

(1) Install a self-adhesive placard of the size and in the location depicted in Figure 4 of ASB No. 31A005 with the following text in white letters on a red background: "CAUTION: ON GROUND OPERATION OF EMERGENCY LANDING GEAR PUMP IS TIME LIMITED—SEE OPERATING LIMITATIONS" and

(2) Revise the Operating Limitations by inserting the following text into the Rotorcraft Flight Manual (RFM):

"(i) Limit the emergency landing gear pump (pump) to 10 minutes of continuous operation.

(ii) When the pump is continuously operated from 1 to 5 minutes, allow it to cool for 15 minutes before further use.

(iii) When the pump is continuously operated from 5 to 10 minutes, allow it to cool for 30 minutes before further use."

Note 1: Modifying the electric wiring covered by Alert Telex No. 31A005R1, dated September 19, 2002, led to inhibiting the protective thermal switch of the electric pump. This resulted in the need for a limitation placard. The purpose of the limitation placard is to remind operators about the on-ground operating limitations that apply to the electric pump.

(c) Within 10 months, modify the wiring and wiring harness by following the Accomplishment Instructions, paragraphs 2.A. and 2.B., of Eurocopter EC155 ASB No. 31A008, dated August 20, 2003 (ASB No. 31A008). If you made the temporary modifications described in paragraph (b) of this AD, remove the placard from the helicopter and the limitations inserted in the RFM as a result of paragraphs (b)(1) and (b)(2) of this AD.

(d) Permanently modifying the wiring and wiring harness following ASB No. 31A008 is terminating action for the requirements of this AD.

(e) To request a different method of compliance or a different compliance time for this AD, follow the procedures in 14 CFR 39.19. Contact the Safety Management Group, Rotorcraft Directorate, FAA, for information about previously approved alternative methods of compliance.

(f) Special flight permits will not be issued.

(g) Cleaning the auxiliary system unit and modifying the wiring and wiring harness shall be done by following Eurocopter EC155 Alert Service Bulletin No. 31A005 and Alert Service Bulletin No. 31A008, both dated August 20, 2003, as applicable. The Director of the Federal Register approved this incorporation by reference in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from American Eurocopter Corporation, 2701 Forum Drive, Grand Prairie, Texas 75053-4005, telephone (972) 641-3460, fax (972) 641-3527. Copies may be inspected at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

(h) This amendment becomes effective on September 3, 2004.

Note 2: The subject of this AD is addressed in Direction Generale De L'Aviation Civile (France) AD Nos. 2002-515(A) R1 and 2003-323(A), both dated September 3, 2003.

Issued in Fort Worth, Texas, on July 16, 2004.

David A. Downey,

Manager, Rotorcraft Directorate, Aircraft Certification Service.

[FR Doc. 04-17219 Filed 7-29-04; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2002-NM-302-AD; Amendment 39-13751; AD 2004-15-17]

RIN 2120-AA64

Airworthiness Directives; Fokker Model F27 Mark 100, 200, 300, 400, 500, 600, and 700 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to certain Fokker Model F27 Mark 100, 200, 300, 400, 500, 600, and 700 series airplanes, that requires a one-time inspection to determine the part number of the engine mounting frames, brace struts, and attachment fittings; and related corrective action. This action is necessary to ensure the structural integrity of the engine-to-wing load path and prevent possible separation of the engine from the airplane. This action is intended to address the identified unsafe condition.

DATES: Effective September 3, 2004.

The incorporation by reference of a certain publication listed in the regulations is approved by the Director of the Federal Register as of September 3, 2004.

ADDRESSES: The service information referenced in this AD may be obtained from Fokker Services B.V., P.O. Box 231, 2150 AE Nieuw-Vennep, the Netherlands. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call (202) 741-6030, or go to: http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

FOR FURTHER INFORMATION CONTACT: Tom Groves, Aerospace Engineer, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-1503; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to certain Fokker Model F27 Mark 100, 200, 300, 400, 500, 600, and 700 series airplanes was published in the **Federal Register** on June 2, 2004 (69 FR 31053). That action proposed to require a one-time inspection to determine the part number of the engine mounting frames, brace struts, and attachment fittings; and related corrective action.

Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were submitted in response to the proposal or the FAA's determination of the cost to the public.

