

**Inspection**

(a) Within 30 days after the effective date of this AD: Perform a general visual inspection of bearings having part numbers (P/N) 3326653-1 and 3326653-2 for evidence of a groove greater than 0.060 inch deep. Perform the inspection per paragraph 2.A. of the Accomplishment Instructions specified in Hercules Service Bulletin 382-52-9, dated July 5, 2000.

**Note 2:** For the purposes of this AD, a general visual inspection is defined as: "A visual examination of an interior or exterior area, installation, or assembly to detect obvious damage, failure, or irregularity. This level of inspection is made from within touching distance unless otherwise specified. A mirror may be necessary to enhance visual access to all exposed surfaces in the inspection area. This level of inspection is made under normally available lighting conditions such as daylight, hangar lighting, flashlight, or droplight and may require removal or opening of access panels or doors. Stands, ladders, or platforms may be required to gain proximity to the area being checked."

(1) If evidence of a groove greater than 0.060-inch deep is not found: Repeat the inspection at 30-day intervals until accomplishment of the terminating action required by paragraph (b) of this AD.

(2) If evidence of a groove greater than 0.060-inch deep is found: Before further flight, repair the bearings per paragraph (b) of this AD.

**Repair**

(b) Within 90 days after the effective date of this AD: Repair bearings having P/N 3326653-1 and P/N 3326653-2 per paragraph 2.B. of the Accomplishment Instructions specified in Hercules Service Bulletin 382-52-9, dated July 5, 2000. Accomplishment of this repair terminates the requirements of paragraph (a) of this AD.

**Alternative Methods of Compliance**

(c) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Atlanta Aircraft Certification Office (ACO), FAA. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Atlanta ACO.

**Note 3:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Atlanta ACO.

**Special Flight Permits**

(d) Special flight permits may be issued in accordance with §§ 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

**Incorporation by Reference**

(e) The actions shall be done in accordance with Hercules Service Bulletin 382-52-9, dated July 5, 2000. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C.

552(a) and 1 CFR part 51. Copies may be obtained from Lockheed Martin Corporation/Lockheed Martin Aeronautics Company, Airworthiness Office, Dept. 6A0M, Zone 0252, Column P-58, 86 S. Cobb Drive, Marietta, Georgia 30063. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, Atlanta Aircraft Certification Office, One Crown Center, 1895 Phoenix Boulevard, suite 450, Atlanta, Georgia; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**Effective Date**

(f) This amendment becomes effective on August 25, 2003.

Issued in Renton, Washington, on July 8, 2003.

**Ali Bahrami,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 03-17771 Filed 7-18-03; 8:45 am]

**BILLING CODE 4910-13-P**

**DEPARTMENT OF TRANSPORTATION****Federal Aviation Administration****14 CFR Part 39**

**[Docket No. 2002-SW-39-AD; Amendment 39-13237; AD 2003-14-18]**

**RIN 2120-AA64**

**Airworthiness Directives; Sikorsky Aircraft Corporation Model S76A, B, and C Helicopters**

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD) for the specified Sikorsky Aircraft Corporation (Sikorsky) model helicopters that requires removing non-conforming main landing gear brake discs (discs) and replacing them with different part-numbered airworthy discs. It also requires revising the Rotorcraft Flight Manual (RFM) to adjust takeoff and landing distances until the discs are replaced. This amendment is prompted by the manufacture of some discs using inferior materials. The actions specified by this AD are intended to prevent reduced braking performance and subsequent loss of control of the helicopter.

**DATES:** Effective August 25, 2003.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of August 25, 2003.

**ADDRESSES:** The service information referenced in this AD may be obtained

from Sikorsky Aircraft Corporation, Attn: Manager, Commercial Tech Support, 6900 Main Street, Stratford, Connecticut 06614, phone (203) 386-3001, fax (203) 386-5983. This information may be examined at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**FOR FURTHER INFORMATION CONTACT:**

Terry Fahr, Aviation Safety Engineer, Boston Aircraft Certification Office, 12 New England Executive Park, Burlington, MA 01803, telephone (781) 238-7155, fax (781) 238-7199.

**SUPPLEMENTARY INFORMATION: A**

proposal to amend 14 CFR part 39 to include an AD for Sikorsky Model S76A, B, and C helicopters was published in the **Federal Register** on February 7, 2003 (68 FR 6382). That action proposed to require, within 60 days, determining if discs, part number (P/N) 5014067, are installed. If so, replacing them with discs, P/N 5007672, and re-identifying brake assembly, P/N 5007555 and P/N 5007555-1, as brake assembly P/N 5007555-3, and brake assembly, P/N 5007555-2, as brake assembly, P/N 5007555-4, was proposed to be required within 90 days. The action also proposed to require revising the RFM to adjust the Category A rejected takeoff distance, the Category A landing distance, and the Category B landing distance by multiplying the distance by 1.67 to obtain the corrected distance until the discs are replaced.

Sikorsky has issued Alert Service Bulletin (ASB) No. 76-32-27, dated April 30, 2002, which contains Aircraft Braking Systems Corporation ASB S76-32-A24, dated April 10, 2002; and Sikorsky Aircraft Corporation ASB No. 76-32-28, dated May 17, 2002, which contains Aircraft Braking Systems Corporation ASB S76-32-A25, dated May 15, 2002. The ASBs describe procedures for replacing any non-conforming discs, reidentifying brake assemblies, and revising takeoff and landing distances in the RFM until the discs are replaced.

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due consideration has been given to the comment received.

The one commenter, the manufacturer, states that the AD should state that a manufacturer's warranty exists. The commenter states that the service information issued by Aircraft Braking Systems Corporation states that the replacement discs, P/N 5007672, are

available at no cost to the owners or operators. The FAA agrees, and that information is added to the economic impact paragraph of the AD.

After careful review of the available data, including the comments noted above, the FAA has determined that air safety and the public interest require the adoption of the rule with the change described previously. The FAA has determined that this change will neither increase the economic burden on any operator nor increase the scope of the AD.

On July 10, 2002, the FAA issued a new version of 14 CFR part 39 (67 FR 47997, July 22, 2002), which governs the FAA's AD system. The regulation now includes material that relates to altered products, special flight permits, and alternative methods of compliance. However, for clarity and consistency in this final rule, we have retained the language of the NPRM regarding that material.

The FAA estimates that 180 helicopters of U.S. registry will be affected by this AD, that it will take approximately 0.5 work hour per helicopter to determine if non-conforming discs are installed, and 1.25 work hours per helicopter to remove, replace and re-identify any non-conforming discs. The average labor rate is \$60 per work hour. Required parts will cost approximately \$1,902 per disc, and there are two discs per helicopter. The Aircraft Braking Systems Corporation ASB's state that "operators should contact their usual supply source for replacement of 5014067 Rotating Discs with free of charge 5007672 Rotating Discs on an exchange basis." Based on these figures, the total cost impact of this AD on U.S. operators is estimated to be \$703,620 (\$18,900 assuming brake discs are provided free of charge) to replace the discs throughout the fleet.

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities

under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

#### Adoption of the Amendment

■ Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

#### PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

■ 2. Section 39.13 is amended by adding a new airworthiness directive to read as follows:

**2003-14-18 Sikorsky Aircraft Corporation:** Amendment 39-13237. Docket No. 2002-SW-39-AD.

**Applicability:** Model S76A, B, and C helicopters, with main landing gear brake assembly (brake assembly), part number (P/N) 5007555, 5007555-1, or 5007555-2 installed, certificated in any category.

**Note 1:** This AD applies to each helicopter identified in the preceding applicability provision, regardless of whether it has been otherwise modified, altered, or repaired in the area subject to the requirements of this AD. For helicopters that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (d) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

**Compliance:** Required as indicated, unless accomplished previously.

To prevent reduced braking performance and subsequent loss of control of the helicopter, accomplish the following:

(a) Within 60 days, determine if a main landing gear brake disc (disc), part number (P/N) 5014067, is installed in the braking assembly in accordance with:

(1) Section III—Accomplishment Instructions, paragraph 1.A. through 1.D., of Aircraft Braking Systems Corporation Alert Service Bulletin S76-32-A24, dated April 10, 2002 (ASB A24) for braking assembly, P/N 5007555 and P/N 5007555-1, and

(2) Section III—Accomplishment Instructions, paragraph 1.A. and 1.B., of

Aircraft Braking Systems Corporation Alert Service Bulletin S76-32-A25, dated May 15, 2002 (ASB A25), for braking assembly, P/N 5007555-2.

(b) If disc, P/N 5014067, is installed, within 90 days, remove that disc and replace it with disc, P/N 5007672, and re-identify:

(1) Brake assembly, P/N 5007555 and P/N 5007555-1, as brake assembly, P/N 5007555-3, in accordance with the conversion of brake assembly instructions on page 6 of ASB A24, and

(2) Brake assembly, P/N 5007555-2, as brake assembly, P/N 5007555-4, in accordance with the conversion of brake assembly instructions on page 6 of ASB A25.

**Note 2:** Sikorsky Aircraft Corporation ASB No. 76-32-27, dated April 30, 2002, contains Aircraft Braking Systems Corporation ASB S76-32-A24, dated April 10, 2002, and Sikorsky Aircraft Corporation ASB No. 76-32-28, dated May 17, 2002, contains Aircraft Braking Systems Corporation ASB S76-32-A25, dated May 15, 2002.

(c) Until all installed discs, P/N 5014067, on the helicopter are replaced with disc, P/N 5007672, and all brake assemblies are re-identified in accordance with paragraph (b) of this AD, before further flight, increase the Category A—Rejected Takeoff Distance, the Category A—Landing Distance, and the Category B—Landing Distance as stated in the current Rotorcraft Flight Manual (RFM) by multiplying these rejected takeoff and landing distances by a factor of 1.67.

**Note 3:** There are temporary revisions to the RFM available from the helicopter manufacturer that documents increased rejected takeoff and landing distances.

(d) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Boston Aircraft Certification Office, Engine and Propeller Directorate, FAA. Operators shall submit their requests through an FAA Principal Maintenance Inspector, who may concur or comment and then send it to the Manager, Boston Aircraft Certification Office.

**Note 4:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Boston Aircraft Certification Office.

(e) Special flight permits may be issued in accordance with 14 CFR 21.197 and 21.199 to operate the helicopter to a location where the requirements of this AD can be accomplished.

(f) Identifying, removing and replacing the discs shall be done in accordance with Aircraft Braking Systems Corporation Alert Service Bulletin No. S76-32-A24, dated April 10, 2002, and Aircraft Braking Systems Corporation Alert Service Bulletin S76-32-A25, dated May 15, 2002. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Sikorsky Aircraft Corporation, Attn: Manager, Commercial Tech Support, 6900 Main Street, Stratford, Connecticut 06614, phone (203) 386-3001, fax (203) 386-5983. Copies may be inspected at the FAA, Office of the Regional Counsel,

Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

(g) This amendment becomes effective on August 25, 2003.

Issued in Fort Worth, Texas, on July 8, 2003.

**Mark R. Schilling,**

*Acting Manager, Rotorcraft Directorate, Aircraft Certification Service.*

[FR Doc. 03-17946 Filed 7-18-03; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 2003-CE-04-AD; Amendment 39-13239; AD 2003-14-20]

RIN 2120-AA64

#### **Airworthiness Directives; AeroSpace Technologies of Australia Pty Ltd. Models N22B and N24A Airplanes**

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment supersedes Airworthiness Directive (AD) 82-12-06, which applies to certain AeroSpace Technologies of Australia Pty Ltd. (ASTA) Models N22B and N24A airplanes. AD 82-12-06 currently requires repetitive visual inspections of all rudder control lever shaft assemblies for cracks and discrepancies, and, if cracks or discrepancies are found, it requires replacement with new or serviceable rudder control shafts, and a check of the fit of all rod end bearings in lever shafts. AD 82-12-06 also allows you to inspect all lever shafts by magnetic particle inspection or dye penetrant methods as terminating action for the repetitive visual inspections. This AD is the result of recent reports of failures of the upper control lever torque shaft due to fatigue loading on the affected airplanes, including those that included the terminating actions. This AD requires more detailed repetitive inspections (than there are in AD 82-12-06) of the upper and lower rudder pedal torque shafts and a one-time inspection for discrepancies in the thickness of the lever shaft side plates with appropriate follow-up action. The actions specified by this AD are intended to detect and correct cracks in the rudder control lever torque shafts and discrepancies in the thickness of the lever shaft side plates, which could result in failure of the rudder control lever torque shaft. Such failure could

lead to reduced controllability of the airplane.

**DATES:** This AD becomes effective on September 8, 2003.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in the regulations as of September 8, 2003.

**ADDRESSES:** You may get the service information referenced in this AD from Nomad Operations, Aerospace Support Division, Boeing Australia, PO Box 767, Brisbane, QLD 4000 Australia; telephone 61 7 3306 3366; facsimile 61 7 3306 3111. You may view this information at the Federal Aviation Administration (FAA), Central Region, Office of the Regional Counsel, Attention: Rules Docket No. 2003-CE-04-AD, 901 Locust, Room 506, Kansas City, Missouri 64106; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Ron Atmur, Aerospace Engineer, FAA, Los Angeles Aircraft Certification Office, 3960 Paramount Boulevard, Lakewood, California 90712; telephone (562) 627-5224; facsimile (562) 627-5210.

#### **SUPPLEMENTARY INFORMATION:**

##### **Discussion**

*Has FAA taken any action to this point?* Reports of cracking and other discrepancies on rudder control lever shaft assemblies on certain ASTA Models N22B and N24A airplanes caused us to issue AD 82-12-06, Amendment 39-4399. AD 82-12-06 currently requires the following on certain ASTA Models N22B and N24A airplanes:

- Repetitively inspecting visually all rudder control lever shafts for cracking;
- If cracks are found, before further flight, replacing with new or serviceable rudder control shafts;
- Checking for clearance of the fit of all rod end bearings in lever shafts; and
- Discontinuing the repetitive visual inspections when lever shafts are inspected either by magnetic particle inspection or dye penetrant methods.

*What has happened since AD 82-12-06 to initiate this proposed action?* The Civil Aviation Safety Authority (CASA), which is the airworthiness authority for Australia, recently notified FAA of the need to change AD 82-12-06. The CASA reports failures of the rudder control lever shaft. All the failures have occurred during ground operations and nosewheel steering/rudder loads are now considered the primary cause of the failure.

Some of the failures occurred on airplanes where the terminating action of AD 82-12-06 was incorporated.

What is the potential impact if FAA took no action? This condition, if not detected and corrected, could result in failure of the rudder control lever torque shaft. Such failure could lead to reduced controllability of the airplane.

*Has FAA taken any action to this point?* We issued a proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an AD that would apply to certain ASTA Models N22B and N24A. This proposal was published in the **Federal Register** as a notice of proposed rulemaking (NPRM) on April 29, 2003 (68 FR 22636). The NPRM proposed to require you to repetitively inspect, using either dye penetrant or magnetic particle methods and measurements, rudder control lever shafts for cracks; inspect (one-time) all lever shaft side plates by measuring the thickness; and if cracks or discrepancies in thickness are found, replace unserviceable parts with new or serviceable parts.

*Was the public invited to comment?* The FAA encouraged interested persons to participate in the making of this amendment. We did not receive any comments on the proposed rule or on our determination of the cost to the public.

#### **FAA's Determination**

*What is FAA's final determination on this issue?* After careful review of all available information related to the subject presented above, we have determined that air safety and the public interest require the adoption of the rule as proposed except for minor editorial corrections. We have determined that these minor corrections:

- Provide the intent that was proposed in the NPRM for correcting the unsafe condition; and
- do not add any additional burden upon the public than was already proposed in the NPRM.

*How does the revision to 14 CFR part 39 affect this AD?* On July 10, 2002, FAA published a new version of 14 CFR part 39 (67 FR 47997, July 22, 2002), which governs FAA's AD system. This regulation now includes material that relates to special flight permits, alternative methods of compliance, and altered products. This material previously was included in each individual AD. Since this material is included in 14 CFR part 39, we will not include it in future AD actions.