

the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

#### Adoption of the Amendment

■ Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

#### PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

■ 2. Section 39.13 is amended by adding the following new airworthiness directive:

**2004-19-07 Bombardier, Inc. (Formerly de Havilland, Inc.):** Amendment 39-13801. Docket 2003-NM-185-AD.

**Applicability:** Model DHC-8-102 airplanes, serial numbers 023 through 392 inclusive; certificated in any category; equipped with an RDS86 Weather Radar System, excluding those airplanes equipped with option CR834CH00284.

**Compliance:** Required as indicated, unless accomplished previously.

To prevent component failure in the radar indicator, resulting in an overcurrent condition and consequent overheating or burning of an internal component or the ribbon cable, which could lead to smoke in the cockpit, resulting in incapacitation of the crew and loss of control of the airplane; accomplish the following:

#### Modification

(a) Within 12 months after the effective date of this AD, modify the electrical power

circuit by accomplishing all the actions in the Accomplishment Instructions of Bombardier Modification Summary Package (ModSum) IS8Q3450000, Revision A, dated October 15, 2002; as applicable. Do the actions per the ModSum.

#### Alternative Methods of Compliance

(b) In accordance with 14 CFR 39.19, the Manager, New York Aircraft Certification Office (ACO), FAA, is authorized to approve alternative methods of compliance for this AD.

#### Incorporation by Reference

(c) The actions shall be done in accordance with Bombardier Modification Summary Package IS8Q3450000, Revision A, dated October 15, 2002. (The date of the Modification Summary Package only appears on the first page of the document.) This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Bombardier, Inc., Bombardier Regional Aircraft Division, 123 Garratt Boulevard, Downsview, Ontario M3K 1Y5, Canada. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, New York Aircraft Certification Office (ACO), 1600 Stewart Ave., suite 410, Westbury, New York; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call (202) 741-6030, or go to: [http://www.archives.gov/federal\\_register/code\\_of\\_federal\\_regulations/ibr\\_locations.html](http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html).

**Note 1:** The subject of this AD is addressed in Canadian airworthiness directive CF-2003-13, effective June 20, 2003.

#### Effective Date

(d) This amendment becomes effective on October 27 2004.

Issued in Renton, Washington, on September 14, 2004.

**Ali Bahrami,**

*Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 04-21174 Filed 9-21-04; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 2003-NE-57-AD; Amendment 39-13798; AD 2004-19-04]

RIN 2120-AA64

#### Airworthiness Directives; Rolls-Royce plc RB211-22B, RB211-524, and RB211-535 Series Turbofan Engines

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Final rule.

**SUMMARY:** The FAA is adopting a new airworthiness directive (AD) for Rolls-Royce plc (RR) RB211-22B, RB211-524, and RB211-535 series turbofan engines. This AD requires revising the Time Limits Manual for RR RB211-22B, RB211-524, and RB211-535 series turbofan engines. These revisions include required enhanced inspection of selected critical life-limited parts at each piece-part exposure. This AD results from the need to require enhanced inspection of selected critical life-limited parts of RB211-22B, RB211-524, and RB211-535 series turbofan engines. We are issuing this AD to prevent failure of critical life-limited rotating engine parts, which could result in an uncontained engine failure and damage to the airplane.

**DATES:** This AD becomes effective October 27, 2004.

**ADDRESSES:** You can get the service information identified in this AD from Rolls-Royce plc, P.O. Box 31, Derby, DE24 8BJ, United Kingdom; telephone: 011-44-1332-242424; fax: 011-44-1332-249936.

You may examine the AD docket at the FAA, New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA.

**FOR FURTHER INFORMATION CONTACT:** Christopher Spinney, Aerospace Engineer, Engine Certification Office, FAA, Engine and Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803-5299; telephone (781) 238-7175, fax (781) 238-7199.

**SUPPLEMENTARY INFORMATION:** The FAA proposed to amend 14 CFR part 39 with a proposed airworthiness directive (AD). The proposed AD applies to Rolls-Royce plc RB211-22B, RB211-524, and RB211-535 series turbofan engines. We published the proposed AD in the **Federal Register** on March 12, 2004 (69 FR 11821). That action proposed to require revisions to the Time Limits Manual for RR RB211-22B, RB211-524, and RB211-535 series turbofan engines to include required enhanced inspection of selected critical parts at each piece-part exposure.

