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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2005-20107; Directorate Identifier 2005-SW-02-AD; Amendment 39-13981; AD 2005-04-09]

RIN 2120-AA64

Airworthiness Directives; Bell Helicopter Textron Canada Model 222, 222B, 222U, 230, and 430 Helicopters

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule; request for comments.

SUMMARY: This amendment supersedes an existing airworthiness directive (AD) for the specified Bell Helicopter Textron Canada (BHTC) model helicopters that currently requires certain checks and inspections of the tail rotor blades. If a crack is found, the existing AD requires replacing the tail rotor blade (blade) with an airworthy blade before further flight. This amendment requires the same checks and inspections as the existing AD, but expands the applicability with the addition of two BHTC Model 430 helicopter serial numbers. This amendment is prompted by the manufacturer issuing revised service information that includes the additional two serial numbers. The actions specified by this AD are intended to detect a crack in the blade, and to prevent loss of a blade and subsequent loss of control of the helicopter.

DATES: Effective March 4, 2005.

Comments for inclusion in the Rules Docket must be received on or before April 18, 2005.

ADDRESSES: Use one of the following addresses to submit comments on this AD:

- DOT Docket Web site: Go to <http://dms.dot.gov> and follow the instructions for sending your comments electronically;

- Government-wide rulemaking Web site: Go to <http://www.regulations.gov> and follow the instructions for sending your comments electronically;

- *Mail:* Docket Management Facility; U.S. Department of Transportation, 400 Seventh Street SW., Nassif Building, Room PL-401, Washington, DC 20590;
- *Fax:* (202) 493-2251; or
- *Hand Delivery:* Room PL-401 on the plaza level of the Nassif Building, 400 Seventh Street SW., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

You may get the service information identified in this AD from Bell Helicopter Textron Canada, 12,800 Rue de l'Avenir, Mirabel, Quebec J7J1R4, telephone (450) 437-2862 or (800) 363-8023, fax (450) 433-0272. You may examine this information at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call (202) 741-6030, or go to: http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

EXAMINING THE DOCKET: You may examine the docket that contains the AD, any comments, and other information on the Internet at <http://dms.dot.gov>, or in person at the Docket Management System (DMS) Docket Offices between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The Docket Office (telephone (800) 647-5227) is located on the plaza level of the Department of Transportation Nassif Building at the street address stated in the **ADDRESSES** section. Comments will be available in the AD docket shortly after the DMS receives them.

FOR FURTHER INFORMATION CONTACT: Sharon Miles, Aviation Safety Engineer, FAA, Rotorcraft Directorate, Regulations and Policy Group, Fort Worth, Texas 76193-0111, telephone (817) 222-5122, fax (817) 222-5961.

SUPPLEMENTARY INFORMATION: On December 23, 2004, the FAA issued AD 2004-26-11, Amendment 39-13923 (70 FR 7; January 3, 2005), to require certain checks and inspections of the blades. If a crack is found, that AD requires replacing the blade with an airworthy blade before further flight. That action

was prompted by three reports of cracked blades that were found during scheduled inspections. That condition, if not corrected, could result in loss of a blade and subsequent loss of control of the helicopter.

Since issuing that AD, the alert service bulletin (ASB) that is applicable to BHTC Model 430 helicopters has been revised by the manufacturer to include two additional helicopter serial numbers. Further, we discovered two typographical errors in the AD—the word “Canada” is inadvertently omitted from the manufacturer’s name in the Summary section, and in Note 1 of the AD, the number for the Model 430 helicopter ASB is incorrectly stated as 430-04-32 instead of 430-04-31—as well as some minor editorial errors, which have been corrected in this AD.

Transport Canada, the airworthiness authority for Canada, notified the FAA that an unsafe condition may exist on the specified BHTC model helicopters. Transport Canada advises of the discovery of cracked blades during scheduled inspections on three occasions. Two cracks originated from the outboard feathering bearing bore underneath the flanged sleeves. The third crack started from the inboard feathering bearing bore. Investigation found that the cracks originated from either a machining burr or a corrosion site in the bearing bore underneath the flanged sleeves.

