

TABLE 1—APPLICABILITY

Boeing model—	As identified in Boeing Alert Service Bulletin—
737–100, –200, and –200C series airplanes .....	737–53A1197, dated August 25, 2006.
737–300, –400, and –500 series airplanes .....	737–53A1188, Revision 2, dated May 9, 2007, or 737–53A1197, dated August 25, 2006.

**Unsafe Condition**

(d) This AD results from reports of cracks found at the cutout in the web of body station frame 303.9 inboard of stringer 16L. We are issuing this AD to detect and correct such cracking, which could prevent the left forward entry door from sealing correctly, and could cause in-flight decompression of the airplane.

**Compliance**

(e) You are responsible for having the actions required by this AD performed within the compliance times specified, unless the actions have already been done.

**Repetitive Inspections: Service Bulletin 737–53A1188**

(f) For airplanes identified in Boeing Alert Service Bulletin 737–53A1188, Revision 2, dated May 9, 2007, including airplanes modified by the repair/preventive change specified in the original version, dated April 9, 1998; or Revision 1, dated March 18, 1999; of the service bulletin: Do detailed and high frequency eddy current (HFEC) inspections in the web and doubler around the slotted holes in the frame web at stringers 15L and 16L, in accordance with the Accomplishment Instructions of Revision 2 of the service bulletin. Do the inspections at the applicable time specified in paragraph 1.E. of Revision 2 of the service bulletin, except as provided by paragraph (h) of this AD. Do all applicable corrective actions before further flight in accordance with Revision 2 of the service bulletin, except as provided by paragraph (i) of this AD. Repeat the inspections at intervals not to exceed 4,500 flight cycles until accomplishment of the repair/preventive change in accordance with Revision 2 of the service bulletin, which terminates the repetitive inspection requirements. A repair/preventive change done in accordance with the original version or Revision 1 of the service bulletin does not terminate the repetitive inspections, but the repetitive inspections may be terminated after the existing kit is replaced with a new kit in accordance with paragraph 3.B., Part II, step 3, or Part III, step 3, of Revision 2 of the service bulletin.

**Repetitive Inspections: Service Bulletin 737–53A1197**

(g) For airplanes identified in Boeing Alert Service Bulletin 737–53A1197, dated August 25, 2006: Do an ultrasound inspection of the slot-shaped cutout in the web for the door stop strap at stringer 16L, an HFEC inspection of the web along the upper and lower edges of the doubler around the doorstop strap at stringer 16L, and a detailed inspection of the web around the doubler for the cutout at stringer 16L, in accordance with the Accomplishment Instructions of the

service bulletin. Do the inspections at the applicable time specified in paragraph 1.E. of the service bulletin, except as provided by paragraph (h) of this AD. Do all applicable corrective actions before further flight in accordance with the service bulletin, except as provided by paragraph (i) of this AD. Repeat the inspections at intervals not to exceed 4,500 flight cycles, until accomplishment of the repair/preventive change in accordance with the service bulletin, which terminates the repetitive inspections.

**Exceptions to Service Bulletin Specifications**

(h) Where Boeing Alert Service Bulletin 737–53A1188, Revision 2, dated May 9, 2007, and Boeing Alert Service Bulletin 737–53A1197, dated August 25, 2006, specify a compliance time after release of the service bulletin, this AD requires compliance within the specified time after the effective date of this AD. For the initial inspection, the grace period for airplanes that have exceeded the specified threshold is extended to 4,500 flight cycles after the effective date of this AD.

(i) Where Boeing Alert Service Bulletin 737–53A1188, Revision 2, dated May 9, 2007, and Boeing Alert Service Bulletin 737–53A1197, dated August 25, 2006, specify to contact Boeing for appropriate action, including repair of damage outside the scope of the service bulletin, repair using a method approved in accordance with the procedures specified in paragraph (j) of this AD.

