

determination with the Office of Information and Regulatory Affairs, Office of Management and Budget.

List of Subjects in 10 CFR Part 72

Administrative practice and procedure, Criminal penalties, Manpower training programs, Nuclear materials, Occupational safety and health, Penalties, Radiation protection, Reporting and recordkeeping requirements, Security measures, Spent fuel, Whistleblowing.

■ For the reasons set out in the preamble and under the authority of the Atomic Energy Act of 1954, as amended; the Energy Reorganization Act of 1974, as amended; and 5 U.S.C. 552 and 553; the NRC is adopting the following amendments to 10 CFR Part 72.

PART 72—LICENSING REQUIREMENTS FOR THE INDEPENDENT STORAGE OF SPENT NUCLEAR FUEL AND HIGH-LEVEL RADIOACTIVE WASTE

■ 1. The authority citation for Part 72 continues to read as follows:

Authority: Secs. 51, 53, 57, 62, 63, 65, 69, 81, 161, 182, 183, 184, 186, 187, 189, 68 Stat. 929, 930, 932, 933, 934, 935, 948, 953, 954, 955, as amended, sec. 234, 83 Stat. 444, as amended (42 U.S.C. 2071, 2073, 2077, 2092, 2093, 2095, 2099, 2111, 2201, 2232, 2233, 2234, 2236, 2237, 2238, 2282); sec. 274, Pub. L. 86–373, 73 Stat. 688, as amended (42 U.S.C. 2021); sec. 201, as amended, 202, 206, 88 Stat. 1242, as amended, 1244, 1246 (42 U.S.C. 5841, 5842, 5846); Pub. L. 95–601, sec. 10, 92 Stat. 2951 as amended by Pub. L. 102–486, sec. 7902, 106 Stat. 3123 (42 U.S.C. 5851); sec. 102, Pub. L. 91–190, 83 Stat. 853 (42 U.S.C. 4332); secs. 131, 132, 133, 135, 137, 141, Pub. L. 97–425, 96 Stat. 2229, 2230, 2232, 2241, sec. 148, Pub. L. 100–203, 101 Stat. 1330–235 (42 U.S.C. 10151, 10152, 10153, 10155, 10157, 10161, 10168).

Section 72.44(g) also issued under secs. 142(b) and 148(c), (d), Pub. L. 100–203, 101 Stat. 1330–232, 1330–236 (42 U.S.C. 10162(b), 10168(c),(d)). Section 72.46 also issued under sec. 189, 68 Stat. 955 (42 U.S.C. 2239); sec. 134, Pub. L. 97–425, 96 Stat. 2230 (42 U.S.C. 10154). Section 72.96(d) also issued under sec. 145(g), Pub. L. 100–203, 101 Stat. 1330–235 (42 U.S.C. 10165(g)). Subpart J also issued under secs. 2(2), 2(15), 2(19), 117(a), 141(h), Pub. L. 97–425, 96 Stat. 2202, 2203, 2204, 2222, 2244 (42 U.S.C. 10101, 10137(a), 10161(h)). Subparts K and L are also issued under sec. 133, 98 Stat. 2230 (42 U.S.C. 10153) and sec. 218(a), 96 Stat. 2252 (42 U.S.C. 10198).

■ 2. In § 72.214, Certificate of Compliance 1025 is revised to read as follows:

§ 72.214 List of approved spent fuel storage casks.

* * * * *

Certificate Number: 1025.

Initial Certificate Effective Date: April 10, 2000.

Amendment Number 1 Effective Date: November 13, 2001.

Amendment Number 2 Effective Date: May 29, 2002.

Amendment Number 3 Effective Date: October 1, 2003.

SAR Submitted by: NAC International, Inc.

SAR Title: Final Safety Analysis Report for the NAC-Multipurpose Canister System (NAC-MPC System).

Docket Number: 72–1025.

Certificate Expiration Date: April 10, 2020.

Model Number: NAC-MPC.

* * * * *

For the Nuclear Regulatory Commission.

Dated at Rockville, Maryland, this 7th day of July, 2003.

William D. Travers,

Executive Director for Operations.

[FR Doc. 03–18260 Filed 7–17–03; 8:45 am]

BILLING CODE 7590–01–P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2001–NM–395–AD; Amendment 39–13228; AD 2003–14–09]

RIN 2120–AA64

Airworthiness Directives; Boeing Model 767 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment supersedes an existing airworthiness directive (AD), applicable to certain Boeing Model 767 series airplanes, that currently requires repetitive detailed inspections to detect cracked, corroded, or stained collar fittings on both inboard trailing edge flaps; and follow-on corrective actions, if necessary. This amendment expands the applicability in the existing AD, and adds repetitive inspections for discrepancies of the collar fittings, torque tube, and splined bushings on both inboard trailing edge flaps; and follow-on and corrective actions, if necessary. The actions specified by this AD are intended to prevent failure of the collar fittings, which could result in separation of the inboard trailing edge flap and consequent reduced controllability of the airplane. This action is intended to address the identified unsafe condition.

