

Weekly Compilation of Presidential Documents.

■ For the reasons discussed in the preamble, the Administrative Committee of the Federal Register, with the approval of the Archivist of the United States and the Attorney General, is amending part 11 of chapter I of title 1 of the Code of Federal Regulations as set forth below:

#### PART 11—SUBSCRIPTIONS

■ 1. The authority citation for part 11 continues to read as follows:

**Authority:** 44 U.S.C. 1506; sec. 6, E.O. 10530, 19 FR 2709, 3 CFR, 1954–1958 Comp., p. 189.

■ 2. In § 11.2, revise paragraph (a) to read as follows:

##### § 11.2 Federal Register.

(a) The subscription price for the paper edition of the daily **Federal Register** is \$749 per year. A combined subscription to the daily **Federal Register**, the monthly **Federal Register** Index, and the monthly LSA (List of CFR Sections Affected) is \$808 per year for the paper edition, or \$165 per year for the microfiche edition. Six-month subscriptions for the paper and microfiche editions are also available at one-half the annual rate. Those prices exclude postage. The prevailing postal rates will be applied to orders according to the delivery method requested. The price of a single copy of the daily **Federal Register**, including postage, is based on the number of pages: \$11 for an issue containing less than 200 pages; \$22 for an issue containing 200 to 400 pages; and \$33 for an issue containing more than 400 pages. Single issues of the microfiche edition may be purchased for \$3 per copy, including postage.

\* \* \* \* \*

■ 3. In § 11.3, revise paragraph (a) to read as follows:

##### § 11.3 Code of Federal Regulations.

(a) The subscription price for a complete set of the Code of Federal Regulations is \$1,019 per year for the bound, paper edition, or \$247 per year for the microfiche edition. Those prices exclude postage. The prevailing postal rates will be applied to orders according to the delivery method requested. The Government Printing Office sells individual volumes of the paper edition of the Code of Federal Regulations at prices determined by the Superintendent of Documents under the general direction of the Administrative Committee. The price of a single volume

of the microfiche edition is \$4 per copy, including postage.

\* \* \* \* \*

■ 4. In § 11.6, revise paragraph (a) to read as follows:

##### § 11.6 Weekly Compilation of Presidential Documents.

(a) The subscription price for the paper edition of the Weekly Compilation of Presidential Documents is \$113 per year, excluding postage. The prevailing postal rates will be applied to orders according to the delivery method requested. The price of an individual copy is \$5, including postage.

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■ 5. Revise § 11.7 to read as follows:

##### § 11.7 Federal Register Index.

The annual subscription price for the monthly **Federal Register** Index, purchased separately, in paper form, is \$29. The price excludes postage. The prevailing postal rates will be applied to orders according to the delivery method requested.

■ 6. Revise § 11.8 to read as follows:

##### § 11.8 LSA (List of CFR Sections Affected).

The annual subscription price for the monthly LSA (List of CFR Sections Affected), purchased separately, in paper form, is \$30. The price excludes postage. The prevailing postal rates will be applied to orders according to the delivery method requested.

**John W. Carlin,**

*Chairman, Administrative Committee of the Federal Register.*

**Bruce R. James,**

*Member, Administrative Committee of the Federal Register.*

**Rosemary Hart,**

*Member, Administrative Committee of the Federal Register.*

Approved by:

**James B. Comey,**

*Deputy Attorney General.*

**John W. Carlin,**

*Archivist of the United States.*

[FR Doc. 04–6198 Filed 3–17–04; 8:45 am]

BILLING CODE 1505–01–P

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 2003–NE–56–AD; Amendment 39–13525; AD 2004–05–30]

RIN 2120–AA64

#### Airworthiness Directives; Rolls-Royce plc RB211 Trent 500 Series Turbofan Engines

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Final rule; request for comments.

**SUMMARY:** The FAA is adopting a new airworthiness directive (AD) for Rolls-Royce plc (RR) RB211–Trent 500 series turbofan engines. This AD requires revising the Time Limits Manual for RR RB211 Trent 500 series turbofan engines. These revisions include required enhanced inspection of selected critical life-limited parts at each piece-part exposure. This AD results from the need to require enhanced inspection of selected critical life-limited parts of RR Trent 500 series turbofan engines. We are issuing this AD to prevent failure of critical life-limited rotating engine parts, which could result in an uncontained engine failure and damage to the airplane.

**DATES:** Effective April 2, 2004.

We must receive any comments on this AD by May 17, 2004.

**ADDRESSES:** Use one of the following addresses to submit comments on this AD:

- By mail: the Federal Aviation Administration (FAA), New England Region, Office of the Regional Counsel, Attention: Rules Docket No. 2003–NE–56–AD, 12 New England Executive Park, Burlington, MA 01803–5299.

- By fax: (781) 238–7055.

- By e-mail: [9-ane-adcomment@faa.gov](mailto:9-ane-adcomment@faa.gov).

You may examine the AD docket, by appointment, at the FAA, New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA.

#### FOR FURTHER INFORMATION CONTACT:

Christopher Spinney, Aerospace Engineer, Engine Certification Office, FAA, Engine and Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803–5299; telephone (781) 238–7175, fax (781) 238–7199.

**SUPPLEMENTARY INFORMATION:** A recent FAA study analyzing 15 years of accident data for transport category airplanes identified several root causes for a failure mode that can result in

serious safety hazards to transport category airplanes. This study identified uncontained failure of critical life-limited rotating engine parts as the leading engine-related safety hazard to airplanes. Uncontained engine failures have resulted from undetected cracks in rotating parts that started and grew to failure. Cracks can start from causes such as unintended excessive stress from the original design, or they may start from stresses induced from material flaws, handling, or damage from machining operations. The failure of a rotating part can present a significant safety hazard to the airplane by release of high-energy fragments that could injure passengers or crew by penetration of the cabin, damage flight control surfaces, sever flammable fluid lines, or otherwise compromise the airworthiness of the airplane.

Based on these findings, the FAA, with the concurrence of the Civil Aviation Authority (CAA), which is the Airworthiness Authority for the United Kingdom (U.K.), has developed an intervention strategy to significantly reduce uncontained engine failures. This intervention strategy was developed after consultation with industry and will be used as a model for future initiatives. The intervention strategy is to conduct enhanced, nondestructive inspections of rotating parts, which could most likely result in a safety hazard to the airplane in the event of a part fracture. We are considering the need for additional rulemaking. We might issue future ADs to introduce additional intervention strategies to further reduce or eliminate uncontained engine failures.

Properly focused enhanced inspections require identification of the parts whose failure presents the highest safety hazard to the airplane, identifying the most critical features to inspect on these parts, and utilizing inspection procedures and techniques that improve crack detection. The CAA, with close cooperation of RR, has completed a detailed analysis that identifies the most safety significant parts and features, and the most appropriate inspection methods.

Critical life-limited high-energy rotating parts are currently subject to some form of recommended crack inspection when exposed during engine maintenance or disassembly. The inspections currently recommended by the manufacturer will become mandatory for those parts listed in the compliance section as a result of this AD. Furthermore, we intend that additional mandatory enhanced inspections resulting from this AD will serve as an adjunct to the existing

inspections. We have determined that the enhanced inspections will significantly improve the probability of crack detection on disassembled parts during maintenance. All mandatory inspections must be conducted in accordance with detailed inspection procedures prescribed in the manufacturer's Engine Manual.

Additionally, this AD will:

- Allow air carriers that operate under the provisions of 14 CFR part 121 with an FAA-approved continuous airworthiness maintenance program, and maintenance facilities to verify completion of the enhanced inspections.

- Allow the air carrier or maintenance facility to retain the maintenance records that include the inspections resulting from this AD, if the records include the date and signature of the person who performed the maintenance action.

- Require retaining the records with the maintenance records of the part, engine module, or engine until the task is repeated.

- Establish a method of record preservation and retrieval typically used in existing continuous airworthiness maintenance programs.

- Require adding instructions in an air carrier's maintenance manual on how to implement and integrate this record preservation and retrieval system into the air carrier's record keeping system.

For engines or engine modules that are approved for return to service by an authorized FAA-certificated entity, and that are acquired by an operator after the effective date of the AD, you will not need to perform the mandatory enhanced inspections until the next piece-part opportunity. For example, you will not have to disassemble to piece-part level, an engine or module returned to service by an FAA-certificated facility simply because that engine or module was previously operated by an entity not required to comply with this AD. Furthermore, we intend that operators perform the enhanced inspections of these parts at the next piece-part opportunity after the initial acquisition, installation, and removal of the part after the effective date of this AD. For piece parts not approved for return to service before the effective date of this AD, the AD requires that you perform the mandatory enhanced inspections before approval of those parts for return to service. The AD allows installation of piece parts approved for return to service before the effective date of this AD. However, the AD requires an enhanced inspection at the next piece-part opportunity.

This AD requires, within the next 40 days after the effective date of this AD, revisions to the Time Limits Manual.

#### **FAA's Determination and Requirements of This AD**

Although no airplanes that are registered in the United States use these engines, the possibility exists that the engines could be used on airplanes that are registered in the United States in the future. The unsafe condition described previously is likely to exist or develop on other RR RB211 Trent 500 series turbofan engines of the same type design. We are issuing this AD to prevent critical life-limited rotating engine part failure, which could result in an uncontained engine failure and damage to the airplane.

#### **FAA's Determination of the Effective Date**

Since there are currently no domestic operators of this engine model, notice and opportunity for public comment before issuing this AD are unnecessary. Therefore, a situation exists that allows the immediate adoption of this regulation.

#### **Changes to 14 CFR Part 39—Effect on the AD**

On July 10, 2002, we issued a new version of 14 CFR part 39 (67 FR 47998, July 22, 2002), which governs our AD system. This regulation now includes material that relates to special flight permits, alternative methods of compliance, and altered products. This material previously was included in each individual AD. Since this material is included in 14 CFR part 39, we will not include it in future AD actions.

#### **Comments Invited**

This AD is a final rule that involves requirements affecting flight safety and was not preceded by notice and an opportunity for public comment; however, we invite you to submit any written relevant data, views, or arguments regarding this AD. Send your comments to an address listed under **ADDRESSES**. Include "AD Docket No. 2003-NE-56-AD" in the subject line of your comments. If you want us to acknowledge receipt of your mailed comments, send us a self-addressed, stamped postcard with the docket number written on it; we will date-stamp your postcard and mail it back to you. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of the rule that might suggest a need to modify it. If a person contacts us through a verbal communication, and that contact relates to a substantive part

of this AD, we will summarize the contact and place the summary in the docket. We will consider all comments received by the closing date and may amend the AD in light of those comments.

We are reviewing the writing style we currently use in regulatory documents. We are interested in your comments on whether the style of this document is clear, and your suggestions to improve the clarity of our communications with you. You may get more information about plain language at <http://www.faa.gov/language> and <http://www.plainlanguage.gov>.

**Examining the AD Docket**

You may examine the AD Docket (including any comments and service information), by appointment, between 8 a.m. and 4:30 p.m., Monday through Friday, except Federal holidays. See ADDRESSES for the location.

**Regulatory Findings**

We have determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that the regulation:

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
3. Will not have a significant economic impact, positive or negative,

on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a summary of the costs to comply with this AD and placed it in the AD Docket. You may get a copy of this summary by sending a request to us at the address listed under ADDRESSES. Include "AD Docket No. 2003-NE-56-AD" in your request.

**List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Safety.

**Adoption of the Amendment**

Accordingly, under the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

**PART 39—AIRWORTHINESS DIRECTIVES**

- 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

**§ 39.13 [Amended]**

- 2. The FAA amends § 39.13 by adding the following new airworthiness directive:

**2004-05-30 Rolls-Royce plc:** Amendment 39-13525. Docket No. 2003-NE-56-AD.

**Effective Date**

- (a) This airworthiness directive (AD) becomes effective April 2, 2004.

**Affected ADs**

- (b) None.