#### Examining the AD Docket

You may examine the AD Docket (including any comments and service information), by appointment, between 8 a.m. and 4:30 p.m., Monday through Friday, except Federal holidays. See **ADDRESSES** for the location.

#### Comments

We provided the public the opportunity to participate in the development of this AD. We received no

comments on the proposal or on the determination of the cost to the public.

**Conclusion**

We have carefully reviewed the available data and determined that air safety and the public interest require adopting the AD as proposed.

**Costs of Compliance**

There are about 882 RB211-22B and RB211-524 series engines and about 1,160 RB211-535 series engines of the affected design in the worldwide fleet. We estimate that 30 RB211-22B and RB211-524 series engines and 620 RB211-535 series engines installed on airplanes of U.S. registry will be affected by this AD. We also estimate that it will take about 75 work hours per engine to perform the inspections, and that the average labor rate is \$65 per work hour. Since this is an added inspection requirement, included as part of the normal maintenance cycle, no additional part costs are involved. Based on these figures, the total cost impact of this AD on U.S. operators is estimated to be \$3,169,000.

**Regulatory Findings**

We have determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866;
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and

(3) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a summary of the costs to comply with this AD and placed it in the AD Docket. You may get a copy of this summary by sending a request to us at the address listed under **ADDRESSES**. Include "AD Docket No. 2003-NE-57-AD" in your request.

**List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Safety.

**Adoption of the Amendment**

■ Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

**PART 39—AIRWORTHINESS DIRECTIVES**

■ 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

**§ 39.13 [Amended]**

■ 2. The Federal Aviation Administration (FAA) amends § 39.13 by adding the following new airworthiness directive (AD):

**2004-19-04 Rolls-Royce plc:** Docket No. 2003-NE-57-AD.

**Effective Date**

(a) This AD becomes effective October 27, 2004.

**Affected ADs**

(b) None.

**Applicability**

(c) This AD applies to Rolls-Royce plc (RR) RB211-22B, RB211-524, and RB211-535

series turbofan engines. These engines are installed on, but not limited to, Boeing 747, 757, 767, Lockheed L-1011, and Tupolev Tu204 airplanes.

**Unsafe Condition**

(d) This AD results from the need to require enhanced inspection of selected critical life-limited parts of RB211-22B, RB211-524, and RB211-535 series turbofan engines. The actions specified in this AD are intended to prevent failure of critical life-limited rotating engine parts, which could result in an uncontained engine failure and damage to the airplane.

**Compliance**

(e) You are responsible for having the actions required by this AD performed within the compliance times specified unless the actions have already been done.

(f) Within the next 40 days after the effective date of this AD, revise the Time Limits Manual (TLM), and for air carrier operations revise the approved continuous airworthiness maintenance program, by adding the following text and the applicable table determined by engine model number:

**"GROUP A PARTS MANDATORY INSPECTION**

(1) Inspections referred to as 'Focus Inspect' in the applicable Engine Manual inspection Task are mandatory inspections for the components given below, when the conditions that follow are satisfied:

(i) When the component has been completely disassembled to piece-part level as given in the applicable disassembly procedures contained in the Engine Manual; and

(ii) The part has more than 100 recorded flight cycles in operation since the last piece-part inspection; or

(iii) The component removal was for damage or a cause directly related to its removal; or

(iv) Where serviceable used components, for which the inspection history is not fully known, are to be used again.

(2) The list of Group A Parts for RB211-22B engines is specified below:

Part nomenclature (RB211-22B series engines)	Part number	Inspected per overhaul manual task
Low Pressure Compressor Rotor Disc .....	All .....	72-31-12-200-006
Low Pressure Compressor Rotor Shaft .....	All .....	72-31-20-200-000
Intermediate Pressure Compressor Rotor Shaft Stages 1 to 5 .....	All .....	72-32-31-200-000
Intermediate Pressure Compressor Rotor Shaft Stages 6 to 7 .....	All .....	72-32-31-200-001
Intermediate Pressure Compressor Rotor Rear Stubshaft .....	All .....	72-33-31-200-000
High Pressure Compressor Rotor Stage 1 to 2 Disc Shaft .....	All .....	72-41-31-200-000
High Pressure Compressor Rotor Stage 3 Disc .....	All .....	72-41-31-200-001
High Pressure Compressor Rear Rotor Shaft Assembly .....	All .....	72-41-31-200-002
Compressor/Turbine Joint Flange Support Disc .....	All .....	72-41-31-200-003
High Pressure Turbine Disc .....	All .....	72-41-51-200-000
Intermediate Pressure Turbine Disc .....	All .....	72-51-31-200-000
Intermediate Pressure Turbine Shaft .....	All .....	72-51-33-200-000
Low Pressure Turbine Stage 1 Disc .....	All .....	72-51-61-200-000
Low Pressure Turbine Stage 2 Disc .....	All .....	72-51-61-200-001
Low Pressure Turbine Stage 3 Disc .....	All .....	72-51-61-200-002
Low Pressure Turbine Shaft .....	All .....	72-51-63-200-000

(3) The list of Group A Parts for RB211-535 series engines is specified below:

Part nomenclature (RB211-535 series engines)	Part number	Inspected per overhaul manual task
Low Pressure Compressor Rotor Disc .....	All .....	72-31-12-200-000
Low Pressure Compressor Rotor Shaft .....	All .....	72-31-20-200-000
Intermediate Pressure Compressor Rotor Shaft .....	All .....	72-32-31-200-001
Intermediate Pressure Compressor Rotor Rear Stubshaft .....	All .....	72-33-21-200-000
High Pressure Compressor Rotor Stage 1 & 2 Disc .....	All .....	72-41-31-200-000
High Pressure Compressor Rotor Stage 3 Disc .....	All .....	72-41-31-200-001
High Pressure Compressor Rear Rotor Shaft Assembly .....	All .....	72-41-31-200-002
Compressor/Turbine Joint Flange Support Disc (applicable to -535C only) .....	All .....	72-41-31-200-003
High Pressure Turbine Disc .....	All .....	72-41-51-200-000
Intermediate Pressure Turbine Rotor Disc .....	All .....	72-51-31-200-000
Intermediate Pressure Turbine Shaft .....	All .....	72-51-33-200-000
Low Pressure Turbine Stage 1 Disc .....	All .....	72-51-61-200-000
Low Pressure Turbine Stage 2 Disc .....	All .....	72-51-61-200-001
Low Pressure Turbine Stage 3 Disc .....	All .....	72-51-61-200-002
Low Pressure Turbine Shaft .....	All .....	72-51-63-200-000

(4) The list of Group A Parts for RB211-524B, -524B3, and -524B4 series engines is specified below:

Part nomenclature (RB211-524B, -524B3, and -524B4 series engines)	Part number	Inspected per overhaul manual task
Low Pressure Compressor Rotor Disc .....	All .....	<sup>1</sup> 72-31-12-200-005 <sup>2</sup> 72-31-12-200-013
Low Pressure Compressor Rotor Shaft .....	All .....	72-31-20-200-000
Intermediate Pressure Compressor Stage 1 Disc .....	All .....	72-32-31-200-000
Intermediate Pressure Compressor Stage 2 Disc .....	All .....	72-32-31-200-000
Intermediate Pressure Compressor Stage 3 Disc .....	All .....	72-32-31-200-000
Intermediate Pressure Compressor Stage 4 Disc .....	All .....	72-32-31-200-000
Intermediate Pressure Compressor Stage 5 Disc .....	All .....	72-32-31-200-001
Intermediate Pressure Compressor Rotor Shaft Stages 6 to 7 .....	All .....	72-32-31-200-001
Intermediate Pressure Compressor Front Stubshaft Drive Cone .....	All .....	72-32-31-200-008
Intermediate Pressure Compressor Rotor Rear Stubshaft .....	All .....	72-33-21-200-010
High Pressure Compressor Rotor Stage 1 to 2 Disc .....	All .....	72-41-31-200-000
High Pressure Compressor Rotor Stage 3 Disc .....	All .....	72-41-31-200-001
High Pressure Compressor Rear Rotor Shaft Assembly .....	All .....	72-41-31-200-002
High Pressure Compressor/Turbine Joint Flange Support Disc .....	All .....	72-41-31-200-006
High Pressure Turbine Bearing Inner Race Support Panel .....	All .....	72-41-51-200-005
High Pressure Turbine Disc .....	All .....	72-41-51-200-019
High Pressure Turbine Conical Shaft .....	All .....	72-41-51-200-021
Intermediate Pressure Turbine Disc .....	All .....	72-51-31-200-003
Intermediate Pressure Turbine Shaft .....	All .....	72-51-33-200-005
Low Pressure Turbine Stage 1 Disc .....	All .....	<sup>1</sup> 72-51-61-200-000 <sup>2</sup> 72-51-61-200-007
Low Pressure Turbine Stage 2 Disc .....	All .....	<sup>1</sup> 72-51-61-200-001 <sup>2</sup> 72-51-61-200-008
Low Pressure Turbine Stage 3 Disc .....	All .....	<sup>1</sup> 72-51-61-200-002 <sup>2</sup> 72-51-61-200-009
Low Pressure Turbine Shaft .....	All .....	<sup>1</sup> 72-51-63-200-000 <sup>2</sup> 72-51-63-200-003

