

**Installation of Bonding Jumper**

(g) For any composite gray water drain mast identified during the inspection or records check required by paragraph (f) of this AD: Within 60 months after the effective date of this AD, install a bonding jumper between a ground and the clamp on the tube of the gray water composite drain mast, in accordance with the Accomplishment Instructions of Boeing Special Attention Service Bulletin 777-30-0014, dated July 24, 2006.

**Installation of Bonding Jumper Not Necessary for Aluminum Drain Masts**

(h) For airplanes on which the forward composite drain mast has been replaced with an aluminum drain mast per Boeing Service Bulletin 777-38-0026: Installation of the bonding jumper specified in paragraph (g) of this AD is not required for the forward gray water drain mast, as specified in Part 1 of the Accomplishment Instructions of Boeing Special Attention Service Bulletin 777-30-0014, dated July 24, 2006.

**Parts Installation**

(i) As of the effective date of this AD, no person may install, on any airplane, a composite gray water drain mast, unless a bonding jumper is also installed, as specified in paragraph (g) of this AD.

**Alternative Methods of Compliance (AMOCs)**

(j)(1) The Manager, Seattle Aircraft Certification Office, FAA, has the authority to approve AMOCs for this AD, if requested in accordance with the procedures found in 14 CFR 39.19.

(2) To request a different method of compliance or a different compliance time for this AD, follow the procedures in 14 CFR 39.19. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

**Material Incorporated by Reference**

(k) You must use Boeing Special Attention Service Bulletin 777-30-0014, dated July 24, 2006, to do the actions required by this AD, unless the AD specifies otherwise.

(1) The Director of the Federal Register approved the incorporation by reference of this service information under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) For service information identified in this AD, contact Boeing Commercial Airplanes, P.O. Box 3707, Seattle, Washington 98124-2207.

(3) You may review copies of the service information incorporated by reference at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: [http://www.archives.gov/federal\\_register/code\\_of\\_federal\\_regulations/ibr\\_locations.html](http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html).

Issued in Renton, Washington, on April 17, 2008.

**Ali Bahrami,**

*Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. E8-9113 Filed 5-1-08; 8:45 am]

**BILLING CODE 4910-13-P**

**DEPARTMENT OF TRANSPORTATION****Federal Aviation Administration****14 CFR Part 39**

**[Docket No. FAA-2008-0015; Directorate Identifier 2007-NM-328-AD; Amendment 39-15498; AD 2008-09-17]**

**RIN 2120-AA64**

**Airworthiness Directives; McDonnell Douglas Model DC-10-10, DC-10-10F, DC-10-15, and MD-10-10F Airplanes**

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Final rule.

**SUMMARY:** We are adopting a new airworthiness directive (AD) for certain McDonnell Douglas Model DC-10-10, DC-10-10F, DC-10-15, and MD-10-10F airplanes. This AD requires repetitive inspections for the presence of stray nickel or chrome plating deposits on the air filler valve bore of certain main landing gear (MLG) shock strut cylinders, and if necessary, related investigative and corrective actions. Doing the corrective action terminates the repetitive inspections. This AD results from a report of a left MLG collapse during landing rollout. We are issuing this AD to detect and correct stray nickel and chrome plating deposits, corrosion, and cracking of the air filler valve bore on the MLG cylinder, which could result in landing gear failure, significant damage to the airplane, and injury to personnel.

**DATES:** This AD is effective June 6, 2008.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in this AD as of June 6, 2008.

**ADDRESSES:** For service information identified in this AD, contact Boeing Commercial Airplanes, Long Beach Division, 3855 Lakewood Boulevard, Long Beach, California 90846, Attention: Data and Service Management, Dept. C1-L5A (D800-0024).

**Examining the AD Docket**

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between

9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (telephone 800-647-5527) is the Document Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

**FOR FURTHER INFORMATION CONTACT:**

Maureen Moreland, Aerospace Engineer, Airframe Branch, ANM-120L, FAA, Los Angeles Aircraft Certification Office, 3960 Paramount Boulevard, Lakewood, California 90712-4137; telephone (562) 627-5238; fax (562) 627-5210.

**SUPPLEMENTARY INFORMATION:****Discussion**

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an airworthiness directive (AD) that would apply to certain McDonnell Douglas Model DC-10-10, DC-10-10F, DC-10-15, and MD-10-10F airplanes. That NPRM was published in the **Federal Register** on January 14, 2008 (73 FR 2206). That NPRM proposed to require repetitive inspections for the presence of stray nickel or chrome plating deposits on the air filler valve bore of certain main landing gear (MLG) shock strut cylinders, and if necessary, related investigative and corrective actions. Doing the corrective action would terminate the repetitive inspections.

**Comments**

We gave the public the opportunity to participate in developing this AD. We considered the comment received from the commenter.

**Support for the NPRM**

The Air Line Pilots Association, International (ALPA), supports the intent of the NPRM.

