

Comments Due Date

(a) The Federal Aviation Administration (FAA) must receive comments on this airworthiness directive (AD) action by December 28, 2004.

Affected ADs

(b) This AD supersedes AD 2004-13-03, Amendment 39-13684.

Applicability

(c) This AD applies to Rolls-Royce (1971) Limited, Bristol Engine Division (RR) Model Viper Mk.601-22 turbojet engines. These engines are installed on, but not limited to, Raytheon HS.125 Series 600 and BH.125 Series 600 airplanes.

Unsafe Condition

(d) This AD results from comments received on AD 2004-13-03, that the AD is unnecessarily more restrictive than the requirements in the associated RR Alert Service Bulletin (ASB) No. 72-A184. We are proposing this AD to prevent multiple failures of 1st stage turbine rotor blades that could result in a dual-engine shutdown. The actions specified in this AD are intended to prevent multiple failures of 1st stage turbine rotor blades that could result in a dual-engine shutdown.

Compliance

(e) You are responsible for having the actions required by this AD performed within

the compliance times specified unless the actions have already been done.

New Reduced Life Limit

(f) Change the RR Time Limits Manual life limit for the 1st stage turbine rotor blades, P/Ns V926000, V926293, and V926319, from 7,000 hours time-in-service (TIS) to 4,600 hours TIS.

(g) Limit the number of installed engines with 1st stage turbine rotor blades that exceed 4,600 hours TIS on the effective date of this AD as specified in the following Table 1:

TABLE 1.—INSTALLED ENGINES

On the effective date of this AD, if	Then:
(1) Both engines installed on the airplane have 1st stage turbine rotor blades that exceed 5,800 hours TIS.	Replace the 1st stage turbine rotor blades in the engine that has the higher blade life within 50 hours TIS or 6 weeks after the effective date of this AD, whichever occurs first.
(2) One engine installed on the airplane has 1st stage turbine rotor blades that exceed turbine 5,800 hours TIS, and the other engine has 1st stage turbine rotor blades that exceed 4,600 hours TIS.	Replace the 1st stage turbine rotor blades in the engine that has the higher blade life within 100 hours TIS or 4 months after the effective date of this AD, whichever occurs first.
(3) One engine installed on the airplane has 1st stage turbine rotor blades that exceed 5,800 hours TIS, and the other engine has 1st stage turbine rotor blades with fewer than 4,600 hours TIS.	Replace the 1st stage turbine rotor blades in the engine that has the higher blade life within 200 hours TIS or 6 months after the effective date of this AD, whichever occurs first.
(4) One engine installed on the airplane has 1st stage turbine rotor blades that exceed 4,600 hours TIS, but have fewer than 5,800 hours TIS, and the other engine has 1st turbine stage turbine rotor blades with fewer than 4,600 hours TIS.	Replace the 1st stage rotor blades in the engine that has the higher blade life at 5,800 hours TIS or 6 months after the effective date of this AD, whichever occurs later.

(h) No engine may operate with a blade life exceeding 5,800 hours TIS, applicable beginning 6 months from the effective date of this AD.

(i) No engine may operate with a blade life exceeding 4,600 hours TIS, applicable beginning 3 years from the effective date of this AD.

Installation of Engines After the Effective Date of This AD

(j) After the effective date of this AD, do not install any engine that has 1st stage turbine rotor blades, P/Ns V926000, V926293, and V926319, that exceed 4,600 hours TIS, except as allowed in Table 1 of this AD.

Alternative Methods of Compliance

(k) The Manager, Engine Certification Office, has the authority to approve alternative methods of compliance for this AD if requested using the procedures found in 14 CFR 39.19

Material Incorporated by Reference

(l) None.

Related Information

(m) Civil Aviation Authority airworthiness directive AD 004-01-2001, dated January 2001, also addresses the subject of this AD.

Issued in Burlington, Massachusetts, on October 25, 2004.

Francis A. Favara,

Acting Manager, Engine and Propeller Directorate, Aircraft Certification Service.

