

Cost Impact

The FAA estimates that the proposed installation would be required to be accomplished on 45 Model CL-600-2B19 (Regional Jet Series 100 & 440) airplanes of U.S. registry, that it would take approximately 15 work hours per airplane to accomplish the proposed installation, and that the average labor rate is \$60 per work hour. Required parts would cost approximately \$10,273 per airplane. Based on these figures, the cost impact of the proposed installation on U.S. operators is estimated to be \$502,785, or \$11,173 per airplane.

The FAA estimates that the proposed inspection would be required to be accomplished on 43 Model CL-600-2B19 (Regional Jet Series 100 & 440) airplanes of U.S. registry, that it would take approximately 1 work hour per airplane to accomplish the proposed inspection, and that the average labor rate is \$60 per work hour. Based on these figures, the cost impact of the proposed inspection on U.S. operators is estimated to be \$2,580, or \$60 per airplane.

The cost impact figures discussed above are based on assumptions that no operator has yet accomplished any of the proposed requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up, planning time, or time necessitated by other administrative actions.

Regulatory Impact

The regulations proposed herein would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this proposal would not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this proposed regulation: (1) Is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft

regulatory evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption ADDRESSES.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

Bombardier, Inc. (Formerly Canadair):

Docket 2001-NM-328-AD.

Applicability: Model CL-600-2B19 (Regional Jet Series 100 & 440) airplanes having serial numbers 7003 through 7067 inclusive and 7069 through 7109 inclusive, certificated in any category; excluding those airplanes on which the actions specified in Bombardier Service Bulletin 601R-28-024, dated May 21, 1996, have been accomplished. (This applicability includes airplanes informally identified as "Series 200.")

Compliance: Required as indicated, unless accomplished previously.

To prevent a fire hazard due to fuel spillage, accomplish the following:

Installation

(a) Within 180 days after the effective date of this AD, install new vent tube assemblies for the main fuel tanks, per Part A of paragraph 2.B. of the Accomplishment Instructions of Bombardier Service Bulletin 601R-28-024, Revision 'A', dated November 11, 1998.

Inspection and Corrective Action

(b) For airplanes having serial numbers 7003 through 7035 inclusive, and 7048 through 7057 inclusive: Before further flight after installing the vent tube assemblies as required by paragraph (a) of this AD, perform a general visual inspection to measure the clearance between the vent system tubing and the applicable wing rib, per Part B of paragraph 2.B. of the Accomplishment Instructions of Bombardier Service Bulletin 601R-28-024, Revision 'A', dated November 11, 1998.

Note 1: For the purposes of this AD, a general visual inspection is defined as: "A visual examination of an interior or exterior area, installation, or assembly to detect

obvious damage, failure, or irregularity. This level of inspection is made from within touching distance unless otherwise specified. A mirror may be necessary to enhance visual access to all exposed surfaces in the inspection area. This level of inspection is made under normally available lighting conditions such as daylight, hangar lighting, flashlight, or droplight and may require removal or opening of access panels or doors. Stands, ladders, or platforms may be required to gain proximity to the area being checked."

(1) If the clearance between the vent system tubing and the applicable wing rib is 0.125 inch or more, no further action is required by this paragraph.

(2) If the clearance between the vent system tubing and the applicable wing rib is less than 0.125 inch, prior to further flight, install the bracket assemblies in accordance with paragraphs B.(8) through B.(10) of the Accomplishment Instructions of the service bulletin.

Alternative Methods of Compliance

(c) In accordance with 14 CFR 39.19, the Manager, New York Aircraft Certification Office (ACO), FAA, is authorized to approve alternative methods of compliance for this AD.

Note 2: The subject of this AD is addressed in Canadian airworthiness directive CF-2001-31, dated August 7, 2001.

Issued in Renton, Washington, on June 11, 2003.

Kalene C. Yanamura,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.
[FR Doc. 03-15326 Filed 6-17-03; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2001-NM-246-AD]

RIN 2120-AA64

Airworthiness Directives; Boeing Model 737-200, -200C, -300, -400, and -500 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: This document proposes the adoption of a new airworthiness directive (AD) that is applicable to certain Boeing Model 737-200, -200C, -300, -400, and -500 series airplanes. This proposal would require repetitive inspections to find fatigue cracking of certain upper and lower skin panels of the fuselage, and follow-on and corrective actions, if necessary. This proposal also includes terminating action for the repetitive inspections of

certain modified or repaired areas only. This action is necessary to find and fix fatigue cracking of the skin panels, which could result in sudden fracture and failure of the skin panels of the fuselage, and consequent rapid decompression of the airplane. This action is intended to address the identified unsafe condition.

DATES: Comments must be received by August 4, 2003.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 2001-NM-246-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056. Comments may be inspected at this location between 9 a.m. and 3 p.m., Monday through Friday, except Federal holidays. Comments may be submitted via fax to (425) 227-1232. Comments may also be sent via the Internet using the following address: *9-anm-nprmcomment@faa.gov*. Comments sent via fax or the Internet must contain "Docket No. 2001-NM-246-AD" in the subject line and need not be submitted in triplicate. Comments sent via the Internet as attached electronic files must be formatted in Microsoft Word 97 for Windows or ASCII text.

The service information referenced in the proposed rule may be obtained from Boeing Commercial Airplane Group, PO Box 3707, Seattle, Washington 98124-2207. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington.

FOR FURTHER INFORMATION CONTACT: Duong Tran, Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 917-6452; fax (425) 917-6590.

SUPPLEMENTARY INFORMATION:

Comments Invited

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified above. All communications received on or before the closing date for comments, specified above, will be considered before taking action on the proposed rule. The proposals contained in this action may be changed in light of the comments received.

Submit comments using the following format:

- Organize comments issue-by-issue. For example, discuss a request to change the compliance time and a request to change the service bulletin reference as two separate issues.

- For each issue, state what specific change to the proposed AD is being requested.

- Include justification (*e.g.*, reasons or data) for each request.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report summarizing each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this action must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket Number 2001-NM-246-AD." The postcard will be date stamped and returned to the commenter.

Availability of NPRMs

Any person may obtain a copy of this NPRM by submitting a request to the FAA, Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 2001-NM-246-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056.

Discussion

The FAA has received reports indicating that cracks were found along the edges of the chem-milled pockets in the upper skin at stringer S-12, and above the S-4, S-10, and S-14 lap joints, on several Boeing Model 737 series airplanes. The cracks were up to 6 inches long and multiple adjacent bays were found to be cracked along the same stringers on three of the airplanes. The airplanes had accumulated between 34,574 and 56,949 total flight cycles. Additionally, skin cracks up to 4 inches long located below the S-14 lap joint along the bonded skin doublers were reported on 25 other airplanes which had accumulated between 22,786 and 80,113 total flight cycles.

Analysis by the manufacturer revealed that these cracks are caused by fatigue due to high bending stresses at the edge of chem-milled pockets or bonded skin doublers. Such fatigue cracking could result in sudden fracture and failure of the skin panels of the fuselage, and consequent rapid decompression of the airplane.

Related Rulemaking

This proposed AD is related to AD 2002-07-08, amendment 39-12702 (67 FR 17917, April 12, 2002). That AD references Boeing Service Bulletin 737-53A1177, Revision 6, dated May 31, 2001, as the appropriate source of service information for accomplishment of the specified actions. (The AD also referenced, for accomplishment of certain actions, Boeing Alert Service Bulletin 737-53A1177, Revision 1, dated September 19, 1996; Revision 2, dated July 24, 1997; Revision 3, dated September 18, 1997; Revision 4, dated September 2, 1999; and Revision 5, dated February 15, 2001.) That AD is applicable to certain Boeing Model 737 series airplanes and requires repetitive inspections to find cracking of the lower skin at the lower row of fasteners in the lap joints of the fuselage, and repair of any cracking found. That AD also requires modification of the fuselage lap joints at certain locations, which constitutes terminating action for repetitive inspections of the modified areas. Additionally, that AD requires replacement of a preventive modification with an improved modification.

Explanation of Relevant Service Information

We have reviewed and approved Boeing Alert Service Bulletin 737-53A1210, Revision 1, including Appendix A and Evaluation Form, dated October 25, 2001. The service bulletin describes procedures for repetitive external detailed and eddy current inspections to find fatigue cracking of the upper and lower skin panels of the fuselage (crown area and lower lobe area) at stringer S-12, and above the S-4, S-10, and S-14 lap joints, and repair of any cracking with either a permanent or time-limited repair.

For airplanes on which a time-limited repair is done, Part 4 of the service bulletin describes procedures for a subsequent permanent repair within 10,000 flight cycles after installation of the time-limited repair. Doing a permanent repair eliminates the need for the repetitive inspections for the repaired area only.

For Group 3, 5, 6, and 8 airplanes only, on which no cracking is found, Part 5 of the service bulletin provides procedures for a preventive modification of the chem-milled pockets in the upper skins at stringer S-12, between body station (BS) 500D and BS 520, which would end the repetitive inspections for the modified area only.

The service bulletin also describes procedures for repetitive follow-on visual inspections for cracking of the lower lobe skins from S-15L to S-15R between stations 360 and 1016 and in section 41; replacement of any loose fasteners with new fasteners; an internal eddy current inspection of the skin, tear straps, and lap joint in each adjacent bay for cracking; and repair of any cracking found.

Accomplishment of the actions specified in Service Bulletin 737-53A1210, Revision 1, is intended to adequately address the identified unsafe condition.