**Request To Reduce the Compliance Time**

ALPA requests that we reduce the 24-month compliance time for the airplanes identified in paragraph (f)(2) of the NPRM. ALPA requests the reduction in compliance time due to the stated severity of a landing gear failure, the relatively short inspection times, and the low estimated inspection costs.

We do not agree to reduce the compliance time specified in paragraph (f)(2) of this AD. In developing the compliance time for this AD action, we considered not only the safety implications of the identified unsafe

condition, but the average utilization rate of the affected fleet, the practical aspects of an orderly inspection of the fleet during regular maintenance periods, and the availability of replacement parts. In addition, we also considered the manufacturer's recommendation for an appropriate compliance time. After considering all the available information, we determined that the 24-month compliance time represents an

appropriate interval of time in which the required actions can be performed in a timely manner within the affected fleet, while still maintaining an adequate level of safety. However, if additional data are presented that would justify a shorter compliance time, we might consider further rulemaking on this issue. We have not changed this AD in this regard.

**Conclusion**

We reviewed the relevant data, considered the comment received, and determined that air safety and the public interest require adopting the AD as proposed.

**Costs of Compliance**

We estimate that this AD affects 75 airplanes of U.S. registry. The following table provides the estimated costs for U.S. operators to comply with this AD.

ESTIMATED COSTS

Action	Work hours	Average labor rate per hour	Cost per product	Number of U.S.-registered airplanes	Fleet cost
Inspection .....	4	\$80	\$320, per inspection cycle .....	75	\$24,000, per inspection cycle.

**Authority for This Rulemaking**

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

**Regulatory Findings**

This AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866,
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979), and
- (3) Will not have a significant economic impact, positive or negative, on a substantial number of small entities

under the criteria of the Regulatory Flexibility Act.

You can find our regulatory evaluation and the estimated costs of compliance in the AD Docket.

**List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

**Adoption of the Amendment**

■ Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

**PART 39—AIRWORTHINESS DIRECTIVES**

■ 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

**§ 39.13 [Amended]**

■ 2. The FAA amends § 39.13 by adding the following new AD:

**2008-09-17 McDonnell Douglas:**  
Amendment 39-15498. Docket No. FAA-2008-0015; Directorate Identifier 2007-NM-328-AD.

**Effective Date**

(a) This airworthiness directive (AD) is effective June 6, 2008.

**Affected ADs**

(b) None.

**Applicability**

(c) This AD applies to McDonnell Douglas Model DC-10-10, DC-10-10F, DC-10-15, and MD-10-10F airplanes, certificated in any category; as identified in Boeing Alert Service Bulletin DC10-32A259, dated October 30, 2007.

**Unsafe Condition**

(d) This AD results from a report of a left main landing gear (MLG) collapse during landing rollout. We are issuing this AD to detect and correct stray nickel and chrome plating deposits, corrosion, and cracking of the air filler valve bore on the MLG cylinder, which could result in landing gear failure, significant damage to the airplane, and injury to personnel.

**Compliance**

(e) Comply with this AD within the compliance times specified, unless already done.

**Inspections and Corrective Actions**

(f) At the applicable time specified in paragraph (f)(1), (f)(2), or (f)(3) of this AD, do a video probe inspection for the presence of stray nickel or chrome plating deposits on the air filler valve bore of the MLG shock strut cylinders, and before further flight, do all applicable related investigative and corrective actions, in accordance with the Accomplishment Instructions of Boeing Alert Service Bulletin DC10-32A259, dated October 30, 2007. Repeat the video probe inspection thereafter at intervals not to exceed 2,400 flight cycles or 20 months, whichever occurs first. Accomplishment of the repair specified in Part 2 of the Accomplishment Instructions of the service bulletin or the replacement specified in Part 3 of the Accomplishment Instructions of the service bulletin terminates the repetitive inspections for that MLG shock strut cylinder.

(1) For passenger airplanes: Within 24 months after the effective date of this AD.

(2) For freighter airplanes with MLG cylinders that have accumulated fewer than 7,200 flight cycles in a freighter configuration as of the effective date of this AD: Within 24 months after the effective date of this AD.

(3) For freighter airplanes with MLG cylinders that have accumulated 7,200 flight cycles or more in a freighter configuration as of the effective date of this AD: Within 6 months after the effective date of this AD.

**Parts Installation**

(g) As of the effective date of this AD, no person may install a MLG shock strut cylinder assembly, part number ARG7002-1, -501, -503, or -505, on any airplane, unless the air filler valve bore hole has been oversized and closing action has been accomplished in accordance with Boeing Alert Service Bulletin DC10-32A259, dated October 30, 2007, and the MLG shock strut cylinder assembly has been permanently identified with part number SB10320259-3 adjacent to the existing ARG7002 part number.