[FR Doc. 04-24230 Filed 10-28-04; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2004-19470; Directorate Identifier 2003-NM-268-AD]

RIN 2120-AA64

Airworthiness Directives; Boeing Model 747-100B SUD, -300, -400, and -400D Series Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: The FAA proposes to adopt a new airworthiness directive (AD) for certain Boeing Model 747-100B SUD, -300, -400, and -400D series airplanes. This proposed AD would require a one-time inspection for discrepancies of the fuselage frame to tension tie joints at body stations (BS) 1120 through 1220

and to determine if steel splice plates are installed on the fuselage frames, and related investigative and corrective actions. This proposed AD is prompted by reports indicating that severed tension ties were found at the fuselage frame joints at BS 1120 and 1140. We are proposing this AD to prevent fatigue cracking of the fuselage frame to tension tie joints, which could result in severing of the tension ties and consequent rapid decompression of the airplane fuselage.

DATES: We must receive comments on this proposed AD by December 13, 2004.

ADDRESSES: Use one of the following addresses to submit comments on this proposed AD.

- DOT Docket Web site: Go to <http://dms.dot.gov> and follow the instructions for sending your comments electronically.
- Government-wide rulemaking web site: Go to <http://www.regulations.gov> and follow the instructions for sending your comments electronically.
- Mail: Docket Management Facility, U.S. Department of Transportation, 400 Seventh Street, SW., Nassif Building, room PL-401, Washington, DC 20590.
- By fax: (202) 493-2251.
- Hand Delivery: Room PL-401 on the plaza level of the Nassif Building, 400 Seventh Street, SW., Washington,

DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this proposed AD, contact Boeing Commercial Airplanes, P.O. Box 3707, Seattle, Washington 98124-2207.

You can examine the contents of this AD docket on the Internet at <http://dms.dot.gov>, or at the Docket Management Facility, U.S. Department of Transportation, 400 Seventh Street, SW., room PL-401, on the plaza level of the Nassif Building, Washington, DC.

FOR FURTHER INFORMATION CONTACT:

Technical Information: Ivan Li, Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 917-6437; fax (425) 917-6590.

Plain language information: Marcia Walters, marcia.walters@faa.gov.

SUPPLEMENTARY INFORMATION:

Docket Management System (DMS)

The FAA has implemented new procedures for maintaining AD dockets electronically. As of May 17, 2004, new AD actions are posted on DMS and assigned a docket number. We track each action and assign a corresponding directorate identifier. The DMS AD docket number is in the form "Docket No. FAA-2004-99999." The Transport Airplane Directorate identifier is in the form "Directorate Identifier 2004-NM-999-AD." Each DMS AD docket also lists the directorate identifier ("Old Docket Number") as a cross-reference for searching purposes.

Comments Invited

We invite you to submit any relevant written data, views, or arguments regarding this proposed AD. Send your comments to an address listed under **ADDRESSES**. Include "Docket No. FAA-2004-19470; Directorate Identifier 2003-NM-268-AD" in the subject line of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of the proposed AD. We will consider all comments submitted by the closing date and may amend the proposed AD in light of those comments.

We will post all comments we receive, without change, to <http://dms.dot.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact with FAA personnel concerning this proposed AD. Using the search function of that Web site, anyone can find and read the comments in any of our dockets,

including the name of the individual who sent the comment (or signed the comment on behalf of an association, business, labor union, etc.). You can review DOT's complete Privacy Act Statement in the **Federal Register** published on April 11, 2000 (65 FR 19477-78), or you can visit <http://dms.dot.gov>.

We are reviewing the writing style we currently use in regulatory documents. We are interested in your comments on whether the style of this document is clear, and your suggestions to improve the clarity of our communications that affect you. You can get more information about plain language at <http://www.faa.gov/language> and <http://www.plainlanguage.gov>.

Examining the Docket

You can examine the AD docket on the Internet at <http://dms.dot.gov>, or in person at the Docket Management Facility office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The Docket Management Facility office (telephone (800) 647-5227) is located on the plaza level of the Nassif Building at the DOT street address stated in the **ADDRESSES** section. Comments will be available in the AD docket shortly after the DMS receives them.

Discussion

We have received a report indicating that severed tension ties at the fuselage frame joints at body stations (BS) 1120 and 1140 were found on a Model 747-400 series airplane. The cause of the severed ties was fatigue cracking due to incorrect splice plate installation during the manufacturing process. The splice plates that were installed were made of aluminum instead of steel. When the severed tension ties were found, the airplane had accumulated approximately 6,505 total flight cycles and 52,334 total flight hours. Inspection of another Model 747-400 series airplane revealed loose fasteners at the frame to tension tie joints. Loose fasteners in the joints increase the potential for fatigue cracking. That airplane had accumulated approximately 5,875 total flight cycles and 49,426 total flight hours. Inadequate installation procedures have been determined as the root cause of these incidents. Fatigue cracking of the fuselage frame to tension tie joints, if not found and fixed, could result in severing of the tension ties and consequent rapid decompression of the airplane fuselage.