**Alternative Methods of Compliance (AMOCs)**

(j)(1) The Manager, Seattle Aircraft Certification Office, FAA, ATTN: Howard Hall, Aerospace Engineer, Airframe Branch, ANM–120S; telephone (425) 917–6430; fax (425) 917–6590; has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19.

(2) To request a different method of compliance or a different compliance time for this AD, follow the procedures in 14 CFR 39.19. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

(3) An AMOC that provides an acceptable level of safety may be used for any repair required by this AD, if it is approved by an Authorized Representative for the Boeing Commercial Airplanes Delegation Option Authorization Organization who has been authorized by the Manager, Seattle ACO, to make those findings. For a repair method to be approved, the repair must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

**Material Incorporated by Reference**

(k) You must use Boeing Alert Service Bulletin 737–53A1188, Revision 2, dated May 9, 2007; or Boeing Alert Service Bulletin 737–53A1197, dated August 25, 2006; as applicable; to do the actions required by this AD; unless the AD specifies otherwise.

(1) The Director of the Federal Register approved the incorporation by reference of this service information under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) For service information identified in this AD, contact Boeing Commercial Airplanes, P.O. Box 3707, Seattle, Washington 98124–2207.

(3) You may review copies of the service information incorporated by reference at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202–741–6030, or go to [http://www.archives.gov/federal\\_register/code\\_of\\_federal\\_regulations/ibr\\_locations.html](http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html).

Issued in Renton, Washington, on August 6, 2008.

**Ali Bahrami,**

*Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. E8–18812 Filed 8–18–08; 8:45 am]

**BILLING CODE 4910–13–P**

**DEPARTMENT OF TRANSPORTATION**

**Federal Aviation Administration**

**14 CFR Part 39**

**[Docket No. FAA–2008–0622; Directorate Identifier 2008–NM–064–AD; Amendment 39–15642; AD 2008–17–04]**

**RIN 2120–AA64**

**Airworthiness Directives; BAE Systems (Operations) Limited (Jetstream) Model 4101 Airplanes**

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Final rule.

**SUMMARY:** We are adopting a new airworthiness directive (AD) for the products listed above. This AD results from mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country to identify and correct an unsafe condition on an aviation

product. The MCAI describes the unsafe condition as:

Resulting from the assessment of fuel tank wiring installations required by SFAR 88 (Special Federal Aviation Regulation 88) and equivalent JAA/EASA (Joint Aviation Authorities/European Aviation Safety Agency) policy, BAE Systems identified \* \* \* features in the Jetstream 4100 where the need for design changes was apparent.  
\* \* \*

Internal fuel tank wiring chafing damage, if not corrected, could lead to ignition of fuel vapours and subsequent fuel tank explosion.  
\* \* \* \* \*

We are issuing this AD to require actions to correct the unsafe condition on these products.

**DATES:** This AD becomes effective September 23, 2008.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in this AD as of September 23, 2008.

**ADDRESSES:** You may examine the AD docket on the Internet at <http://www.regulations.gov> or in person at the U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Todd Thompson, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue, SW., Renton, Washington 98057-3356; telephone (425) 227-1175; fax (425) 227-1149.

#### SUPPLEMENTARY INFORMATION:

#### Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that would apply to the specified products. That NPRM was published in the **Federal Register** on June 9, 2008 (73 FR 32488). That NPRM proposed to correct an unsafe condition for the specified products. The MCAI states:

Resulting from the assessment of fuel tank wiring installations required by SFAR 88 (Special Federal Aviation Regulation 88) and equivalent JAA/EASA (Joint Aviation Authorities/European Aviation Safety Agency) policy, BAE Systems identified two features in the Jetstream 4100 where the need for design changes was apparent. One of these is addressed by Service Bulletin (SB) J41-28-014 which introduces changes to the wiring harness installations to the left (LH) and right (RH) fuel boost pumps, identified by modification number JM41672. In addition, to detect excessive cable lengths and evidence of chafing damage, SB J41-28-014 provides instructions to inspect and correct, as necessary, the internal fuel tank wiring routed to the LH and RH high level sensors.

