

determined that a stricter compliance schedule is necessary. The European Aviation Safety Agency issued AD 2004-0007, on December 15, 2004, which required compliance by December 14, 2005. However, due to insufficient parts availability, we propose to require compliance no later than June 30, 2007.

FAA's Determination and Requirements of the Proposed AD

We have evaluated all pertinent information and identified an unsafe condition that is likely to exist or develop on other engines of this same type design. We are proposing this AD, which would require reworking certain CF6-50C forward fan stator cases and installing secondary containment shields at the next engine shop visit after the effective date of the proposed AD, but no later than June 30, 2007.

The proposed AD would require you to use the service information described previously to perform these actions.

Costs of Compliance

We estimate that this proposed AD would affect 226 CF6-50C series turbofan engines installed on airplanes of U.S. registry. We also estimate that it would take about 2.5 work hours per engine to perform the proposed actions, and that the average labor rate is \$80 per work hour. Required parts would cost about \$9,451 per engine. Based on these figures, we estimate the total cost of the proposed AD to U.S. operators to be \$2,181,126.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in subtitle VII, part A, subpart III, section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We have determined that this proposed AD would not have federalism

implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that the proposed regulation:

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
3. Would not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this proposed AD. See the **ADDRESSES** section for a location to examine the regulatory evaluation.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Under the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. The FAA amends § 39.13 by adding the following new airworthiness directive:

General Electric Company: Docket No. FAA-2006-24171; Directorate Identifier 2006-NE-08-AD.

Comments Due Date

- (a) The Federal Aviation Administration (FAA) must receive comments on this airworthiness directive (AD) action by June 16, 2006.

Affected ADs

- (b) None.

Applicability

- (c) This AD applies to General Electric Company (GE) CF6-50C, CF6-50C1, CF6-50C2, CF6-50C2B, CF6-50C2F, and CF6-50C2R turbofan engines, with a forward fan stator case, part number 9064M53G04, GO5, G06, G07, G08, G09, G10, G12, or G13, installed. These engines are installed on, but not limited to, Airbus A300, McDonnell Douglas DC-10 series, and DC-10-30F (KC-10A, KDC-10) airplanes.

Unsafe Condition

(d) This AD results from reports of uncontained fan blade failures causing damage and separation of airplane hydraulic lines. We are issuing this AD to prevent uncontained fan blade failures, which can result in separation of airplane hydraulic lines, damage to critical airplane systems, and possible loss of airplane control.

Compliance

(e) You are responsible for having the actions required by this AD performed within the compliance times specified unless the actions have already been done.

(f) At the next engine shop visit after the effective date of this AD, but no later than June 30, 2007, rework the forward fan stator case and install the fan module secondary containment shield.

(1) For engines on Airbus 300 series airplanes, use paragraph 3, Accomplishment Instructions, of GE SB No. CF6-50 S/B 72-0985, dated December 2, 1991 or Revision 1, dated September 15, 1998, to do the rework and installation.

(2) For engines on McDonnell Douglas airplanes, use paragraph 3, Accomplishment Instructions, of GE Service Bulletin (SB) No. CF6-50 S/B 72-0986, dated December 2, 1991 or Revision 1, dated September 15, 1998, to do the rework and installation.

(g) The rework and installation specified in paragraphs (f)(1) through (f)(2) of this AD can also be done on-wing.

Alternative Methods of Compliance

(h) The Manager, Engine Certification Office, has the authority to approve alternative methods of compliance for this AD if requested using the procedures found in 14 CFR 39.19.

Related Information

(i) European Aviation Safety Agency airworthiness directive 2004-0007, dated December 15, 2004, also addresses the subject of this AD.

Issued in Burlington, Massachusetts, on April 11, 2006.

Francis A. Favara,

Manager, Engine and Propeller Directorate, Aircraft Certification Service.

[FR Doc. E6-5645 Filed 4-14-06; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 97-ANE-09]

Airworthiness Directives; Rolls-Royce plc RB211 Trent 800 Series Turbofan Engines

AGENCY: Federal Aviation Administration, DOT.

ACTION: Proposed rule; withdrawal.

SUMMARY: This action withdraws a notice of proposed rulemaking (NPRM).

That NPRM proposed to revise the existing airworthiness directive (AD) applicable to Rolls-Royce plc (RR) Model RB211 Trent 892, 884, 877, 875, and 892B series turbofan engines. That proposed rule would have required initial and repetitive inspections of the angled drive upper shroud, the intermediate gearbox housing (IGH), and the external gearbox lower bevel box (LBB) housing, and initial and repetitive master magnetic chip detector (MCD) inspections. Since we issued that proposed rule, RR notified us that after reviewing the service experience and the original actions taken, the unsafe condition no longer exists and mandatory actions required by the proposed rule are no longer required. Accordingly, we withdraw the proposed rule.

