

**PART 39—AIRWORTHINESS DIRECTIVES**

■ 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

**§ 39.13 [Amended]**

■ 2. FAA amends § 39.13 by adding a new AD to read as follows:

**2005–08–06 Centrair:** Amendment 39–14058; Docket No. FAA–2004–19616; Directorate Identifier 2004–CE–38–AD.

Actions	Compliance	Procedures
(1) Replace any installed elevator and aileron hinge pins that are not part number (P/N) SY991A hinge pins with P/N SY991A hinge pins.	Within the next 25 hours time-in-service (TIS) after June 2, 2005 (the effective date of this AD), unless already done.	Follow Société Nouvelle Centrair Service Bulletin No. 101–22, dated March 13, 2001.
(2) Do not install any elevator and aileron hinge pins that are not P/N SY991A hinge pins as specified in paragraph (e)(1) of this AD.	As of June 2, 2005 (the effective date of this AD).	Not Applicable.

**May I Request an Alternative Method of Compliance?**

(f) You may request a different method of compliance or a different compliance time for this AD by following the procedures in 14 CFR 39.19. Unless FAA authorizes otherwise, send your request to your principal inspector. The principal inspector may add comments and will send your request to the Manager, Small Airplane Directorate, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329–4130; facsimile: (816) 329–4090.

**Is There Other Information That Relates to This Subject?**

(g) French AD Number 2001–247(A), dated June 27, 2001, also addresses the subject of this AD.

**Does This AD Incorporate Any Material by Reference?**

(h) You must do the actions required by this AD following the instructions in Société Nouvelle Centrair Service Bulletin No. 101–22, dated March 13, 2001. The Director of the Federal Register approved the incorporation by reference of this service bulletin in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. To get a copy of this service information, contact CENTRAIR, Aerodome B.P.N. 44, 36300 Le Blanc, France; telephone: 02.54.37.07.96; facsimile: 02.54.37.48.64. To review copies of this service information, go to the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, go to: [http://www.archives.gov/federal\\_register/code\\_of\\_federal\\_regulations/ibr\\_locations.html](http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html) or call (202) 741–6030. To view the AD docket, go to the Docket Management Facility; U.S. Department of Transportation, 400 Seventh Street, SW., Nassif Building, Room PL–401, Washington, DC 20590–001 or on the Internet at <http://dms.dot.gov>.

**When Does This AD Become Effective?**

(a) This AD becomes effective on June 2, 2005.

**What Other ADs Are Affected by This Action?**

(b) None.

**What Gliders Are Affected by This AD?**

(c) This AD affects Models 101, 101A, 101AP, and 101P gliders, all serial numbers, without elevator and aileron part number SY991A hinge pins installed, certificated in any category.

**What Is the Unsafe Condition Presented in This AD?**

(d) This AD is the result of mandatory continuing airworthiness information (MCAI) issued by the airworthiness authority for France. The actions specified in this AD are intended to replace incorrectly heat-treated elevator or aileron hinge pins, which could result in failure of the elevator or ailerons. Such failure during takeoff, landing, or flight operations could lead to loss of glider control.

(e) To address this problem, you must do the following:

[dms.dot.gov](http://dms.dot.gov). The docket number is FAA–2004–19616.

Issued in Kansas City, Missouri, on April 11, 2005.

**Nancy C. Lane,**

*Acting Manager, Small Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 05–7564 Filed 4–18–05; 8:45 am]

**BILLING CODE 4910–13–P**

**DEPARTMENT OF TRANSPORTATION****Federal Aviation Administration****14 CFR Part 39**

**[Docket No. FAA–2005–20136; Directorate Identifier 2004–NM–185–AD; Amendment 39–14061; AD 2005–08–09]**

**RIN 2120–AA64**

**Airworthiness Directives; Boeing Model 747–200B, –200C, –200F, and –400F Series Airplanes**

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Final rule.

**SUMMARY:** The FAA is adopting a new airworthiness directive (AD) for certain Boeing Model 747 series airplanes. This AD requires repetitive detailed inspections for cracks in the crease beam and adjacent structure of the fuselage, and related investigative and corrective actions if necessary. This AD is prompted by fatigue cracks found in the crease beam during a follow-on inspection of a previously installed modification. We are issuing this AD to find and fix fatigue cracking of the fuselage frame, which could result in

reduced structural integrity of the frame and consequent rapid decompression of the airplane.

**DATES:** This AD becomes effective May 24, 2005.

The incorporation by reference of a certain publication listed in the AD is approved by the Director of the Federal Register as of May 24, 2005.

**ADDRESSES:** The service information referenced in this AD may be obtained from Boeing Commercial Airplanes, P.O. Box 3707, Seattle, Washington 98124–2207.

**Docket:** The AD docket contains the proposed AD, comments, and any final disposition. You can examine the AD docket on the Internet at <http://dms.dot.gov>, or in person at the Docket Management Facility office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The Docket Management Facility office (telephone (800) 647–5227) is located on the plaza level of the Nassif Building at the U.S. Department of Transportation, 400 Seventh Street SW., room PL–401, Washington, DC. This docket number is FAA–2005–20136; the directorate identifier for this docket is 2005–NM–185–AD.

**FOR FURTHER INFORMATION CONTACT:** Nick Kusz, Aerospace Engineer, Airframe Branch, ANM–120S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98055–4056; telephone (425) 917–6432; fax (425) 917–6590.

**SUPPLEMENTARY INFORMATION:** The FAA proposed to amend 14 CFR Part 39 with an AD for certain Boeing Model 747 series airplanes. That action, published in the **Federal Register** on January 28,

2005 (70 FR 4048), proposed to require repetitive detailed inspections for cracks in the crease beam and adjacent structure of the fuselage, and related investigative and corrective actions if necessary.

### Comments

We provided the public the opportunity to participate in the development of this AD. We have considered the single comment that has been submitted on the proposed AD. The commenter supports the proposed AD.

### Conclusion

We have carefully reviewed the available data, including the comment that has been submitted, and determined that air safety and the public interest require adopting the AD as proposed.

### Costs of Compliance

There are about 163 airplanes of the affected design in the worldwide fleet. This AD will affect about 30 airplanes of U.S. registry. The inspection will take about 8 work hours per airplane, at an average labor rate of \$65 per work hour. Based on these figures, the estimated cost of the inspection for U.S. operators is \$15,600, or \$520 per airplane, per inspection cycle.

### Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, Section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

### Regulatory Findings

We have determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and

responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866;
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
- (3) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this AD. See the **ADDRESSES** section for a location to examine the regulatory evaluation.

### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

### Adoption of the Amendment

■ Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

### PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

■ 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

**2005-08-09 Boeing:** Amendment 39-14061.  
Docket No. FAA-2005-20136;  
Directorate Identifier 2004-NM-185-AD.

#### Effective Date

(a) This AD becomes effective May 24, 2005.

#### Affected ADs

(b) None.

#### Applicability

(c) This AD applies to Boeing Model 747-200B, -200C, -200F, and -400F series airplanes, line numbers 604 and subsequent, certificated in any category; as listed in Boeing Alert Service Bulletin 747-53A2504, dated August 19, 2004.

#### Unsafe Condition

(d) This AD was prompted by fatigue cracks found in the crease beam during a follow-on inspection of a previously installed modification. We are issuing this AD to find and fix fatigue cracking of the fuselage frame, which could result in reduced structural integrity of the frame and consequent rapid decompression of the airplane.

### Compliance

(e) You are responsible for having the actions required by this AD performed within the compliance times specified, unless the actions have already been done.

### Repetitive Inspections

(f) Accomplish a detailed inspection for cracks in the crease beam and adjacent structure of the fuselage by doing all the applicable actions in accordance with the Accomplishment Instructions of Boeing Alert Service Bulletin 747-53A2504, dated August 19, 2004; at the applicable time specified in paragraph (f)(1) or (f)(2) of this AD. Repeat the inspection thereafter at intervals not to exceed 6,000 flight cycles.

(1) For Groups 1 and 2 airplanes as identified in the service bulletin: Before the accumulation of 10,000 total flight cycles, or within 1,500 flight cycles after the effective date of this AD, whichever is later.

(2) For Groups 3 and 4 airplanes as identified in the service bulletin: Before the accumulation of 14,000 total flight cycles, or within 1,500 flight cycles after the effective date of this AD, whichever is later.

### Related Investigative and Corrective Actions

(g) If any crack is found during any inspection required by paragraph (f) of this AD: Before further flight, repair the cracking in accordance with the Accomplishment Instructions of Boeing Alert Service Bulletin 747-53A2504, dated August 19, 2004. If cracking of the crease beam or outer tee chord attachment is found: Before further flight, do a high frequency eddy current inspection for additional cracking, and repair any cracking found, in accordance with the service bulletin. Where the service bulletin specifies contacting the manufacturer for disposition of certain repair conditions, repair before further flight in accordance with a method approved by the Manager, Seattle Aircraft Certification Office (ACO), FAA; or by an Authorized Representative for the Boeing Delegation Option Authorization (DOA) Organization, who has been authorized by the Manager, Seattle ACO, to make those findings. For a repair method to be approved, the repair must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

### No Reporting Required

(h) For certain airplanes, the service bulletin referenced in this AD recommends reporting any discrepancies to the manufacturer, but this AD does not include that requirement.

### Alternative Methods of Compliance (AMOCs)

(i)(1) The Manager, Seattle ACO, has the authority to approve AMOCs for this AD, if requested in accordance with the procedures found in 14 CFR 39.19.

(2) An AMOC that provides an acceptable level of safety may be used for a repair required by this AD, if it is approved by an Authorized Representative for the Boeing DOA Organization who has been authorized by the Manager, Seattle ACO, to make such findings.

