

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up, planning time, or time necessitated by other administrative actions.

Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption ADDRESSES.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

2001-15-16 Airbus Industrie: Amendment 39-12350. Docket 2000-NM-421-AD.

Applicability: Model A319, A320, and A321 series airplanes, up to and including manufacturer's serial number (MSN) 1261, manufactured in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (b) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent blockage of the outer door handle flap in an intermediate pushed-in position, which may prevent a passenger door from opening from the inside of the airplane, thereby delaying an emergency evacuation, accomplish the following:

Inspection and Corrective Action

(a) Within 500 flight hours after the effective date of this AD, perform a one-time general visual inspection of the outer handle flap mechanisms of the passenger doors for the presence of corrosion inhibitor and for correct operation; remove any corrosion inhibitor, grease the doors, and check that the flap comes back correctly, flush with the door skin, when the handle is in the closed position; in accordance with Airbus All Operators Telex (AOT) A320-52A1106, dated September 28, 2000.

Note 2: For the purposes of this AD, a general visual inspection is defined as: "A visual examination of an interior or exterior area, installation, or assembly to detect obvious damage, failure, or irregularity. This level of inspection is made under normally available lighting conditions such as daylight, hangar lighting, flashlight, or drop-light, and may require removal or opening of access panels or doors. Stands, ladders, or platforms may be required to gain proximity to the area being checked."

Alternative Methods of Compliance

(b) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM-116.

Note 3: Information concerning the existence of other approved alternative

methods of compliance with this AD, if any, may be obtained from the International Branch, ANM-116.

Special Flight Permits

(c) Special flight permits may be issued in accordance with §§ 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Incorporation by Reference

(d) The actions shall be done in accordance with Airbus All Operators Telex A320-52A1106, dated September 28, 2000. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Note 4: The subject of this AD is addressed in French airworthiness directive 2000-519-158(B), dated December 13, 2000.

Effective Date

(e) This amendment becomes effective on September 4, 2001.

Issued in Renton, Washington, on July 19, 2001.

Donald L. Riggan,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 01-18470 Filed 7-30-01; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2000-NM-271-AD; Amendment 39-12349; AD 2001-15-15]

RIN 2120-AA64

Airworthiness Directives; Boeing Model 747 Series Airplanes Powered By Pratt & Whitney JT9D-7 Series Engines

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment supersedes an existing airworthiness directive (AD), applicable to certain Boeing Model 747 series airplanes powered by Pratt & Whitney JT9D-7 series engines, that currently requires detailed visual inspections of the lugs on the bulkhead fitting of the rear engine mount, and corrective action, if necessary. The existing AD also specifies optional

ultrasonic inspections, which, if accomplished, extend the repetitive interval for the required detailed visual inspections. This amendment requires accomplishment of the previously optional ultrasonic inspections and, for certain airplanes, rework of the bulkhead fitting of the rear engine mount. The actions specified by this AD are intended to detect and correct bushing migration, corrosion, or cracking of the lugs on the bulkhead fitting of the rear engine mount, which could result in fracture of the lugs and separation of the engine from the airplane. This action is intended to address the identified unsafe condition.

DATES: Effective September 4, 2001.

The incorporation by reference of Boeing Service Bulletin 747-54A2200, Revision 1, dated February 15, 2001, as listed in the regulations, is approved by the Director of the Federal Register as of September 4, 2001.

The incorporation by reference of Boeing Alert Service Bulletin 747-54A2200, dated July 7, 2000, as listed in the regulations, was approved previously by the Director of the Federal Register as of September 18, 2000 (65 FR 53161, September 1, 2000).