The fuselage frame to tension tie joints on certain Model 747-100B SUD, -300, and -400D series airplanes are

identical to those on the affected Model 747-400 series airplanes. Therefore, all of these models may be subject to the same unsafe condition.

Relevant Service Information

We have reviewed Boeing Special Attention Service Bulletin 747-53-2483, Revision 1, dated August 28, 2003. The service bulletin describes procedures for a one-time detailed visual inspection for discrepancies (cracks; loose, missing, or damaged fasteners or collars) of the fuselage frame to tension tie joints at BS 1120 through 1220, and to determine if steel splice plates are installed on the fuselage frame, and related investigative and corrective actions. The investigative and corrective actions include performing an open-hole eddy current inspection of the fastener holes if loose, missing, or damaged fasteners or collars are found, and installing new fasteners if necessary.

The service bulletin also specifies that operators may contact the manufacturer for disposition of certain repair instructions, and recommends that the manufacturer be contacted if an aluminum splice plate is installed.

Accomplishing the actions specified in the service information is intended to adequately address the unsafe condition.

FAA's Determination and Requirements of the Proposed AD

We have evaluated all pertinent information and identified an unsafe condition that is likely to exist or develop on other airplanes of this same type design. Therefore, we are proposing this AD, which would require you to use the service information described previously to perform the actions, except as discussed under "Differences Between the Proposed AD and Service Bulletin."

Differences Between the Proposed AD and Service Bulletin

The service bulletin provides the following information in Note 5 of the Accomplishment Instructions: "For the purposes of this service bulletin, do not count flight-cycles with a cabin pressure differential of 2.0 [pounds per square inch (psi)] or less. However, any flight-cycle with momentary spikes in cabin pressure differential above 2.0 psi must be included as a full-pressure flight-cycle. Cabin pressure records must be maintained for each airplane. Fleet averaging of cabin pressure is not allowed." We have determined that an adjustment of flight cycles due to a lower cabin differential pressure is not substantiated and will not be allowed

for use in determining the flight cycle threshold for this proposed AD.

The service bulletin also specifies that, if repair requirements exceed allowable repair criteria, operators may contact the manufacturer for disposition of repair instructions. This proposed AD would require operators to repair these conditions in accordance with a method approved by the FAA or in accordance with data meeting the type certification basis of the airplane approved by a Boeing Company Designated Engineering Representative who has been authorized by the FAA to make such findings.

The service bulletin also recommends that the manufacturer be contacted for repair instructions if an aluminum splice plate is installed, but this proposed AD would require replacement of the plate with a new plate in accordance with a method approved by the FAA, or in accordance with data meeting the type certification basis of the airplane approved by a Boeing Company Designated Engineering Representative who has been authorized by the FAA to make those findings.

The service bulletin specifies to submit certain information to the manufacturer, but this proposed AD does not include that requirement.

Although the service bulletin defines a "detailed visual" inspection for discrepancies, we have determined that the procedures in the service bulletin should be described as a "detailed inspection." A note has been added to define that type of inspection.

Costs of Compliance

This proposed AD would affect about 67 airplanes of U.S. registry and 537 airplanes worldwide. The proposed inspection would take about 2 work hours per airplane, at an average labor rate of \$65 per work hour. Based on these figures, the estimated cost of the proposed AD for U.S. operators is \$8,710, or \$130 per airplane.

Regulatory Findings

We have determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that the proposed regulation:

1. Is not a "significant regulatory action" under Executive Order 12866;

2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and

3. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this proposed AD. See the **ADDRESSES** section for a location to examine the regulatory evaluation.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

Boeing: Docket No. FAA-2004-19470; Directorate Identifier 2003-NM-268-AD.

Comments Due Date

(a) The Federal Aviation Administration (FAA) must receive comments on this AD action by December 13, 2004.

Affected ADs

(b) None.

