

FR 11034, February 26, 1979). If it is determined that this emergency regulation otherwise would be significant under DOT Regulatory Policies and Procedures, a final regulatory evaluation will be prepared and placed in the Rules Docket. A copy of it, if filed, may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

#### Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

#### PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

2. Section 39.13 is amended by adding a new airworthiness directive to read as follows:

**2002-13-06 Eurocopter Deutschland GMBH (ECD):** Amendment 39-12794. Docket No. 2002-SW-07-AD.

**Applicability:** Model BO-105A, BO-105C, BO-105 C-2, BO-105 CB-2, BO-105 CB-4, BO-105S, BO-105 CS-2, BO-105 CBS-2, BO-105 CBS-4, and BO-105LS A-1 helicopters, with main rotor head assembly, part number (P/N) 105-14101, and tension-torsion (T-T) straps, P/N 2602559 or P/N 2606576, installed, certificated in any category.

**Note 1:** This AD applies to each helicopter identified in the preceding applicability provision, regardless of whether it has been otherwise modified, altered, or repaired in the area subject to the requirements of this AD. For helicopters that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (e) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

**Compliance:** Required as indicated, unless accomplished previously.

To prevent fatigue failure of a T-T strap, loss of a main rotor blade, and subsequent loss of control of the helicopter, accomplish the following:

(a) Before further flight:

(1) Create a component log card or equivalent record for each T-T strap.

(2) Review the history of each affected helicopter and each T-T strap. For each T-T strap, determine the number of months since initial installation on any helicopter (age), the number of flights, and the number of flight hours time-in-service (TIS). Enter the age, the number of flights, and the number of flight hours TIS for each T-T strap on the component log card or equivalent record.

(i) If the number of flights is unknown, multiply the number of hours TIS by 5 and use this result as the number of flights.

(ii) If a T-T strap has been previously used at any time on Model BO-105LS A-3 "SUPER LIFTER", BO-105 CB-5, BO-105 CBS-5, BO-105 DBS-5, or any MBB-BK 117 series helicopter, multiply the number of flights accumulated on those other models by a factor of 1.6 and then add that result to the number of flights accumulated on the helicopters affected by this AD.

(3) Remove any T-T strap from service if the total hours TIS or number of flights and age cannot be determined.

(b) Before further flight, remove from service and replace with an airworthy T-T strap any T-T strap that has been in service 120 months since initial installation on any helicopter, accumulated 15,600 flights (a flight is a takeoff and a landing), or has accumulated 2,400 hours TIS on any helicopter.

(c) This AD revises the Airworthiness Limitations Section of the maintenance manual by establishing a life limit for the T-T strap, P/N 2602559 and P/N 2606576, of 120 months or 15,600 flights, or 2,400 hours TIS, whichever occurs first.

**Note 2:** T-T straps, P/N 2602559 and P/N 2606576, are no longer in production. T-T straps, P/N 2604067 or P/N J17322-1, may be used as alternate replacements if necessary.

(d) Before T-T straps, P/N 2604067 or P/N J17322-1, are installed, modify any main rotor head P/N 105-14101 configuration to a main rotor head P/N 105-141081 configuration in accordance with paragraph 2, Accomplishment Instructions, and Figure 1 of Eurocopter Service Bulletin No. SB-BO 105-10-100, Revision 1, dated July 16, 2001.

**Note 3:** AD 2001-17-08 (65 FR 52010, August 28, 2000) established the life limits for T-T straps, P/N 2604067 and P/N J17322-1.

(e) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Regulations Group, Rotorcraft Directorate. Operators shall submit their requests through an FAA Principal Maintenance Inspector, who may concur or comment and then send it to the Manager, Regulations Group.

**Note 4:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Regulations Group.

(f) Special flight permits may be issued in accordance with 14 CFR 21.197 and 21.199 to operate the helicopter to a location where the requirements of this AD can be accomplished.

