

**§ 39.13 [Amended]**

2. Section 39.13 is amended by adding the following new airworthiness directive:

**2002-03-14 Bombardier, Inc. (Formerly Canadair):** Amendment 39-12655. Docket 2001-NM-155-AD.

*Applicability:* Model CL-600-2B19 series airplanes, serial numbers 7003 through 7999 inclusive, certificated in any category.

**Note 1:** This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (d) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

*Compliance:* Required as indicated, unless accomplished previously.

To find and fix cracking of the left and right lower wing planks, which could result in reduced structural integrity of the wing, accomplish the following:

**Repetitive Inspections**

(a) Perform an external detailed visual inspection for cracking of the left and right lower wing planks in the area of the rear spar and wing station 148.019, according to Part 2, Accomplishment Instructions, of Bombardier Alert Service Bulletin A601R-57-031, Revision "A," including Appendix A, dated March 28, 2001. Do the initial inspection at the time shown in paragraph (a)(1), (a)(2), or (a)(3) of this AD, as applicable; and repeat the inspection at least every 5,000 flight cycles.

**Note 2:** For the purposes of this AD, a detailed visual inspection is defined as: "An intensive visual examination of a specific structural area, system, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at intensity deemed appropriate by the inspector. Inspection aids such as mirror, magnifying lenses, etc., may be used. Surface cleaning and elaborate access procedures may be required."

**Compliance Times for Inspection**

(1) For airplanes that have accumulated 6,500 total flight cycles or less as of the effective date of this AD: Inspect prior to the accumulation of 7,000 total flight cycles.

(2) For airplanes that have accumulated 6,501 total flight cycles, but fewer than 13,500 total flight cycles, as of the effective date of this AD: Inspect prior to the accumulation of 13,700 total flight cycles, or within 500 flight cycles after the effective date of this AD, whichever occurs first.

(3) For airplanes that have accumulated 13,500 total flight cycles or more as of the

effective date of this AD: Inspect within 200 flight cycles after the effective date of this AD.

**Note 3:** Inspections accomplished prior to the effective date of this AD in accordance with Bombardier Alert Service Bulletin A601R-57-031, dated March 22, 2001, are considered acceptable for compliance with paragraph (a) of this AD.

**Note 4:** There is no terminating action available at this time for the repetitive inspections required by paragraph (a) of this AD.

**Repair**

(b) If any crack is found during any inspection according to paragraph (a) of this AD: Before further flight, repair per a method approved by either the Manager, New York Aircraft Certification Office (ACO), FAA; or Transport Canada Civil Aviation (or its delegated agent).

**Reporting Requirement**

(c) Submit a report of inspection findings (both positive and negative) to Bombardier Aerospace Technical Help Desk, fax (514) 855-8500, at the applicable time specified in paragraph (c)(1) or (c)(2) of this AD. Information collection requirements contained in this regulation have been approved by the Office of Management and Budget (OMB) under the provisions of the Paperwork Reduction Act of 1980 (44 U.S.C. 3501 *et seq.*) and have been assigned OMB Control Number 2120-0056.

(1) For airplanes on which the inspection is accomplished after the effective date of this AD: Submit the report within 30 days after performing the inspection required by paragraph (a) of this AD.

(2) For airplanes on which the inspection has been accomplished prior to the effective date of this AD: Submit the report within 30 days after the effective date of this AD.

**Alternative Methods of Compliance**

(d) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, New York ACO. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, New York ACO.

**Note 5:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the New York ACO.

**Special Flight Permits**

(e) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

**Incorporation by Reference**

(f) The inspections shall be done in accordance with Bombardier Alert Service Bulletin A601R-57-031, Revision "A," including Appendix A, dated March 28, 2001. This incorporation by reference was

approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Bombardier, Inc., Canadair, Aerospace Group, P.O. Box 6087, Station Centre-ville, Montreal, Quebec H3C 3G9, Canada. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, New York Aircraft Certification Office, 10 Fifth Street, Third Floor, Valley Stream, New York; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**Note 6:** The subject of this AD is addressed in Canadian airworthiness directive CF-2001-15, dated March 30, 2001.

**Effective Date**

(g) This amendment becomes effective on March 27, 2002.

Issued in Renton, Washington, on February 8, 2002.

**Vi L. Lipski,**

*Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 02-3611 Filed 2-19-02; 8:45 am]

**BILLING CODE 4910-13-U**

**DEPARTMENT OF TRANSPORTATION****Federal Aviation Administration****14 CFR Part 39**

**[Docket No. 2001-NM-143-AD; Amendment 39-12654; AD 2002-03-13]**

**RIN 2120-AA64**

**Airworthiness Directives; Short Brothers Model SD3-60, SD3-60 SHERPA, and SD3-SHERPA Series Airplanes**

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD), applicable to all Short Brothers Model SD3-60, SD3-60 SHERPA, and SD3-SHERPA series airplanes, that requires a one-time inspection of the wiring harness and power cables of the heated windshield to detect inadequate clearance, inadequate support, or chafing. This amendment also requires corrective action (including re-routing for adequate clearance or replacing damaged cables, as applicable), if necessary. The actions specified by this AD are intended to prevent chafing or damage of the power cables of the heated windshield, which could cause arcing and result in smoke and fire in the cockpit. This action is intended to address the identified unsafe condition.

**DATES:** Effective March 27, 2002.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of March 27, 2002.

**ADDRESSES:** The service information referenced in this AD may be obtained from Short Brothers, Airworthiness & Engineering Quality, P.O. Box 241, Airport Road, Belfast BT3 9DZ, Northern Ireland. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Todd Thompson, Aerospace Engineer, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington, 98055-4056; telephone (425) 227-1175; fax (425) 227-1149.

**SUPPLEMENTARY INFORMATION:** A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to all Short Brothers Model SD3-60, SD3-60 SHERPA, and SD3-SHERPA series airplanes was published in the **Federal Register** on November 23, 2001 (66 FR 58680). That action proposed to require a one-time inspection of the two power cables to the heated windshield to detect inadequate clearance, chafing, and inadequate support. The action also proposed to require corrective action, if necessary, including increasing the clearance, providing additional support, re-routing, and replacing power cables, as applicable.

#### Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were submitted in response to the proposal or the FAA's determination of the cost to the public.

#### Explanation of Changes to Final Rule

The statement of unsafe condition in the "Summary" section and subsequent sections of this AD have been changed for clarification to include a reference to the wiring harness of the power cables and to clarify certain wording.

Paragraph (a) of the proposed rule specifies a general visual inspection of the power cables to the heated windshield to detect inadequate clearance, chafing, and inadequate support. However, paragraph (a) of the proposed rule does not make clear that the inspection includes inspecting the

wiring harness and ensuring that the harness is securely supported. Therefore, for clarification, we have changed paragraph (a) of this final rule to require a general visual inspection of the wiring harness and power cables of the heated windshield to detect inadequate clearance between the cables and the flight instruments, and to ensure that the harness is securely supported and no chafing of the protective cover on the cables is evident.

In addition, we have changed paragraphs (b), (c), and (d) of the proposed rule to paragraphs (a)(1), (a)(2), and (a)(3) in this final rule, as they are subparagraphs of paragraph (a), and we have clarified certain wording in those paragraphs. We find that these changes do not expand the scope of the proposed AD but merely provide clarification of the requirements of this AD.

#### Conclusion

After careful review of the available data, the FAA has determined that air safety and the public interest require the adoption of the rule with the changes described previously. The FAA has determined that these changes will neither increase the economic burden on any operator nor increase the scope of the AD.

#### Cost Impact

The FAA estimates that 78 airplanes of U.S. registry will be affected by this AD, that it will take approximately 1 work hour per airplane to accomplish the required inspection, and that the average labor rate is \$60 per work hour. Based on these figures, the cost impact of the AD on U.S. operators is estimated to be \$4,680, or \$60 per airplane.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up, planning time, or time necessitated by other administrative actions.

#### Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and

responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

#### Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

#### PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

#### 2002-03-13 Short Brothers PLC:

Amendment 39-12654. Docket 2001-NM-143-AD.

*Applicability:* All Model SD3-60, SD3-60 SHERPA, and SD3-SHERPA series airplanes; certificated in any category.

**Note 1:** This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (b) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

*Compliance:* Required as indicated, unless accomplished previously.

To prevent chafing or damage of the power cables of the heated windshield, which could cause arcing and result in smoke and fire in the cockpit, accomplish the following:

#### Inspection and Corrective Action

(a) Within 90 days after the effective date of this AD: Perform a general visual inspection of the wiring harness and power cables of the heated windshield to detect inadequate clearance between the cables and the flight instruments, and to ensure that the harness is securely supported and no chafing of the protective cover on the cables is evident, in accordance with Short Brothers Service Bulletin SD3 SHERPA-30-2 (for Model SD3 SHERPA series airplanes), SD360 SHERPA-30-2 (for Model SD3-60 SHERPA series airplanes), or SD360-30-26 (for Model SD3-60 series airplanes); all dated April 2, 2001; as applicable. If no evidence of chafing is found and clearance and support of the power cables is adequate, no further action is needed.

**Note 2:** For the purposes of this AD, a general visual inspection is defined as: "A visual examination of an interior or exterior area, installation, or assembly to detect obvious damage, failure, or irregularity. This level of inspection is made under normally available lighting conditions such as daylight, hangar lighting, flashlight, or drop-light, and may require removal or opening of access panels or doors. Stands, ladders, or platforms may be required to gain proximity to the area being checked."

(1) If no evidence of chafing is found, but clearance or support of the harness is inadequate: Prior to further flight, re-route the power cable for adequate clearance or provide additional support of the cable harness, as applicable, in accordance with Short Brothers Service Bulletin SD3 SHERPA-30-2 (for Model SD3 SHERPA series airplanes), SD360 SHERPA-30-2 (for Model SD3-60 SHERPA series airplanes), or SD360-30-26 (for Model SD3-60 series airplanes); all dated April 2, 2001; as applicable.

(2) If evidence of chafing is found, but there is no damage to the outer nylon protective cover resulting in exposure of the glass fiber braid: Prior to further flight, re-route the power cables for adequate clearance, in accordance with Short Brothers Service Bulletin SD3 SHERPA-30-2 (for Model SD3 SHERPA series airplanes), SD360 SHERPA-30-2 (for Model SD3-60 SHERPA series airplanes), or SD360-30-26 (for Model SD3-60 series airplanes); all dated April 2, 2001; as applicable.

(3) If evidence of chafing is found, and there is damage to the outer nylon protective cover resulting in exposure of the glass fiber braid: Prior to further flight, replace the damaged power cable with a new cable, in accordance with Short Brothers Service Bulletin SD3 SHERPA-30-2 (for Model SD3 SHERPA series airplanes), SD360 SHERPA-30-2 (for Model SD3-60 SHERPA series airplanes), or SD360-30-26 (for Model SD3-60 series airplanes); all dated April 2, 2001; as applicable.

#### Alternative Methods of Compliance

(b) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM-116.

**Note 3:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM-116.

#### Special Flight Permits

(c) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

#### Incorporation by Reference

(d) The actions shall be done in accordance with Short Brothers Service Bulletin SD3 SHERPA-30-2, dated April 2, 2001; Short Brothers Service Bulletin SD360 SHERPA-30-2, dated April 2, 2001; and Short Brothers Service Bulletin SD360-30-26, dated April 2, 2001; as applicable. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Short Brothers, Airworthiness & Engineering Quality, P.O. Box 241, Airport Road, Belfast BT3 9DZ, Northern Ireland. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**Note 4:** The subject of this AD is addressed in British airworthiness directive 001-04-2001.

#### Effective Date

(e) This amendment becomes effective on March 27, 2002.

Issued in Renton, Washington, on February 7, 2002.

**Vi L. Lipski,**

*Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 02-3587 Filed 2-19-02; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 2001-NM-185-AD; Amendment 39-12656; AD 2002-03-15]

RIN 2120-AA64

#### Airworthiness Directives; Dornier Model 328-100 and -300 Series Airplanes

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD), applicable to certain Dornier Model 328-100 and -300 series airplanes, that requires testing of the left- and right-hand potentiometer levers of the aileron flight control system, and follow-on or corrective action, as applicable. This amendment is necessary to prevent detachment of an aileron potentiometer lever, which could result in jamming of the elevator and/or aileron flight control systems and consequent reduced controllability of the airplane. This action is intended to address the identified unsafe condition.

**DATES:** Effective March 27, 2002.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of March 27, 2002.

**ADDRESSES:** The service information referenced in this AD may be obtained from **Fairchild Dornier, Dornier Luftfahrt GmbH**, P.O. Box 1103, D-82230 Wessling, Germany. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Tom Groves, Aerospace Engineer, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-1503; fax (425) 227-1149.

**SUPPLEMENTARY INFORMATION:** A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to certain Dornier Model 328-100 and -300 series airplanes was published in the **Federal Register** on September 25, 2001 (66 FR 48989). That action proposed to require testing of the left- and right-hand potentiometer levers of the aileron flight