Applicability

(c) This AD applies to Model 747-100B SUD, -300, -400, and -400D series airplanes; certificated in any category; as listed in Boeing Special Attention Service Bulletin 747-53-2483, Revision 1, dated August 28, 2003.

Unsafe Condition

(d) This AD was prompted by reports indicating that severed tension ties were found at the fuselage frame joints at body stations (BS) 1120 and 1140. We are issuing this AD to prevent fatigue cracking of the fuselage frame to tension tie joints, which could result in severing of the tension ties and consequent rapid decompression of the airplane fuselage.

Compliance

(e) You are responsible for having the actions required by this AD performed within the compliance times specified, unless the actions have already been done.

One-Time Inspection/Investigative and Corrective Actions

(f) Before the accumulation of 4,000 total flight cycles or within 1,000 flight cycles after the effective date of this AD, whichever is later: Perform a detailed inspection for discrepancies of the fuselage frame to tension tie joints at BS 1120 through BS 1220, and to determine if steel splice plates are installed on the fuselage frames. Do the inspection in accordance with the Accomplishment Instructions of Boeing Special Attention Service Bulletin 747-53-2483, Revision 1, dated August 28, 2003. Do any applicable investigative and corrective actions before further flight in accordance with the service bulletin, except as provided by paragraph (h) of this AD.

Note 1: For the purposes of this AD, a detailed inspection is: "An intensive visual examination of a specific structural area, system, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at intensity deemed appropriate by the inspector. Inspection aids such as mirror, magnifying lenses, etc., may be used. Surface cleaning and elaborate access procedures may be required."

Determining Number of Flight Cycles for Compliance Time

(g) For the purposes of calculating the compliance threshold for the actions required by paragraph (f) of this AD, all pressurized flight cycles, including the number of flight cycles in which cabin differential pressure is at 2.0 pounds per square inch (psi) or less, must be counted when determining the number of flight cycles that have occurred on the airplane. Where the service bulletin and this AD differ, the AD prevails.

Repair Requirements

(h) For any repairs outside the limits of Boeing Special Attention Service Bulletin 747-53-2483, Revision 1, dated August 28, 2003, or if any aluminum splice plate is installed on the fuselage frames: Repair or replace, as applicable, in accordance with a method approved by the Manager, Seattle Aircraft Certification Office (ACO), FAA; or in accordance with data meeting the type certification basis of the airplane approved by a Boeing Company Designated Engineering Representative (DER) who has been authorized by the Manager, Seattle ACO, to make such findings. For a repair or replacement method to be approved, as required by this paragraph, the approval must specifically reference this AD.

Actions Accomplished Per Previous Issue of Service Bulletin

(i) Inspections and corrective actions accomplished before the effective date of this AD in accordance with Boeing Special Attention Service Bulletin 747-53-2483, dated October 24, 2002, are considered acceptable for compliance with the corresponding actions specified in this AD.

No Reporting Requirements

(j) Although the Accomplishment Instructions of Boeing Special Attention

Service Bulletin 747-53-2483, Revision 1, dated August 28, 2003; describe procedures for submitting certain information to the manufacturer, this AD does not require that action.

Alternative Methods of Compliance (AMOCs)

(k)(1) The Manager, Seattle ACO, has the authority to approve AMOCs for this AD, if requested in accordance with the procedures found in 14 CFR 39.19.

(2) An AMOC that provides an acceptable level of safety may be used for any repair required by this AD, if it is approved by a Boeing Company Designated Engineering Representative who has been authorized by the Manager, Seattle ACO, to make those findings. For a repair method to be approved, the approval must specifically refer to this AD.

Issued in Renton, Washington, on October 21, 2004.

Kalene C. Yanamura,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 04-24220 Filed 10-28-04; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 71

[Docket No. FAA-2002-13247; Airspace Docket No. 02-AAL-5]

RIN 2120-AA66

Proposed Modification and Revocation of Federal Airways; AK

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM); withdrawal.

SUMMARY: This action withdraws the notice of proposed rulemaking (NPRM) published in the **Federal Register** on November 7, 2002. In that action, the FAA proposed to modify one jet route (J-133); and revoke one jet route 711 (J-711) in Alaska. The FAA has decided to withdraw the proposed rule since the Hinchinbrook Nondirectional Radio Beacon (NDB) is being decommissioned. The replacement of the Hinchinbrook NDB and the revision of several airways in Alaska will be reflected in a subsequent NPRM.