**Alternative Methods of Compliance (AMOCs)**

(h)(1) The Manager, Los Angeles Aircraft Certification Office (ACO), FAA, ATTN: Maureen Moreland, Aerospace Engineer, Airframe Branch, ANM-120L, Los Angeles ACO, 3960 Paramount Boulevard, Lakewood, California 90712-4137; telephone (562) 627-5238; fax (562) 627-5210; has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19.

(2) To request a different method of compliance or a different compliance time for this AD, follow the procedures in 14 CFR 39.19. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

**Material Incorporated by Reference**

(i) You must use Boeing Alert Service Bulletin DC10-32A259, dated October 30, 2007, to do the actions required by this AD, unless the AD specifies otherwise.

(1) The Director of the Federal Register approved the incorporation by reference of this service information under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) For service information identified in this AD, contact Boeing Commercial Airplanes, Long Beach Division, 3855 Lakewood Boulevard, Long Beach, California 90846, Attention: Data and Service Management, Dept. C1-L5A (D800-0024).

(3) You may review copies of the service information incorporated by reference at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, Washington; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: [http://www.archives.gov/federal\\_register/code\\_of\\_federal\\_regulations/ibr\\_locations.html](http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html).

Issued in Renton, Washington, on April 18, 2008.

**Ali Bahrami,**

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. E8-9439 Filed 5-1-08; 8:45 am]

**BILLING CODE 4910-13-P**

**DEPARTMENT OF TRANSPORTATION****Federal Aviation Administration****14 CFR Part 39**

[Docket No. FAA-2008-0266; Directorate Identifier 2008-NM-013-AD; Amendment 39-15506; AD 2008-09-25]

**RIN 2120-AA64**

**Airworthiness Directives; Bombardier Model DHC-8-102, DHC-8-103, DHC-8-106, DHC-8-201, DHC-8-202, DHC-8-301, DHC-8-311, and DHC-8-315 Airplanes**

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Final rule.

**SUMMARY:** We are adopting a new airworthiness directive (AD) for the products listed above. This AD results from mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as:

Bombardier Aerospace has completed a system safety review of the aircraft fuel system against fuel tank safety standards introduced in Chapter 525 of the Airworthiness Manual through Notice of Proposed Amendment (NPA) 2002-043. The identified non-compliances were then assessed using Transport Canada Policy Letter No. 525-001, to determine if mandatory corrective action is required.

The assessment showed that it is necessary to introduce Critical Design Configuration Control Limitations (CDCCL), in order to preserve critical fuel tank system ignition source prevention features during configuration changes such as modifications and repairs, or during maintenance actions. Failure to preserve critical fuel tank system ignition source prevention features could result in a fuel tank explosion. \* \* \*

We are issuing this AD to require actions to correct the unsafe condition on these products.

**DATES:** This AD becomes effective June 6, 2008.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD as of June 6, 2008.

**ADDRESSES:** You may examine the AD docket on the Internet at <http://www.regulations.gov> or in person at the U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Mazdak Hobbi, Aerospace Engineer,

Airframe and Propulsion Branch, ANE-171, FAA, New York Aircraft Certification Office, 1600 Stewart Avenue, Suite 410, Westbury, New York 11590; telephone (516) 228-7330; fax (516) 794-5531.

**SUPPLEMENTARY INFORMATION:****Discussion**

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that would apply to the specified products. That NPRM was published in the **Federal Register** on March 11, 2008 (73 FR 12912). That NPRM proposed to correct an unsafe condition for the specified products. The MCAI states:

Bombardier Aerospace has completed a system safety review of the aircraft fuel system against fuel tank safety standards introduced in Chapter 525 of the Airworthiness Manual through Notice of Proposed Amendment (NPA) 2002-043. The identified non-compliances were then assessed using Transport Canada Policy Letter No. 525-001, to determine if mandatory corrective action is required.

The assessment showed that it is necessary to introduce Critical Design Configuration Control Limitations (CDCCL), in order to preserve critical fuel tank system ignition source prevention features during configuration changes such as modifications and repairs, or during maintenance actions. Failure to preserve critical fuel tank system ignition source prevention features could result in a fuel tank explosion. Revisions have been made to Part 2 "Airworthiness Limitations List" of the Maintenance Program Manuals of the affected aircraft models to introduce the required CDCCL.

The corrective action is revising the Airworthiness Limitations Section of the Instructions for Continued Airworthiness to include the CDCCL data. You may obtain further information by examining the MCAI in the AD docket.

**Comments**

We gave the public the opportunity to participate in developing this AD. We received no comments on the NPRM or on the determination of the cost to the public.

**Conclusion**

We reviewed the available data and determined that air safety and the public interest require adopting the AD as proposed.

**Differences Between This AD and the MCAI or Service Information**

We have reviewed the MCAI and related service information and, in general, agree with their substance. But we might have found it necessary to use different words from those in the MCAI