DATES: Effective August 22, 2003.

The incorporation by reference of Boeing Alert Service Bulletin 767–57A0066, Revision 3, including Appendices A and B, dated December 19, 2001, is approved by the Director of the Federal Register as of August 22, 2003.

The incorporation by reference of Boeing Alert Service Bulletin 767–57A0066, Revision 1, dated August 6, 1998, as listed in the regulations, was approved previously by the Director of the Federal Register as of November 12, 1998 (63 FR 57577, October 28, 1998).

ADDRESSES: The service information referenced in this AD may be obtained from Boeing Commercial Airplane Group, P.O. Box 3707, Seattle, Washington 98124–2207. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT:

Suzanne Masterson, Aerospace Engineer, Airframe Branch, ANM–120S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98055–4056; telephone (425) 917–6441; fax (425) 917–6590.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) by superseding AD 98–22–12, amendment 39–10859 (63 FR 57577, October 28, 1998), which is applicable to certain Boeing Model 767 series airplanes, was published in the **Federal Register** on January 3, 2003 (68 FR 324). The action proposed to continue to require repetitive detailed inspections to detect cracked, corroded, or stained collar fittings on both inboard trailing edge flaps; and follow-on corrective actions, if necessary. The new action proposed to expand the applicability in the existing AD, and would add repetitive inspections for discrepancies of the collar fittings, torque tube, and splined bushings on both inboard trailing edge flaps; and follow-on and corrective actions, if necessary.

Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due consideration has been given to the comments received. One commenter concurs with the contents of the proposed AD.

and reassemble the parts in accordance with the service bulletin.”

We agree with the commenter that paragraph (d) of the proposed AD should be changed. We have changed paragraph (d) of this final rule, for clarification and consistency, to state, “* * * refinish and reassemble the parts with liberal coatings of corrosion-inhibiting compound (CIC) BMS 3-27 or BMS 3-38, in accordance with the service bulletin.”

- Change paragraph (h) of the proposed AD to explicitly specify line numbers 704, 719, and 720, and change the end of the last sentence, for clarification, to read, at the time specified in paragraph (g) of this AD.” The commenter states that this change is necessary for clarification of the applicability and compliance time specified in this paragraph. The commenter notes that paragraph (h) of the proposed AD is applicable only to those three airplanes which were assembled with BMS 3-27, not the MIL-G-23827 grease, after the proposed AD was issued. The commenter adds that without this change operators may be confused as to the applicability and the exact compliance time.

We agree with the commenter that paragraph (i) of the final rule (paragraph (h) of the proposed AD) should be changed, for the reasons specified. Paragraph (i) of this final rule has been changed to specify line numbers 704, 719, and 720, and to add, “at the time specified in paragraph (h) of this AD” (paragraph (g) of the proposed AD) at the end of the last sentence, for clarity.

Conclusion

After careful review of the available data, including the comments noted above, the FAA has determined that air safety and the public interest require the adoption of the rule with the changes previously described. The FAA has determined that these changes will neither increase the economic burden on any operator nor increase the scope of the AD.

Cost Impact

There are approximately 691 airplanes of the affected design in the worldwide fleet. The FAA estimates that 293 airplanes of U.S. registry will be affected by this AD.

The actions that are currently required by AD 98-22-12 take approximately 2 work hours per airplane to accomplish, at an average labor rate of \$60 per work hour. Based on these figures, the cost impact of the currently required actions is estimated to be \$120 per airplane, per inspection cycle.

The new inspections and refinishing that are required by this AD action will take approximately 127 work hours (including access and close up) per airplane to accomplish, at an average labor rate of \$60 per work hour. Based on these figures, the cost impact of the inspections and refinishing required by this AD on U.S. operators is estimated to be \$2,232,660, or \$7,620 per airplane, per cycle.

The cost impact figures discussed above are based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions actually required by the AD. These figures typically do not include incidental costs, such as planning time, or time necessitated by other administrative actions.

Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a “significant regulatory action” under Executive Order 12866; (2) is not a “significant rule” under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption “ADDRESSES.”

