

premiums will be reduced based on your new, lower level of coverage. The Carrier will refund or credit any portion of premium paid in advance for the period following the date on which you decrease your coverage.

(c) You may cancel your coverage at any time.

(1) If you cancel during the free look period, your premiums will be refunded to you.

(2) If you cancel your coverage at any time other than during the free look period, cancellation will take effect on your requested cancellation date or at the end of the period covered by your last premium payment, whichever occurs first. You will not receive any refund of premiums paid, other than any premiums paid in advance for the period following the effective date of your cancellation of coverage, and you will not have to pay any more premiums unless you owed retroactive premiums.

§ 875.407 Who makes insurability decisions?

The Carrier determines the insurability of all applicants. The Carrier's decision may not be appealed to OPM.

§ 875.408 What is the significance of incontestability?

(a) *Incontestability* means coverage issued based on an erroneous application may remain in effect. Such coverage will not remain in effect, and your claim may be denied, under any of the following conditions:

(1) If your coverage has been in force for less than 6 months, the Carrier may void your coverage or deny a claim upon a showing that information on your signed application that was material to your approval for coverage is different than what is shown in your medical records.

(2) If your coverage has been in force for at least 6 months but less than 2 years, the Carrier may void your coverage or deny a claim upon a showing that information on your signed application that was material to your approval for coverage is different than what is shown in your medical records, and pertains to the condition for which benefits are sought.

(3) After your coverage has been in effect for 2 years, the Carrier may void your coverage only upon a showing that you knowingly and intentionally made a false or misleading statement or omitted information in your signed application for coverage regarding your health status.

(b) Your coverage can be contested at any time when the Carrier finds that you were not an eligible individual at the

time you applied and were approved for coverage.

(c) If the Carrier voids coverage after it has paid benefits, it cannot recover the benefits already paid.

(d) Incontestability does not apply when you have not paid your premiums on a timely basis.

§ 875.409 Must I provide an authorization to release medical information?

You must provide the Carrier with an authorization to release medical information when requested. The Carrier may deny a claim for benefits or void your coverage if the Carrier does not receive an authorization to release medical information within 3 weeks after its request (4 weeks for those outside the United States).

§ 875.410 May I continue my coverage when I leave Federal or military service?

If you are a Federal civilian or Postal employee or member of the uniformed services, your coverage will automatically continue when you leave active service, as long as the Carrier continues to receive the required premium when due. However, once you leave active service, you are no longer eligible for any abbreviated underwriting provided during any future open season.

§ 875.411 May I continue my coverage when I am no longer a qualified relative?

If you are already enrolled as a qualified relative, you may continue your FLTCIP coverage if you subsequently lose qualified relative status (such as upon divorce), as long as the Carrier receives the required premium when due.

§ 875.412 When will my coverage terminate?

Your coverage will terminate on the earliest of the following dates:

(a) The date you specify to the Carrier that you wish your coverage to end;

(b) The date of your death;

(c) The end of the period covered by your last premium payment if you do not pay the required premiums when due, after a grace period of 30 days; or

(d) The date you have exhausted your maximum lifetime benefit. (However, in this event, care coordination services will continue.)

§ 875.413 Is it possible to have coverage reinstated?

(a) Under certain circumstances, your coverage can be reinstated. The Carrier will reinstate your coverage if it receives proof satisfactory to it, within 6 months from the termination date, that you suffered from a cognitive impairment or loss of functional capacity, before the

grace period ended, that caused you to miss making premium payments. In that event, you will not be required to submit to underwriting. Your coverage will be reinstated retroactively to the termination date but you must pay back premiums for that period. The premium will be the same as it was prior to termination.

(b) If your coverage has terminated because you did not pay premiums or because you requested cancellation, the Carrier may reinstate your coverage within 12 months from the termination date at your request. You will be required to reapply based on full underwriting, and the Carrier will determine whether you are still insurable. If you are insurable, your coverage will be reinstated retroactively to the termination date and you must pay back premiums for that period. The premium will be the same as it was prior to termination.

§ 875.414 Will benefits be coordinated with other coverage?

Yes, benefits will be coordinated with other plans, following the coordination of benefits (COB) guidelines set by the National Association of Insurance Commissioners. The total benefits from all plans that pay a long term care benefit to you should not exceed the actual costs you incur. The other plans that are considered for COB purposes include government programs, group medical benefits, and other employer-sponsored long term care insurance plans. Medicaid, individual insurance policies, and association group insurance policies are not taken into consideration under this provision.

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 23

[Docket No. CE194, Special Condition 23-134-SC]

Special Conditions; Cirrus Design Corporation SR22; Protection of Systems for High Intensity Radiated Fields (HIRF)

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final special conditions; request for comments.

SUMMARY: These special conditions are issued to Cirrus Design Corporation, 4515 Taylor Circle, Duluth, Minnesota 55811, for a Type Design Change for the

Cirrus Design Corporation Model SR22 airplane. This airplane will have novel and unusual design features when compared to the state of technology envisaged in the applicable airworthiness standards. These novel and unusual design features include the installation of electronic flight instrument system (EFIS) displays Model 700-00006-XXX-() manufactured by Avidyne Corporation for which the applicable regulations do not contain adequate or appropriate airworthiness standards for the protection of these systems from the effects of high intensity radiated fields (HIRF). These special conditions contain the additional safety standards that the Administrator considers necessary to establish a level of safety equivalent to the airworthiness standards applicable to these airplanes.

DATES: The effective date of these special conditions is January 24, 2003. Comments must be received on or before March 6, 2003.

ADDRESSES: Comments may be mailed in duplicate to: Federal Aviation Administration, Regional Counsel, ACE-7, Attention: Rules Docket Clerk, Docket No. CE194, Room 506, 901 Locust, Kansas City, Missouri 64106. All comments must be marked: Docket No. CE194. Comments may be inspected in the Rules Docket weekdays, except Federal holidays, between 7:30 a.m. and 4 p.m.

FOR FURTHER INFORMATION CONTACT: Wes Ryan, Aerospace Engineer, Standards Office (ACE-110), Small Airplane Directorate, Aircraft Certification Service, Federal Aviation Administration, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone (816) 329-4127.

SUPPLEMENTARY INFORMATION: The FAA has determined that notice and opportunity for prior public comment hereon are impracticable because these procedures would significantly delay issuance of the design approval and thus delivery of the affected aircraft. In addition, the substance of these special conditions has been subject to the public comment process in several prior instances with no substantive comments received. The FAA, therefore, finds that good cause exists for making these special conditions effective upon issuance.

Comments Invited

Interested persons are invited to submit such written data, views, or arguments as they may desire. Communications should identify the regulatory docket or notice number and be submitted in duplicate to the address

specified above. All communications received on or before the closing date for comments will be considered by the Administrator. The special conditions may be changed in light of the comments received. All comments received will be available in the Rules Docket for examination by interested persons, both before and after the closing date for comments. A report summarizing each substantive public contact with FAA personnel concerning this rulemaking will be filed in the docket. Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this notice must include a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket No. CE194." The postcard will be date stamped and returned to the commenter.

Background

On July 8, 2002, Cirrus Design Corporation, 4515 Taylor Circle, Duluth, Minnesota 55811, made an application to the FAA for a Type Design Change for the Cirrus Design Corporation Model SR22 airplane. The Model SR22 is currently approved under TC No. A00009CH. The proposed modification incorporates a novel or unusual design feature, such as digital avionics consisting of an EFIS, that is vulnerable to HIRF external to the airplane.

Type Certification Basis

Under the provisions of 14 CFR part 21, § 21.101, Cirrus Design Corporation must show that the Cirrus Design Corporation Model SR22 aircraft meets the following provisions, or the applicable regulations in effect on the date of application for the change to the Cirrus Design Corporation Model SR22: Part 23 of the Federal Aviation Regulations effective February 1, 1965, as amended by 23-1 through 23-53, except as follows: § 23.301 through Amendment 47; §§ 23.855, 23.1326, 23.1359, not applicable. 14 CFR 36 dated December 1, 1969, as amended by current amendment as of the date of type Certification.

Equivalent Levels of Safety finding (ACE-96-5) made per the provisions of 14 CFR part 23, § 23.221; Refer to FAA ELOS letter dated June 10, 1998 for models SR20, SR22. Equivalent Levels Of Safety finding (ACE-00-09) made per the provisions of 14 CFR part 23, §§ 23.1143(g) and 23.1147(b); Refer to FAA ELOS letter dated September 11, 2000 for model SR22.

Special Condition (23-ACE-88) for ballistic parachute; Refer to FAA letter

November 25, 1997 for models SR20, SR22.

Discussion

If the Administrator finds that the applicable airworthiness standards do not contain adequate or appropriate safety standards because of novel or unusual design features of an airplane, special conditions are prescribed under the provisions of § 21.16.

Special conditions, as appropriate, as defined in § 11.19, are issued in accordance with § 11.38 after public notice and become part of the type certification basis in accordance with § 21.101 (b)(2).

Special conditions are initially applicable to the model for which they are issued. Should the applicant apply for a supplemental type certificate to modify any other model already included on the same type certificate to incorporate the same novel or unusual design feature, the special conditions would also apply to the other model under the provisions of § 21.101.

Novel or Unusual Design Features

Cirrus Design Corporation plans to incorporate certain novel and unusual design features into an airplane for which the airworthiness standards do not contain adequate or appropriate safety standards for protection from the effects of HIRF. These features include EFIS, which are susceptible to the HIRF environment, that were not envisaged by the existing regulations for this type of airplane.

Protection of Systems From High Intensity Radiated Fields (HIRF)

Recent advances in technology have given rise to the application in aircraft designs of advanced electrical and electronic systems that perform functions required for continued safe flight and landing. Due to the use of sensitive solid state advanced components in analog and digital electronics circuits, these advanced systems are readily responsive to the transient effects of induced electrical current and voltage caused by the HIRF. The HIRF can degrade electronic systems performance by damaging components or upsetting system functions.

Furthermore, the HIRF environment has undergone a transformation that was not foreseen when the current requirements were developed. Higher energy levels are radiated from transmitters that are used for radar, radio, and television. Also, the number of transmitters has increased significantly. There is also uncertainty concerning the effectiveness of airframe

shielding for HIRF. Furthermore, coupling to cockpit-installed equipment through the cockpit window apertures is undefined.

The combined effect of the technological advances in airplane design and the changing environment has resulted in an increased level of vulnerability of electrical and electronic systems required for the continued safe flight and landing of the airplane. Effective measures against the effects of exposure to HIRF must be provided by the design and installation of these systems. The accepted maximum energy

levels in which civilian airplane system installations must be capable of operating safely are based on surveys and analysis of existing radio frequency emitters. These special conditions require that the airplane be evaluated under these energy levels for the protection of the electronic system and its associated wiring harness. These external threat levels, which are lower than previous required values, are believed to represent the worst case to which an airplane would be exposed in the operating environment.

These special conditions require qualification of systems that perform critical functions, as installed in aircraft, to the defined HIRF environment in paragraph 1 or, as an option to a fixed value using laboratory tests, in paragraph 2, as follows:

(1) The applicant may demonstrate that the operation and operational capability of the installed electrical and electronic systems that perform critical functions are not adversely affected when the aircraft is exposed to the HIRF environment defined below:

Frequency	Field strength (volts per meter)	
	Peak	Average
10 kHz–100 kHz	50	50
100 kHz–500 kHz	50	50
500 kHz–2 MHz	50	50
2 MHz–30 MHz	100	100
30 MHz–70 MHz	50	50
70 MHz–100 MHz	50	50
100 MHz–200 MHz	100	100
200 MHz–400 MHz	100	100
400 MHz– 700 MHz	700	50
700 MHz–1 GHz	700	100
1 GHz–2 GHz	2000	200
2 GHz–4 GHz	3000	200
4 GHz–6 GHz	3000	200
6 GHz–8 GHz	1000	200
8 GHz–12 GHz	3000	300
12 GHz–18 GHz	2000	200
18 GHz–40 GHz	600	200

Note.—The field strengths are expressed in terms of peak root-mean-square (rms) values.

or,

(2) The applicant may demonstrate by a system test and analysis that the electrical and electronic systems that perform critical functions can withstand a minimum threat of 100 volts per meter, electrical field strength, from 10 kHz to 18 GHz. When using this test to show compliance with the HIRF requirements, no credit is given for signal attenuation due to installation.

A preliminary hazard analysis must be performed by the applicant, for approval by the FAA, to identify either electrical or electronic systems that perform critical functions. The term "critical" means those functions whose failure would contribute to, or cause, a failure condition that would prevent the continued safe flight and landing of the airplane. The systems identified by the hazard analysis that perform critical functions are candidates for the application of HIRF requirements. A system may perform both critical and non-critical functions. Primary electronic flight display systems, and their associated components, perform critical functions such as attitude,

altitude, and airspeed indication. The HIRF requirements apply only to critical functions.

Compliance with HIRF requirements may be demonstrated by tests, analysis, models, similarity with existing systems, or any combination of these. Service experience alone is not acceptable since normal flight operations may not include an exposure to the HIRF environment. Reliance on a system with similar design features for redundancy as a means of protection against the effects of external HIRF is generally insufficient since all elements of a redundant system are likely to be exposed to the fields concurrently.

Applicability

As discussed above, these special conditions are applicable to Cirrus Design Corporation Model SR22 airplane. Should Cirrus Design Corporation apply at a later date for a type design change to modify any other model on the same type certificate to incorporate the same novel or unusual design feature, the special conditions would apply to that model as well under the provisions of § 21.101.

Conclusion

This action affects only certain novel or unusual design features on one model of airplane. It is not a rule of general applicability and affects only the applicant who applied to the FAA for approval of these features on the airplane.

The substance of these special conditions has been subjected to the notice and comment period in several prior instances and has been derived without substantive change from those previously issued. It is unlikely that prior public comment would result in a significant change from the substance contained herein. For this reason, and because a delay would significantly affect the certification of the airplane, which is imminent, the FAA has determined that prior public notice and comment are unnecessary and impracticable, and good cause exists for adopting these special conditions upon issuance. The FAA is requesting comments to allow interested persons to submit views that may not have been submitted in response to the prior

opportunities for comment described above.

List of Subjects in 14 CFR Part 23

Aircraft, Aviation safety, Signs and symbols.

Citation

The authority citation for these special conditions is as follows:

Authority: 49 U.S.C. 106(g), 40113 and 44701; 14 CFR 21.16 and 21.101; and 14 CFR 11.38 and 11.19.

The Special Conditions

Accordingly, pursuant to the authority delegated to me by the Administrator, the following special conditions are issued as part of the type certification basis for Cirrus Design Corporation SR22 airplane modified by Cirrus Design Corporation to add an EFIS.

1. *Protection of Electrical and Electronic Systems from High Intensity Radiated Fields (HIRF).* Each system that performs critical functions must be designed and installed to ensure that the operations, and operational capabilities of these systems to perform critical functions, are not adversely affected when the airplane is exposed to high intensity radiated electromagnetic fields external to the airplane.

2. For the purpose of these special conditions, the following definition applies:

Critical Functions: Functions whose failure would contribute to, or cause, a failure condition that would prevent the continued safe flight and landing of the airplane.

Issued in Kansas City, Missouri, on January 24, 2003.

Michael Gallagher,

Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. 03-2524 Filed 2-3-03; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2001-NE-48-AD; Amendment 39-13045; AD 2003-03-20]

RIN 2120-AA64

Airworthiness Directives; Hartzell Propeller Inc., Model HC-C2YR-4CF Propellers

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), that is applicable to Hartzell Propeller Inc. model HC-C2YR-4CF propellers. This amendment requires the reduction of the original hub and blades certified service (fatigue) life from unlimited hours to 2,000 hours. This amendment is prompted by a reevaluation by Hartzell Propeller Inc. of the original hub and blades service life certification calculations. The actions specified by this AD are intended to prevent fatigue failure of the original propeller hub and blades which may result in loss of airplane control.

DATES: Effective March 11, 2003.

ADDRESSES: Information regarding this action may be examined, by appointment, at the Federal Aviation Administration (FAA), New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA.

FOR FURTHER INFORMATION CONTACT: Tomaso DiPaolo, Aerospace Engineer, Chicago Aircraft Certification Office, FAA, Small Airplane Directorate, 2300 East Devon Avenue, Des Plaines, IL 60018, telephone (847) 294-7031; fax (847) 294-7834.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an AD that is applicable to Hartzell Propeller Inc. model HC-C2YR-4CF propellers was published in the **Federal Register** on September 19, 2002 (67 FR 59026). That action proposed to require the reduction of the propeller hubs part number (P/N) D-6522-1 or D-2201-16 and blades P/N FC8477A-4 certified service (fatigue) life from unlimited hours to 2,000 hours. The FAA and Hartzell Propeller Inc. have received reports of several engine crankshaft failures on Sky International Inc. (Pitts) S-2S and S-2B airplanes, which are manufactured by Aviat Aircraft Inc. of Afton, WY. Hartzell Propeller Inc. reevaluated the service (fatigue) life of the original propeller hubs P/N D-6522-1 or D-2201-16 and blades P/N FC8477A-4 installed in the model HC-C2YR-4CF propellers. Hartzell has reduced the certified service (fatigue) life of these original propeller hubs and blades from unlimited hours to 2,000 hours. Exceeding these life limits could result in fatigue failure of the hubs or blades which may result in loss of airplane control. The 2,000-hour life limit is documented in the Airworthiness Limitations section of Hartzell Manual 113B.

Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due consideration has been given to the comment received.

Risk if Life of a Component Is Not Known

One commenter states that the proposal introduces a life limit where there was none previously required. The commenter also states that there is a risk that operators or maintenance organizations may not know the current life of the applicable parts, and that the NPRM does not include any proposal to estimate usage or factoring where the life of a component is not known.

The FAA does not agree. Under 14 CFR 91.417(a)(2)(i), each registered owner or operator must keep records of the total time in service of each propeller. The propellers affected by this AD are flown on aircraft used in part 91 operations. Moreover, 14 CFR 91.417(b)(2) requires that the records must denote the total time, must be retained for an unlimited time, and must be transferred with the aircraft. Therefore, if a propeller's total time is unknown, then the propeller and the registered owner or operator are not in compliance with the regulations. Presently, the FAA will not pursue policy to approve a general formula for calculating total time on propellers with unknown total times. Please note that the final rule allows for the submittal of data to request and to justify an alternate method of compliance to the AD or an adjustment of the compliance time in the AD.

After careful review of the available data, including the comment noted above, the FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

Economic Analysis

There are approximately 377 propellers of the affected design in the worldwide fleet. The FAA estimates that 300 propellers installed on airplanes of U.S. registry would be affected by this AD, that it would take approximately 6 work hours per propeller to do the actions, and that the average labor rate is \$60 per work hour. The approximate cost of a new hub and blades is \$9,000. Based on these figures, the total cost of the AD to U.S. operators is estimated to be \$2,808,000.

Regulatory Analysis

This final rule does not have federalism implications, as defined in Executive Order 13132, because it would not have a substantial direct