BHTC has issued ASB No. 222-04-100 for Model 222 and 222B helicopters; ASB No. 222U-04-71 for Model 222U helicopters; and ASB No. 230-04-31 for Model 230 helicopters, all dated August 27, 2004; and, ASB No. 430-04-31, Revision A, dated November 29, 2004, for Model 430 helicopters. The ASBs specify a visual inspection of the blade root end around the feather bearings for a crack, not later than at the next scheduled inspection, and thereafter at intervals not to exceed 3 flight hours. Further, they describe a visual inspection for a crack, to include removing the blade from the helicopter, within 50 flight hours, and thereafter at intervals not to exceed 50 flight hours. Transport Canada classified these ASBs as mandatory and issued AD CF-2004-21R1, dated December 9, 2004, to ensure the continued airworthiness of these helicopters in Canada. This AD differs from those ASBs in that it requires an initial visual check, which may be

performed by a pilot, within 3 hours time-in-service (TIS) rather than a visual inspection not later than at the next scheduled inspection and every 3 flight hours maximum thereafter as stated in the ASBs.

These helicopter models are manufactured in Canada and are type certificated for operation in the United States under the provisions of 14 CFR 21.29 and the applicable bilateral agreement. Pursuant to the applicable bilateral agreement, Transport Canada has kept the FAA informed of the situation described above. The FAA has examined the findings of Transport Canada, reviewed all available information, and determined that AD action is necessary for products of these type designs that are certificated for operation in the United States.

This previously described unsafe condition is likely to exist or develop on other helicopters of the same type design. Therefore, this AD supersedes AD 2004-26-11 to require the following:

- Within 3 hours TIS, and thereafter at intervals not to exceed 3 hours TIS, clean and visually check both sides of each blade for a crack in the area around the tail rotor feathering bearing. An owner/operator (pilot) may perform this check. Pilots may perform the checks required by paragraph (a) of this AD because they require no tools, can be done by observation, and can be done equally well by a pilot or a mechanic. However, the pilot must enter compliance with these requirements into the helicopter maintenance records by following 14 CFR 43.11 and 91.417(a)(2)(v).

- Within 50 hours TIS, and thereafter at intervals not to exceed 50 hours TIS, clean and inspect both sides of each blade for a crack using a 10X or higher magnifying glass.

- If a crack is found in the blade paint during a visual check or inspection, further inspect the blade as follows, before further flight:

- Remove the blade. Remove the paint to the bare metal in the area of the suspected crack by using plastic metal blasting (PMB) or a nylon web abrasive pad and abrading the blade surface in a span-wise direction only.

- Using a 10X or higher power magnifying glass, inspect the blade for a crack.

- If a crack is found, replace the blade with an airworthy blade before further flight.

- If no crack is found in the blade surface, refinish the blade by applying one coat of epoxy polyamide primer, MIL-P-23377 or MIL-P-85582, so that the primer overlaps the existing coats

just beyond the abraded area. Let the area dry for 30 minutes to 1 hour. Then, apply one sealer coat of polyurethane, MILC85285 TYI CL2, color number 27925 (semi-gloss white), per Fed. Std. 595, and reinstall the blade.

This AD is an interim action, pending release of additional service information from the manufacturer concerning instructions for inspecting and reworking the affected blades. We expect that service information to eliminate the recurring inspections required by this AD.

The short compliance time involved is required because the previously described critical unsafe condition can adversely affect the controllability and structural integrity of the helicopter. Therefore, checking the blade for a crack within 3 hours TIS, and thereafter at intervals not to exceed 3 hours TIS, is required, and this AD must be issued immediately.

Since a situation exists that requires the immediate adoption of this regulation, it is found that notice and opportunity for prior public comment hereon are impracticable, and that good cause exists for making this amendment effective in less than 30 days.

We estimate that this AD will affect 156 helicopters and will require:

- 0.25 work hour for a pilot check, and 2 work hours for a maintenance inspection, at an average labor rate of \$65 per work hour; and
- Parts, which will cost an estimated \$13,410 per helicopter.

Based on these figures, the estimated total cost impact of the AD on U.S. operators is \$2,842,320 per year, assuming each helicopter will require 200 pilot checks, 12 maintenance inspections, and one blade replacement per year.

Comments Invited

This AD is a final rule that involves requirements that affect flight safety and was not preceded by notice and an opportunity for public comment; however, we invite you to submit any written data, views, or arguments regarding this AD. Send or deliver your comments to an address listed under **ADDRESSES**. Include "Docket No. FAA-2005-20107; Directorate Identifier 2005-SW-02-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of the AD. We will consider all comments received by the closing date and may amend the AD in light of those comments.