FOR FURTHER INFORMATION CONTACT:

Christopher Spinney, Aerospace Engineer, Engine Certification Office, FAA, Engine and Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803; telephone (781) 238-7175; fax (781) 238-7199.

SUPPLEMENTARY INFORMATION: The FAA proposed to amend 14 CFR part 39 with a proposed AD. The proposed AD applies to RR Model RB211 Trent 892, 884, 877, 875, and 892B series turbofan engines. We published the proposed AD in the **Federal Register** on November 2, 1999 (64 FR 59137). That proposed action would have required initial and repetitive inspections of the angled drive upper shroud, the intermediate gearbox housing (IGH), and the external gearbox lower bevel box (LBB) housing. In addition, that proposed AD would have required initial and repetitive master magnetic chip detector (MCD) inspections. Also, that proposed AD would have eliminated the repetitive inspections of the IGH, external gearbox LBB housing, and the angled drive upper shroud, if the engines have incorporated modifications described in certain RR service bulletins. Also, that proposed AD would have increased the inspection interval for repetitive master MCD inspections. That proposed AD resulted from service experience since publication of AD 97-06-13. We proposed that AD to prevent loss of oil, which could cause an engine fire, and in-flight engine shutdowns and airplane diversions caused by oil loss and bearing failures.

Since we issued that proposed AD, RR notified us that after reviewing the service experience and the original actions taken, the unsafe condition no longer exists and mandatory actions required by the proposed AD are no longer required.

Upon further consideration, we hereby withdraw the proposed rule based on RR's analysis and conclusion stated above.

Withdrawal of this notice of proposed rulemaking constitutes only such action, and does not preclude the agency from issuing another notice in the future, nor does it commit the agency to any course of action in the future.

Since this action only withdraws a notice of proposed rulemaking, it is neither a proposed nor a final rule. Executive Order 12866, the Regulatory Flexibility Act, or DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979) do not cover this withdrawal.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Withdrawal

Accordingly, we withdraw the notice of proposed rulemaking, Docket No. 97-ANE-09, published in the **Federal Register** on November 2, 1999 (64 FR 59137).

Issued in Burlington, Massachusetts, on April 11, 2006.

Francis A. Favara,

Manager, Engine and Propeller Directorate, Aircraft Certification Service.

[FR Doc. E6-5666 Filed 4-14-06; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. **FAA-2005-21242**; Directorate Identifier **2005-NE-09-AD**]

RIN 2120-AA64

Airworthiness Directives; Turbomeca Arriel 1B, 1D, 1D1, and 1S1 Turboshaft Engines

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: The FAA proposes to revise an existing airworthiness directive (AD) for certain Turbomeca Arriel 1B, 1D, 1D1, and 1S1 turboshaft engines. That AD currently requires initial and repetitive position checks of the gas generator 2nd stage turbine blades on all Turbomeca Arriel 1B, 1D, 1D1, and 1S1 turboshaft engines. That AD also currently requires initial and repetitive replacements of 2nd stage turbines on

1B, 1D, and 1D1 engines only. This proposed AD revision would require the same actions, but would relax the compliance times for initially replacing 2nd stage turbines in Arriel 1B, 1D, and 1D1 turboshaft engines. This proposed AD revision results from a request by Turbomeca to clarify the compliance times for 2nd stage turbine initial replacement on Arriel 1D, 1D1, and 1B turboshaft engines. We are proposing this AD revision to clarify and relax the AD compliance times for 2nd stage turbine initial replacement on Arriel 1B, 1D, and 1D1 turboshaft engines. We are also proposing this AD revision to prevent inflight engine shutdown and subsequent forced autorotation landing or accident.

DATES: We must receive any comments on this proposed AD revision by June 16, 2006.

ADDRESSES: Use one of the following addresses to comment on this proposed AD revision.

- DOT Docket Web site: Go to <http://dms.dot.gov> and follow the instructions for sending your comments electronically.

- Government-wide rulemaking Web site: Go to <http://www.regulations.gov> and follow the instructions for sending your comments electronically.

- Mail: Docket Management Facility; U.S. Department of Transportation, 400 Seventh Street, SW., Nassif Building, Room PL-401, Washington, DC 20590-0001.

- Fax: (202) 493-2251.
- Hand Delivery: Room PL-401 on the plaza level of the Nassif Building, 400 Seventh Street, SW., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

Contact Turbomeca, 40220 Tarnos, France; telephone +33 05 59 74 40 00, fax +33 05 59 74 45 15, for the service information identified in this proposed AD revision.

FOR FURTHER INFORMATION CONTACT:

Christopher Spinney, Aerospace Engineer, Engine Certification Office, FAA, Engine and Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803; telephone (781) 238-7175, fax (781) 238-7199.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to send any written relevant data, views, or arguments regarding this proposal. Send your comments to an address listed under **ADDRESSES**. Include "Docket No. **FAA-2005-21242**; Directorate Identifier **2005-NE-09-AD**" in the subject line of your comments. We specifically invite comments on the overall regulatory,