Explanation of Requirements of Proposed Rule

Since an unsafe condition has been identified that is likely to exist or develop on other products of this same type design, the proposed AD would require accomplishment of the actions specified in the service bulletin described previously, except as discussed below.

Differences Between Proposed AD and Service Bulletin 737-53A1210, Revision 1

The service bulletin specifies that the manufacturer may be contacted for disposition of certain repair conditions, but this proposed AD would require the repair of those conditions to be done per a method approved by the FAA, or per data meeting the type certification basis of the airplane approved by a Boeing Company Designated Engineering Representative who has been authorized by the FAA to make such findings.

The service bulletin recommends that, after installation of a time-limited repair, an internal eddy current inspection should be done at the first "C-check" or within 4,000 flight cycles, whichever is last. Because "C-check" schedules vary among operators, such a nonspecific interval would provide no assurance that operators would do the inspection within the prescribed schedule. This proposed AD would require that the inspection be done within 4,000 flight cycles after the repair installation. We find that a 4,000-flight-cycle interval is appropriate for affected airplanes to continue to operate without compromising safety.

Although the service bulletin recommends that operators report inspection results to the manufacturer, this proposed AD does not contain such a reporting requirement.

Interim Action

This is considered to be interim action for Group 7 airplanes. Although the service bulletin described

previously does not include the inspection of the crown area (upper lobe) for Group 7 airplanes, as specified in paragraph (a) of this proposed AD, the manufacturer has advised that it currently is developing a new service bulletin to address those airplanes. Once the FAA has reviewed and approved the service bulletin, we may consider additional rulemaking to mandate those inspections.

Cost Impact

There are approximately 2,200 airplanes of the affected design in the worldwide fleet. The FAA estimates that 903 airplanes of U.S. registry would be affected by this proposed AD.

It would take approximately 94 work hours per airplane to accomplish the proposed inspections of the crown area, at an average labor rate of \$60 per work hour. Based on these figures, the cost impact of these proposed inspections on U.S. operators is estimated to be \$5,092,920, or \$5,640 per airplane, per inspection cycle.

It would take approximately 96 work hours per airplane to accomplish the proposed inspections of the lower lobe area, at an average labor rate of \$60 per work hour. Based on these figures, the cost impact of these proposed inspections on U.S. operators is estimated to be \$5,201,280, or \$5,760 per airplane, per inspection cycle.

The cost impact figures discussed above are based on assumptions that no operator has yet accomplished any of the proposed requirements of this AD action, and that no operator would accomplish those actions in the future if this proposed AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up, planning time, or time necessitated by other administrative actions.

Should an operator elect to install the preventive modification, it would take approximately 108 work hours to accomplish, at an average labor rate of \$60 per work hour. Based on these figures, the cost impact of the preventive modification is estimated to be \$6,480 per airplane.

Regulatory Impact

The regulations proposed herein would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore,

it is determined that this proposal would not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this proposed regulation (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption

ADDRESSES.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

Boeing: Docket 2001-NM-246-AD.

Applicability: Model 737-200, -200C, -300, -400, and -500 series airplanes, as listed in Boeing Alert Service Bulletin 737-53A1210, Revision 1, dated October 25, 2001; certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (g) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To find and fix fatigue cracking of certain upper and lower skin panels of the fuselage, which could result in sudden fracture and failure of the skin panels and consequent rapid decompression of the airplane, accomplish the following:

External Detailed and Eddy Current Inspections

(a) For Groups 1 through 6 and Group 8 airplanes: Before the accumulation of 35,000 total flight cycles, or within 4,500 flight cycles after the effective date of this AD, whichever is later, do external detailed and eddy current inspections of the crown area skin panels of the fuselage for cracking, per Part 1 and Figure 1 of the Work Instructions of Boeing Alert Service Bulletin 737-53A1210, Revision 1, including Appendix A and excluding Evaluation Form, dated October 25, 2001. Repeat the inspections at least every 4,500 flight cycles until paragraph (c) or (d)(1)(ii) of this AD has been done, as applicable. Although paragraph 1.D. of the service bulletin references a reporting requirement, such reporting is not required by this AD.

Note 2: For the purposes of this AD, a detailed inspection is defined as: "An intensive visual examination of a specific structural area, system, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at intensity deemed appropriate by the inspector. Inspection aids such as mirror, magnifying lenses, etc., may be used. Surface cleaning and elaborate access procedures may be required."

(b) For all airplanes: Before the accumulation of 40,000 total flight cycles, or within 4,500 flight cycles after the effective date of this AD, whichever is later, do an external detailed inspection of the lower lobe area and section 41 of the fuselage for cracking, per Part 2 and Figure 2 of the Work Instructions of Boeing Alert Service Bulletin 737-53A1210, Revision 1, including Appendix A and excluding Evaluation Form, dated October 25, 2001. Repeat the inspection at least every 9,000 flight cycles until paragraph (d)(2) of this AD has been done, as applicable.

