

in Figure 9 of the service bulletin. Completing the initial inspection required by this paragraph terminates the repetitive

inspections required by paragraph (i) of this AD. For airplanes on which paragraph (i) of this AD has not been done, doing the initial

inspection required by this paragraph at the specified compliance time eliminates the need to comply with paragraph (i) of this AD.

TABLE 3.—COMPLIANCE TIMES FOR INITIAL INSPECTION REQUIRED BY PARAGRAPH (L)

For the inspections identified in the following figures referenced in figure 9 of the service bulletin—	For these airplanes—	Do the inspection—
Figure 10 or 11	Airplanes not inspected previously in accordance with paragraph (i) of this AD.	Within 15,000 flight cycles after doing the modification or permanent repair.
Figure 10 or 11	Airplanes inspected previously in accordance with paragraph (i) of this AD using the surface HFEC method for the most recent inspection.	Within 1,000 flight cycles after the most recent inspection.
Figure 10 or 11	Airplanes inspected previously in accordance with paragraph (i) of this AD using the open-hole HFEC method for the most recent inspection.	Within 3,000 flight cycles after the most recent inspection.
Figure 12 or 13	All airplanes	Within 6,000 flight cycles after doing the modification or permanent repair, or within 1,000 flight cycles after the effective date of this AD, whichever is later.

Repair

(m) If any crack is found during any inspection required by paragraph (j), (k), or (l) of this AD: Before further flight, repair in accordance with the Accomplishment Instructions of Boeing Service Bulletin 747–53A2459, Revision 1, dated March 11, 2004; except where the service bulletin specifies to contact Boeing for appropriate action, repair in accordance with a method approved by the Manager, Seattle ACO; or in accordance with data meeting the type certification basis of the airplane approved by a Boeing Company DER who has been authorized by the Manager, Seattle ACO, to make such findings. For a repair method to be approved, the approval must specifically reference this AD.

Reporting Not Required

(n) Although Boeing Service Bulletin 747–53A2459, Revision 1, dated March 11, 2004, specifies to report certain body frame cracks on certain airplanes, this AD does not include that requirement.

Alternative Methods of Compliance (AMOCs)

(o)(1) The Manager, Seattle ACO, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19.

(2) An AMOC that provides an acceptable level of safety may be used for any repair required by this AD, if it is approved by a Boeing Company Designated Engineering Representative who has been authorized by the Manager, Seattle ACO, to make such findings.

(3) AMOCs approved previously in accordance with AD 2002–18–04, amendment 39–12878, are approved as alternative methods of compliance with paragraphs (f), (g), (h), and (i) of this AD.

Issued in Renton, Washington, on October 26, 2004.

Ali Bahrami,

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 04–24723 Filed 11–4–04; 8:45 am]

BILLING CODE 4910–13–P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA–2004–19534; Directorate Identifier 2004–NM–99–AD]

RIN 2120–AA64

Airworthiness Directives; Airbus Model A300 B2 and B4 Series Airplanes; Model A300 B4–600, B4–600R, and F4–600R Series Airplanes, and Model A300 C4–605R Variant F Airplanes (Collectively Called A300–600); and Model A310 Series Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: The FAA proposes to adopt a new airworthiness directive (AD) for certain Airbus airplane models, as specified above. This proposed AD would require modifying the thermal insulation system of certain fuselage frames, and modifying the fuselage drainage system. This proposed AD would also require revising the FAA-approved maintenance inspection program to include inspections for corrosion or cracking in the subject areas. This proposed AD is prompted by reports of corrosion in the lower part of

the pressure bulkhead at certain fuselage frames. We are proposing this AD to prevent accumulation of condensation in the insulation blankets of certain fuselage frames, which could cause corrosion that could result in reduced structural integrity of the fuselage and consequent rapid decompression of the airplane.

DATES: We must receive comments on this proposed AD by December 6, 2004.

ADDRESSES: Use one of the following addresses to submit comments on this proposed AD.

- *DOT Docket Web Site:* Go to <http://dms.dot.gov> and follow the instructions for sending your comments electronically.

- *Government-wide Rulemaking Web Site:* Go to <http://www.regulations.gov> and follow the instructions for sending your comments electronically.

- *Mail:* Docket Management Facility, U.S. Department of Transportation, 400 Seventh Street, SW., Nassif Building, room PL–401, Washington, DC 20590.

- *By Fax:* (202) 493–2251.