**Applicability**

- (c) This AD applies to Rolls-Royce plc (RR) Trent 500 series turbofan engines. These

engines are installed on, but not limited to, Airbus A340 series airplanes.

**Unsafe Condition**

(d) This AD results from the need to require enhanced inspection of selected critical life-limited parts of RR Trent 500 series turbofan engines. We are issuing this AD to prevent critical life-limited rotating engine part failure, which could result in an uncontained engine failure and damage to the airplane.

**Compliance**

(e) You are responsible for having the actions required by this AD performed within the compliance times specified unless the actions have already been done.

(f) Within the next 40 days after the effective date of this AD, revise the Time Limits Manual (TLM), and for air carrier operations revise the approved continuous airworthiness maintenance program, by adding the following:

"GROUP A PARTS MANDATORY INSPECTION

B. Inspections referred to as 'Focus Inspect' in the applicable Engine Manual inspection Task are mandatory inspections for the components given below, when the conditions that follow are satisfied:

(1) When the component has been completely disassembled to piece part level as given in the applicable disassembly procedures contained in the Engine Manual; and

(2) The part has more than 100 recorded flight cycles in operation since the last piece part inspection; or

(3) The component removal was for damage or a cause directly related to its removal; or

(4) Where serviceable used components, for which the inspection history is not fully known, are to be used again.

C. The list of Group A Parts is specified below:

Part nomenclature	Part No.	Inspected per overhaul manual task
Low Pressure Compressor Rotor Disk .....	All .....	72-31-16-200-801
Low Pressure Compressor Rotor Shaft .....	All .....	72-31-20-200-801
Intermediate Pressure Compressor Rotor Shaft .....	All .....	72-32-31-200-801
Intermediate Pressure Rear Shaft .....	All .....	72-33-21-200-801
High Pressure Compressor Stage 1 to 4 Rotor Disks Shaft .....	All .....	72-41-31-200-803
High Pressure Compressor Stage 5 & 6 Disks and Cone .....	All .....	72-41-31-200-801
High Pressure Turbine Rotor Disk .....	All .....	72-41-51-200-801
High Pressure Turbine Front Coverplate .....	All .....	72-41-51-200-806
Intermediate Pressure Turbine Rotor Disk .....	All .....	72-51-31-200-801
Intermediate Pressure Turbine Rotor Shaft .....	All .....	72-51-33-200-801
Low Pressure Turbine Stage 1 Rotor Disk .....	All .....	72-52-31-200-801
Low Pressure Turbine Stage 2 Rotor Disk .....	All .....	72-52-31-200-802
Low Pressure Turbine Stage 3 Rotor Disk .....	All .....	72-52-31-200-803
Low Pressure Turbine Stage 4 Rotor Disk .....	All .....	72-52-31-200-804
Low Pressure Turbine Stage 5 Rotor Disk .....	All .....	72-52-31-200-805
Low Pressure Turbine Rotor Shaft .....	All .....	72-52-33-200-801"

**Alternative Methods of Compliance**

(g) You must perform these mandatory inspections using the TLM and the applicable Engine Manual unless you receive

approval to use an alternative method of compliance under paragraph (h) of this AD. Section 43.16 of the Federal Aviation Regulations (14 CFR 43.16) may not be used

to approve alternative methods of compliance or adjustments to the times in which these inspections must be performed.

(h) The Manager, Engine Certification Office, has the authority to approve alternative methods of compliance for this AD if requested using the procedures found in 14 CFR 39.19.

#### Maintaining Records of the Mandatory Inspections

(i) You have met the requirements of this AD by using a TLM changed as specified in paragraph (f) of this AD, and, for air carriers operating under part 121 of the Federal Aviation Regulations (14 CFR part 121), by modifying your continuous airworthiness maintenance plan to reflect those changes. You must maintain records of the mandatory inspections that result from those changes to the TLM according to the regulations governing your operation. You do not need to record each piece-part inspection as compliance to this AD. For air carriers operating under part 121, you may use either the system established to comply with section 121.369 or use an alternative system that your principal maintenance inspector has accepted if that alternative system:

(1) Includes a method for preserving and retrieving the records of the inspections resulting from this AD; and

(2) Meets the requirements of section 121.369(c); and

(3) Maintains the records either indefinitely or until the work is repeated.

