

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. Section 39.13 is amended by adding the following new airworthiness directive:

2003-14-05 McDonnell Douglas:

Amendment 39-13224. Docket 2003-NM-156-AD.

Applicability: All Model 717-200 airplanes, certificated in any category.

Compliance: Required as indicated, unless accomplished previously.

To find and correct cracking of the support fitting assemblies of the main spoiler actuators, which could result in damage of adjacent structure such as the rear spar or upper skin panel, and consequent reduced structural integrity of the airplane; accomplish the following:

Repetitive Detailed Inspections

(a) Prior to the accumulation of 2,000 total flight hours, or within 550 flight hours after the effective date of this AD: Perform a detailed inspection for cracking of the support fitting assemblies and stop pads of the main spoiler actuators, per the Accomplishment Instructions of Boeing Alert Service Bulletin (ASB) 717-57A0016, dated May 29, 2003. Thereafter, repeat the detailed inspections at intervals not to exceed 550 flight hours.

Note 1: For the purposes of this AD, a detailed inspection is defined as: "An intensive visual examination of a specific structural area, system, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at intensity deemed appropriate by the inspector. Inspection aids such as mirror, magnifying lenses, etc., may be used. Surface cleaning and elaborate access procedures may be required."

No Cracking Found: Follow-on Action

(b) If no cracking is found during any inspection required by paragraph (a) of this AD, before further flight, lubricate the spoiler system and ensure that grease "squeeze-out" occurs at the locations indicated in Figure 3 of the Accomplishment Instructions of Boeing Alert Service Bulletin 717-57A0016, dated May 29, 2003.

If Any Cracking Found:

(c) If any cracking is found, before further flight, repair and perform follow-on inspections per a method approved by the Manager, Los Angeles Aircraft Certification Office (ACO), FAA; or per data meeting the type certification basis of the airplane approved by a Boeing Company Designated Engineering Representative who has been authorized by the Manager, Los Angeles ACO, to make such findings. For a repair method to be approved, the approval must specifically reference this AD.

No Reporting Requirements

(d) Although the Accomplishment Instructions of the ASB referenced in this AD specifies to submit information to the manufacturer, this AD does not include such a requirement.

Alternative Methods of Compliance

(e) In accordance with 14 CFR 39.19, the Manager, Los Angeles ACO, is authorized to approve alternative methods of compliance for this AD.

Incorporation by Reference

(f) Unless otherwise specified in this AD, the actions shall be done in accordance with Boeing Alert Service Bulletin 717-57A0016, dated May 29, 2003. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Boeing Commercial Aircraft Group, Long Beach Division, 3855 Lakewood Boulevard, Long Beach, California 90846, Attention: Data and Service Management, Dept. C1-L5A (D800-0024). Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; at the FAA, Los Angeles Aircraft Certification Office, 3960 Paramount Boulevard, Lakewood, California; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Effective Date

(g) This amendment becomes effective on July 31, 2003.

Issued in Renton, Washington, on July 3, 2003.

Vi L. Lipski,

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 03-17430 Filed 7-15-03; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2002-CE-51-AD; Amendment 39-13226; AD 2003-14-07]

RIN 2120-AA64

Airworthiness Directives; Pilatus Aircraft Ltd. Models PC-12 and PC-12/45 Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD) that applies to all Pilatus Aircraft Ltd. (Pilatus) Models PC-12 and PC-12/45 airplanes. This AD requires you to repetitively replace the nose landing gear (NLG) drag link right-hand part every 4,000 landings until an improved

design NLG drag link right-hand part is installed. This AD also requires you to install an improved design NLG drag link right-hand part as terminating action for the repetitive replacements. This AD is the result of mandatory continuing airworthiness information (MCAI) issued by the airworthiness authority for Switzerland. The actions specified by this AD are intended to prevent structural failure of the nose landing gear (NLG) caused by fatigue damage to the NLG drag link right-hand part that develops over time. Such failure could result in either an unintended NLG extension during flight or the NLG not properly locking upon extension, which could lead to loss of airplane control during landing operations.

DATES: This AD becomes effective on September 5, 2003.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in the regulations as of September 5, 2003.