**Material Incorporated by Reference**

(j) You must use Boeing Alert Service Bulletin 747-53A2504, dated August 19, 2004, to perform the actions that are required by this AD, unless the AD specifies otherwise. The Director of the Federal Register approves the incorporation by reference of this document in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. To get copies of the service information, go to Boeing Commercial Airplanes, P.O. Box 3707, Seattle, Washington 98124-2207. To view the AD docket, go to the Docket Management Facility, U.S. Department of Transportation, 400 Seventh Street SW., room PL-401, Nassif Building, Washington, DC. To review copies of the service information, go to the National Archives and Records Administration (NARA). For information on the availability of this material at the NARA, call (202) 741-6030, or go to [http://www.archives.gov/federal\\_register/code\\_of\\_federal\\_regulations/ibr\\_locations.html](http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html).

Issued in Renton, Washington, on April 11, 2005.

**Ali Bahrami,**

Manager, Transport Airplane Directorate,  
Aircraft Certification Service.

[FR Doc. 05-7683 Filed 4-18-05; 8:45 am]

**BILLING CODE 4910-13-P**

**DEPARTMENT OF TRANSPORTATION****Federal Aviation Administration****14 CFR Part 39**

[Docket No. FAA-2004-19810; Directorate Identifier 2004-NM-119-AD; Amendment 39-14062; AD 2005-08-10]

**RIN 2120-AA64**

**Airworthiness Directives; Boeing Model 737-600, -700, and -800 Series Airplanes**

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Final rule.

**SUMMARY:** The FAA is adopting a new airworthiness directive (AD) for certain Boeing Model 737-600, -700, and -800 series airplanes. This AD requires doing a general visual inspection for sealant at the interface of the upper spar fittings, strut side skins, and the fittings of the thrust reverser strut fairing on the engine struts; and applying an injection seal or silicone sponge rubber with fillet seal if necessary. This AD is prompted by a report that an injection seal in the engine strut area may not have been properly completed or installed during production. We are issuing this AD to prevent flammable fluid (such as fuel or hydraulic fluid) from leaking onto a hot engine exhaust nozzle or into the engine

core fire zone, and consequently causing an uncontrolled fire or explosion.

**DATES:** This AD becomes effective May 24, 2005.

The incorporation by reference of certain publications listed in the AD is approved by the Director of the Federal Register as of May 24, 2005.

**ADDRESSES:** For service information identified in this AD, contact Boeing Commercial Airplanes, P.O. Box 3707, Seattle, Washington 98124-2207.

**Docket:** The AD docket contains the proposed AD, comments, and any final disposition. You can examine the AD docket on the Internet at <http://dms.dot.gov>, or in person at the Docket Management Facility office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The Docket Management Facility office (telephone (800) 647-5227) is located on the plaza level of the Nassif Building at the U.S. Department of Transportation, 400 Seventh Street SW., room PL-401, Washington, DC. This docket number is FAA-2004-19810; the directorate identifier for this docket is 2004-NM-119-AD.

**FOR FURTHER INFORMATION CONTACT:**

Doug Pegors, Aerospace Engineer, Propulsion Branch, ANM-140S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 917-6504; fax (425) 917-6590.

**SUPPLEMENTARY INFORMATION:** The FAA proposed to amend 14 CFR Part 39 with an AD for certain Boeing Model 737-600, -700, and -800 series airplanes. That action, published in the **Federal Register** on December 14, 2004 (69 FR 74465), proposed to require doing a general visual inspection for sealant at the interface of the upper spar fittings, strut side skins, and the fittings of the thrust reverser strut fairing on the engine struts; and applying an injection seal or silicone sponge rubber with fillet seal if necessary.

**Comments**

We provided the public the opportunity to participate in the development of this AD. We have considered the comments that have been submitted on the proposed AD. Three commenters support the proposed AD.

**Request for Shortening the Compliance Time**

A commenter supports the proposed AD, but requests that the compliance time of 18 months or 3,500 flight cycles be shortened. The commenter suggests that, due to the low cost of modifying an airplane, short repair time, and the

potential severity of a failure, the compliance time is too long.

We do not agree with the commenter's suggestion. In developing an appropriate compliance time, we considered the safety implications and normal maintenance schedules for timely accomplishment of the required inspection and repair. Further, we arrived at the compliance time with operator and manufacturer concurrence. In consideration of all of these factors, we determined that the compliance time, represents an appropriate interval in which the engine nacelle struts can be inspected, and repaired if required, in a timely manner within the fleet, while still maintaining an adequate level of safety. Operators are always permitted to accomplish the requirements of an AD at a time earlier than the specified compliance time. If additional data are presented that would justify a shorter compliance time, we may consider further rulemaking on this issue.

**Conclusion**

We have carefully reviewed the available data, including the comments that have been submitted, and determined that air safety and the public interest require adopting the AD as proposed.

**Costs of Compliance**

There are about 257 airplanes worldwide of the affected design. This AD will affect about 99 airplanes of U.S. registry. The inspection will take about 2 work hours per airplane, at an average labor rate of \$65 per work hour. Based on these figures, the estimated cost of the AD for U.S. operators is \$12,870, or \$130 per airplane.

**Authority for This Rulemaking**

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, Section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on