

*Compliance:* Required as indicated, unless accomplished previously.

To prevent uncommanded seat movement during takeoff and/or landing, which could result in interference with the operation of the airplane and consequent temporary loss of control of the airplane, accomplish the following:

#### Service Bulletin References

(a) The term "service bulletin," as used in this AD, means the Accomplishment Instructions of the applicable service bulletins listed in Table 1 of this AD.

#### Inspection for Engagement and Excessive Wear of the Seat Locking Pins

(b) Within 18 months after the effective date of this AD, do the actions specified in paragraphs (b)(1) and (b)(2) of this AD, per the service bulletin.

(1) Do a detailed inspection of the seat locking pin for minimum engagement with the detent holes in the seat track of the captain's and first officer's seat assemblies.

**Note 1:** For the purposes of this AD, a detailed inspection is defined as: "An intensive visual examination of a specific structural area, system, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at intensity deemed appropriate by the inspector. Inspection aids such as mirror, magnifying lenses, etc., may be used. Surface cleaning and elaborate access procedures may be required."

(2) Do a detailed inspection of the seat lock pins for excessive wear.

#### Corrective Actions

(c) If any discrepancy is detected during the inspections required by paragraph (b) of this AD, before further flight, do the corrective action(s), as applicable, per the service bulletin. Those corrective actions include adjusting/replacing the seat locking pin with a new pin and/or adjusting/repairing/replacing the seat track with a new track.

#### Alternative Methods of Compliance

(d) In accordance with 14 CFR 39.19, the Manager, Los Angeles Aircraft Certification Office, FAA, is authorized to approve alternative methods of compliance for this AD.

#### Incorporation by Reference

(e) The actions shall be done in accordance with Boeing Alert Service Bulletin DC9-25A350, Revision 01, dated June 14, 2002; or Boeing Alert Service Bulletin MD90-25A009, Revision 01, dated July 1, 2002; as applicable. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Boeing Commercial Airplanes, Long Beach Division, 3855 Lakewood Boulevard, Long Beach, California 90846, Attention: Data and Service Management, Dept. C1-L5A (D800-0024). Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, Los Angeles Aircraft Certification

Office, 3960 Paramount Boulevard, Lakewood, California; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

#### Effective Date

(f) This amendment becomes effective on April 19, 2004.

Issued in Renton, Washington, on March 3, 2004.

**Ali Bahrami,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*  
[FR Doc. 04-5431 Filed 3-12-04; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 2002-NM-238-AD; Amendment 39-13522; AD 2004-05-27]

**RIN 2120-AA64**

#### **Airworthiness Directives; Boeing Model 737-200 Series Airplanes Modified by Supplemental Type Certificate ST00516AT**

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD), applicable to all Boeing Model 737-200 series airplanes modified by Supplemental Type Certificate (STC) ST00516AT, that requires removal of the in-flight entertainment (IFE) system installed per that STC. This action is necessary to eliminate the possibility that the airplane crew could be unable to remove power from the IFE system during a non-normal or emergency situation, which could result in the airplane crew's inability to control smoke or fumes in the airplane flight deck or cabin. This action is intended to address the identified unsafe condition.

**DATES:** Effective April 19, 2004.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of April 19, 2004.

**ADDRESSES:** The service information referenced in this AD may be obtained from Kosola and Associates, Inc., 5601 Newton Road, P.O. Box 3529, Albany, Georgia 31706. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, Atlanta

Aircraft Certification Office, One Crown Center, 1895 Phoenix Boulevard, suite 450, Atlanta, Georgia; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

#### **FOR FURTHER INFORMATION CONTACT:**

Myles Jalalian, Aerospace Engineer, Systems and Flight Test Branch, ACE-116A, FAA, Atlanta Aircraft Certification Office, One Crown Center, 1895 Phoenix Boulevard, suite 450, Atlanta, Georgia 30349; telephone (770) 703-6073; fax (770) 703-6097.

#### **SUPPLEMENTARY INFORMATION:** A

proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to all Boeing Model 737-200 series airplanes modified by Supplemental Type Certificate (STC) ST00516AT was published in the **Federal Register** on September 4, 2003 (68 FR 52539). That action proposed to require removal of the in-flight entertainment (IFE) system installed per STC ST00516AT.

#### **Comments**

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due consideration has been given to the comment received. The commenter supports the proposed rule.

#### **Conclusion**

After careful review of the available data, including the comment noted above, the FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

#### **Cost Impact**

There are approximately 4 Model 737-200 series airplanes of the affected design in the worldwide fleet. The FAA estimates that 2 airplanes of U.S. registry will be affected by this AD, that it will take approximately 1 work hour per airplane to accomplish the required actions, and that the average labor rate is \$65 per work hour. Based on these figures, the cost impact of the AD on U.S. operators is estimated to be \$130, or \$65 per airplane.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up,

planning time, or time necessitated by other administrative actions.

### Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

### Adoption of the Amendment

■ Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

### PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

■ 2. Section 39.13 is amended by adding the following new airworthiness directive:

**2004-05-27 Boeing:** Amendment 39-13522. Docket 2002-NM-238-AD.

**Applicability:** Model 737-200 series airplanes modified by Supplemental Type Certificate (STC) ST00516AT, certificated in any category.

**Compliance:** Required as indicated, unless accomplished previously.

To eliminate the possibility that the airplane crew could be unable to remove power from the in-flight entertainment (IFE) system during a non-normal or emergency situation, which could result in the airplane crew's inability to control smoke or fumes in

the airplane flight deck or cabin, accomplish the following:

### Removal of IFE System

(a) Within 18 months after the effective date of this AD, remove the IFE system installed by STC ST00516AT per the procedure in Kosola and Associates Service Bulletin 2002-1, dated July 16, 2003. The procedure includes disconnecting the power line that leads from the IFE system control unit to the P6 panel, capping and stowing all related wiring or removing related wiring from the airplane, removing the IFE system circuit breaker from the P6 panel, and removing all components of the IFE system from the airplane.

### Inspections Accomplished per Previous Issue of Service Bulletin

(b) Removal of the IFE system installed by STC ST00516AT before the effective date of this AD per Kosola and Associates Service Bulletin 2002-1, dated June 5, 2002, is considered acceptable for compliance with paragraph (a) of this AD.

### Parts Installation

(c) As of the effective date of this AD, no person may install an IFE system approved by STC ST00516AT on any airplane.

### Alternative Methods of Compliance

(d) In accordance with 14 CFR 39.19, the Manager, Atlanta Aircraft Certification Office (ACO), FAA, is authorized to approve alternative methods of compliance (AMOCs) for this AD.

### Incorporation by Reference

(e) Unless otherwise specified by this AD, the actions shall be done in accordance with Kosola and Associates Service Bulletin 2002-1, dated July 16, 2003. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Kosola and Associates, Inc., 5601 Newton Road, P.O. Box 3529, Albany, Georgia 31706. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, Atlanta Aircraft Certification Office, One Crown Center, 1895 Phoenix Boulevard, suite 450, Atlanta, Georgia; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

### Effective Date

(f) This amendment becomes effective on April 19, 2004.

Issued in Renton, Washington, on March 2, 2004.

**Ali Bahrami,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*  
[FR Doc. 04-5432 Filed 3-12-04; 8:45 am]

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## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 2003-SW-12-AD; Amendment 39-13524, AD 2004-05-29]

**RIN 2120-AA64**

### Airworthiness Directives; Eurocopter France Model EC 155B Helicopters

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD) for the specified Eurocopter France (Eurocopter) model helicopters that requires installing a tail rotor blade (blade)-to-torsion bar attachment tuning weight assembly on each blade of the Quiet Fenestron tail rotor and replacing each blade attachment bushing. This amendment is prompted by the discovery of tail rotor induced vibrations during flight tests. The actions specified by this AD are intended to prevent vibration in the tail rotor and the pilot's anti-torque pedals, blade pitch control failure, and subsequent loss of control of the helicopter.

**DATES:** Effective April 19, 2004.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of April 19, 2004.

**ADDRESSES:** The service information referenced in this AD may be obtained from American Eurocopter Corporation, 2701 Forum Drive, Grand Prairie, Texas 75053-4005, telephone (972) 641-3460, fax (972) 641-3527. This information may be examined at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Richard Monschke, Aviation Safety Engineer, FAA, Rotorcraft Directorate, Rotorcraft Standards Staff, Fort Worth, Texas 76193-0110, telephone (817) 222-5116, fax (817) 222-5961.

**SUPPLEMENTARY INFORMATION:** A proposal to amend 14 CFR part 39 to include an AD for the specified model helicopters was published in the **Federal Register** on November 24, 2003 (68 FR 65856). That action proposed to require removing each tail rotor attachment bushing, part number (P/N)