- *Hand Delivery:* Room PL–401 on the plaza level of the Nassif Building, 400 Seventh Street, SW., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this proposed AD, contact Airbus, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France.

You can examine the contents of this AD docket on the Internet at <http://dms.dot.gov>, or in person at the Docket Management Facility, U.S. Department of Transportation, 400 Seventh Street, SW., room PL–401, on the plaza level of the Nassif Building, Washington, DC.

FOR FURTHER INFORMATION CONTACT:

Technical Information: Dan Rodina, Aerospace Engineer, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-2125; fax (425) 227-1149.

Plain Language Information: Marcia Walters, marcia.walters@faa.gov.

SUPPLEMENTARY INFORMATION:

Docket Management System (DMS)

The FAA has implemented new procedures for maintaining AD docket electronically. As of May 17, 2004, new AD actions are posted on DMS and assigned a docket number. We track each action and assign a corresponding directorate identifier. The DMS AD docket number is in the form "Docket No. FAA-2004-99999." The Transport Airplane Directorate identifier is in the form "Directorate Identifier 2004-NM-999-AD." Each DMS AD docket also lists the directorate identifier ("Old Docket Number") as a cross-reference for searching purposes.

Comments Invited

We invite you to submit any relevant written data, views, or arguments regarding this proposed AD. Send your comments to an address listed under **ADDRESSES**. Include "Docket No. FAA-2004-19534; Directorate Identifier 2004-NM-99-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of the proposed AD. We will consider all comments submitted by the closing date and may amend the

proposed AD in light of those comments.

We will post all comments we receive, without change, to <http://dms.dot.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact with FAA personnel concerning this proposed AD. Using the search function of our docket Web site, anyone can find and read the comments in any of our dockets, including the name of the individual who sent the comment (or signed the comment on behalf of an association, business, labor union, etc.). You can review the DOT's complete Privacy Act Statement in the **Federal Register** published on April 11, 2000 (65 FR 19477-78), or you can visit <http://dms.dot.gov>.

We are reviewing the writing style we currently use in regulatory documents. We are interested in your comments on whether the style of this document is clear, and your suggestions to improve the clarity of our communications that affect you. You can get more information about plain language at <http://www.faa.gov/language> and <http://www.plainlanguage.gov>.

Examining the Docket

You can examine the AD docket on the Internet at <http://dms.dot.gov>, or in person at the Docket Management Facility office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The Docket Management Facility office (telephone (800) 647-5227) is on the plaza level of the Nassif Building at the DOT street

address stated in the **ADDRESSES** section. Comments will be available in the AD docket shortly after the DMS receives them.

Discussion

The Direction Générale de l'Aviation Civile (DGAC), which is the airworthiness authority for France, notified us that an unsafe condition may exist on certain Airbus Model A300 B2 and B4 series airplanes; Model A300 B4-600, B4-600R, and F4-600R series airplanes, and Model A300 C4-605R Variant F airplanes (collectively called A300-600); and Model A310 series airplanes. The DGAC advises that corrosion has been found in the lower part of the pressure bulkhead at fuselage frame (FR) 54 and, on certain airplanes, FR 39. Investigation revealed that the rear face of FR 54, and the front face of FR 39 on certain airplanes, are completely thermally insulated; no drainage is provided at the insulation blankets. The lack of drainage results in condensation being trapped. This condition, if not corrected, could result in corrosion that could result in reduced structural integrity of the fuselage and consequent rapid decompression of the airplane.

Relevant Service Information

Airbus has issued the service bulletins listed in the table below, which describe procedures for modifying the thermal insulation system using the specific procedures identified in the table below.

SERVICE BULLETINS FOR MODIFYING THE THERMAL INSULATION SYSTEM

Airbus model	Airbus service bulletin	Specific procedures described
A300 B2 and B4 series	A300-21-0116, Revision 02, dated June 13, 2003.	Permanently removing insulation blankets from the front face of FR 39 and the rear face of FR 54 between the cargo compartment floor and the fuselage lower wall and removing insulation blankets between FR 54 and FR 54.1 and stringers 50 through 56 inclusive on the left and right sides of the airplane; installing bracket assemblies to raise the blankets in this area; and re-installing these insulation blankets.
A300 B4-600, B4-600R, and F4-600R series; and A300 C4-605R Variant F (collectively called A300-600).	A300-21-6025, Revision 01, dated June 13, 2003.	Permanently removing insulation blankets from the rear face of FR 54 between the cargo compartment floor and the fuselage lower wall and removing insulation blankets between FR 54 and FR 54.1 and stringers 50 through 56 inclusive on the left and right sides of the airplane; installing bracket assemblies to raise the blankets in this area; and re-installing these insulation blankets.