Internal fuel tank wiring chafing damage, if not corrected, could lead to ignition of fuel vapours and subsequent fuel tank explosion.

For the reason stated above, this EASA Airworthiness Directive (AD) requires the replacement of the (LH and RH) fuel boost pump metallic conduit assemblies with loom assemblies and the inspection of internal fuel tank high level sensor wiring, including corrective actions, as necessary.

Corrective actions include replacing any damaged internal fuel tank high level sensor wiring and removing excess wiring. You may obtain further information by examining the MCAI in the AD docket.

#### Comments

We gave the public the opportunity to participate in developing this AD. We received no comments on the NPRM or on the determination of the cost to the public.

#### Conclusion

We reviewed the available data and determined that air safety and the public interest require adopting the AD as proposed.

#### Differences Between This AD and the MCAI or Service Information

We have reviewed the MCAI and related service information and, in general, agree with their substance. But we might have found it necessary to use different words from those in the MCAI to ensure the AD is clear for U.S. operators and is enforceable. In making these changes, we do not intend to differ substantively from the information provided in the MCAI and related service information.

We might also have required different actions in this AD from those in the MCAI in order to follow our FAA policies. Any such differences are highlighted in a NOTE within the AD.

#### Costs of Compliance

We estimate that this AD will affect 7 products of U.S. registry. We also estimate that it will take 47 work-hours per product to comply with the basic requirements of this AD. The average labor rate is \$80 per work-hour. Required parts will cost about \$7,000 per product. Where the service information lists required parts costs that are covered under warranty, we have assumed that there will be no charge for these parts. As we do not control warranty coverage for affected parties, some parties may incur costs higher than estimated here. Based on these figures, we estimate the cost of this AD to the U.S. operators to be \$75,320, or \$10,760 per product.

#### Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

#### Regulatory Findings

We determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

*For the reasons discussed above, I certify this AD:*

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
3. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this AD and placed it in the AD docket.

#### Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Operations office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains the NPRM, the regulatory evaluation, any comments received, and other information. The street address for the Docket Operations office (telephone (800) 647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

**List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

**Adoption of the Amendment**

■ Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

**PART 39—AIRWORTHINESS DIRECTIVES**

■ 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

**§ 39.13 [Amended]**

■ 2. The FAA amends § 39.13 by adding the following new AD:

**2008-17-04 BAE Systems (Operations) Limited (Formerly British Aerospace Regional Aircraft):** Amendment 39-15642. Docket No. FAA-2008-0622; Directorate Identifier 2008-NM-064-AD.

**Effective Date**

(a) This airworthiness directive (AD) becomes effective September 23, 2008.

**Affected ADs**

(b) None.

**Applicability**

(c) This AD applies to all BAE Systems (Operations) Limited Model Jetstream 4100 airplanes, certificated in any category, all serial numbers.

**Subject**

(d) Air Transport Association (ATA) of America Code 28: Fuel.

**Reason**

(e) The mandatory continuing airworthiness information (MCAI) states:

Resulting from the assessment of fuel tank wiring installations required by SFAR 88 (Special Federal Aviation Regulation 88) and equivalent JAA/EASA (Joint Aviation Authorities/European Aviation Safety Agency) policy, BAE Systems identified two features in the Jetstream 4100 where the need for design changes was apparent. One of these is addressed by Service Bulletin (SB) J41-28-014 which introduces changes to the wiring harness installations to the left (LH) and right (RH) fuel boost pumps, identified by modification number JM41672. In addition, to detect excessive cable lengths and evidence of chafing damage, SB J41-28-014 provides instructions to inspect and correct, as necessary, the internal fuel tank wiring routed to the LH and RH high level sensors.

Internal fuel tank wiring chafing damage, if not corrected, could lead to ignition of fuel vapours and subsequent fuel tank explosion.

For the reason stated above, this EASA Airworthiness Directive (AD) requires the replacement of the (LH and RH) fuel boost pump metallic conduit assemblies with loom assemblies and the inspection of internal fuel

tank high level sensor wiring, including corrective actions, as necessary.

Corrective actions include replacing any damaged internal fuel tank high level sensor wiring and removing excess wiring.

**Actions and Compliance**

(f) Unless already done: Within 24 months after the effective date of this AD, do the following actions.

(1) Modify the LH and RH wing fuel boost pump wiring in accordance with paragraphs 2.B. and 2.C. of the Accomplishment Instructions of BAE Systems (Operations) Limited Service Bulletin J41-28-014, Revision 1, dated December 21, 2007.

(2) Inspect the LH and RH wing fuel high level sensor wiring in accordance with paragraph 2.D. of the Accomplishment Instructions of BAE Systems (Operations) Limited Service Bulletin J41-28-014, Revision 1, dated December 21, 2007.

(3) When excess wiring and/or damaged wiring is found during the inspection required by paragraph (f)(2) of this AD, before next flight, accomplish the corrective actions as specified in paragraph 2.D. of the Accomplishment Instructions of BAE Systems (Operations) Limited Service Bulletin J41-28-014, Revision 1, dated December 21, 2007.

**FAA AD Differences**

*Note:* This AD differs from the MCAI and/or service information as follows: No differences.

**Other FAA AD Provisions**

(g) The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs):* The Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. Send information to *ATTN:* Todd Thompson, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue, SW., Renton, Washington 98057-3356; telephone (425) 227-1175; fax (425) 227-1149. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

(2) *Airworthy Product:* For any requirement in this AD to obtain corrective actions from a manufacturer or other source, use these actions if they are FAA-approved. Corrective actions are considered FAA-approved if they are approved by the State of Design Authority (or their delegated agent). You are required to assure the product is airworthy before it is returned to service.

(3) *Reporting Requirements:* For any reporting requirement in this AD, under the provisions of the Paperwork Reduction Act, the Office of Management and Budget (OMB) has approved the information collection requirements and has assigned OMB Control Number 2120-0056.

**Related Information**

(h) Refer to MCAI EASA Airworthiness Directive 2008-0041, dated February 27,

2008; and BAE Systems (Operations) Limited Service Bulletin J41-28-014, Revision 1, dated December 21, 2007; for related information.

**Material Incorporated by Reference**

(i) You must use BAE Systems (Operations) Limited Service Bulletin J41-28-014, Revision 1, dated December 21, 2007, to do the actions required by this AD, unless the AD specifies otherwise.

(1) The Director of the Federal Register approved the incorporation by reference of this service information under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) For service information identified in this AD, contact British Aerospace Regional Aircraft American Support, 13850 Mclearn Road, Herndon, Virginia 20171.

(3) You may review copies at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call (202) 741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Renton, Washington, on August 5, 2008.

**Ali Bahrami,**

*Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. E8-18810 Filed 8-18-08; 8:45 am]

**BILLING CODE 4910-13-P**

**DEPARTMENT OF TRANSPORTATION****Federal Aviation Administration****14 CFR Part 39**

[Docket No. FAA-2008-0649; Directorate Identifier 2008-CE-038-AD; Amendment 39-15646; AD 2008-17-08]

**RIN 2120-AA64**

**Airworthiness Directives; DG Flugzeugbau GmbH Model DG-500MB Powered Sailplanes**

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Final rule.

**SUMMARY:** We are adopting a new airworthiness directive (AD) for the products listed above. This AD results from mandatory continuing airworthiness information (MCAI) issued by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as:

A DG-500MB experienced, after the engine shutdown, an uncommanded retraction of its powerplant.

Investigations revealed that some bolts of the extension retraction mechanism had