofan engines with front compressor front hub (fan hub), P/N 5000501-01 installed. We published the proposed AD in the **Federal Register** on December 21, 2009 (74 FR 67831). That action proposed to require cleaning the front compressor front hubs (fan hubs), initial and repetitive ECI and FPI of tierod and counterweight holes for cracks, removal of bushings, cleaning and ECI and FPI of bushed holes for cracks and, if necessary, replacement with serviceable parts. That action also proposed to eliminate the monthly reporting of the number of completed inspections.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Operations office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Operations office (telephone (800) 647-5527) is provided in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

Comments

We provided the public the opportunity to participate in the development of this AD. We have considered the one comment received. The commenter supports the proposal.

Conclusion

We have carefully reviewed the available data, including the comment received, and determined that air safety

and the public interest require adopting the AD as proposed.

Costs of Compliance

We estimate that this AD revision will affect 1,170 JT8D-209, -217, -217C, and -219 turbofan engines installed on airplanes of U.S. registry. We estimate that it will take four work-hours per engine to complete one inspection of the fan hub at piece-part exposure. The average labor rate is \$80 per work-hour. Based on these figures, we estimate the total cost of the AD to U.S. operators to be \$374,400.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, Section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We have determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866;
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
- (3) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a summary of the costs to comply with this AD and placed it in the AD Docket. You may get a copy of this summary at the address listed under **ADDRESSES**.