

reduced controllability of the airplane, accomplish the following:

Modification of Clear-Ice Indication System

(a) For Model EMB-145XR airplanes: Within 24 months or 5,000 flight hours after the effective date of this AD, whichever comes first, perform the actions specified in paragraphs (a)(1) and (a)(2) of this AD, as applicable, in accordance with the Accomplishment Instructions of EMBRAER Service Bulletin 145-30-0035, Revision 03, dated March 3, 2005.

(1) Install complete electrical connections and provisions to add an additional indication device to the clear-ice indication system, as specified in the Accomplishment Instructions, Part I.

(2) Replace the existing clear-ice indication lamp with a new lamp having a new part number, as specified in the Accomplishment Instructions, Part II.

(b) For Model EMB-135BJ airplanes: Within 24 months or 5,000 flight hours after the effective date of this AD, whichever comes first, perform the actions of paragraphs (b)(1), (b)(2), (b)(3), and (b)(4) of this AD, as applicable, in accordance with the Accomplishment Instructions of EMBRAER Service Bulletin 145LEG-30-0002, Revision 01, dated January 4, 2005.

(1) Install complete electrical connections and provisions to add an additional indication device to the clear-ice indication system, as specified in the Accomplishment Instructions, Part I.

(2) Modify the electrical connections of factory-provisioned airplanes to add an additional indication device to the clear-ice indication system, as specified in the Accomplishment Instructions, Part II.

(3) Remove the "Clear-Ice Inoperative" placard and reactivate the clear-ice additional indicator lamp, as specified in the Accomplishment Instructions, Part III.

(4) Replace the existing clear-ice indicator lamp with a new, improved lamp having a new part number, as specified in the Accomplishment Instructions, Part IV or Part V.

Actions Accomplished Per Previous Issues of Service Bulletins

(c) Actions accomplished before the effective date of this AD in accordance with EMBRAER Service Bulletin 145-30-0035, Revision 02, dated January 06, 2005, are considered acceptable for compliance with the corresponding actions specified in this AD.

Alternative Methods of Compliance

(d)(1) In accordance with 14 CFR 39.19, the Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate, is authorized to approve alternative methods of compliance for this AD.

(2) Before using any AMOC approved in accordance with § 39.19 on any airplane to which the AMOC applies, notify the appropriate principal inspector in the FAA Flight Standards Certificate Holding District Office.

Note 1: The subject of this AD is addressed in Brazilian airworthiness directive 2004-01-01, dated January 27, 2004.

Issued in Renton, Washington, on May 15, 2006.

Kevin M. Mullin,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. E6-8117 Filed 5-25-06; 8:45 am]

BILLING CODE 4910-13-P

DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

Contact Boeing Commercial Airplanes, P.O. Box 3707, Seattle, Washington 98124-2207, for the service information identified in this proposed AD.

FOR FURTHER INFORMATION CONTACT: Gary Oltman, Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 917-6443; fax (425) 917-6590.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to submit any relevant written data, views, or arguments regarding this proposed AD. Send your comments to an address listed in the **ADDRESSES** section. Include the docket number "FAA-2006-24891; Directorate Identifier 2006-NM-080-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of the proposed AD. We will consider all comments received by the closing date and may amend the proposed AD in light of those comments.

We will post all comments we receive, without change, to <http://dms.dot.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact with FAA personnel concerning this proposed AD. Using the search function of that Web site, anyone can find and read the comments in any of our dockets, including the name of the individual who sent the comment (or signed the comment on behalf of an association, business, labor union, etc.). You may review DOT's complete Privacy Act Statement in the **Federal Register** published on April 11, 2000 (65 FR 19477-78), or you may visit <http://dms.dot.gov>.

Examining the Docket

You may examine the AD docket on the Internet at <http://dms.dot.gov>, or in person at the Docket Management Facility office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The Docket Management Facility office (telephone (800) 647-5227) is located on the plaza level of the Nassif Building at the DOT street address stated in the **ADDRESSES** section. Comments will be available in the AD docket shortly after the Docket Management System receives them.

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2006-24891; Directorate Identifier 2006-NM-080-AD]

RIN 2120-AA64

Airworthiness Directives; Boeing Model 777-200, -300, and -300ER Series Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: The FAA proposes to adopt a new airworthiness directive (AD) for certain Boeing Model 777-200, -300, and -300ER series airplanes. This proposed AD would require replacement of the gimbal plates of the left and right outboard trailing edge flaps with improved gimbal plates and other specified actions. This proposed AD results from a broken pivot link found on the inboard support for the outboard trailing edge flap. We are proposing this AD to prevent disconnection of the drive arm from its drive gimbal, due to a broken pivot link on an outboard flap support, which could result in unexpected roll of the airplane and loss of control of the airplane.

DATES: We must receive comments on this proposed AD by July 10, 2006.