(g) The main rotor head modification shall be done in accordance with paragraph 2 of

the Accomplishment Instructions and Figure 1 of Eurocopter Service Bulletin No. SB-BO 105-10-100, Revision 1, dated July 16, 2001. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR 51. Copies may be obtained from American Eurocopter Corporation, 2701 Forum Drive, Grand Prairie, Texas 75053-4005, telephone (972) 641-3460, fax (972) 641-3527. Copies may be inspected at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

(h) This amendment becomes effective on July 15, 2002.

**Note 5:** The subject of this AD is addressed in Luftfahrt-Bundesamt (Federal Republic of Germany) AD 2001-281, effective October 18, 2001.

Issued in Fort Worth, Texas, on June 18, 2002.

**Eric Bries,**

*Acting Manager, Rotorcraft Directorate, Aircraft Certification Service.*

[FR Doc. 02-16056 Filed 6-27-02; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 99-NE-39-AD; Amendment 39-12791; AD 99-27-16R1]

**RIN 2120-AA64**

#### Airworthiness Directives; CFE Company Model CFE738-1-1B Turbofan Engines

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment revises an existing airworthiness directive (AD), that is applicable to CFE Company Model CFE738-1-1B turbofan engines. That AD currently requires a one-time visual inspection of stage 2 high pressure turbine (HPT) aft cooling plates for nicks, dents, raised metal, and scratches, and if necessary, repair of the cooling plates or replacement with serviceable parts. This amendment reduces the number of stage 2 HPT aft cooling plates affected by AD 99-27-16, and identifies the applicable engines by engine serial numbers (SN's). This amendment is prompted by an updated alert service bulletin (ASB) that reduces the number of stage 2 HPT aft cooling plates affected by AD 99-27-16 and identifies the applicable engines by engine SN's. The actions specified by this AD are intended to prevent stage 2

HPT aft cooling plate failure, which could result in an uncontained engine failure and damage to the airplane.

**DATES:** Effective date August 2, 2002. The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of August 2, 2002.

**ADDRESSES:** The service information referenced in this AD may be obtained from CFE Company, Data Distribution, MS 64-03/2101-201, PO Box 29003, Phoenix, AZ 85038-9003; telephone (602) 365-2493, fax (602) 365-5577. This information may be examined, by appointment, at the Federal Aviation Administration (FAA), New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**FOR FURTHER INFORMATION CONTACT:**

Keith Mead, Aerospace Engineer, Engine Certification Office, FAA, Engine and Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803-5299; telephone: (781) 238-7744; fax: (781) 238-7199.

**SUPPLEMENTARY INFORMATION:** A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) by revising AD 99-27-16, Amendment 39-11497 (65 FR 691, January 6, 2000), which is applicable to CFE Company model CFE738-1-1B turbofan engines, was published in the **Federal Register** on June 6, 2001 (66 FR 30341). That action proposed to reduce the number of stage 2 HPT aft cooling plates affected by AD 99-27-16, and identifies the applicable engines by engine SN's, in accordance with CFE Company Alert Service Bulletin (ASB) No. CFE738-A72-8031, Revision 2, dated October 17, 2000. Since the proposal was published, CFE Company has issued ASB No. CFE738-A72-8031, Revision 4, dated March 27, 2002, which includes the engine SN's that have the affected gas generator modules installed. This final rule references ASB Revision 4 instead of ASB Revision 2 which was referenced in the proposal.

**Comments**

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were received on the proposal or the FAA's determination of the cost to the public. After careful review of the available data, including the ASB reference change noted above, the FAA has determined that air safety and the public interest require the adoption of the rule with the change

described previously. The FAA has determined that this change will neither increase the economic burden on any operator nor increase the scope of the AD.

**Economic Analysis**

There are approximately ten engines of the affected design in the worldwide fleet. The FAA estimates that ten engines installed on airplanes of US registry would be affected by this AD, that it would take approximately four work hours per engine to accomplish the inspection if the inspection did not take place during scheduled maintenance, and that the average labor rate is \$60 per work hour. Required parts would cost approximately \$1,536 per engine. Based on these figures, the total cost of the AD on US operators is estimated to be \$17,760.

**Regulatory Analysis**

This final rule does not have federalism implications, as defined in Executive Order 13132, because it would not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Accordingly, the FAA has not consulted with state authorities prior to publication of this final rule.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption **ADDRESSES**.

**List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

**Adoption of the Amendment**

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

**PART 39—AIRWORTHINESS DIRECTIVES**

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

**§ 39.13 [Amended]**

2. Section 39.13 is amended by removing Amendment 39-11497 (65 FR 691, January 6, 2000) and by adding a new airworthiness directive, Amendment 39-12791, to read as follows:

**99-27-16R1 CFE Company:** Amendment 39-12791. Docket No. 99-NE-39-AD. Revises AD 99-27-16, Amendment 39-11497.

*Applicability:* This airworthiness directive (AD) is applicable to CFE Model CFE738-1-1B turbofan engines, part number (P/N) 3050000-5, with gas generator modules P/N 6091T09G01, serial numbers (SN's) 800421, 800422, 800423, 800424, 800425, 800426, 800427, 800428, 800429, and 800430 installed. These engines are installed on, but not limited to Dassault-Breguet Falcon 2000 series airplanes.

**Note 1:** This AD applies to each engine identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For engines that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (b) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

*Compliance:* Compliance with this AD is required as indicated, unless already done.

To prevent stage 2 high pressure turbine (HPT) aft cooling plate failure, which could result in an uncontained engine failure and damage to the airplane, do the following:

**Inspections and Follow-On Actions**

(a) At the next engine shop visit after the effective date of this AD where the HPT assembly is sufficiently disassembled to afford access to the stage 2 HPT aft cooling plate, but not later than 4,500 part cycles-since-new (CSN), do the following:

(1) Inspect the stage 2 HPT aft cooling plate for nicks, dents, and scratches on surface D in accordance with the requirements of CFE Alert Service Bulletin (ASB) No. CFE738-A72-8031, Revision 4, dated March 27, 2002, paragraph 2.B.(1).

(2) Repair those stage 2 HPT aft cooling plates with indentation 0.003 inch deep or less in accordance with ASB No. CFE738-A72-8031, Revision 4, dated March 27, 2002, paragraph 2.B.(1).

(3) Remove from service before further flight those stage 2 HPT aft cooling plates that have nicks, dents, and/or scratches that exceed the acceptance limits in accordance

with ASB No. CFE738-A72-8031, Revision 4, dated March 27, 2002, paragraph 2.B.(1), and replace with serviceable parts.

(4) Inspect the stage 2 HPT rotor disk post aft mating surface for raised metal, and remove raised metal if present in accordance with ASB No. CFE738-A72-8031, Revision 4, dated March 27, 2002, paragraph 2.B.(2).

**Alternative Methods of Compliance**

(b) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Engine

Certification Office (ECO). Operators must submit their request through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, ECO.

**Note 2:** Information concerning the existence of approved alternative methods of compliance with this airworthiness directive, if any, may be obtained from the ECO.

**Special Flight Permits**

(c) Special flight permits may be issued in accordance with §§ 21.197 and 21.199 of the

Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be done.

**Documents That Have Been Incorporated By Reference**

(d) The inspections and follow-on actions must be done in accordance with the following CFE Company Alert Service Bulletin (ASB):

Document No.	Pages	Revision	Date
ASB CFE738-A72-8031 .....	1	4 .....	March 27, 2002.
	2-5	Original .....	May 17, 1999.
Total pages: 5.			

This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from CFE Company, Data Distribution, MS 64-03/2101-201, PO Box 29003, Phoenix, AZ 85038-9003; telephone (602) 365-2493, fax (602) 365-5577. Copies may be inspected, by appointment, at the FAA, New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**Effective Date**

(e) This amendment becomes effective on August 2, 2002.

Issued in Burlington, Massachusetts, on June 17, 2002.

**Jay J. Pardee,**

*Manager, Engine and Propeller Directorate, Aircraft Certification Service.*

[FR Doc. 02-16176 Filed 6-27-02; 8:45 am]

**BILLING CODE 4910-13-P**

**DEPARTMENT OF TRANSPORTATION**

**Federal Aviation Administration**

**14 CFR Part 97**

[Docket No. 30316; Amdt. No. 3011]

**Standard Instrument Approach Procedures; Miscellaneous Amendments**

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment establishes, amends, suspends, or revokes Standard Instrument Approach Procedures (SIAPs) for operations at certain airports. These regulatory actions are needed because of the adoption of new or revised criteria, or because of changes occurring in the National Airspace System, such as the commissioning of

new navigational facilities, addition of new obstacles, or changes in air traffic requirements. These changes are designed to provide safe and efficient use of the navigable airspace and to promote safe flight operations under instrument flight rules at the affected airports.

**EFFECTIVE DATES:** An effective date for each SIAP is specified in the amendatory provisions.

Incorporation by reference approved by the Director of the Federal Register on December 31, 1980, and reapproved as of January 1, 1982.

**ADDRESSES:** Availability of matters incorporated by reference in the amendment is as follows:

*For Examination—*

1. FAA Rules Docket, FAA Headquarters Building, 800 Independence Avenue, SW., Washington, DC 20591;

2. The FAA Regional Office of the region in which the affected airport is located; or

3. The Flight Inspection Area Office which originated the SIAP.

*For Purchase—*Individual SIAP copies may be obtained from:

1. FAA Public Inquiry Center (APA-200), FAA Headquarters Building, 800 Independence Avenue, SW., Washington, DC 20591; or

2. The FAA Regional Office of the region in which the affected airport is located.

*By Subscription—*Copies of all SIAPs, mailed once every 2 weeks, are for sale by the Superintendent of Documents, U.S. Government Printing Office, Washington, DC 20402.

**FOR FURTHER INFORMATION CONTACT:**

Donald P. Pate, Flight Procedure Standards Branch (AMCAFS-420), Flight Technologies and Programs Division, Flight Standards Service Federal Aviation Administration, Mike

Monroney Aeronautical Center, 6500 South MacArthur Blvd. Oklahoma City, OK. 73169 (Mail Address: P.O. Box 25082 Oklahoma City, OK. 73125) telephone: (405) 954-4164.

**SUPPLEMENTARY INFORMATION:** This amendment to part 97 of the Federal Aviation Regulations (14 CFR part 97) establishes, amends, suspends, or revokes Standard Instrument Approach Procedures (SIAPs). The complete regulatory description of each SIAP is contained in official FAA form documents which are incorporated by reference in this amendment under 5 U.S.C. 552(a), 1 CFR part 51, and § 97.20 of the Federal Aviation Regulations (FAR). The applicable FAA Forms are identified as FAA Forms 8260-3, 8260-4, and 8260-5. Materials incorporated by reference are available for examination or purchase as stated above.

The large number of SIAPs, their complex nature, and the need for a special format make their verbatim publication in the **Federal Register** expensive and impractical. Further, airmen do not use the regulatory text of the SIAPs, but refer to their graphic depiction on charts printed by publishers of aeronautical materials. Thus, the advantages of incorporation by reference are realized and publication of the complete description of each SIAP contained in FAA form documents is unnecessary. The provisions of this amendment state the affected CFR (and FAR) sections, with the types and effective dates of the SIAPs. This amendment also identifies the airport, its location, the procedure identification and the amendment number.

**The Rule**

This amendment to part 97 is effective upon publication of each separate SIAP