Preventive Modification

(c) For Groups 3, 5, 6, and 8 airplanes: If no cracking is found during any inspection required by paragraph (a) of this AD, doing the preventive modification of the chem-milled pockets in the upper skin as specified in Part 5 of the Work Instructions of Boeing Alert Service Bulletin 737-53A1210, Revision 1, including Appendix A and excluding Evaluation Form, dated October 25, 2001, ends the repetitive inspections for the modified area only.

Corrective Actions

(d) If any cracking is found during any inspection required by paragraph (a) or (b) of this AD, before further flight, do the actions specified in paragraphs (d)(1) and (d)(2) of this AD, as applicable, per the Work Instructions of Boeing Alert Service Bulletin 737-53A1210, Revision 1, including Appendix A and excluding Evaluation Form,

dated October 25, 2001. Where the service bulletin specifies to contact Boeing for repair instructions, before further flight, repair per a method approved by the Manager, Seattle Aircraft Certification Office (ACO), FAA; or per data meeting the type certification basis of the airplane approved by a Boeing Company Designated Engineering Representative (DER) who has been authorized by the Manager, Seattle ACO, to make such findings. For a repair method to be approved by the Manager, Seattle ACO, as required by this paragraph, the approval letter must specifically reference this AD.

(1) For cracking of the crown area, do the repair specified in either paragraph (d)(1)(i) or (d)(1)(ii) of this AD. Installation of the lap joint repair specified in paragraph (g) of AD 2002-07-08, amendment 39-12702, is considered acceptable for compliance with the corresponding action specified in this paragraph for the lap joint areas only.

(i) Do a time-limited repair per Part 4 of the Work Instructions of the service bulletin, then do the actions required by paragraph (e) of this AD at the times specified in that paragraph.

(ii) Do a permanent repair per Part 3 of the Work Instructions of the service bulletin. Installation of a permanent repair ends the repetitive inspections required by paragraph (a) of this AD for the repaired area only.

(2) For cracking of the lower lobe area and Section 41, repair per Part 2 of the Work Instructions of the service bulletin. Accomplishment of this repair ends the repetitive inspections required by paragraph (b) of this AD for the repaired area only.

Follow-on and Corrective Actions

(e) If a time-limited repair is done, as specified in paragraph (d)(1)(i) of this AD: Do the actions specified in paragraphs (e)(1), (e)(2), and (e)(3) of this AD, at the times specified, per the Work Instructions of Boeing Alert Service Bulletin 737-53A1210, Revision 1, including Appendix A and excluding Evaluation Form, dated October 25, 2001.

(1) Within 3,000 flight cycles after doing the repair: Do a general visual inspection of the repaired area for loose fasteners per Part 4 of the Work Instructions of the service bulletin. If any loose fastener is found, before further flight, replace with a new fastener per the service bulletin. Then repeat the inspection at least every 3,000 flight cycles until permanent rivets are installed in the repaired area, which ends the repetitive inspections for this paragraph.

(2) Within 4,000 flight cycles after doing the repair: Do an internal eddy current inspection of the skin, tear straps, and lap joint in each adjacent bay of the repaired area for cracking, per Part 4 of the Work Instructions of the service bulletin. If any cracking is found, before further flight, repair per a method approved by the Manager, Seattle ACO, or per data meeting the type certification basis of the airplane approved by a Boeing Company DER who has been authorized by the FAA to make such findings. For a repair method to be approved by the Manager, Seattle ACO, as required by this paragraph, the approval letter must specifically reference this AD.

(3) Within 10,000 flight cycles after doing the repair: Make the repair permanent per Part 4 and Figure 20 of the Work Instructions of the service bulletin, which ends the repetitive inspections for the repaired area only.

Credit for Actions Done per Previous Service Bulletin

(f) Inspections, repairs, and preventive modifications done before the effective date of this AD per Boeing Alert Service Bulletin 737-53A1210, dated December 14, 2000, are acceptable for compliance with the corresponding actions required by this AD.

Alternative Methods of Compliance

(g) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Seattle ACO. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Seattle ACO.

Note 3: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Seattle ACO.

Special Flight Permit

(h) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Issued in Renton, Washington, on June 11, 2003.

Ali Bahrami,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 03-15327 Filed 6-17-03; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2000-NM-169-AD]

RIN 2120-AA64

Airworthiness Directives; McDonnell Douglas Model DC-9-81 (MD-81), DC-9-82 (MD-82), DC-9-83 (MD-83), DC-9-87 (MD-87), and Model MD-88 Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: This document proposes the adoption of a new airworthiness directive (AD) that is applicable to certain McDonnell Douglas airplanes. This proposal would require reversing the ground stud installation of the main