

List of Subjects in 14 CFR Part 25

Aircraft, Aviation safety, Reporting and recordkeeping requirements.

The authority citation for these special conditions is as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701, 44702, 44704.

The Special Conditions

Accordingly, pursuant to the authority delegated to me by the Administrator, the following special conditions are issued as part of the supplemental type certification basis for the Boeing Model 727-100/-200 series airplanes as modified by Aircraft Systems & Manufacturing.

1. *Protection from Unwanted Effects of High-Intensity Radiated Fields (HIRF)*. Each electrical and electronic system that performs critical functions must be designed and installed to ensure that the operation and operational capability of these systems to perform critical functions are not adversely affected when the airplane is exposed to high-intensity radiated fields.

2. For the purpose of these special conditions, the following definition applies: *Critical Functions*: Functions whose failure would contribute to or cause a failure condition that would prevent the continued safe flight and landing of the airplane.

Issued in Renton, Washington, on October 19, 2001.

Ali Bahrami,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 01-27160 Filed 10-26-01; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION**Federal Aviation Administration****14 CFR Part 39**

[Docket No. 2001-NM-300-AD; Amendment 39-12481; AD 2001-22-02]

RIN 2120-AA64

Airworthiness Directives; Airbus Model A300 B2 and B4 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule; request for comments.

SUMMARY: This amendment adopts a new airworthiness directive (AD) that is applicable to certain Airbus Model A300 B2 and B4 series airplanes. This action requires determining the part and amendment numbers of the variable

lever arm (VLA) of the rudder control system to verify the parts were installed using the correct standard, and corrective actions, if necessary. This action is necessary to prevent failure of both spring boxes of the VLA due to corrosion damage, which could result in loss of rudder control and consequent reduced controllability of the airplane. This action is intended to address the identified unsafe condition.

DATES: Effective November 13, 2001.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of November 13, 2001.

Comments for inclusion in the Rules Docket must be received on or before November 28, 2001.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 2001-NM-300-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056. Comments may be inspected at this location between 9 a.m. and 3 p.m., Monday through Friday, except Federal holidays. Comments may be submitted via fax to (425) 227-1232. Comments may also be sent via the Internet using the following address: 9-anm-iarcomment@faa.gov. Comments sent via the Internet must contain "Docket No. 2001-NM-300-AD" in the subject line and need not be submitted in triplicate. Comments sent via fax or the Internet as attached electronic files must be formatted in Microsoft Word 97 for Windows or ASCII text.

The service information referenced in this AD may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Dan Rodina, Aerospace Engineer, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-2125; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION: The Direction Générale de l'Aviation Civile (DGAC), which is the airworthiness authority for France, recently notified the FAA that an unsafe condition may exist on certain Airbus Model A300 B2 and B4 series airplanes. The DGAC advises that two reports were received which indicated that, during regularly

scheduled maintenance, damage to the variable lever arm (VLA) of the rudder control system was found. Further investigation revealed that the VLA spring box mountings, the mounting trunnion, and a tie rod also were damaged. Such damage was attributed to corrosion of the spring boxes. Both affected spring boxes were installed per the pre-vendor service bulletin (VSB) 27-21-1H standard, causing stiff operation of the springs and subsequent damage. Failure of one spring box of the VLA does not affect safety of flight, but failure of both spring boxes could result in loss of rudder control and consequent reduced controllability of the airplane.

Explanation of Relevant Service Information

Airbus has issued All Operators Telex (AOT) A300-27A0196, dated September 20, 2001, which describes procedures for determining the part and amendment numbers of the variable lever arm (VLA) of the rudder control system to verify the spring boxes were installed using the post-VSB 27-21-1H standard, and corrective actions, if necessary. The corrective actions include a detailed visual inspection of the VLA tie rod for damage (bent or ruptured rod) if the part and amendment numbers of the VLA are incorrect, and replacement of any damaged tie rod with a new tie rod.

The DGAC classified this AOT as mandatory and issued French telegraphic airworthiness directive T 2001-447(B), dated September 24, 2001, in order to assure the continued airworthiness of these airplanes in France.

FAA's Conclusions

These airplane models are manufactured in France and are type certificated for operation in the United States under the provisions of section 21.29 of the Federal Aviation Regulations (14 CFR 21.29) and the applicable bilateral airworthiness agreement. Pursuant to this bilateral airworthiness agreement, the DGAC has kept the FAA informed of the situation described above. The FAA has examined the findings of the DGAC, reviewed all available information, and determined that AD action is necessary for products of this type design that are certificated for operation in the United States.

Explanation of Requirements of Rule

Since an unsafe condition has been identified that is likely to exist or develop on other airplanes of the same type design registered in the United States, this AD is being issued to

prevent failure of both spring boxes of the variable lever arm (VLA) due to corrosion damage, which could result in jammed rudder pedals, loss of rudder control, and consequent reduced controllability of the airplane. This AD requires a one-time inspection to determine the part number of the VLA of the rudder control system, and follow-on actions, if necessary. The actions are required to be accomplished in accordance with the AOT described previously.

Interim Action

This is considered to be interim action. The manufacturer is gathering data that will enable it to obtain better insight into the nature, cause, and extent of the corrosion damage, and eventually to develop final action to address the unsafe condition. Once final action has been identified, the FAA may consider further rulemaking.

Determination of Rule's Effective Date

Since a situation exists that requires the immediate adoption of this regulation, it is found that notice and opportunity for prior public comment hereon are impracticable, and that good cause exists for making this amendment effective in less than 30 days.

Comments Invited

Although this action is in the form of a final rule that involves requirements affecting flight safety and, thus, was not preceded by notice and an opportunity for public comment, comments are invited on this rule. Interested persons are invited to comment on this rule by submitting such written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified under the caption **ADDRESSES**. All communications received on or before the closing date for comments will be considered, and this rule may be amended in light of the comments received. Factual information that supports the commenter's ideas and suggestions is extremely helpful in evaluating the effectiveness of the AD action and determining whether additional rulemaking action would be needed.

Submit comments using the following format:

- Organize comments issue-by-issue. For example, discuss a request to change the compliance time and a request to change the service bulletin reference as two separate issues.
- For each issue, state what specific change to the AD is being requested.

- Include justification (*e.g.*, reasons or data) for each request.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the rule that might suggest a need to modify the rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report that summarizes each FAA-public contact concerned with the substance of this AD will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this rule must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket Number 2001-NM-300-AD." The postcard will be date stamped and returned to the commenter.

Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

The FAA has determined that this regulation is an emergency regulation that must be issued immediately to correct an unsafe condition in aircraft, and that it is not a "significant regulatory action" under Executive Order 12866. It has been determined further that this action involves an emergency regulation under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979). If it is determined that this emergency regulation otherwise would be significant under DOT Regulatory Policies and Procedures, a final regulatory evaluation will be prepared and placed in the Rules Docket. A copy of it, if filed, may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

2001-22-02 Airbus Industrie: Amendment 39-12481. Docket 2001-NM-300-AD.

Applicability: Model A300 B2 and B4 series airplanes, certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (b) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent failure of both spring boxes of the variable lever arm (VLA) due to corrosion damage, which could result in loss of rudder control and consequent reduced controllability of the airplane, accomplish the following:

Inspection/Corrective Actions

(a) Within 10 days after the effective date of this AD: Determine the part and amendment numbers of the VLA of the rudder control system to verify the parts were installed using the correct standard, per Airbus All Operators Telex (AOT) A300-27A0196, dated September 20, 2001.

(1) If the part and amendment numbers shown are not correct, as specified in the AOT, before further flight, do a detailed visual inspection of the VLA tie rod for damage (bent or ruptured rod) per the AOT.

(i) If the tie rod is damaged, replace the VLA with a new VLA per the AOT. Such replacement ends the requirements of this AD.

(ii) If the tie rod is not damaged, no further action is required by this AD.

(2) If the part and amendment numbers shown are correct, no further action is required by this AD.

Note 2: For the purposes of this AD, a detailed visual inspection is defined as: "An intensive visual examination of a specific structural area, system, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at intensity deemed appropriate by the inspector. Inspection aids such as mirror, magnifying lenses, etc., may be used. Surface cleaning and elaborate access procedures may be required."

Alternative Methods of Compliance

(b) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM-116.

Note 3: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM-116.

Special Flight Permits

(c) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Incorporation by Reference

(d) The actions shall be done in accordance with Airbus All Operators Telex A300-27A0196, dated September 20, 2001. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Note 4: The subject of this AD is addressed in French telegraphic airworthiness directive T 2001-447(B), dated September 24, 2001.

Effective Date

(e) This amendment becomes effective on November 13, 2001.

Issued in Renton, Washington, on October 18, 2001.

Ali Bahrami,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 01-26860 Filed 10-26-01; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2001-SW-28-AD; Amendment 39-12479; AD 2001-22-01]

RIN 2120-AA64

Airworthiness Directives; Enstrom Helicopter Corporation Model F-28, F-28A, F-28C, F-28F, 280, 280C, 280F and 280FX Helicopters

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule; request for comments.

SUMMARY: This amendment supersedes an existing airworthiness directive (AD) for Enstrom Helicopter Corporation (EHC) Model F-28, F-28A, and 280 helicopters. That AD currently requires inspecting the main rotor shaft (shaft) for a crack or other evidence of damage until appropriately modifying or replacing the shaft with an airworthy shaft at specified time intervals. This amendment adds EHC Model F-28C, F-28F, 280C, 280F, and 280FX helicopters and establishes life limits after which all unmodified shafts must be retired. This amendment requires determining the radius of the shaft fillet, certain visual and dye-penetrant inspections before further flight, and replacing certain main rotor transmissions. This amendment is prompted by the failure of a shaft on an EHC Model F-28A helicopter due to a fatigue crack. The actions specified by this AD are intended to prevent shaft failure and subsequent loss of control of the helicopter.

DATES: Effective November 13, 2001.

Comments for inclusion in the Rules Docket must be received on or before December 28, 2001.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Office of the Regional Counsel, Southwest Region, Attention: Rules Docket No. 2001-SW-28-AD, 2601 Meacham Blvd., Room 663, Fort Worth, Texas 76137. You may also send comments electronically to the Rules Docket at the following address: 9-asw-adcomments@faa.gov.

FOR FURTHER INFORMATION CONTACT: Joseph McGarvey, Fatigue Specialist, FAA, Chicago Aircraft Certification Office, Airframe and Administrative Branch, 2300 East Devon Ave., Des Plaines, Illinois 60018, telephone (847) 294-7136, fax (847) 294-7834.

SUPPLEMENTARY INFORMATION: On August 16, 1976, the FAA issued AD 76-17-08, Amendment 39-2700 (41 FR 36015, August 26, 1976). On September 16, 1976, the FAA revised that AD issuing AD 76-17-08 R1, Amendment 39-3043 (42 FR 51563, September 29, 1977), for EHC Model F-28, F-28A, and 280 helicopters to establish service time limits after which all unmodified shafts must be replaced. That AD was prompted by FAA's determination, after a review of the service experience, that shaft crack sites may be introduced by allowing the shafts to remain in service for extended periods without modification. That condition, if not corrected, could result in shaft failure

and subsequent loss of control of the helicopter.

Since the issuance of that AD, EHC has issued Service Directive Bulletin No. 0094, Revision 1, dated May 31, 2001, specifying certain inspections for a crack in certain shafts due to failure of a shaft on an EHC Model F-28A helicopter.

The FAA and the National Transportation Safety Board (NTSB) investigated the accident involving the failure of a shaft on the EHC Model F28A helicopter. The shaft was designed with a small upper fillet radius of 0.13 inch and failed due to a fatigue crack. Such a shaft design causes a high stress concentration. That, coupled with the occurrence of more frequent than anticipated high flight load conditions, can accelerate the development of fatigue cracks. Preliminary investigation revealed that the shaft installed in the transmission, P/N 28-13101-1-R, failed because of a fatigue crack in the fillet area of the shaft directly beneath the main rotor hub. The FAA concluded, based on its investigation and after reviewing NTSB Report 01-052, dated April 13, 2001, that an inspection should be made of such shafts for cracks before further flight. The FAA determined that since inspections for cracks are imprecise and detection of all existing cracks is uncertain, a shaft with a small radius fillet should be replaced with an airworthy shaft with a large radius fillet on certain model helicopters within 300 hours time-in-service (TIS).

We have identified an unsafe condition that is likely to exist or develop on other helicopters of the same type designs. This AD supersedes AD 76-17-08 and 76-17-08 R1 for EHC Model F-28, F-28A, and 280 helicopters to add Model F-28C, F-28F, 280C, 280F, and 280FX helicopters, to require the following:

- Before further flight, determine the transmission P/N and the radius of the shaft fillet.
- For certain models, replace any transmission having a shaft with a small radius fillet with an airworthy transmission before further flight.
- For certain other models, replace the transmission having a small radius shaft fillet that is not P/N 28-13101-1 or -1-R with an airworthy transmission before further flight.
- For certain models with transmission, P/N 28-13101-1 or -1-R, having a small radius shaft fillet installed:
 - Before further flight and at recurring intervals, visually inspect the shaft for a crack. If a crack is suspected,