ADDRESSES: Use one of the following addresses to submit comments on this proposed AD.

- DOT Docket Web site: Go to <http://dms.dot.gov> and follow the instructions for sending your comments electronically.

- Government-wide rulemaking Web site: Go to <http://www.regulations.gov> and follow the instructions for sending your comments electronically.

- Mail: Docket Management Facility, U.S. Department of Transportation, 400 Seventh Street SW., Nassif Building, room PL-401, Washington, DC 20590.

- Fax: (202) 493-2251.

- Hand Delivery: Room PL-401 on the plaza level of the Nassif Building, 400 Seventh Street SW., Washington,

Discussion

We have received a report indicating that a broken pivot link was found on the inboard support for the outboard trailing edge flap on a Boeing Model 777–300 series airplane. That broken pivot link was found after an incident where, during approach, the flightcrew received the FLAPS PRIMARY and FLAPS SKEW messages, and the airplane rolled slightly to the left. The flightcrew was able to land the airplane without difficulty. Investigation revealed that a broken pivot link on outboard flap support number 2 caused an increased load on the drive arm assembly of support number 1. The increased load caused the drive arm gimbal plates to disconnect from the drive gimbal, which led to a skewed outboard flap. This condition, if not corrected, could result in unexpected roll of the airplane and loss of control of the airplane.

Relevant Service Information

We have reviewed Boeing Alert Service Bulletin 777–27A0073, dated March 30, 2006. The service bulletin describes procedures for replacing the gimbal plates of the left and right outboard trailing edge flaps with improved gimbal plates and doing other specified actions. The other specified actions include adjusting the gimbal plate shims, rotating the upper gimbal bushing, installing a new grease fitting, adjusting the bulkhead fitting shim, changing the flap skew detection bracket assembly, lubricating the outboard transmission, ballscrews, and gimbal of the outboard flaps, and doing the adjustment/test of the trailing edge flap system. Accomplishing the actions specified in the service information is intended to adequately address the unsafe condition.

FAA's Determination and Requirements of the Proposed AD

We have evaluated all pertinent information and identified an unsafe condition that is likely to exist or develop on other airplanes of this same type design. For this reason, we are proposing this AD, which would require accomplishing the actions specified in the service information described previously.

Costs of Compliance

There are about 546 airplanes of the affected design in the worldwide fleet. This proposed AD would affect about 145 airplanes of U.S. registry. The proposed actions would take about 153 work hours per airplane, at an average labor rate of \$80 per work hour. Required parts would cost about

\$69,850 per airplane. Based on these figures, the estimated cost of the proposed AD for U.S. operators is \$11,903,050, or \$82,090 per airplane.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, Section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We have determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that the proposed regulation:

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
3. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this proposed AD and placed it in the AD docket. See the **ADDRESSES** section for a location to examine the regulatory evaluation.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. The Federal Aviation Administration (FAA) amends § 39.13 by adding the following new airworthiness directive (AD):

BOEING: Docket No. FAA-2006-24891; Directorate Identifier 2006-NM-080-AD.

Comments Due Date

(a) The FAA must receive comments on this AD action by July 10, 2006.

Affected ADs

(b) None.

Applicability

(c) This AD applies to Boeing Model 777–200, –300, and –300ER series airplanes, certificated in any category; as identified in Boeing Alert Service Bulletin 777–27A0073, dated March 30, 2006.

Unsafe Condition

(d) This AD results from a broken pivot link found on the inboard support for the outboard trailing edge flap. We are issuing this AD to prevent disconnection of the drive arm from its drive gimbal, due to a broken pivot link on an outboard flap support, which could result in unexpected roll of the airplane and loss of control of the airplane.

Compliance

(e) You are responsible for having the actions required by this AD performed within the compliance times specified, unless the actions have already been done.

Replacement of Gimbal Plates

(f) Within 24 months after the effective date of this AD, replace the gimbal plates of the left and right outboard trailing edge flaps with improved gimbal plates, and do the other specified actions before further flight after the replacement, by accomplishing all the actions specified in the Accomplishment Instructions of Boeing Alert Service Bulletin 777–27A0073, dated March 30, 2006.

Parts Installation

(g) As of the effective date of this AD, no person may install a gimbal plate, part numbers 113W1112–3, 113W1112–4, 113W1212–3, and 113W1212–4, on any airplane, unless it has been modified in accordance with paragraph (f) of this AD.

Alternative Methods of Compliance (AMOCs)

(h)(1) The Manager, Seattle Aircraft Certification Office (ACO), FAA, has the authority to approve AMOCs for this AD, if requested in accordance with the procedures found in 14 CFR 39.19.

(2) Before using any AMOC approved in accordance with § 39.19 on any airplane to which the AMOC applies, notify the appropriate principal inspector in the FAA

Flight Standards Certificate Holding District Office.