Conclusion

The FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

Cost Impact

The FAA estimates that 41 airplanes of U.S. registry will be affected by this AD, that it will take approximately 4 work hours per airplane to accomplish the required actions, and that the average labor rate is \$65 per work hour. Based on these figures, the cost impact of the AD on U.S. operators is estimated to be \$10,660, or \$260 per airplane.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up, planning time, or time necessitated by other administrative actions.

Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various

levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

■ Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. Section 39.13 is amended by adding the following new airworthiness directive:

2004-15-17 Fokker Services B.V.:

Amendment 39-13751. Docket 2002-NM-302-AD.

Applicability: Model F27 Mark 100, 200, 300, 400, 500, 600, and 700 series airplanes; certificated in any category; on which one or more of the modifications specified in paragraph 1.A.(1) of Fokker Service Bulletin F27/54-53, dated February 15, 2002, has been done.

Compliance: Required as indicated, unless accomplished previously.

To ensure the structural integrity of the engine-to-wing load path and prevent possible separation of the engine from the airplane, accomplish the following:

One-Time Inspection

(a) Within 24 months after the effective date of this AD: Do a one-time general visual inspection to determine the part numbers of the engine mounting frames, brace struts, and attachment fittings; per the Accomplishment Instructions of Fokker Service Bulletin F27/54-53, dated February 15, 2002. Do the inspection and corrective action per the

Accomplishment Instructions of the service bulletin. Do the related corrective action before further flight.

Note 1: For the purposes of this AD, a general visual inspection is defined as: "A visual examination of an interior or exterior area, installation, or assembly to detect obvious damage, failure, or irregularity. This level of inspection is made from within touching distance unless otherwise specified. A mirror may be necessary to enhance visual access to all exposed surfaces in the inspection area. This level of inspection is made under normally available lighting conditions such as daylight, hangar lighting, flashlight, or droplight and may require removal or opening of access panels or doors. Stands, ladders, or platforms may be required to gain proximity to the area being checked."

Related Service Information

Note 2: Fokker Service Bulletin F27/54-53, dated February 15, 2002, references Fokker Service Bulletin 51-24, dated December 1, 1971, as the appropriate source of service information for installing a new, improved engine mounting frame; and Fokker Service Bulletin F27/54-26, Revision 5, dated September 30, 2001, as the appropriate source of service information for installing new, improved, stronger brace struts and brackets.

Parts Installation

(b) As of the effective date of this AD, no person may install on any airplane an engine mounting frame, brace strut, or attachment fitting unless that part has been identified as appropriate for the airplane configuration, as specified in the Accomplishment Instructions of Fokker Service Bulletin F27/54-53, dated February 15, 2002.

Alternative Methods of Compliance

(c) In accordance with 14 CFR 39.19, the Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate, is authorized to approve alternative methods of compliance for this AD.

Incorporation by Reference

(d) The actions shall be done in accordance with Fokker Service Bulletin F27/54-53, dated February 15, 2002. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Fokker Services B.V., P.O. Box 231, 2150 AE Nieuw-Vennep, the Netherlands. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call (202) 741-6030, or go to: http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

Note 3: The subject of this AD is addressed in Dutch airworthiness directive 2002-067, dated May 31, 2002.

Effective Date

(e) This amendment becomes effective on September 3, 2004.

Issued in Renton, Washington, on July 19, 2004.

Kevin M. Mullin,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 04-17220 Filed 7-29-04; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2002-NM-344-AD; Amendment 39-13750; AD 2004-15-16]

RIN 2120-AA64

Airworthiness Directives; Airbus Model A310 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to certain Airbus Model A310 series airplanes, that requires modification of certain wires in the right-hand wing. This action is necessary to ensure that fuel quantity indication wires are properly separated from wires carrying 115-volt alternating current (AC). Improper separation of such wires, in the event of wire damage, could lead to a short circuit and a possible ignition source, which could result in a fire in the airplane. This action is intended to address the identified unsafe condition.

DATES: Effective September 3, 2004.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of September 3, 2004.

ADDRESSES: The service information referenced in this AD may be obtained from Airbus, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call (202) 741-6030, or go to: http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

FOR FURTHER INFORMATION CONTACT: Tim Backman, Aerospace Engineer, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington