

(2) You may get copies from APEX AIRCRAFT, Direction Technique, Route de Troyes, F21121 Darois, France; telephone: +33 (380) 356 510; facsimile: +33 (380) 356 515. You may view copies at the FAA, Central Region, Office of the Regional Counsel, 901 Locust, Room 506, Kansas City, Missouri, or at the Office of the Federal Register, 800 North Capitol Street, NW, suite 700, Washington, DC.

(i) *Does this AD action affect any existing AD actions?* This amendment supersedes AD 98-12-10, Amendment 39-10566 and AD 99-21-23, Amendment 39-11368.

Note 3: The subject of this AD is addressed in French AD Number 2001-616(A) R1, dated May 29, 2002.

(j) *When does this amendment become effective?* This amendment becomes effective on April 4, 2003.

Issued in Kansas City, Missouri, on February 4, 2003.

Michael Gallagher,
 Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. 03-3450 Filed 2-18-03; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2002-CE-47-AD; Amendment 39-13056; AD 2003-04-08]

RIN 2120-AA64

Airworthiness Directives; Piaggio Aero Industries S.p.A. Model P-180 Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD) that applies to certain Piaggio Aero Industries S.p.A. (Piaggio) Model P-180 airplanes. This AD requires you to install a placard on the inside of the lavatory door that prohibits occupying the lavatory seat during takeoff and landing. This AD also requires you to incorporate a temporary revision into the Limitations Section of the pilot operating handbook/airplane flight

manual (POH/AFM). This AD is the result of mandatory continuing airworthiness information (MCAI) issued by the airworthiness authority for Italy. The actions specified by this AD are intended to prevent passengers from occupying the lavatory seat during takeoff and landing. The lavatory/cabin partition could fail and lead to passenger injury in an emergency situation.

DATES: This AD becomes effective on April 11, 2003.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in the regulations as of April 11, 2003.

ADDRESSES: You may get the service information referenced in this AD from Piaggio Aero Industries S.p.A, Via Cibrario 4, 16154 Genoa, Italy; telephone: +39 010 6481 856; facsimile: +39 010 6481 374. You may view this information at the Federal Aviation Administration (FAA), Central Region, Office of the Regional Counsel, Attention: Rules Docket No. 2002-CE-47-AD, 901 Locust, Room 506, Kansas City, Missouri 64106; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Doug Rudolph, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4059; facsimile: (816) 329-4090.

SUPPLEMENTARY INFORMATION:

Discussion

What events have caused this AD? The Ente Nazionale per l'Aviazione Civile (ENAC), which is the airworthiness authority for Italy, recently notified FAA of a manufacturing/installation defect on the lavatory/cabin partitions on certain Piaggio Model P-180 airplanes. The lavatory/cabin partitions were installed improperly and are not of sufficient strength. This condition was found during a quality control inspection.

What is the potential impact if FAA took no action? Occupying the lavatory seat during takeoff or landing could result in failure of the lavatory/cabin

partition. Such failure could result in passenger injury in an emergency situation.

Has FAA taken any action to this point? We issued a proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an AD that would apply to certain Piaggio Model P-180 airplanes. This proposal was published in the **Federal Register** as a notice of proposed rulemaking (NPRM) on November 21, 2002 (67 FR 70187). The NPRM proposed to require you to install a placard on the inside of the lavatory door that prohibits occupying the lavatory seat during takeoff and landing; and incorporate a temporary revision into the Limitations Section of the pilot operating handbook/airplane flight manual (POH/AFM).

Was the public invited to comment? The FAA encouraged interested persons to participate in the making of this amendment. We did not receive any comments on the proposed rule or on our determination of the cost to the public.

FAA's Determination

What is FAA's final determination on this issue? After careful review of all available information related to the subject presented above, we have determined that air safety and the public interest require the adoption of the rule as proposed except for minor editorial corrections. We have determined that these minor corrections:

- Provide the intent that was proposed in the NPRM for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the NPRM.

Cost Impact

How many airplanes does this AD impact? We estimate that this AD affects 12 airplanes in the U.S. registry.

What is the cost impact of this AD on owners/operators of the affected airplanes? We estimate the following costs to accomplish the placard installation:

Labor cost	Parts cost	Total cost per airplane	Total cost on U.S. operators
1 workhour × \$60 = \$60	\$20	\$80	12 × \$80 = \$960

Compliance Time of This AD

What is the compliance time of this AD? The compliance time of this AD is "within the next 30 days after the

effective date of this AD, unless already accomplished."

Why is the compliance time presented in calendar time instead of hours time-in-service (TIS)? The compliance of this

AD is presented in calendar time instead of hours TIS because the lavatory/cabin partitions are unsafe as a result of an improper installation. The unsafe condition has the same chance of

occurring on an airplane with 50 hours TIS as it does for an airplane with 1,000 hours TIS. Therefore, we believe that a compliance time of 30 days will:

- Ensure that the unsafe condition does not go undetected for a long period of time on the affected airplanes; and
- Not inadvertently ground any of the affected airplanes.