We will post all comments we receive, without change, to <http://dms.dot.gov>, including any personal

information you provide. We will also post a report summarizing each substantive verbal contact with FAA personnel concerning this AD. Using the search function of our docket web site, you can find and read the comments to any of our dockets, including the name of the individual who sent the comment. You may review the DOT's complete Privacy Act Statement in the **Federal Register** published on April 11, 2000 (65 FR 19477-78), or you may visit <http://dms.dot.gov>.

Regulatory Findings

We have determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that the regulation:

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
3. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared an economic evaluation of the estimated costs to comply with this AD. See the DMS to examine the economic evaluation.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in subtitle VII, part A, subpart III, section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

Adoption of the Amendment

■ Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. Section 39.13 is amended by removing Amendment 39–13923 (70 FR 7, January 3, 2005), and by adding a new airworthiness directive (AD), Amendment 39–13981, to read as follows:

2005–04–09 Bell Helicopter Textron
Canada: Amendment 39–13981. Docket

No. FAA–2005–20107; Directorate Identifier 2005–SW–02–AD. Supersedes AD 2004–26–11, Amendment 39–13923, Docket No. FAA–2004–19969, Directorate Identifier 2004–SW–43–AD.

Applicability: The following helicopter models, identified by serial number, with one of the following part numbered tail rotor blades installed, certificated in any category.

Model	Serial No.	Tail rotor blade (blade) part no.
222	47006 through 47089	222–016–001–123, –127, –131, and –135.
222B	47131 through 47156	222–016–001–123, –127, –131, and –135.
222U	47501 through 47574	222–016–001–123, and –131.
230	23001 through 23038	222–016–001–123, and –131.
430	49001 through 49107	222–016–001–123, and –131.

Compliance: Required as indicated.
 To detect a crack in the blade and to prevent loss of the blade and subsequent loss of control of the helicopter, accomplish the following:

(a) Within 3 hours time-in-service (TIS), and thereafter at intervals not to exceed 3 hours TIS, clean and visually check both sides of each blade for a crack in the paint in the areas shown in Figure 1 of this AD. An owner/operator (pilot), holding at least a

private pilot certificate, may perform this visual check and must enter compliance with this paragraph into the helicopter maintenance records by following 14 CFR 43.11 and 91.417(a)(2)(v).

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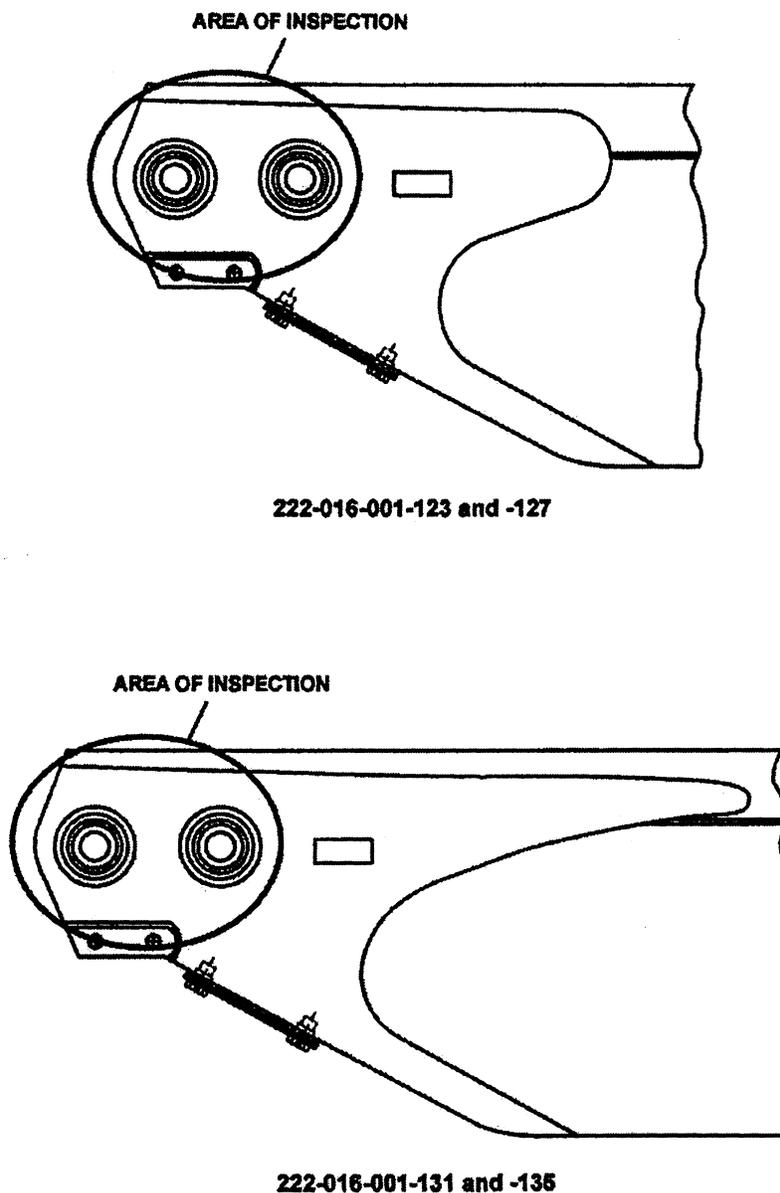