SERVICE BULLETINS FOR MODIFYING THE THERMAL INSULATION SYSTEM—Continued

Airbus model	Airbus service bulletin	Specific procedures described
A310 series	A310–21–2041, Revision 02, dated June 13, 2003.	Permanently removing insulation blankets from the rear face of FR 54 between the cargo compartment floor and the fuselage lower wall and removing insulation blankets between FR 54 and FR 54.1 and stringers 50 through 56 inclusive on the left and right sides of the airplane; installing bracket assemblies to raise the blankets in this area; and reinstalling these insulation blankets and testing the additional center fuel tank, if installed.

Airbus has also issued the service bulletins listed in the table below,

which describe procedures for modifying the fuselage drainage system

using the specific procedures identified in the table below.

SERVICE BULLETINS FOR MODIFYING THE FUSELAGE DRAINAGE SYSTEM

Airbus model	Airbus service bulletin	Specific procedures described
A300 B2 and B4 series	A300–53–0201, Revision 04, dated May 2, 2003.	Replacing all existing drain valves in the lower fuselage and upper deck cargo door with new, improved (plastic) drain valves; installing one additional drain valve; and drilling additional drain holes in the stringers.
A300 B4–600, B4–600R, and F4–600R series; and A300 C4–605R Variant F (collectively called A300–600).	A300–53–6008, Revision 05, dated July 15, 2004.	Replacing all existing drain valves in the lower fuselage and upper deck cargo door with new, improved (plastic) drain valves; installing one additional drain valve; and drilling additional drain holes in the stringers.
A310 series	A310–53–2027, Revision 04, dated July 15, 2004.	Replacing all existing drain valves in the lower fuselage with new, improved (plastic) drain valves; installing additional drain valves; and drilling additional drain holes in the stringers.

Airbus has also issued Maintenance Planning Documents containing Task Numbers 538295–0603–01 (for Airbus Model A300 B2 and B4 series airplanes), and 541531–01–1 and 531533–01–1 (for Airbus Model A300–600 and Model A310 series airplanes). These task numbers describe procedures for repetitive detailed visual inspections for corrosion or cracking of fuselage structure from FR 38.2 to 39, and at FR 54, as applicable. These tasks are necessary for airplanes modified in accordance with the service information described previously.

Accomplishing the actions specified in the applicable service information is intended to adequately address the unsafe condition. The DGAC mandated the service information and issued French airworthiness directive 2003–317(B), dated August 20, 2003, to ensure the continued airworthiness of these airplanes in France.

FAA’s Determination and Requirements of the Proposed AD

These airplane models are manufactured in France and are type certificated for operation in the United States under the provisions of section 21.29 of the Federal Aviation Regulations (14 CFR 21.29) and the applicable bilateral airworthiness agreement. Pursuant to this bilateral airworthiness agreement, the DGAC has kept the FAA informed of the situation described above. We have examined the DGAC’s findings, evaluated all pertinent information, and determined that we need to issue an AD for products of this type design that are certificated for operation in the United States.

Therefore, we are proposing this AD, which would require accomplishing the actions specified in the service information described previously. This proposed AD would also require revising the FAA-approved maintenance inspection program to include detailed inspections for corrosion and cracking in the area of FRs 39 and 54, as described in the Maintenance Planning

Document task numbers identified previously.

Differences Between the Proposed AD and French Airworthiness Directive

The applicability of French airworthiness directive 2003–317(B) excludes airplanes that accomplished Airbus Service Bulletin A300–53–0201, A300–53–6008, or A310–53–2027 in service. However, we have not excluded those airplanes in the applicability of this proposed AD; rather, this proposed AD includes a requirement to accomplish the actions specified in those service bulletins. This requirement would ensure that the actions specified in the service bulletins and that would be required by this proposed AD are accomplished on all affected airplanes. Operators must continue to operate the airplane in the configuration that would be required by this proposed AD unless we approve an alternative method of compliance.