ADDRESSES: You may get the service information referenced in this AD from Pilatus Business Aircraft Ltd., Product Support Department, 11755 Airport Way, Broomfield, Colorado 80021; telephone: (303) 465-9099; facsimile: (303) 465-6040. You may view this information at the Federal Aviation Administration (FAA), Central Region, Office of the Regional Counsel, Attention: Rules Docket No. 2002-CE-51-AD, 901 Locust, Room 506, Kansas City, Missouri 64106; or at the Office of the Federal Register, 800 North Capitol Street, NW, suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Doug Rudolph, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4059; facsimile: (816) 329-4090.

SUPPLEMENTARY INFORMATION:

Discussion

What events have caused this AD?
The Federal Office for Civil Aviation (FOCA), which is the airworthiness authority for Switzerland, recently notified FAA that an unsafe condition may exist on certain Pilatus Models PC-12 and PC-12/45 airplanes. The FOCA reports that 3 aircraft experienced a failure of the nose landing gear (NLG) drag link assembly during cruise flight. The actuator attachment levers on the right-hand upper drag link part failed. In all cases, the NLG fell out due to gravity, and the emergency spring pack extended it forward and allowed safe landings.

What is the potential impact if FAA took no action? Structural failure of the

NLG drag link right-hand part could result in either an unintended NLG extension during flight or the NLG not properly locking upon extension. This could lead to loss of airplane control during landing operations.

Has FAA taken any action to this point? We issued a proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an AD that would apply to all Pilatus Models PC-12 and PC-12/45 airplanes. This proposal was published in the **Federal Register** as a supplemental notice of proposed rulemaking (NPRM) on April 23, 2003 (68 FR 19963). The supplemental NPRM proposed to require you to repetitively replace the nose landing gear (NLG) drag link right-hand part every 4,000 landings until an improved design NLG drag link right-hand part is installed. The NPRM also proposed to require you to install an improved design NLG drag link right-hand part as terminating action for the repetitive replacements.

Was the public invited to comment? The FAA encouraged interested persons to participate in the making of this amendment. The following presents the comments received on the proposal and FAA's response to each comment:

Comment Issue No. 1: Temporary Revision Incorporated in Aircraft Maintenance Manual

What is the commenter's concern? A commenter states that when the NPRM and the supplemental NPRM were issued that the reference to Temporary Revision No. 32-14, dated June 4, 2002,

to Pilatus PC-12 Maintenance Manual 32-20-06, was correct. However, since issuance of the NPRMs, Pilatus has incorporated the temporary revision into the Pilatus PC-12 Maintenance Manual. The commenter recommends that FAA revise the reference to read the Pilatus PC-12 Maintenance Manual 32-20-06 instead of the temporary revision.

What is FAA's response to the concern? We concur with the commenter and for clarity and completeness will change the final rule AD action to incorporate this change.

Comment Issue No. 2: Use of Correct Service Bulletin

What is the commenter's concern? A commenter states that Pilatus has issued PC12 Service Bulletin No. 32-014, Revision No. 1, dated May 13, 2003, which includes minor changes. However, the commenter points out that compliance following either the original service bulletin or the revised service bulletin should be acceptable. Further, operators who do the work following the revised service bulletin should not need to request an alternative method of compliance (AMOC). The commenter recommends the final AD action include references to the original service bulletin and the revised service bulletin.

What is FAA's response to the concern? The FAA agrees and we are changing the final rule AD to provide for owners/operators who accomplish the work under either the original service bulletin or Revision 1 of the service bulletin.

FAA's Determination

What is FAA's final determination on this issue? We carefully reviewed all available information related to the subject presented above and determined that air safety and the public interest require the adoption of the rule as proposed except for the changes discussed above and minor editorial corrections. We have determined that these changes and minor corrections:

- Provide the intent that was proposed in the NPRM for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the NPRM.

How does the revision to 14 CFR part 39 affect this AD? On July 10, 2002, FAA published a new version of 14 CFR part 39 (67 FR 47997, July 22, 2002), which governs FAA's AD system. This regulation now includes material that relates to special flight permits, alternative methods of compliance, and altered products. This material previously was included in each individual AD. Since this material is included in 14 CFR part 39, we will not include it in future AD actions.

Cost Impact

How many airplanes does this AD impact? We estimate that this AD affects 265 airplanes in the U.S. registry.

What is the cost impact of this AD on owners/operators of the affected airplanes? We estimate the following costs to accomplish the replacement with the same design part:

Labor cost	Parts cost	Total cost per airplane	Total cost on U.S. operators
6 workhours × \$60 per hour = \$360	\$1,000	\$1,360	\$1,360 × 265 = \$360,400.

We estimate the following costs to accomplish the replacement with the improved design part:

Labor cost	Parts cost	Total cost per airplane	Total cost on U.S. operators
6 workhours × \$60 per hour = \$360	\$2,200	\$2,560	\$2,560 × 265 = \$678,400.

Compliance Time of This AD

What is the compliance time of this AD? The compliance time of this AD is based on the number of landings rather than hours TIS.

Why is the compliance time of this AD presented in landings? The reason for this type of compliance is that the area that is showing fatigue is the NLG drag link right-hand part. This area of the

airplane is used during the landing operation. We have determined to base the compliance time for this AD upon the number of landings.

Since airplane operators are not required to keep track of landings, we will provide a method of calculating hours TIS into landings.

Regulatory Impact

Does this AD impact various entities? The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not

have federalism implications under Executive Order 13132.

Does this AD involve a significant rule or regulatory action? For the reasons discussed above, I certify that this action (1) is not a “significant regulatory action” under Executive Order 12866; (2) is not a “significant rule” under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the final evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

■ Accordingly, under the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. FAA amends § 39.13 by adding a new AD to read as follows:

2003–14–07 Pilatus Aircraft Ltd.:

Amendment 39–13226; Docket No. 2002–CE–51–AD.

(a) *What airplanes are affected by this AD?* This AD affects Models PC–12 and PC–12/45 airplanes, all serial numbers, that are certificated in any category.

(b) *Who must comply with this AD?* Anyone who wishes to operate any of the airplanes identified in paragraph (a) of this AD must comply with this AD.

(c) *What problem does this AD address?* The actions specified by this AD are intended to prevent structural failure of the nose landing gear (NLG) caused by fatigue damage to the NLG drag link right-hand part that develops over time. Such failure could result in either an unintended NLG extension during flight or the NLG not properly locking upon extension, which could lead to loss of airplane control during landing operations.

(d) *What actions must I accomplish to address this problem?* To address this problem, you must accomplish the following:

Actions	Compliance	Procedures
(1) Replace the nose landing gear (NLG) drag link righthand part, part number (P/N) 532.20.12.140 with:	Upon the accumulation of 4,000 landings on the nose landing gear (NLG) drag link right hand part or within the next 100 landings after September 5, 2003 (the effective date of this AD), whichever part occurs later. Incorporation of the improved-design NLG drag link brace is terminating action for this AD.	In accordance with the Pilatus PC–12 the Maintenance Manual 32–20-06, pages 401 through 405, dated October 1, 2002.
(i) The same P/N 532.20.12.140 or FAA-approved equivalent part number; or.	Upon the accumulation of 4,000 landings. Incorporation of improved-design NLG drag link brace is terminating action for this AD.	In accordance with the Pilatus PC–12 Maintenance Manual 32–20-06, pages 401 through 405, dated October 1, 2002.
(ii) Improved design NLG drag link right-hand part, P/N 532.20.12.289.	At the third replacement required in paragraph (d)(2) of this AD.	In accordance with either Pilatus Aircraft Ltd. PC12 Service Bulletin No. 32–014, dated August 13, 2002, or Pilatus Aircraft Ltd. PC12 Service Bulletin No. 32–014, Revision No. 1, dated May 13, 2003, and the Pilatus PC–12 Maintenance Manual.
(2) If replacement in paragraph (d)(1) is with the original style part, replace with:	When an improved P/N 532.20.12.289 NLG drag link part is installed after the effective date of this AD.	Not Applicable.
(i) The same P/N 532.20.12.140 or FAA-approved equivalent part number; or.		
(ii) Improved design NLG drag link right-hand part, P/N 532.20.12.289.		
(3) Unless already accomplished per paragraph (d)(1) or (d)(2), replace the NLG drag link right-hand part, P/N 532.20.12.140, with an improved design NLG drag link right-hand part, P/N 532.20.12.289 or FAA-approved equivalent part number. Installing the improved part number terminates the repetitive replacement requirements of paragraph (d)(2) of this AD.		
(4) Do not install, on any affected airplane, an NLG drag link right-hand part that is not P/N 532.20.12.289 or FAA-approved equivalent part number.		

(e) *What if I do not keep track of landings?* The compliance times of this AD are presented in landings instead of hours time-in-service (TIS). If landings are not known, hours TIS may be used by dividing the numbers of hours TIS by the unknown landings factor (0.75).

Note 1: For the purposes of this AD, 3,000 hours TIS would be equivalent to 4,000 landings (3,000 hours/0.75 = 4,000 landings).

(f) *Can I comply with this AD in any other way?* To use an alternative method of compliance or adjust the compliance time, use the procedures in 14 CFR 39.19. Send these requests to the Standards Office Manager, Small Airplane Directorate. Contact Doug Rudolph, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106;

telephone: (816) 329–4059; facsimile: (816) 329–4090 for information on any already approved alternative methods of compliance.

(g) *Are any service bulletins incorporated into this AD by reference?* Actions required by this AD must be done in accordance with either Pilatus Aircraft Ltd. PC12 Service Bulletin No. 32–014, dated August 13, 2002, or Pilatus Aircraft Ltd. PC12 Service Bulletin No. 32–014, Revision No. 1, dated May 13, 2003; and Pilatus PC–12 Maintenance Manual 32–20–06, pages 401 through 405, dated October 1, 2002. The Director of the Federal Register approved this incorporation by reference under 5 U.S.C. 552(a) and 1 CFR part 51. You may get copies from Pilatus Aircraft Ltd., Customer Liaison Manager, CH–6371 Stans, Switzerland; telephone: +41 41 619 63 19; facsimile: +41 41 619 6224; or from Pilatus Business Aircraft Ltd., Product

Support Department, 11755 Airport Way, Broomfield, Colorado 80021; telephone: (303) 465–9099; facsimile: (303) 465–6040. You may view copies at the FAA, Central Region, Office of the Regional Counsel, 901 Locust, Room 506, Kansas City, Missouri, or at the *Office of the Federal Register*, 800 North Capitol Street, NW., suite 700, Washington, DC.

Note 2: The subject of this AD is addressed in Swiss AD Number HB 2002–271, dated June 17, 2002.

(h) *When does this amendment become effective?* This amendment becomes effective on September 5, 2003.

Issued in Kansas City, Missouri, on July 7, 2003.

Michael Gallagher,

Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. 03-17566 Filed 7-15-03; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2003-NE-23-AD; Amendment 39-13210; AD 2003-13-10]

RIN 2120-AA64

Airworthiness Directives; Rolls-Royce Corporation (formerly Allison Engine Company, Allison Gas Turbine Division, and Detroit Diesel Allison) Models 250-C30R/3, -C30R/3M, -C47B, and -C47M Turboshaft Engines; Correction

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule; request for comments, correction.

SUMMARY: This document makes a correction to Airworthiness Directive (AD) 2003-13-10, applicable to Rolls-Royce Corporation (formerly Allison Engine Company, Allison Gas Turbine Division, and Detroit Diesel Allison) Models 250-C30R/3, -C30R/3M, -C47B, and -C47M turboshaft engines. AD 2003-13-10 was published in the **Federal Register** on June 30, 2003 (68 FR 38590). In the compliance section, paragraph (f) incorrectly references a compliance date of July 15, 2003 and should reference a compliance date of July 31, 2003. This document corrects that date. In all other respects, the original document remains the same.

EFFECTIVE DATE: July 15, 2003.

FOR FURTHER INFORMATION CONTACT: Khailaa Hosny, Aerospace Engineer, Chicago Aircraft Certification Office, FAA, 2300 East Devon Avenue, Des Plaines, IL 60018-4696; telephone (847) 294-7134; fax (847) 294-7834.