Figure 1. Blade inspection area

BILLING CODE 4910-13-C

Note 1: Bell Helicopter Textron Alert Service Bulletin (ASB) No. 222-04-100, No. 222U-04-71, and No. 230-04-31, all dated August 27, 2004, and ASB No. 430-04-31, Revision A, dated November 29, 2004, pertain to the subject of this AD.

(b) If the visual check required by paragraph (a) reveals a crack in the paint, before further flight, remove the blade and follow the requirements in paragraphs (c)(2) through (c)(3)(ii) of this AD.

(c) Within the next 50 hours TIS, unless accomplished previously, and thereafter at

intervals not to exceed 50 hours TIS, clean the blade by wiping down both surfaces of each blade in the inspection area depicted in Figure 1 of this AD using aliphatic naphtha (C-305) or detergent (C-318) or an equivalent. Using a 10X or higher power magnifying glass, visually inspect both sides of the blade in the areas depicted in Figure 1 of this AD.

(1) If a crack is found, even if only in the paint, before further flight, remove the blade from the helicopter and proceed with the following:

(2) Remove the paint on the blade down to the bare metal in the area of the suspected crack by using plastic metal blasting (PMB) or a nylon web abrasive pad. Abrade the blade surface in a span-wise direction only.

Note 2: PMB may cause damage to helicopter parts if untrained personnel perform the paint removal. BHT-ALL-SPM, chapter 3, paragraph 3-24, pertains to the subject of this AD.

(3) Using a 10X or higher power magnifying glass, inspect the blade for a crack.

(i) If a crack is found, replace the blade with an airworthy blade before further flight.

(ii) If no crack is found in the blade surface, refinish the blade by applying one coat of epoxy polyamide primer, MIL-P-23377 or MIL-P-85582, so that the primer overlaps the existing coats just beyond the abraded area. Let the area dry for 30 minutes to 1 hour. Then, apply one sealer coat of polyurethane, MILC85285 TYI CL2, color number 27925 (semi-gloss white), per Fed. Std. 595. Reinstall the blade.

Note 3: BHT-ALL-SPM, chapter 4, pertains to painting the blade.

(d) To request a different method of compliance or a different compliance time for this AD, follow the procedures in 14 CFR 39.19. Contact the Safety Management Group, FAA, for information about previously approved alternative methods of compliance.

(e) Special flight permits may be issued by following 14 CFR 21.197 and 21.199 to operate the helicopter to a location where the requirements of this AD can be accomplished provided you do not find a crack in the blade paint during a check or inspection.

(f) This amendment becomes effective March 4, 2005.

Note 4: The subject of this AD is addressed in Transport Canada (Canada) Airworthiness Directive CF-2004-21R1, dated December 9, 2004.

Issued in Fort Worth, Texas, on February 10, 2005.

Kim Smith,

Acting Manager, Rotorcraft Directorate, Aircraft Certification Service.

[FR Doc. 05-3049 Filed 2-16-05; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2005-20276; Directorate Identifier 2005-NM-023-AD; Amendment 39-13979; AD 2005-04-07]

RIN 2120-AA64

Airworthiness Directives; Bombardier Model CL-600-2B19 (Regional Jet Series 100 & 440) Airplanes and Model CL-600-1A11 (CL-600), CL-600-2A12 (CL-601), and CL-600-2B16 (CL-601-3A, CL-601-3R, and CL-604) Series Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule; request for comments.