French airworthiness directive 2003–317(B) specifies that airplanes modified in accordance with the service

information described previously must be inspected in accordance with the Maintenance Planning Document tasks described previously. However, no compliance time is given for revising the maintenance program. This proposed AD would require revising the maintenance program to include these tasks within 90 days after doing the modifications specified in the service information described previously, or within 90 days after the effective date of the AD, whichever is later. We find that this represents an appropriate interval

of time in which this action may be accomplished by all affected operators without jeopardizing continued flight safety.

Clarification of Inspection Terminology

In this proposed AD, the “detailed visual inspections” specified in Maintenance Planning Document Task Numbers 538295–0603–01 (for Airbus Model A300 B2 and B4 series airplanes), and 541531–01–1 and 531533–01–1 (for Airbus Model A300 B4–600, B4–600R, and F4–600R series airplanes, and A300

C4–605R Variant F airplanes (collectively called A300–600); and Model A310 series airplanes), are referred to as “detailed inspections.” We have included the definition for a detailed inspection in a note in the proposed AD.

Costs of Compliance

The following table provides the estimated costs for U.S. operators to comply with this proposed AD, at an average labor rate of \$65 per work hour.

ESTIMATED COSTS

Action	Models	Work hours	Parts	Cost per air-plane	Number of U.S.-registered airplanes	Fleet cost
Modifying the Thermal Insulation System.	A300 B2/B4	5	\$567	\$892	23	\$20,516
Modifying the Thermal Insulation System.	A300 B4–600, B4–600R, and F4–600R series; and A300 C4–605R Variant F (collectively called A300–600).	4	567	827	116	95,932
Modifying the Thermal Insulation System.	A310	4	567	827	47	38,869
Modifying the Fuselage Drainage System.	A300 B2/B4	38	1,857	4,327	23	99,521
Modifying the Fuselage Drainage System.	A300 B4–600, B4–600R, and F4–600R series; and A300 C4–605R Variant F (collectively called A300–600).	36	1,378	3,718	116	431,288
Modifying the Fuselage Drainage System.	A310	27	1,451	3,206	47	150,682
Maintenance Program Revision.	All	1	None	65	186	12,090

Regulatory Findings

We have determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that the proposed regulation:

1. Is not a “significant regulatory action” under Executive Order 12866;
2. Is not a “significant rule” under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
3. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this proposed AD. See the ADDRESSES section for a location to examine the regulatory evaluation.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

Airbus: Docket No. FAA–2004–19534; Directorate Identifier 2004–NM–99–AD.

Comments Due Date

(a) The Federal Aviation Administration must receive comments on this AD action by December 6, 2004.

Affected ADs

(b) None.

Applicability: (c) This AD applies to Airbus Model A300 B2 and B4 series airplanes; A300 B4–600, B4–600R, and F4–600R series airplanes, and A300 C4–605R Variant F airplanes (collectively called A300–600); and A310 series airplanes; certificated in any category; except those on which Airbus Modification 5946 was done during production.

Unsafe Condition

(d) This AD was prompted by reports of corrosion in the lower part of the pressure bulkhead at fuselage frames (FR) 39 and 54. We are issuing this AD to prevent accumulation of condensation in the insulation blankets of certain fuselage FRs, which could cause corrosion that could result in reduced structural integrity of the fuselage and consequent rapid decompression of the airplane.

Compliance: (e) You are responsible for having the actions required by this AD performed within the compliance times specified, unless the actions have already been done.

Modification of Thermal Insulation and Fuselage Drainage Systems

(f) Within 22 months after the effective date of this AD, modify the thermal

insulation system of applicable fuselage frames and modify the fuselage drainage system, by doing all actions in the Accomplishment Instructions of the

applicable service bulletins specified in Table 1 of this AD.

TABLE 1.—RELEVANT SERVICE BULLETINS

For Airbus models	Modify the thermal insulation according to Airbus Service Bulletin	And modify the fuselage drainage system according to Airbus Service Bulletin
A300 B2 and B4 series	A300–21–0116, Revision 02, dated June 13, 2003.	A300–53–0201, Revision 04, dated May 2, 2003.
A300 B4–600, B4–600R, and F4–600R series; A300 C4–605R Variant F (collectively called A300–600).	A300–21–6025, Revision 01, dated June 13, 2003.	A300–53–6008, Revision 05, dated July 15, 2004.
A310 series	A310–21–2041, Revision 02, dated June 13, 2003.	A310–53–2027, Revision 04, dated July 15, 2004.