(j) These record keeping requirements apply only to the records used to document the mandatory inspections required as a result of revising the Time Limits Manual as specified in paragraph (f) of this AD, and do not alter or amend the record keeping requirements for any other AD or regulatory requirement.

#### Related Information

(k) CAA airworthiness directive G-2003-0005, dated September 18, 2003, also addresses the subject of this AD.

Issued in Burlington, Massachusetts, on March 5, 2004.

**Jay J. Pardee,**

*Manager, Engine and Propeller Directorate, Aircraft Certification Service.*

[FR Doc. 04-5620 Filed 3-17-04; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 2002-NM-18-AD; Amendment 39-13528; AD 2004-06-02]

RIN 2120-AA64

#### Airworthiness Directives; Airbus Model A319, A320, and A321 Series Airplanes

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD);

applicable to certain Airbus Model A319, A320, and A321 series airplanes; that requires replacing the lower guide rod fittings at the rear passenger doors with improved fittings. This action is necessary to prevent failure of a lower guide rod fitting, which could cause a rear passenger door to jam during opening, delaying an emergency evacuation and resulting in injury to passengers or crew members. This action is intended to address the identified unsafe condition.

**DATES:** Effective April 22, 2004.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of April 22, 2004.

**ADDRESSES:** The service information referenced in this AD may be obtained from Airbus, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Dan Rodina, Aerospace Engineer, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-2125; fax (425) 227-1149.

**SUPPLEMENTARY INFORMATION:** A

proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to certain Airbus Model A319, A320, and A321 series airplanes, was published in the **Federal Register** on December 17, 2003 (68 FR 70213). That action proposed to require replacing the upper guide rod fittings at the rear passenger doors with improved fittings.

#### Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due consideration has been given to the comments received.

#### Support for the Proposed AD

Three commenters support the proposed AD.

#### Request To Revise Identification of Affected Parts

One commenter requests that the FAA revise the proposed AD to refer to the correct parts to be replaced. The commenter notes that, while the proposed AD states that it is the upper

guide rod fitting on each rear passenger door that must be replaced, Airbus Service Bulletin A320-53-1154, Revision 2, dated March 7, 2003 (which is the applicable source of service information referenced in the proposed AD), refers to the lower guide rod fitting.

We concur with the commenter's request to revise this AD to refer to the lower guide rod fitting instead of the upper. The references to "upper guide rod fitting" in the proposed AD are consistent with the terminology in French airworthiness directive 2001-634(B), dated December 26, 2001, which refers to the original issue of the service bulletin, Airbus Service Bulletin A320-53-1154, dated July 12, 2001. The original issue of the service bulletin erroneously referred to the upper guide rod fitting instead of the lower. We have revised the preamble and body of this final rule to contain the correct terminology. Also, we have revised paragraph (b) of this AD to clarify that replacements, of the lower guide rod fitting only, accomplished per previous revisions of the service bulletin are acceptable for compliance with this AD. We find that these changes do not increase the scope of the AD because the service information referenced in the proposed AD, Airbus Service Bulletin A320-53-1154, Revision 2, contains the correct instructions for accomplishing the required actions.

#### Request To Extend Compliance Time

One commenter requests that we revise the compliance time for the requirements of this AD from 22 months to 5 years, and suggests that we add repetitive inspections for cracking of the lower guide arm fittings as an interim action until the modification is accomplished. The commenter would like to incorporate this modification into the 5-year heavy maintenance visit for its fleet, and extending the compliance time for the proposed AD would accommodate the commenter's schedule. The commenter states that repetitive inspections for cracking at intervals not to exceed 600 flight hours, and accomplishment of the replacement of the lower guide arm fittings within 5 years, would significantly reduce the possibility of door failure and would not compromise safety.

We do not concur with the commenter's request. The operator did not submit appropriate inspection procedures to justify that its request would adequately ensure an acceptable level of safety. An affected operator may request approval of an alternative method of compliance or adjustment of the compliance time for this AD if the