SUMMARY: The FAA is adopting a new airworthiness directive (AD) for all Bombardier Model CL-600-2B19 (Regional Jet Series 100 & 440) airplanes and Model CL-600-1A11 (CL-600), CL-

600-2A12 (CL-601), and CL-600-2B16 (CL-601-3A, CL-601-3R, and CL-604) series airplanes. This AD requires revising the airplane flight manuals to include a new cold weather operations limitation. This AD is prompted by a report that even small amounts of frost, ice, snow, or slush on the wing leading edges or forward upper wing surfaces can cause an adverse change in the stall speeds, stall characteristics, and the protection provided by the stall protection system. We are issuing this AD to prevent possible loss of control on take-off resulting from even small amounts of frost, ice, snow, or slush on the wing leading edges or forward upper wing surfaces.

DATES: Effective February 22, 2005.

The incorporation by reference of certain publications listed in the AD is approved by the Director of the Federal Register as of February 22, 2005.

We must receive comments on this AD by April 18, 2005.

ADDRESSES: Use one of the following addresses to submit comments on this AD.

- DOT Docket Web site: Go to <http://dms.dot.gov> and follow the instructions for sending your comments electronically.

- Government-wide rulemaking Web site: Go to <http://www.regulations.gov> and follow the instructions for sending your comments electronically.

- Mail: Docket Management Facility; U.S. Department of Transportation, 400 Seventh Street SW., Nassif Building, room PL-401, Washington, DC 20590.

- Fax: (202) 493-2251.

- Hand Delivery: Room PL-401 on the plaza level of the Nassif Building, 400 Seventh Street SW., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For the temporary revisions identified in this AD, contact Bombardier, Inc., Canadair, Aerospace Group, P.O. Box 6087, Station Centre-ville, Montreal, Quebec H3C 3G9, Canada. You can examine this information at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call (202) 741-6030, or go to: http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

You can examine the contents of this AD docket on the Internet at <http://dms.dot.gov>, or in person at the Docket Management Facility, U.S. Department of Transportation, 400 Seventh Street SW., room PL-401, on the plaza level of the Nassif Building, Washington, DC. This docket number is FAA-2005-20276; the directorate identifier for this docket is 2005-NM-023-AD.

Examining the Docket

You can examine the AD docket on the Internet at <http://dms.dot.gov>, or in person at the Docket Management Facility office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The Docket Management Facility office (telephone (800) 647-5227) is located on the plaza level of the Nassif Building at the DOT street address stated in the **ADDRESSES** section. Comments will be available in the AD docket shortly after the DMS receives them.

FOR FURTHER INFORMATION CONTACT:

Bruce Valentine, Aerospace Engineer, Systems and Flight Test Branch, ANE-172, FAA, New York Aircraft Certification Office, 1600 Stewart Avenue, Suite 410, Westbury, New York 11590; telephone (516) 228-7328; fax (516) 794-5531.

SUPPLEMENTARY INFORMATION: Transport Canada Civil Aviation (TCCA), which is the airworthiness authority for Canada, notified the FAA that an unsafe condition may exist under certain operating conditions on all Bombardier Model CL-600-2B19 (Regional Jet Series 100 & 440) airplanes and Model CL-600-1A11 (CL-600), CL-600-2A12 (CL-601), and CL-600-2B16 (CL-601-3A, CL-601-3R, and CL-604) series airplanes. TCCA advises that even small amounts of frost, ice, snow, or slush on the wing leading edges or forward upper wing surfaces of these airplanes can cause an unsafe condition where an adverse change in the stall speeds, stall characteristics, and the protection provided by the stall protection system may result in reduced controllability of the airplane. TCCA advises that cold weather operational requirements for the subject airplane flight manuals should include wing leading edge and upper wing surface inspections using visual and tactile means in identifying potential contamination by frost, ice, snow, or slush.

Relevant Temporary Revision Information

Bombardier has issued temporary revisions (TRs) to the applicable Bombardier airplane flight manuals (AFMs) as listed in the following table. The TRs include a new take-off limitation to emphasize the requirement for an aerodynamically clean airplane during cold weather operations. The TRs specify that, in addition to a visual check, a tactile check must be done to determine that the wing is free from frost, ice, snow, or slush when certain weather conditions exist.