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

■ Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. Section 39.13 is amended by removing amendment 39-10859 (63 FR 57577, October 28, 1998), and by adding a new airworthiness directive (AD), amendment 39-13228, to read as follows:

2003-14-09 Boeing: Amendment 39-13228. Docket 2001 NM-395-AD. Supersedes AD 98-22-12, Amendment 39-10859.

Applicability: Model 767 series airplanes, line numbers 1 through 749 inclusive, certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (k)(1) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent failure of the collar fittings on the inboard trailing edge flaps, which could result in separation of the flap and consequent reduced controllability of the airplane, accomplish the following:

Restatement of Requirements of AD 98-22-12

Detailed Inspections/Corrective Actions

(a) For airplanes having line numbers 1 through 721 inclusive, except as provided by paragraphs (c) and (e) of this AD: Within 8 years since the date of manufacture of the airplane, or within 90 days after November 12, 1998 (the effective date of AD 98-22-12, amendment 39-10859), whichever occurs later; perform a detailed inspection of the collar fittings of both inboard trailing edge flaps to detect cracks, corrosion, or staining, in accordance with Part 1 of the Accomplishment Instructions of Boeing Alert Service Bulletin 767-57A0066, Revision 1, dated August 6, 1998; or Revision 3, including Appendices A and B, dated December 19, 2001. As of the effective date of this AD, only Revision 3 shall be used.

Note 2: For the purposes of this AD, a detailed inspection is defined as: “An intensive visual examination of a specific structural area, system, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at intensity deemed appropriate by

the inspector. Inspection aids such as mirror, magnifying lenses, etc., may be used. Surface cleaning and elaborate access procedures may be required."

(1) If no cracked, corroded, or stained collar fitting is found, repeat the detailed inspection required by paragraph (a) of this AD thereafter at intervals not to exceed 120 days until accomplishment of paragraph (d) or (e) of this AD, as applicable.

(2) If any cracked collar fitting is found, prior to further flight, install a new or serviceable collar fitting in accordance with Part 2 of the Accomplishment Instructions of the alert service bulletin.

(3) If any corroded collar fitting is found, prior to further flight, repair the corrosion in accordance with Figure 4 of the Accomplishment Instructions of Revision 3 of the service bulletin; or in accordance with a method approved by the Manager, Seattle Aircraft Certification Office (ACO), FAA.

(4) If any stained collar fitting is found, accomplish the requirements of paragraphs (a)(4)(i) and (a)(4)(ii) of this AD at the compliance times specified.

(i) Repeat the detailed inspection required by paragraph (a) of this AD thereafter at intervals not to exceed 45 days; and

(ii) Within 18 months after finding the stained collar fitting, accomplish Part 2 of the Accomplishment Instructions of the alert service bulletin. If any corroded collar fitting is found, before further flight, repair the corrosion in accordance with Part 3 and Part 4 of the Accomplishment Instructions of Revision 3 of the service bulletin; or in accordance with a method approved by the Manager, Seattle ACO.

New Requirements of This AD

Detailed Inspection

(b) For airplane line number 723: Within 8 years since the date of manufacture of the airplane, or within 90 days after the effective date of this AD, whichever is later; do a detailed inspection of the collar fittings of both inboard trailing edge flaps to detect cracks, corrosion, or staining, as specified in paragraph (a) of this AD, in accordance with Part 1 of the Accomplishment Instructions of Boeing Alert Service Bulletin 767-57A0066, Revision 3, including Appendices A and B, dated December 19, 2001. Then do the applicable actions specified in paragraphs (a)(1), (a)(2), (a)(3), and (a)(4) of this AD.

Repetitive Inspections/Follow-On and Corrective Actions

(c) For airplanes having line numbers 1 through 703 inclusive, 705 through 715 inclusive, 717, 718, 721, and 723; and for the right-hand side of the airplane on line number 716: Within 10 years since the date of manufacture of the airplane, or within 4 years after the effective date of this AD, whichever is later; do a spline inspection of the collar fittings, torque tube, and splined bushings for discrepancies (including cracks, fractures, corrosion, corrosion pits, and light wear), in accordance with Part 3 of the Accomplishment Instructions of Boeing Alert Service Bulletin 767-57A0066, Revision 3, including Appendices A and B, dated December 19, 2001. Accomplishment of the inspections required by this paragraph,

before the initial inspection required by paragraph (a) of this AD, meets the inspection requirements in paragraph (a) of this AD.