SUPPLEMENTARY INFORMATION: A final rule; request for comments airworthiness directive FR DOC. 03-15993, applicable to Rolls-Royce Corporation (formerly Allison Engine Company, Allison Gas Turbine Division, and Detroit Diesel Allison) Models 250-C30R/3, -C30R/3M, -C47B, and -C47M turboshaft engines, was published in the **Federal Register** on June 30, 2003 (68 FR 38590). The following correction is needed:

On page 38592, in the first column, under Initial Inspection heading, paragraph (f), fifth line, which reads “no later than July 15, 2003, in accordance * * *” is corrected to read “no later than July 31, 2003, in accordance * * *”.

Issued in Burlington, MA, on July 10, 2003.

Francis A. Favara,

Acting Manager, Engine and Propeller Directorate, Aircraft Certification Service.

[FR Doc. 03-17950 Filed 7-15-03; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF THE TREASURY

Internal Revenue Service

26 CFR Parts 1 and 602

[TD 9076]

RIN 1545-AX34

Special Rules Under Section 417(a)(7) for Written Explanations Provided by Qualified Retirement Plans After Annuity Starting Dates

AGENCY: Internal Revenue Service (IRS), Treasury.

ACTION: Final regulations.

SUMMARY: This document contains final regulations relating to the special rule added by the Small Business Job Protection Act of 1996 which permits the required written explanations of certain benefits to be provided by qualified retirement plans to plan participants after the annuity starting date. These final regulations affect sponsors and administrators of qualified retirement plans, and participants in those plans.

DATES: Effective Date: These regulations are effective July 16, 2003.

Applicability Date: These regulations apply to plan years beginning on or after January 1, 2004.

FOR FURTHER INFORMATION CONTACT: Robert Walsh (202) 622-6090 (not a toll-free number).

SUPPLEMENTARY INFORMATION:

Paperwork Reduction Act

The collection of information contained in these final regulations has been reviewed and approved by the Office of Management and Budget in accordance with the Paperwork Reduction Act of 1995 (44 U.S.C. 3507(d)) under control number 1545-1724.

The collection of information in this final regulation is in § 1.417(e)-1(b)(3)(iv)(B) and § 1.417(e)-1(b)(3)(v)(A). This collection of

information is required by the IRS to ensure that the participant and the participant's spouse consent to a form of distribution from a qualified retirement plan that may result in reduced periodic payments.

An agency may not conduct or sponsor, and a person is not required to respond to, a collection of information unless it displays a valid control number assigned by the Office of Management and Budget.

Comments concerning the accuracy of this burden estimate and suggestions for reducing this burden should be sent to the Internal Revenue Service, Attn: IRS Reports Clearance Officer, W:CAR:MP:T:T:SP, Washington, DC 20224, and to the Office of Management and Budget, Attn: Desk Officer for the Department of the Treasury, Office of Information and Regulatory Affairs, Washington, DC 20503.

Books or records relating to a collection of information must be retained as long as their contents might become material in the administration of any internal revenue law. Generally, tax returns and tax return information are confidential, as required by 26 U.S.C. 6103.

Background

This document contains amendments to 26 CFR part 1 under section 417(a)(7). On January 17, 2001, a notice of proposed rulemaking (REG-109481-99) was published in the **Federal Register** (66 FR 3916) under section 417(a)(7) of the Internal Revenue Code. No public hearing was requested or held. Written comments responding to the notice of proposed rulemaking were received. After consideration of all the comments, the proposed regulations are adopted as amended by this Treasury decision.

Section 401(a)(11) of the Internal Revenue Code provides that, subject to certain exceptions, all distributions from a qualified plan must be made in the form of a qualified joint and survivor annuity (QJSA). One such exception is provided in section 417, which allows a participant to elect to waive the QJSA in favor of another form of distribution. Section 417(a)(2) provides that, for the waiver to be valid, the participant's spouse must consent to the waiver. Section 417(a)(3)(A) requires a qualified plan to provide to each participant, within a reasonable period of time before the annuity starting date, a written explanation (QJSA explanation) that describes the QJSA, the right to waive the QJSA, and the rights of the participant's spouse.

Section 417(a)(7), which was added to the Code by section 1451(a) of the Small Business Job Protection Act of 1996,