FOR FURTHER INFORMATION CONTACT: Ken McElroy, Office of System Operations and Safety, Federal Aviation Administration, 800 Independence Avenue, SW., Washington, DC 20591; telephone: (202) 267-8783.

SUPPLEMENTARY INFORMATION: On November 7, 2002, an NPRM was published in the **Federal Register**

proposing to amend 14 Code of Federal Regulations (14 CFR) part 73 (part 73) to revise J-133 between the Sitka, AK, NDB, and the Hinchinbrook, AK, NDB (67 FR 67800). The Hinchinbrook NDB will be decommissioned and replaced with the Orca Bay NDB. The replacement of the Hinchinbrook NDB and the revision of several airways in Alaska will be reflected in a subsequent NPRM.

List of Subjects in 14 CFR part 71

Airspace, Incorporation by reference, Navigation (air).

The Withdrawal

In consideration of the foregoing, the NPRM for FAA Docket No. FAA-2002-13247 and Airspace Docket No. 02-AAL-5, as published in the **Federal Register** on November 7, 2002 (67 FR 67800), is hereby withdrawn.

Authority: 49 U.S.C. 106(g), 40103, 40113, 40120; E.O. 10854; 24 FR 9565, 3 CFR, 1959-1963 Comp., p. 389.

Issued in Washington, DC, on October 21, 2004.

Reginald C. Matthews,

Manager, Airspace and Rules.

[FR Doc. 04-24145 Filed 10-28-04; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF HOMELAND SECURITY

Coast Guard

33 CFR Part 117

[CGD05-04-169]

RIN 1625-AA09

Drawbridge Operation Regulations: New Jersey Intracoastal Waterway, Point Pleasant Canal, NJ

AGENCY: Coast Guard, DHS.

ACTION: Notice of proposed rulemaking.

SUMMARY: The Coast Guard proposes to change the regulations that govern the operation of the Route 13/Lovelandtown Bridge across Point Pleasant Canal, at New Jersey Intracoastal Waterway (NJICW) mile 3.9, in Point Pleasant, NJ. The bridge will be closed to navigation beginning 8 a.m. on January 3, 2005, through 5 p.m. on March 31, 2005. This closure is necessary to facilitate extensive mechanical rehabilitation and to maintain the bridge's operational integrity.

DATES: Comments and related material must reach the Coast Guard on or before December 28, 2004.

ADDRESSES: You may mail comments and related material to Commander

(obr), Fifth Coast Guard District, Federal Building, 4th Floor, 431 Crawford Street, Portsmouth, Virginia 23704-5004, or they may be hand delivered to the same address between 8 a.m. and 4 p.m., Monday through Friday, except Federal holidays. The Commander (obr), Fifth Coast Guard District maintains the public docket for this rulemaking. Comments and material received from the public, as well as documents indicated in this preamble as being available in the docket, will become part of this docket and will be available for inspection or copying at the above address.

FOR FURTHER INFORMATION CONTACT:

Waverly W. Gregory, Jr., Bridge Administrator, Fifth Coast Guard District, at (757) 398-6222.

SUPPLEMENTARY INFORMATION:

Request for Comments

We encourage you to participate in this rulemaking by submitting comments and related material. If you do so, please include your name and address, identify the docket number for this rulemaking CGD05-04-169, indicate the specific section of this document to which each comment applies, and give the reason for each comment. Please submit all comments and related material in an unbound format, no larger than 8½ by 11 inches, suitable for copying. If you would like confirmation to know if they were received, please enclose a stamped, self-addressed postcard or envelope. We will consider all comments and material received during the comment period. We may change this proposed rule in view of those comments.

Public Meeting

We do not now plan to hold a public meeting. But you may submit a request for a meeting by writing to the Commander, Fifth Coast Guard District at the address under **ADDRESSES** explaining why one would be beneficial. If we determine that one would aid this rulemaking, we will hold one at a time at a place announced by a later notice in the **Federal Register**.

Background and Purpose

The New Jersey Department of Transportation (NJDOT) owns and operates the vertical-lift span of the Route 13/Lovelandtown Bridge across Point Pleasant Canal, in Point Pleasant, New Jersey. The bridge has a vertical clearance in the closed and full open position of 30 feet and 65 feet, at mean high water, respectively. The current regulations are outlined under the general regulations at 33 CFR 117.5,