Regulatory Impact

Does this AD impact various entities? The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

Does this AD involve a significant rule or regulatory action? For the reasons discussed above, I certify that this action (1) is not a “significant regulatory action” under Executive Order 12866; (2) is not a “significant rule” under DOT Regulatory Policies and Procedures (44

FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the final evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. FAA amends § 39.13 by adding a new AD to read as follows:

2003-04-08 Piaggio Aero Industries S.p.A.:
Amendment 39-13056; Docket No. 2002-CE-47-AD.

(a) *What airplanes are affected by this AD?* This AD affects Model P-180 airplanes, serial numbers 1002, 1004, 1006 through 1037, 1039, 1040, 1042, 1043, and 1045, that are:

- (1) Equipped with a toilet seat; and
- (2) Are certificated in any category.

(b) *Who must comply with this AD?* Anyone who wishes to operate any of the airplanes identified in paragraph (a) of this AD must comply with this AD.

(c) *What problem does this AD address?* The actions specified by this AD are intended to prevent passengers from occupying the lavatory seat during takeoff and landing. The lavatory/cabin partition could fail and lead to passenger injury in an emergency situation.

(d) *What actions must I accomplish to address this problem?* To address this problem, you must accomplish the following:

Actions	Compliance	Procedures
(1) Fabricate a placard that incorporates the following words (using at least ¼-inch black letters on a white background) and install this placard on the inside of the lavatory door in front of the lavatory seat: “LAVATORY SEAT CANNOT BE OCCUPIED DURING TAKEOFF AND LANDING”.	Within the next 30 days after April 11, 2003 (the effective date of this AD), unless already accomplished.	The owner/operator holding at least a private pilot certificate as authorized by section 43.7 of the Federal Aviation Regulations (14 CFR 43.7) may fabricate and install the placard. Make an entry into the aircraft records showing compliance with these portions of the AD in accordance with section 43.9 of the Federal Aviation Regulations (14 CFR 43.9).
(2) Incorporate into the Limitations Section of the pilot operating handbook/airplane flight manual (POH/AFM), page 4 of Piaggio Alert Service Bulletin No. ASB-80-0164, Original Issue: September 10, 2001.	Within the next 30 days after April 11, 2003 (the effective date of this AD), unless already accomplished.	The owner/operator holding at least a private pilot certificate as authorized by § 43.7 of the Federal Aviation Regulations (14 CFR 43.7) may accomplish the POH/AFM manual insertion of this AD. Make an entry into the aircraft records showing compliance with these portions of the AD in accordance with section 43.9 of the Federal Aviation Regulations (14 CFR 43.9).
(3) As an alternative method of compliance to this AD, you may modify the lavatory/cabin partition.	At any time as terminating action for the placard and POH/AFM requirements of this AD.	In accordance with Piaggio Service Bulletin (Recommended) No. SB-80-0165, Original Issue: September 10, 2001.

Note 1: Information about fabricating and installing the placard and the POH/AFM manual insertion is referenced in Piaggio Alert Service Bulletin No. ASB-80-0164, Original Issued: September 10, 2001.

(e) *Can I comply with this AD in any other way?* You may use an alternative method of compliance or adjust the compliance time if:

- (1) Your alternative method of compliance provides an equivalent level of safety; and
- (2) The Manager, Standards Office, Small Airplane Directorate, approves your alternative. Submit your request through an FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Manager, Standards Office.

Note 2: This AD applies to each airplane identified in paragraph (a) of this AD, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (e) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if you have not eliminated the unsafe condition, specific actions you propose to address it.

(f) *Where can I get information about any already-approved alternative methods of compliance?* Contact Doug Rudolph, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4059; facsimile: (816) 329-4090.

(g) *What if I need to fly the airplane to another location to comply with this AD?* The FAA can issue a special flight permit under §§ 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate your airplane to a location where you can accomplish the requirements of this AD.

(h) *Are any service bulletins incorporated into this AD by reference?* The POH/AM requirements of this AD must be done in

accordance with Piaggio Alert Service Bulletin No. ASB-80-0164, Original Issue: September 10, 2001. The procedures for accomplishing the optional modification of this AD are contained in Piaggio Service Bulletin (Recommended) No. SB-80-0165, Original Issue: September 10, 2001. The Director of the Federal Register approved this incorporation by reference under 5 U.S.C. 552(a) and 1 CFR part 51. You may get copies from Piaggio Aero Industries S.p.A, Via Cibrario 4, 16154 Genoa, Italy; telephone: +39 010 6481 856; facsimile: +39 010 6481. You may view copies at the FAA, Central Region, Office of the Regional Counsel, 901 Locust, Room 506, Kansas City, Missouri, or at the Office of the Federal Register, 800 North Capitol Street, NW., Suite 700, Washington, DC.

Note 3: The subject of this AD is addressed in Italian AD Number 2001-513, dated November 30, 2001.

(i) *When does this amendment become effective?* This amendment becomes effective on April 11, 2003.

Issued in Kansas City, Missouri, on February 10, 2003.

Michael Gallagher,

Manager, Small Airplane Directorate, Aircraft Certification Service.

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2001-SW-45-AD; Amendment 39-13053; AD 2003-04-05]

RIN 2120-AA64

Airworthiness Directives; Robinson Helicopter Company Model R44 Helicopters

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD) for Robinson Helicopter Company (RHC) Model R44 helicopters that requires inspecting the tail rotor pitch control assembly for roughness or binding of the pitch control bearings (bearings) by hand-rotating the pitch control bearing housing (housing). If the housing does not rotate freely, the AD requires replacing the unairworthy pitch control assembly with an airworthy unit. This amendment is prompted by reports of failure of the tail rotor pitch control assembly due to improperly lubricated bearings on the RHC Model R22 helicopters. Although there have been no reported failures on the RHC Model R44 helicopters, the design of the tail

rotor pitch control assembly makes it susceptible to the same failures as have occurred on the Model R22 helicopters. The actions specified by this AD are intended to detect corrosion of the bearings and to prevent bearing failure and subsequent loss of directional control of the helicopter.

DATES: Effective March 26, 2003.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of March 26, 2003.

ADDRESSES: The service information referenced in this AD may be obtained from Robinson Helicopter Company, 2901 Airport Drive, Torrance, California 90505, telephone (310) 539-0508, fax (310) 539-5198. This information may be examined at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC. **FOR FURTHER INFORMATION CONTACT:** Fred Guerin, Aviation Safety Engineer, FAA, Los Angeles Aircraft Certification Office, Airframe Branch, 3960 Paramount Blvd., Lakewood, California 90712, telephone (562) 627-5232, fax (562) 627-5210.

SUPPLEMENTARY INFORMATION: A proposal to amend 14 CFR part 39 to include an AD for the specified helicopters was published in the **Federal Register** on September 10, 2002 (67 FR 57351). That action proposed inspecting the pitch control assembly for roughness or binding of the bearings by hand-rotating the housing and if the housing does not rotate freely, replacing each unairworthy pitch control assembly with an airworthy unit.

The FAA has reviewed RHC Service Bulletin SB-43A, Revision A, dated June 10, 2002 (SB), which describes procedures for inspecting the pitch control assembly for roughness or binding of the bearings by hand-rotating the housing. If the housing does not rotate freely, the SB specifies replacing each unairworthy pitch control assembly, part number (P/N) A031-1, with an airworthy unit in accordance with the maintenance manual.

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due consideration has been given to the comments received.

One commenter, the manufacturer, requests a change to the summary information to indicate that bearing failures have been reported only on RHC Model R22 helicopters, and there are no reports of failed bearings on Model R44

helicopters. The FAA agrees and has changed the preamble information to indicate that the bearing failures have occurred on the RHC Model R22 helicopters only.

The same commenter requests a change to the summary and the discussion sections to revise the failure sequence to indicate that bearing failure could result in loss of tail rotor thrust requiring a power-off landing. The commenter states that in all three of the bearing failures, the failed bearing caused the pitch control linkage to fail and the tail rotor to go to flat pitch but none of the failures resulted in an accident. The commenter also states that there was no breakup of the tail rotor assembly, no tail rotor contact of the tailboom, and no loss of control resulting in an accident. The FAA agrees that the failed bearing has not resulted in breakup of the tail rotor assembly and contact with the tailboom, and the likelihood of such a breakup and contact with the tailboom may be remote. Therefore, we have removed the reference to the breakup of the tail rotor assembly and contact with the tailboom from the failure sequence. We do not agree that the failure sequence should state that bearing failure could result in loss of tail rotor thrust requiring a power-off landing. The loss of directional control associated with this type of failure could result in loss of control of the helicopter, and a successful power-off landing may not be possible. The term "loss of control of the helicopter", however, may be understood to mean an almost certain catastrophic event, such as loss of cyclic or pitch control. That is not our intent when we use the failure sequence in our AD's. That sequence states what could happen not necessarily what will happen. Our intent is to convey the sequence of events that we intend to prevent by issuing the AD to correct the unsafe condition. Therefore, we have changed the failure sequence to state that loss of "directional" control can result.

After careful review of the available data, including the comments noted above, the FAA has determined that air safety and the public interest require the adoption of the rule as proposed with the changes described previously. The FAA has determined that these changes will neither increase the economic burden on any operator nor increase the scope of the AD.

The FAA estimates that this AD will affect 440 helicopters of U.S. registry, that it will take approximately 2.3 work hours per helicopter to inspect and replace each pitch control assembly, and that the average labor rate is \$60 per