(d) If no discrepancy is found during any inspection required by paragraph (c) or (h) of this AD, before further flight, refinish and reassemble the parts with liberal coatings of corrosion-inhibiting compound (CIC) BMS 3-27 or BMS 3-38, in accordance with the Accomplishment Instructions of Boeing Alert Service Bulletin 767-57A0066, Revision 3, including Appendices A and B, dated December 19, 2001; and repeat the inspection every 24,000 flight cycles or 12 years, whichever is first. Accomplishment of this paragraph terminates the repetitive inspections required by paragraph (a) of this AD.

(e) If any discrepancy is found during any inspection required by paragraph (c) or (h) of this AD, before further flight, do the actions specified in either paragraph (e)(1) or (e)(2) of this AD in accordance with the Accomplishment Instructions of Boeing Alert Service Bulletin 767-57A0066, Revision 3, including Appendices A and B, dated December 19, 2001. Accomplishment of this paragraph terminates the repetitive inspections required by paragraph (a) of this AD.

(1) Replace the affected part with a new part, and reassemble the joint with liberal coatings of CIC BMS 3-27 or BMS 3-38, in accordance with the Accomplishment Instructions of the service bulletin. Repeat the applicable inspection specified in paragraph (c) or (h) of this AD every 24,000 flight cycles or 12 years, whichever is first.

(2) Rework the affected part, and reassemble the joint with liberal coatings of CIC BMS 3-27 or BMS 3-38, in accordance with the Accomplishment Instructions of the service bulletin. Repeat the applicable inspection specified in paragraph (c) or (h) of this AD, as specified in paragraph (e)(2)(i), (e)(2)(ii), or (e)(2)(iii) of this AD, as applicable.

(i) If five or fewer spline lengths are reworked per Figure 8 of the service bulletin, repeat the inspection every 24,000 flight cycles or 12 years, whichever is first.

(ii) If more than five spline lengths, but fewer than or equal to the maximum number of spline lengths allowed per Figure 8 of the service bulletin are reworked, repeat the inspection every 12,000 flight cycles or 6 years, whichever is first.

(iii) If more than the maximum number of spline lengths allowed per Figure 8 of the service bulletin are reworked, before further flight, replace the splined component and repeat the inspection every 24,000 flight cycles or 12 years, whichever is first.

Additional Inspections for Airplanes Inspected per Revision 2 of the Service Bulletin

(f) For any airplane on which the spline inspection was done in accordance with Part 3 of the Accomplishment Instructions of Boeing Alert Service Bulletin 767-57A0066, Revision 2, dated February 18, 1999; and on which the CIC Titanine JC5A was used, or the maintenance records are inconclusive of the type of CIC used: Do the applicable

inspection specified in Part 1 or Part 3 of the Accomplishment Instructions of Boeing Alert Service Bulletin 767-57A0066, Revision 3, including Appendices A and B, dated December 19, 2001; at the applicable time specified in paragraph (f)(1) or (f)(2) of this AD.

(1) Within 3 years after the last spline inspection done in accordance with Revision 2 of the service bulletin, or within 90 days after the effective date of this AD, whichever is later, do the applicable inspection specified in either Part 1 or Part 3 of Revision 3 of the service bulletin. Before further flight after accomplishment of the Part 1 inspection, do the applicable follow-on actions specified in paragraphs (a)(1), (a)(2), (a)(3), and (a)(4) of this AD. Before further flight after accomplishment of the Part 3 inspection, do the applicable follow-on actions specified in paragraphs (d) and (e) of this AD.

(2) Within 6 years after the last spline inspection done in accordance with Revision 2 of the service bulletin, do the spline inspection specified in Part 3 of Revision 3 of the service bulletin, unless already accomplished per paragraph (f)(1) of this AD. Before further flight after accomplishment of the inspection, do the applicable follow-on actions specified in paragraphs (d) and (e) of this AD.

Credit for Previously Accomplished Inspections

(g) Accomplishment of the spline inspection of the collar fittings, torque tube, and splined bushings per Part 3 of the Accomplishment Instructions of Boeing Alert Service Bulletin 767-57A0066, Revision 2, dated February 18, 1999; and on which the maintenance records are conclusive that CIC Titanine JC5A was not used, is considered acceptable for the initial inspections required by paragraphs (c) and (h) of this AD.

Airplanes Assembled With BMS 3-27

(h) For airplanes having line numbers 704, 719, 720, 722, and 724 through 749 inclusive; and for the left-hand side of the airplane on line number 716: Within 12 years since the date of manufacture of the airplane, or within 24,000 flight cycles after the effective date of this AD, whichever is first; do a spline inspection of the collar fittings, torque tube, and splined bushings for discrepancies (including cracks, fractures, corrosion, corrosion pits, and light wear). Do the inspection in accordance with Part 3 of the Accomplishment Instructions of Boeing Alert Service Bulletin 767-57A0066, Revision 3, including Appendices A and B, dated December 19, 2001; then, before further flight, do the applicable actions specified in either paragraph (d) or (e) of this AD.

(i) For airplanes having line numbers 704, 719, and 720: If the initial inspection required by paragraph (a) of this AD has not been done as of the effective date of this AD, operators may do the inspection required by paragraph (h) of this AD in lieu of the inspection required by paragraph (a) of this AD, at the time specified in paragraph (h) of this AD.

Use of Titanine JC5A Prohibited

(j) As of the effective date of this AD, no person shall use the CIC Titanine JC5A on the collar fittings, torque tube, and splined bushings on any airplane.

Alternative Methods of Compliance

(k)(1) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Seattle ACO. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Seattle ACO.

(2) Alternative methods of compliance, approved previously in accordance with AD 98-22-12, Amendment 39-10859, are not considered to be approved as alternative methods of compliance with this AD.

Note 3: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Seattle ACO.

Special Flight Permits

(l) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Incorporation by Reference

(m) Unless otherwise provided in this AD, the actions shall be done in accordance with Boeing Alert Service Bulletin 767-57A0066, Revision 1, dated August 6, 1998; or Boeing Alert Service Bulletin 767-57A0066, Revision 3, including Appendices A and B, dated December 19, 2001.

(1) The incorporation by reference of Boeing Alert Service Bulletin 767-57A0066, Revision 3, including Appendices A and B, dated December 19, 2001, is approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51.

(2) The incorporation by reference of Boeing Alert Service Bulletin 767-57A0066, Revision 1, dated August 6, 1998, as listed in the regulations, was approved previously by the Director of the Federal Register as of November 12, 1998 (63 FR 57577, October 28, 1998).

(3) Copies may be obtained from Boeing Commercial Airplane Group, P.O. Box 3707, Seattle, Washington 98124-2207. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Effective Date

(n) This amendment becomes effective on August 22, 2003.

Issued in Renton, Washington, on July 7, 2003.

Ali Bahrami,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 03-17692 Filed 7-17-03; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION**Federal Aviation Administration****14 CFR Part 39**

[Docket No. 2001-NM-401-AD; Amendment 39-13233; AD 2003-14-14]

RIN 2120-AA64

Airworthiness Directives; Aerospatiale Model ATR72 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to certain Aerospatiale Model ATR72 series airplanes, that requires installing brackets and ramps under floor panels between frames 23C and 23D and installing wire bundles on the ramps. The actions specified by this AD are intended to prevent chafing damage to the electrical wire cables, which could lead to an electrical short circuit and potential for a fire under the floor panels. This action is intended to address the identified unsafe condition.

DATES: Effective August 22, 2003.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of August 22, 2003.

ADDRESSES: The service information referenced in this AD may be obtained from Aerospatiale, 316 Route de Bayonne, 31060 Toulouse, Cedex 03, France. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Dan Rodina, Aerospace Engineer, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-2125; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to certain Aerospatiale Model ATR72 series airplanes was published in the **Federal Register** on April 11, 2003 (68 FR 17755). That action proposed to require installing brackets and ramps under floor panels between frames 23C and 23D and installing wire bundles on the ramps.

Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were submitted in response to the proposal or the FAA's determination of the cost to the public.

Conclusion

After careful review of the available data, the FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

Changes to 14 CFR Part 39/Effect on the AD

On July 10, 2002, the FAA issued a new version of 14 CFR part 39 (67 FR 47997, July 22, 2002), which governs the FAA's airworthiness directives system. The regulation now includes material that relates to altered products, special flight permits, and alternative methods of compliance. However, for clarity and consistency in this final rule, we have retained the language of the NPRM regarding that material.

Cost Impact

After the proposed AD was issued, we reviewed the figures we use to calculate the labor rate to do the required actions. To account for various inflationary costs in the airline industry, we find it appropriate to increase the labor rate used in these calculations from \$60 to \$65 per work hour. The economic impact information below has been revised to reflect this increase in the specified hourly labor rate.

The FAA estimates that 65 Aerospatiale Model ATR 72 series airplanes of U.S. registry will be affected by this AD, that it will take approximately 6 work hours per airplane to accomplish the required actions, and that the average labor rate is \$65 per work hour. Required parts will cost approximately \$1,844 per airplane. Based on these figures, the cost impact of the AD on U.S. operators is estimated to be \$145,210, or \$2,234 per airplane.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up, planning time, or time necessitated by other administrative actions.