<sup>1</sup> (Configuration 1).  
<sup>2</sup> (Configuration 2).

(5) The list of Group A Parts for RB211-524B2, -524C2, and -524D4 series engines is specified below:

Part nomenclature (RB211-524B2, -524C2, and -524D4 series engines)	Part number	Inspected per overhaul manual task
Low Pressure Compressor Rotor Disc .....	All .....	72-31-12-200-013
Low Pressure Compressor Rotor Shaft .....	All .....	72-31-20-200-000
Intermediate Pressure Compressor Stage 1 Disc .....	All .....	72-32-31-200-000
Intermediate Pressure Compressor Stage 2 Disc .....	All .....	72-32-31-200-000
Intermediate Pressure Compressor Stage 3 Disc .....	All .....	72-32-31-200-000
Intermediate Pressure Compressor Stage 4 Disc .....	All .....	72-32-31-200-000

Part nomenclature (RB211-524B2, -524C2, and -524D4 series engines)	Part number	Inspected per overhaul manual task
Intermediate Pressure Compressor Stage 5 Disc .....	All .....	72-32-31-200-001
Intermediate Pressure Compressor Rotor Shaft Stages 6 to 7 .....	All .....	72-32-31-200-001
Intermediate Pressure Compressor Front Stubshaft Drive Cone .....	All .....	72-32-31-200-008
Intermediate Pressure Compressor Rotor Rear Stubshaft .....	All .....	72-33-21-200-010
High Pressure Compressor Rotor Stage 1 to 2 Disc .....	All .....	72-41-31-200-000
High Pressure Compressor Rotor Stage 3 Disc .....	All .....	72-41-31-200-001
High Pressure Compressor Rear Rotor Shaft Assembly .....	All .....	72-41-31-200-002
High Pressure Compressor/Turbine Joint Flange Support Disc .....	All .....	72-41-31-200-006
High Pressure Turbine Bearing Inner Race Support Panel .....	All .....	72-41-51-200-005
High Pressure Turbine Disc .....	All .....	72-41-51-200-019
High Pressure Turbine Conical Shaft .....	All .....	72-41-51-200-021
Intermediate Pressure Turbine Rotor Disc .....	All .....	72-51-31-200-003
Intermediate Pressure Turbine Shaft .....	All .....	72-51-33-200-005
Low Pressure Turbine Stage 1 Disc .....	All .....	72-51-61-200-007
Low Pressure Turbine Stage 2 Disc .....	All .....	72-51-61-200-008
Low Pressure Turbine Stage 3 Disc .....	All .....	72-51-61-200-009
Low Pressure Turbine Shaft .....	All .....	72-51-63-200-003

(6) The list of Group A Parts for RB211-524G and -524H series engines is specified below:

Part nomenclature (RB211-524G and -524H series engines)	Part number	Inspected per overhaul manual task
Low Pressure Compressor Rotor Disc .....	All .....	72-31-12-200-000
Low Pressure Compressor Rotor Shaft .....	All .....	72-31-20-200-000
Intermediate Pressure Compressor Stage 1 Disc .....	All .....	72-32-31-200-000
Intermediate Pressure Compressor Stage 2 Disc .....	All .....	72-32-31-200-000
Intermediate Pressure Compressor Stage 3 Disc .....	All .....	72-32-31-200-000
Intermediate Pressure Compressor Stage 4 Disc .....	All .....	72-32-31-200-000
Intermediate Pressure Compressor Stage 5 Disc .....	All .....	72-32-31-200-000
Intermediate Pressure Compressor Rotor Shaft Stages 6 to 7 .....	All .....	72-32-31-200-001
Intermediate Pressure Compressor Front Stubshaft Drive Cone .....	All .....	72-32-31-200-008
Intermediate Pressure Compressor Rotor Rear Stubshaft .....	All .....	72-33-21-200-010
High Pressure Compressor Rotor Stage 1 to 2 Disc .....	All .....	<sup>1</sup> 72-41-31-200-000
High Pressure Compressor Rotor Stage 3 Disc .....	All .....	<sup>1</sup> 72-41-31-200-001
High Pressure Compressor Rear Rotor Shaft Assembly .....	All .....	<sup>1</sup> 72-41-31-200-002
Compressor/Turbine Joint Flange Support Disc .....	All .....	<sup>1</sup> 72-41-31-200-003
High Pressure Compressor Rotor Shaft Assembly .....	All .....	<sup>2</sup> 72-41-31-200-014
High Pressure Turbine Disc .....	All .....	<sup>1</sup> 72-41-51-200-010
		<sup>2</sup> 72-41-51-200-024
Intermediate Pressure Turbine Disc .....	All .....	72-51-31-200-003
Intermediate Pressure Turbine Shaft .....	All .....	72-51-33-200-005
Low Pressure Turbine Stage 1 Disc .....	All .....	72-51-61-200-007
Low Pressure Turbine Stage 2 Disc .....	All .....	72-51-61-200-008
Low Pressure Turbine Stage 3 Disc .....	All .....	72-51-61-200-009
Low Pressure Turbine Shaft .....	All .....	72-51-63-200-003 <sup>1</sup>

<sup>1</sup> (Configuration 1).

<sup>2</sup> (Configuration 2).

**Alternative Methods of Compliance**

(g) You must perform these mandatory inspections using the TLM and the applicable Engine Manual unless you receive approval to use an alternative method of compliance under paragraph (h) of this AD. Section 43.16 of the Federal Aviation Regulations (14 CFR 43.16) may not be used to approve alternative methods of compliance or adjustments to the times in which these inspections must be performed.

(h) The Manager, Engine Certification Office, has the authority to approve alternative methods of compliance for this AD if requested using the procedures found in 14 CFR 39.19.

**Maintaining Records of the Mandatory Inspections**

(i) You have met the requirements of this AD by using a TLM changed as specified in paragraph (f) of this AD, and, for air carriers operating under part 121 of the Federal Aviation Regulations (14 CFR part 121), by modifying your continuous airworthiness maintenance plan to reflect those changes. You must maintain records of the mandatory inspections that result from those changes to the TLM according to the regulations governing your operation. You do not need to record each piece-part inspection as compliance to this AD. For air carriers operating under part 121, you may use either the system established to comply with section 121.369 or use an alternative system

that your principal inspector has accepted if that alternative system:

- (1) Includes a method for preserving and retrieving the records of the inspections resulting from this AD; and
  - (2) Meets the requirements of § 121.369(c); and
  - (3) Maintains the records either indefinitely or until the work is repeated.
- (j) These record keeping requirements apply only to the records used to document the mandatory inspections required as a result of revising the Time Limits Manual as specified in paragraph (f) of this AD, and do not alter or amend the record keeping requirements for any other AD or regulatory requirement.

**Material Incorporated by Reference**

(k) None.

**Related Information**

(l) Civil Aviation Authority (CAA) airworthiness directives No. G-2003-0006, dated September 18, 2003, No. G-2003-0009, dated September 19, 2003, and No. G-2003-0007, dated September 18, 2003 also address the subject of this AD.

Issued in Burlington, Massachusetts, on September 10, 2004.

**Francis A. Favara,**

*Acting Manager, Engine and Propeller Directorate, Aircraft Certification Service.*

[FR Doc. 04-21173 Filed 9-21-04; 8:45 am]

**BILLING CODE 4910-13-P**

**DEPARTMENT OF TRANSPORTATION****Federal Aviation Administration****14 CFR Part 39**

[Docket No. 2003-NE-54-AD; Amendment 39-13802; AD 2004-19-08]

RIN 2120-AA64

**Airworthiness Directives; Rolls-Royce plc RB211 Trent 800 Series Turbofan Engines**

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Final rule.