Modifications Accomplished According to Previous Issues of Service Bulletins

(g) Modifications accomplished before the effective date of this AD according to Airbus Service Bulletin A300–53–6008, Revision 03, dated November 6, 1990, or Revision 04, dated April 28, 2003 (for Model A300 B4–600, B4–600R, and F4–600R series airplanes; and A300 C4–605R Variant F airplanes (collectively called A300–600)); or Airbus Service Bulletin A310–53–2027, Revision 02, dated November 6, 1990, or Revision 03, dated May 2, 2003 (for Model A310 series airplanes); are considered acceptable for compliance with the corresponding action specified in this AD.

Maintenance Program Revision

(h) Within 90 days after doing the actions required by paragraph (f) of this AD, or within 90 days after the effective date of this AD, whichever is later: Incorporate into the FAA-approved maintenance inspection program repetitive detailed inspections for corrosion or cracking of fuselage structure from FR 38.2 to 39, and at FR 54, as applicable, as described in Airbus Maintenance Planning Document Task Numbers 538295–0603–01 (for Airbus Model A300 B2 and B4 series airplanes), and 541531–01–1 and 531533–01–1 (for Airbus Model A300 B4–600, B4–600R, and F4–600R series airplanes, and A300 C4–605R Variant F airplanes (collectively called A300–600); and Model A310 series airplanes). Then, thereafter, comply with the applicable requirements.

Note 1: For the purposes of this AD, a detailed inspection is: “An intensive examination of a specific item, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at an intensity deemed appropriate. Inspection aids such as mirror, magnifying lenses, etc., may be necessary. Surface cleaning and elaborate procedures may be required.”

Alternative Methods of Compliance (AMOCs)

(i) The Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA, has the authority to approve AMOCs for this AD, if requested in accordance with the procedures found in 14 CFR 39.19.

Related Information

(j) French airworthiness directive 2003–317(B), dated August 20, 2003, also addresses the subject of this AD.

Issued in Renton, Washington, on October 26, 2004.

Ali Bahrami,

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 04–24722 Filed 11–4–04; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA–2004–19533; Directorate Identifier 2004–NM–31–AD]

RIN 2120–AA64

Airworthiness Directives; Boeing Model 737–300, –400, and –500 Series Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: The FAA proposes to adopt a new airworthiness directive (AD) for certain Boeing Model 737–300, –400, and –500 series airplanes. This proposed AD would require repetitive inspections for cracking of the crown area of the fuselage skin, and corrective actions, if necessary. This proposed AD is prompted by a 737 fuselage structure test and fatigue analysis that indicate fuselage skin cracking could occur between 21,000 and 42,000 total flight cycles. We are proposing this AD to detect and correct fatigue cracking of the fuselage skin, which could cause the fuselage skin to fracture and fail, and could result in rapid decompression of the airplane.

DATES: We must receive comments on this proposed AD by December 20, 2004.

ADDRESSES: Use one of the following addresses to submit comments on this proposed AD.

- **DOT Docket Web Site:** Go to <http://dms.dot.gov> and follow the instructions for sending your comments electronically.

- **Government-wide Rulemaking Web Site:** Go to <http://www.regulations.gov> and follow the instructions for sending your comments electronically.

- **Mail:** Docket Management Facility, U.S. Department of Transportation, 400 Seventh Street SW., Nassif Building, room PL–401, Washington, DC 20590.

- **By Fax:** (202) 493–2251.

- **Hand Delivery:** Room PL–401 on the plaza level of the Nassif Building, 400 Seventh Street SW., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this proposed AD, contact Boeing Commercial Airplanes, P.O. Box 3707, Seattle, Washington 98124–2207.

You can examine the contents of this AD docket on the Internet at <http://dms.dot.gov>, or in person at the Docket Management Facility, U.S. Department of Transportation, 400 Seventh Street SW., room PL–401, on the plaza level of the Nassif Building, Washington, DC. This docket number is FAA–2004–19533; the directorate identifier for this docket is 2004–NM–31–AD.

FOR FURTHER INFORMATION CONTACT:

Technical Information: Sue Lucier, Aerospace Engineer, Airframe Branch, ANM–120S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98055–4056; telephone (425) 917–6438; fax (425) 917–6590.

Plain Language Information: Marcia Walters, marcia.walters@faa.gov.

SUPPLEMENTARY INFORMATION: