

Service Bulletin 120-32-0077, dated February 7, 1997: Within 100 flight hours after November 13, 2000, conduct a general visual inspection to detect the check valve flow direction in accordance with EMBRAER Service Bulletin 120-32-0077, Change 02, dated December 23, 1997; or Change 03, dated August 3, 2001. If the check valve is installed incorrectly, prior to further flight, reinstall the check valve in the proper position in accordance with Change 02 or Change 03 of the service bulletin.

Note 2: For the purposes of this AD, a general visual inspection is defined as: "A visual examination of an interior or exterior area, installation, or assembly to detect obvious damage, failure, or irregularity. This level of inspection is made from within touching distance unless otherwise specified. A mirror may be necessary to enhance visual access to all exposed surfaces in the inspection area. This level of inspection is made under normally available lighting conditions such as daylight, hangar lighting, flashlight, or droplight and may require removal or opening of access panels or doors. Stands, ladders, or platforms may be required to gain proximity to the area being checked."

(c) For airplanes subject to AD 2000-20-05, that are not equipped with the check valve installed in accordance with EMBRAER Service Bulletin 120-32-0077, dated February 7, 1997; Change 01, dated September 25, 1997; Change 02, dated December 23, 1997; or Change 03, dated August 3, 2001: Within 2,000 flight hours after November 13, 2000, install hydraulic tube assemblies incorporating a check valve in accordance with EMBRAER Service Bulletin 120-32-0077, Change 01, dated September 25, 1997; Change 02, dated December 23, 1997; or Change 03, dated August 3, 2001.

New Requirements of This AD

Note 3: Airplanes not subject to AD 2000-20-05 are those airplanes added by Change 03 of EMBRAER Service Bulletin 120-32-0077, dated August 3, 2001.

Airplane Flight Manual (AFM) Revision

(d) For airplanes not subject to AD 2000-20-05: Within 10 flight hours after the effective date of this AD, revise the "Emergency Procedures" and "Abnormal Procedures" sections of the airplane flight manual (AFM) by inserting into the AFM a copy of EMB-120 AFM 120/794, Revision 45, dated October 14, 1996.

Inspection and Corrective Actions

(e) For airplanes that are not subject to AD 2000-20-05, and on which the check valve has been installed in accordance with EMBRAER Service Bulletin 120-32-0077, dated February 7, 1997: Within 100 flight hours after the effective date of this AD, accomplish all of the applicable actions in accordance with the Accomplishment Instructions of EMBRAER Service Bulletin 120-32-0077, Change 03, dated August 3, 2001.

(f) For airplanes not subject to AD 2000-20-05, on which the check valve has not been installed in accordance with EMBRAER Service Bulletin 120-32-0077, dated

February 7, 1997; or Change 01, dated September 25, 1997; or Change 02, dated December 23, 1997: Within 2,000 flight hours after the effective date of this AD, accomplish all of the applicable actions in accordance with the Accomplishment Instructions of EMBRAER Service Bulletin 120-32-0077, Change 03, dated August 3, 2001.

(g) Accomplishment of the specified actions before the effective date of this AD per EMBRAER Service Bulletin 120-32-0077, Change 01, dated September 25, 1997; or Change 02, dated December 23, 1997; is considered acceptable for compliance with the applicable requirements of paragraphs (e) and (f) of this AD.

Alternative Methods of Compliance

(h) In accordance with 14 CFR 39.19, the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA, is authorized to approve alternative methods of compliance for this AD.

Incorporation by Reference

(i) Unless otherwise specified in this AD, the actions shall be done in accordance with EMBRAER Service Bulletin 120-32-0077, Change 01, dated September 25, 1997; EMBRAER Service Bulletin 120-32-0077, Change 02, dated December 23, 1997; and EMBRAER Service Bulletin 120-32-0077, Change 03, dated August 3, 2001; as applicable.

(1) The incorporation by reference of EMBRAER Service Bulletin 120-32-0077, Change 01, dated September 25, 1997; and EMBRAER Service Bulletin 120-32-0077, Change 03, dated August 3, 2001; is approved by the Director of the Federal Register, in accordance with 5 U.S.C. 552(a) and 1 CFR part 51.

(2) The incorporation by reference of EMBRAER Service Bulletin 120-32-0077, Change 02, dated December 23, 1997, was approved previously by the Director of the Federal Register as of November 13, 2000 (65 FR 59708, October 6, 2000).

(3) Copies may be obtained from Empresa Brasileira de Aeronautica S.A. (EMBRAER), PO Box 343—CEP 12.225, Sao Jose dos Campos—SP, Brazil. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Note 4: The subject of this AD is addressed in Brazilian airworthiness directive 97-05-03R3, dated October 2, 2001.

Effective Date

(j) This amendment becomes effective on February 13, 2004.

Issued in Renton, Washington, on December 29, 2003.

Ali Bahrami,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.
[FR Doc. 04-125 Filed 1-8-04; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2001-NM-267-AD; Amendment 39-13417; AD 97-24-02 R1]

RIN 2120-AA64

Airworthiness Directives; Bombardier Model CL-600-1A11 (CL-600), CL-600-2A12 (CL-601), and CL-600-2B16 (CL-601-3A, CL-601-3R, and CL-604) Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment revises an existing airworthiness directive (AD), applicable to certain Bombardier Model CL-600-1A11 (CL-600), CL-600-2A12 (CL-601), and CL-600-2B16 (CL-601-3A, CL-601-3R, and CL-604) series airplanes, that currently requires repetitive inspections to find cracks of a certain bulkhead web of the fuselage at certain locations, and repair if necessary. The actions specified by that AD are intended to prevent cracking in the pressure bulkhead at frame station (FS) 409.00, which could result in uncontrolled depressurization of the airplane and/or reduced structural integrity of the fuselage. This amendment provides an optional terminating modification for certain airplanes. This action is intended to address the identified unsafe condition.

DATES: Effective February 13, 2004.

The incorporation by reference of certain publications, as listed in the regulations, is approved by the Director of the Federal Register as of February 13, 2004.

The incorporation by reference of certain other publications, as listed in the regulations, was approved previously by the Director of the Federal Register as of December 3, 1997 (62 FR 61436, November 18, 1997).

ADDRESSES: The service information referenced in this AD may be obtained from Bombardier, Inc., Canadair, Aerospace Group, P.O. Box 6087, Station Centre-ville, Montreal, Quebec H3C 3G9, Canada. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, New York Aircraft Certification Office, 1600 Stewart Avenue, Westbury, New York; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT:

Serge Napoleon, Aerospace Engineer, Airframe and Propulsion Branch, ANE-171, FAA, New York Aircraft Certification Office, 1600 Stewart Avenue, Westbury, New York; telephone (516) 256-7512; fax (516) 568-2716.

SUPPLEMENTARY INFORMATION: A

proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) by revising AD 97-24-02, amendment 39-10209 (62 FR 61436, November 18, 1997), which is applicable to certain Bombardier Model CL-600-1A11, -2A12, and -2B16 series airplanes, was published in the **Federal Register** on October 21, 2003 (68 FR 60047). The action proposed to continue to require repetitive inspections to find cracks of a certain bulkhead web of the fuselage at certain locations, and repair if necessary. The action also provides an optional terminating modification for certain airplanes.

Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were submitted in response to the proposal or the FAA's determination of the cost to the public.

Conclusion

The FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

Cost Impact

There are about 237 airplanes of U.S. registry that will be affected by this AD.

The inspections that are required by AD 97-24-02 take about 2 work hours per airplane to accomplish, at an average labor rate of \$65 per work hour. Based on these figures, the cost impact of the currently required actions is estimated to be \$130 per airplane, per inspection cycle.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up, planning time, or time necessitated by other administrative actions.

The optional terminating modification, if done, would take between 125 and 300 work hours per

airplane, depending on the airplane configuration, at an average labor rate of \$65 per work hour. Required parts would be provided by the manufacturer at no cost to operators. Based on these figures, we estimate the cost of the modification to be between \$8,125 and \$19,500 per airplane.

Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

■ Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. Section 39.13 is amended by removing amendment 39-10209 (62 FR 61436, November 18, 1997), and by adding a new airworthiness directive (AD), amendment 39-13417, to read as follows:

97-24-02 R1 Bombardier, Inc. (Formerly Canadair): Amendment 39-13417. Docket 2001-NM-267-AD. Revises AD 97-24-02, Amendment 39-10209.

Applicability: Model CL-600-1A11 (CL-600) series airplanes, serial numbers 1004 through 1085 inclusive; Model CL-600-2A12 (CL-601) series airplanes, serial numbers 3001 through 3066 inclusive; Model CL-600-2B16 (CL-601-3A/-3R) series airplanes, serial numbers 5001 through 5194 inclusive; and Model CL-600-2B16 (CL-604) series airplanes, serial numbers 5301 through 5352 inclusive; certificated in any category.

Compliance: Required as indicated, unless accomplished previously.

To prevent cracking in the pressure bulkhead at frame station (FS) 409.00, which could result in uncontrolled depressurization of the airplane and/or reduced structural integrity of the fuselage, accomplish the following:

Restatement of Requirements of AD 97-24-02**Detailed Inspections/Repair**

(a) For Model CL-600-1A11 (CL-600) airplanes: Prior to the accumulation of 1,900 total landings, or within 100 landings after December 3, 1997 (the effective date of AD 97-24-02, amendment 39-10209), whichever occurs later, perform a detailed inspection to detect cracks at FS 409.00 of the bulkhead web (part number (P/N) 600-32014-71/-95/-105), in accordance with Canadair Challenger Service Bulletin 600-0679, dated September 12, 1997.

Note 1: For the purposes of this AD, a detailed inspection is defined as: "An intensive visual examination of a specific structural area, system, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at intensity deemed appropriate by the inspector. Inspection aids such as mirror, magnifying lenses, etc., may be used. Surface cleaning and elaborate access procedures may be required."

(1) If no crack is detected, repeat the detailed inspection thereafter at intervals not to exceed 600 landings.

(2) If any crack is detected and if all three of the conditions specified in paragraphs (a)(2)(i), (a)(2)(ii), and (a)(2)(iii) of this AD are met, within 600 landings or 12 months after the crack is detected, whichever occurs first, repair the crack in accordance with a method approved by the Manager, New York Aircraft Certification Office (ACO), FAA. Until the repair is accomplished, repeat the detailed inspection at intervals not to exceed 100 landings.

(i) No more than one crack exists at each corner radius, as specified in the service bulletin; and

(ii) No crack extends under the angles having P/N 600-32014-13 and P/N 600-32014-15 on the aft side of the bulkhead web; and

(iii) No crack exists in angles having P/N 600-32014-13 and P/N 600-32014-15 on the aft side of the bulkhead web.

(3) If any crack other than that identified in paragraph (a)(2) of this AD is detected, prior to further flight, repair it in accordance with a method approved by the Manager, New York ACO.

(b) For Model CL-600-2A12 (CL-601), CL-600-2B16 (CL-601-3A/-3R), and CL-600-

2B16 (CL-604) series airplanes: Prior to the accumulation of 1,100 total landings, or within 100 landings after December 3, 1997, whichever occurs later, perform a detailed inspection to detect cracks at FS 409.00 of the bulkhead web (P/N 600-32014-105/-137), in accordance with Canadair Challenger Service Bulletin 601-0501, dated September 12, 1997 (for Model CL-600-2A12 (CL-601) and CL-600-2B16 (CL-601-3A/-3R) series airplanes); or Canadair Challenger Service Bulletin 604-53-007, dated September 30, 1997 (for Model CL-600-2B16 (CL-604) series airplanes); as applicable.

(1) If no crack is detected, repeat the detailed inspection thereafter at intervals not to exceed 600 landings.

(2) If any crack is detected and if all three of the conditions specified in paragraphs (b)(2)(i), (b)(2)(ii), and (b)(2)(iii) of this AD are met, within 600 landings or 12 months after the crack is detected, whichever occurs first, repair the crack in accordance with a method approved by the Manager, New York ACO. Until the repair is accomplished, repeat the detailed inspection at intervals not to exceed 100 landings.

(i) No more than one crack exists at each corner radius, as specified in the service bulletin; and

(ii) No crack extends under the angles having P/N 600-32014-113 and P/N 600-32014-115 on the aft side of the bulkhead web; and

(iii) No crack exists in angles having P/N 600-32014-113 and P/N 600-32014-115 on the aft side of the bulkhead web.

(3) If any crack other than that identified in paragraph (b)(2) of this AD is detected, prior to further flight, repair it in accordance with a method approved by the Manager, New York ACO.

New Requirements of This AD

Optional Terminating Modification

(c) For airplanes on which no crack has been found during accomplishment of any inspection required by AD 97-24-02; or on which the pressure bulkhead was not previously repaired: Modification of the pressure bulkhead at FS 409.00 (including inspection, installation of reinforcing material, and tests) by accomplishing all the actions specified in paragraphs 2.A. through 2.D. of the Accomplishment Instructions of Bombardier Service Bulletin 601-0503 (for Model CL-601 and CL-601-3A/-3R series airplanes), Service Bulletin 600-0680 (for Model CL-600 series airplanes), or Service Bulletin 604-53-006 (for Model CL-604 series airplanes); all dated November 30, 1999; per the applicable service bulletin, terminates the repetitive inspections required by this AD.

Repair

(d) If any crack is found during any inspection specified in paragraph (c) of this AD: Before further flight, repair in accordance with a method approved by the Manager, New York ACO; or Transport Canada Civil Aviation or its delegated agent.

Alternative Methods of Compliance

(e) In accordance with 14 CFR 39.19, the Manager, New York ACO, is authorized to

approve alternative methods of compliance for this AD.

Incorporation by Reference

(f) Unless otherwise specified in this AD, the actions shall be done in accordance with Canadair Challenger Service Bulletin 600-0679, dated September 12, 1997, Canadair Challenger Service Bulletin 601-0501, dated September 12, 1997, or Canadair Challenger Service Bulletin 604-53-007, dated September 30, 1997; and Bombardier Service Bulletin 601-0503, dated November 30, 1999, Bombardier Service Bulletin 600-0680, dated November 30, 1999, or Bombardier Service Bulletin 604-53-006, dated November 30, 1999; as applicable.

(1) The incorporation by reference of Bombardier Service Bulletin 601-0503, dated November 30, 1999; Bombardier Service Bulletin 600-0680, dated November 30, 1999; and Bombardier Service Bulletin 604-53-006, dated November 30, 1999; is approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51.

(2) The incorporation by reference of Canadair Challenger Service Bulletin 600-0679, dated September 12, 1997; Canadair Challenger Service Bulletin 601-0501, dated September 12, 1997; and Canadair Challenger Service Bulletin 604-53-007, dated September 30, 1997; was approved previously by the Director of the Federal Register as of December 3, 1997 (62 FR 61436, November 18, 1997).

(3) Copies may be obtained from Bombardier, Inc., Canadair, Aerospace Group, P.O. Box 6087, Station Centre-ville, Montreal, Quebec H3C 3G9, Canada. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, New York Aircraft Certification Office, 1600 Stewart Avenue, Westbury, New York; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Note 2: The subject of this AD is addressed in Canadian airworthiness directive CF-1997-16R2, dated May 31, 2001.

Effective Date

(g) This amendment becomes effective on February 13, 2004.

Issued in Renton, Washington, on December 29, 2003.

Ali Bahrami,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 04-124 Filed 1-8-04; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2001-NM-120-AD; Amendment 39-13416; AD 2004-01-03]

RIN 2120-AA64

Airworthiness Directives; Airbus Model A319, A320, and A321 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment supersedes an existing airworthiness directive (AD), applicable to certain Airbus Model A320 series airplanes, that currently requires an inspection to detect moisture and migrated bushings of the guide fittings of the safety locking pins of the passenger doors, removal of any moisture, application of grease, and reinstallation of any migrated bushing. That AD also requires installation of a greasing nipple on the guide fitting of the locking pin and on three telescopic rods on the passenger doors. This amendment adds a requirement for modification of the upper guide fitting of the locking pin, and expands the applicability in the existing AD. The actions specified by this AD are intended to prevent jamming of the locking pin of the passenger door, which could result in inability to open the passenger door and delay of evacuation in an emergency, resulting in possible injury to passengers or crew. This action is intended to address the identified unsafe condition.

DATES: Effective February 13, 2004.

The incorporation by reference of a certain publication, as listed in the regulations, is approved by the Director of the Federal Register as of February 13, 2004.

The incorporation by reference of certain other publications, as listed in the regulations, was approved previously by the Director of the Federal Register as of February 17, 1998 (63 FR 1905, January 13, 1998).

ADDRESSES: The service information referenced in this AD may be obtained from Airbus, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Tim Dulin, Aerospace Engineer,