(3) An AMOC that provides an acceptable level of safety may be used for any repair required by this AD, if it is approved by an Authorized Representative for the Boeing Commercial Airplanes Delegation Option Authorization Organization who has been authorized by the Manager, Seattle ACO, to make those findings. For a repair method to be approved, the repair must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

Issued in Renton, Washington, on May 18, 2006.

Kevin M. Mullin,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. E6-8123 Filed 5-25-06; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2005-22812; Directorate Identifier 2005-NM-134-AD]

RIN 2120-AA64

Airworthiness Directives; Airbus Model A330 Airplanes and Model A340-200 and -300 Series Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Supplemental notice of proposed rulemaking (NPRM); reopening of comment period.

SUMMARY: The FAA is revising an earlier NPRM for an airworthiness directive (AD) that applies to certain Airbus Model A330 airplanes and Model A340-200 and -300 series airplanes. The original NPRM would have required repetitive detailed inspections for cracking in the aft web of support rib 6 between certain bottom skin stringers on both wings; high frequency eddy current inspections for cracking of the attachment holes of the fuel pipes, and repair if necessary. The original NPRM also would have provided for an optional modification, which would extend a certain inspection threshold. The original NPRM resulted from a report of significant cracking found in the aft web of support rib 6 on both wings. This action revises the original NPRM by mandating, for certain airplanes, a new modification of support rib 6 on both wings, which would end the repetitive inspection requirement. This action also reduces the applicability in the original NPRM. We are proposing this supplemental NPRM

to prevent cracking in the aft web of support rib 6, which could result in overloading of adjacent ribs and the surrounding wing structure and consequent reduced structural integrity of the wing.

DATES: We must receive comments on this supplemental NPRM by June 20, 2006.

ADDRESSES: Use one of the following addresses to submit comments on this supplemental NPRM.

- DOT Docket Web site: Go to <http://dms.dot.gov> and follow the instructions for sending your comments electronically.

- Government-wide rulemaking Web site: Go to <http://www.regulations.gov> and follow the instructions for sending your comments electronically.

- Mail: Docket Management Facility, U.S. Department of Transportation, 400 Seventh Street SW., Nassif Building, Room PL-401, Washington, DC 20590.

- Fax: (202) 493-2251.

- Hand Delivery: Room PL-401 on the plaza level of the Nassif Building, 400 Seventh Street SW., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

Contact Airbus, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France, for the service information identified in this proposed AD.

FOR FURTHER INFORMATION CONTACT: Tim Backman, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-2797; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to submit any relevant written data, views, or arguments regarding this supplemental NPRM. Send your comments to an address listed in the **ADDRESSES** section. Include the docket number “Docket No. FAA-2005-22812; Directorate Identifier 2005-NM-134-AD” at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this supplemental NPRM. We will consider all comments received by the closing date and may amend this supplemental NPRM in light of those comments.

We will post all comments submitted, without change, to <http://dms.dot.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact with FAA personnel concerning this supplemental NPRM. Using the search function of that Web site, anyone

can find and read the comments in any of our dockets, including the name of the individual who sent the comment (or signed the comment on behalf of an association, business, labor union, etc.). You may review the DOT's complete Privacy Act Statement in the **Federal Register** published on April 11, 2000 (65 FR 19477-78), or you may visit <http://dms.dot.gov>.

Examining the Docket

You may examine the AD docket on the Internet at <http://dms.dot.gov>, or in person at the Docket Management Facility office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The Docket Management Facility office (telephone (800) 647-5227) is located on the plaza level in the Nassif Building at the DOT street address stated in **ADDRESSES**. Comments will be available in the AD docket shortly after the Docket Management System receives them.

Discussion

We proposed to amend 14 CFR part 39 with a notice of proposed rulemaking (NPRM) for an airworthiness directive (AD) (the “original NPRM”). The original NPRM applies to certain Airbus Model A330-200, A330-300, A340-200, and A340-300 series airplanes. The original NPRM was published in the **Federal Register** on October 27, 2005 (70 FR 61927). The original NPRM proposed to require repetitive detailed inspections for cracking in the aft web of support rib 6 between certain bottom skin stringers on both wings; high frequency eddy current inspections for cracking of the attachment holes of the fuel pipes, and repair if necessary. The original NPRM also proposed to provide for an optional modification, which would extend a certain inspection threshold.

The preamble to the original NPRM specified that we considered the requirements “interim action” and that the manufacturer was developing a modification to address the unsafe condition. The preamble also explained that we may consider further rulemaking if a modification is developed, approved, and available. The manufacturer now has developed such a modification, and we have determined that further rulemaking is indeed necessary; this supplemental NPRM follows from that determination.

New Relevant Service Information

Airbus has issued Service Bulletins A330-57-3085 (for Model A330 airplanes) and A340-57-4093 (for Model A340-200 and -300 series airplanes), both Revision 02, both dated