**SUMMARY:** The FAA is adopting a new airworthiness directive (AD) for Rolls-Royce plc (RR) RB211 Trent 800 series turbofan engines. This AD requires revising the Time Limits Manual for RR RB211 Trent 800 series turbofan engines. These revisions include required enhanced inspection of selected critical life-limited parts at each piece-part exposure. This AD results from the need to require enhanced inspection of selected critical life-limited parts of RB211 Trent 800 series turbofan engines. We are issuing this AD to prevent failure of critical life-limited rotating engine parts, which could result in an uncontained engine failure and damage to the airplane.

**DATES:** This AD becomes effective October 27, 2004.

**ADDRESSES:** You can get the service information identified in this AD from Rolls-Royce plc, P.O. Box 31, Derby, DE24 8BJ, United Kingdom; telephone: 011-44-1332-242424; fax: 011-44-1332-249936.

You may examine the AD docket at the FAA, New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA.

**FOR FURTHER INFORMATION CONTACT:** Christopher Spinney, Aerospace

Engineer, Engine Certification Office, FAA, Engine and Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803-5299; telephone (781) 238-7175, fax (781) 238-7199.

**SUPPLEMENTARY INFORMATION:** The FAA proposed to amend 14 CFR part 39 with a proposed AD. The proposed AD applies to RR RB211 Trent 800 series turbofan engines. We published the proposed AD in the **Federal Register** on March 4, 2004 (69 FR 10179). That action proposed to require revising the Time Limits Manual for RR RB211 Trent 800 series turbofan engines to include required enhanced inspection of selected critical life-limited parts at each piece-part exposure.

**Examining the AD Docket**

You may examine the AD Docket (including any comments and service information), by appointment, between 8 a.m. and 4:30 p.m., Monday through Friday, except Federal holidays. See **ADDRESSES** for the location.

**Comments**

We provided the public the opportunity to participate in the development of this AD. We received no comments on the proposal or on the determination of the cost to the public.

**Conclusion**

We have carefully reviewed the available data and determined that air safety and the public interest require adopting the AD as proposed.

**Costs of Compliance**

There are about 350 engines of the affected design in the worldwide fleet. We estimate that 90 engines installed on airplanes of U.S. registry are affected by this AD. We also estimate that it will take about 75 work hours per engine to perform the inspections, and that the average labor rate is \$65 per work hour. Since this is an added inspection requirement, included as part of the normal maintenance cycle, no additional part costs are involved. Based on these figures, the total cost impact of the AD on U.S. operators is estimated to be \$438,750.

**Regulatory Findings**

We have determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866;
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
- (3) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a summary of the costs to comply with this AD and placed it in the AD Docket. You may get a copy of this summary by sending a request to us at the address listed under **ADDRESSES**. Include "AD Docket No. 2003-NE-54-AD" in your request.

**List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Safety.

**Adoption of the Amendment**

■ Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

**PART 39—AIRWORTHINESS DIRECTIVES**

■ 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

**§ 39.13 [Amended]**

■ 2. The Federal Aviation Administration (FAA) amends § 39.13 by adding the following new airworthiness directive (AD):

**2004-19-08 Rolls-Royce plc:** Amendment 39-13802. Docket No. 2003-NE-54-AD.

**Effective Date**

(a) This AD becomes effective October 27, 2004.

**Affected ADs**

(b) None.

**Applicability**

(c) This AD applies to Rolls-Royce plc (RR) RB211 Trent 800 series turbofan engines. These engines are installed on, but not limited to, Boeing 777 airplanes.

**Unsafe Condition**

(d) This AD results from the need to require enhanced inspection of selected critical life-limited parts of RB211 Trent 800 series turbofan engines. The actions specified in this AD are intended to prevent critical life-limited rotating engine part failure, which could result in an uncontained engine failure and damage to the airplane.

**Compliance**

(e) You are responsible for having the actions required by this AD performed within the compliance times specified unless the actions have already been done.

(f) Within the next 40 days after the effective date of this AD, revise the Time