ADDRESSES: The service information referenced in this AD may be obtained from Boeing Commercial Airplane Group, P.O. Box 3707, Seattle, Washington 98124-2207. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Tamara L. Anderson, Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-2771; fax (425) 227-1181.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) by superseding AD 2000-18-01, amendment 39-11886 (65 FR 53161, September 1, 2000), which is applicable to certain Boeing Model 747 series airplanes powered by Pratt & Whitney JT9D-7 series engines, was published in the **Federal Register** on March 21, 2001 (66 FR 15814). The action proposed to continue to require detailed visual inspections of the lugs on the bulkhead fitting of the rear engine mount, and corrective action, if necessary. The action also proposed to require ultrasonic inspections (which were provided as an option in the existing

AD) and, for certain airplanes, rework of the bulkhead fitting of the rear engine mount.

Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due consideration has been given to the comments received.

Clarify Unsafe Condition

One commenter requests that the FAA clarify the unsafe condition as stated in the **SUMMARY** and "Discussion" sections of the proposed AD. The commenter requests that the unsafe condition state that bushing migration, corrosion, or cracking of the lugs on the bulkhead fitting of the rear engine mount could result in fracture of the lugs, which could result in separation of the engine from the airplane.

The FAA concurs with the commenter's request and has made this change in the **SUMMARY** section of this final rule. The "Discussion" section of the proposed AD is not restated in this final rule, so no change to that section is needed.

Revise Cost Impact

One commenter requests that the FAA revise the cost impact information in the proposed AD to include the time needed for gaining access and closing up for the proposed ultrasonic inspection. The commenter notes that, due to the 9-month compliance time, it may be necessary for operators to do this inspection at a time other than a normal scheduled heavy maintenance visit.

The FAA concurs with the commenter's request. We note that the cost analysis in AD rulemaking actions typically does not include incidental costs, such as the time required to gain access and close up, planning time, or time necessitated by other administrative actions. Because incidental costs may vary significantly from operator to operator, they are almost impossible to calculate. However, we acknowledge that it may or may not be possible to accomplish the ultrasonic inspection required by this AD during a normal scheduled maintenance visit due to the compliance times for the initial and repetitive inspections. Therefore, we have revised the cost impact information for the ultrasonic inspections in this final rule from 4 to 36 work hours to include the work hours necessary for gaining access and closing up.

Refer to Specific Part of Referenced Service Bulletin

One commenter requests that the FAA revise paragraph (d) of the proposed AD to refer to Part 2 of Boeing Alert Service Bulletin 747-54A2200, dated July 7, 2000; or Revision 1, dated February 15, 2001. The commenter does not state a reason for its request.

The FAA infers that the commenter's request is to make paragraph (d) consistent with other paragraphs of the AD. The FAA concurs and has revised paragraph (d) of this final rule accordingly. Also, the same change has been made to paragraph (a) of this AD.

Give Credit for Inspections Accomplished Previously

One commenter requests that the FAA revise paragraph (c) of the proposed AD to provide a third compliance time option for airplanes inspected per the ultrasonic method provided as an option in AD 2000-18-01. The commenter states that operators who did the ultrasonic inspection per AD 2000-18-01 would be required to repeat this inspection within 9 months after the effective date of this AD. The commenter emphasizes that such a requirement would impose undue economic and scheduling burdens on affected operators.

The FAA does not concur. Credit for inspections accomplished prior to the effective date of the AD is always provided in an AD by means of the statement at the beginning of the "Compliance" section of each AD: "Required as indicated, unless accomplished previously." No change to the final rule is necessary in this regard.

Remove Inspections From Paragraph (f)

One commenter requests that the FAA revise paragraph (f) to remove the requirement to perform detailed visual and non-destructive test inspections for damage of the upper engine mount during accomplishment of the rework of the lugs on the bulkhead fitting of the rear engine mount. The commenter states that these inspections should be necessary only if there is insufficient clearance between the migrated end of the outer lug plain bushing and the adjacent lug of the aft upper engine mount.

The FAA does not concur. We infer that the commenter assumes that there will be no damage to the upper engine mount if sufficient clearance is maintained between the migrated end of the outer lug plain bushing and the adjacent lug of the aft upper engine mount. However, we have determined that the bushing may migrate in either

direction. For example, the bushing may have migrated to a position of no positive clearance and caused damage, but then subsequently may have migrated inward to a position where there is sufficient clearance. Thus, we find it necessary to require the inspections during the rework according to the service bulletin. No change to the final rule is necessary in this regard.

Reference Alternative Method of Compliance for AD 2000-18-01

One commenter requests that the FAA revise the proposed AD to reference a specific alternative method of compliance (AMOC) that was approved previously for AD 2000-18-01. The commenter states that the AMOC addresses conditions of no positive clearance, which may be found during the rework according to Part 4 of the service bulletin.

The FAA does not concur with the request. Paragraph (h)(2) of the proposed AD allows the use of previously approved AMOCs for AD 2000-18-01 for compliance with corresponding actions in the proposed AD. Listing references for specific AMOCs would unnecessarily complicate this final rule. No change to the final rule is necessary in this regard.

Make Specific Tooling Optional

One commenter requests that the FAA revise the proposed AD to make the use of specific tooling identified in the service bulletin optional for compliance with the proposed AD. The commenter refers to a specific boring fixture called out in the service bulletin, and states that use of this specific tooling should be optional. The commenter states that other tooling capable of producing the desired dimensions and finishes specified in the service bulletin should be acceptable for compliance. The commenter notes that inspection requirements and dimensional checks contained in the service bulletin are sufficient to ensure that lugs are properly reworked and free of damage.

The FAA does not concur with the commenter's request. We find that the tooling used to bore the lugs may affect the unsafe condition addressed by this AD and, therefore, it is inappropriate not to specify the tooling to be used. However, operators may request approval of an AMOC under paragraph (h)(1) of this AD if they can show that tooling other than that identified in the service bulletin will provide an acceptable level of safety. No change to the final rule is necessary in this regard.

Conclusion

After careful review of the available data, including the comments noted above, the FAA has determined that air safety and the public interest require the adoption of the rule with the changes previously described. The FAA has determined that these changes will neither increase the economic burden on any operator nor increase the scope of the AD.

Cost Impact

There are approximately 200 Model 747 series airplanes of the affected design in the worldwide fleet. The FAA estimates that 47 airplanes of U.S. registry will be affected by this AD.

The detailed visual inspections that are currently required by AD 2000-18-01 take approximately 8 work hours per airplane to accomplish, at an average labor rate of \$60 per work hour. Based on these figures, the cost impact of the currently required actions on U.S. operators is estimated to be \$22,560, or \$480 per airplane, per inspection cycle.

The new inspections required by this AD will take approximately 36 work hours per airplane to accomplish (including time for gaining access and closing up), at an average labor rate of \$60 per work hour. Based on these figures, the cost impact of the new requirements of this AD on U.S. operators is estimated to be \$101,520, or \$2,160 per airplane, per inspection cycle.

The cost impact figures discussed above are based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up, planning time, or time necessitated by other administrative actions. However, for the new inspections required by this AD, the time for gaining access and closing up has been included in the figures above because it may not be possible for operators to accomplish these inspections during normal scheduled maintenance due to the compliance times associated with these inspections.

Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between

the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by removing amendment 39-11886 (65 FR 53161, September 1, 2000), and by adding a new airworthiness directive (AD), amendment 39-12349, to read as follows:

2001-15-15 Boeing: Amendment 39-12349.

Docket 2000-NM-271-AD. Supersedes AD 2000-18-01, Amendment 39-11886.

Applicability: Model 747 series airplanes powered by Pratt & Whitney JT9D-7 series engines, as listed in Boeing Alert Service Bulletin 747-54A2200, dated July 7, 2000; certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in

accordance with paragraph (h)(1) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To detect and correct bushing migration, corrosion, or cracking of the lugs on the bulkhead fitting of the rear engine mount, accomplish the following:

Restatement of Requirements of AD 2000-18-01

Repetitive Detailed Visual Inspections

(a) At the later of the times in paragraphs (a)(1) and (a)(2) of this AD, perform a detailed visual inspection for bushing migration, corrosion, or cracking; and a physical measurement inspection using feeler gages for bushing migration; of the lugs on the bulkhead fitting of the rear engine mount, in accordance with Part 2 of Boeing Alert Service Bulletin 747-54A2200, dated July 7, 2000; or Revision 1, dated February 15, 2001. Thereafter, repeat the inspection at intervals not to exceed 90 days, until the inspections required by paragraphs (c) and (d) of this AD have been accomplished.

(1) Prior to the accumulation of 10,000 total flight cycles, or within 15 years since the date of manufacture of the airplane, whichever occurs first.

(2) Within 90 days after September 18, 2000 (the effective date of AD 2000-18-01, amendment 39-11886).

Note 2: For the purposes of this AD, a detailed visual inspection is defined as: "An intensive visual examination of a specific structural area, system, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at intensity deemed appropriate by the inspector. Inspection aids such as mirror, magnifying lenses, etc., may be used. Surface cleaning and elaborate access procedures may be required."

Corrective Actions

(b) During any inspection accomplished in accordance with paragraph (a), (c), or (d) of this AD; if bushing migration, corrosion, or cracking is detected, accomplish paragraph (b)(1) or (b)(2) of this AD, as applicable.

(1) If light corrosion or bushing migration is found: Prior to further flight, do interim rework in accordance with Part 4 of Boeing Alert Service Bulletin 747-54A2200, dated July 7, 2000; or Revision 1, dated February 15, 2001; EXCEPT where the service bulletin specifies to contact Boeing, prior to further flight, repair in accordance with a method approved by the Manager, Seattle Aircraft Certification Office (ACO), FAA; or in accordance with data meeting the type certification basis of the airplane approved by a Boeing Company Designated Engineering Representative (DER) who has been authorized by the Manager, Seattle ACO, to make such findings. For a repair method to be approved by the Manager, Seattle ACO, as required by this paragraph,

the approval letter must specifically reference this AD.

(2) If moderate to severe corrosion or any cracking is found: Prior to further flight, rework the lugs on the bulkhead fitting of the rear engine mount in accordance with Part 5 of Boeing Service Bulletin 747-54A2200, Revision 1, dated February 15, 2001, except as provided by paragraph (g) of this AD; or in accordance with a method approved by the Manager, Seattle ACO; or in accordance with data meeting the type certification basis of the airplane approved by a Boeing Company DER who has been authorized by the Manager, Seattle ACO, to make such findings. For a repair method to be approved by the Manager, Seattle ACO, as required by this paragraph, the approval letter must specifically reference this AD. Such rework resets the compliance threshold for the inspections per paragraphs (c) and (d) of this AD to 15 years or 10,000 flight cycles since rework, whichever is earlier.

New Requirements of This AD

Ultrasonic Inspection—Initial and Repetitive Inspections

(c) At the later of the times in paragraphs (c)(1) and (c)(2) of this AD, except as provided by paragraph (f) of this AD, perform an ultrasonic inspection to detect corrosion or cracking of the lugs on the bulkhead fitting of the rear engine mount, per Part 3 of Boeing Alert Service Bulletin 747-54A2200, dated July 7, 2000; or Revision 1, dated February 15, 2001. Thereafter, repeat the ultrasonic inspection described in this paragraph at intervals not to exceed 1,400 flight cycles or 18 months, whichever occurs first.

(1) Prior to the accumulation of 10,000 total flight cycles, or within 15 years since the date of manufacture of the airplane, whichever occurs first.

(2) Within 9 months after the effective date of this AD.

Repetitive Detailed Visual and Physical Measurement Inspections

(d) After initial accomplishment of the inspections required by paragraph (c) of this AD, perform repetitive detailed visual inspections for bushing migration, corrosion, or cracking; and physical measurement inspections using feeler gages for bushing migration; of the lugs on the bulkhead fitting of the rear engine mount; per Part 2 of Boeing Alert Service Bulletin 747-54A2200, dated July 7, 2000; or Revision 1, dated February 15, 2001. Perform the inspections at the interval stated in paragraph (d)(1) or (d)(2) of this AD, except as provided by paragraph (f) of this AD. Accomplishment of repetitive inspections per this paragraph constitutes terminating action for the inspections required by paragraph (a) of this AD.

(1) If no bushing migration is found during any inspection per this AD, the repetitive interval is not to exceed 1,400 flight cycles or 18 months, whichever occurs first.

(2) If any bushing migration is found during any inspection per this AD, the repetitive interval is not to exceed 180 days, until paragraph (e) of this AD has been done.

On-Condition Rework

(e) If any bushing migration is found during any inspection per this AD, within 30

months after finding the migrated bushing, or within 18 months after the effective date of this AD, whichever occurs later, do rework of the lugs on the bulkhead fitting of the rear engine mount (including a detailed visual inspection of the aft upper engine mount for damage; a Non-Destructive Testing inspection and repair of the aft upper engine mount, as applicable; and rework of the lugs, on the bulkhead fitting of the rear engine mount) per Part 5 of Boeing Alert Service Bulletin 747-54A2200, Revision 1, dated February 15, 2001. Such rework resets the compliance threshold for the inspections per paragraphs (c) and (d) of this AD to 15 years or 10,000 flight cycles since rework, whichever is earlier.

Optional Rework

(f) Rework of the lugs on the bulkhead fitting of the rear engine mount (including a detailed visual inspection of the aft upper engine mount for damage; a Non-Destructive Testing inspection and repair of the aft upper engine mount, as applicable; and rework of the lugs, and installation of new bushings in the lug, on the bulkhead fitting of the rear engine mount) per Part 5 of Boeing Alert Service Bulletin 747-54A2200, Revision 1, dated February 15, 2001, resets the compliance threshold for the inspections per paragraphs (c) and (d) of this AD to 15 years or 10,000 flight cycles since rework, whichever is earlier.

Exception to Repair Requirement

(g) Where Boeing Alert Service Bulletin 747-54A2200, dated July 7, 2000; or Revision 1, dated February 15, 2001; says to contact Boeing for repair instructions: Before further flight, repair per a method approved by the Manager, Seattle ACO, or per data meeting the type certification basis of the airplane approved by a Boeing Company DER who has been authorized by the Manager, Seattle ACO, to make such findings. For a repair method to be approved by the Manager, Seattle ACO, as required by this paragraph, the approval letter must specifically reference this AD.

Alternative Methods of Compliance

(h)(1) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Seattle ACO. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Seattle ACO.

(2) Alternative methods of compliance, approved previously in accordance with AD 2000-18-01, amendment 39-11886, are approved as alternative methods of compliance for corresponding actions in this AD.

Note 3: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Seattle ACO.

Special Flight Permits

(i) Special flight permits may be issued in accordance with §§ 21.197 and 21.199 of the

Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Incorporation by Reference

(j) Except as provided by paragraphs (b) and (g) of this AD, the actions shall be done in accordance with Boeing Alert Service Bulletin 747-54A2200, dated July 7, 2000; or Boeing Service Bulletin 747-54A2200, Revision 1, dated February 15, 2001; as applicable.

(1) The incorporation by reference of Boeing Service Bulletin 747-54A2200, Revision 1, dated February 15, 2001, is approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51.

(2) The incorporation by reference of Boeing Alert Service Bulletin 747-54A2200, dated July 7, 2000, was approved previously by the Director of the Federal Register as of September 18, 2000 (65 FR 53161, September 1, 2000).

(3) Copies of these service bulletins may be obtained from Boeing Commercial Airplane Group, P.O. Box 3707, Seattle, Washington 98124-2207. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Effective Date

(k) This amendment becomes effective on September 4, 2001.

Issued in Renton, Washington, on July 19, 2001.

Vi L. Lipski,

*Manager, Transport Airplane Directorate,
Airframe Certification Service.*

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BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 99-NM-234-AD; Amendment 39-12347; AD 2001-15-13]

RIN 2120-AA64

Airworthiness Directives; Airbus Model A310 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to all Airbus Model A310 series airplanes, that requires repetitive inspections of the metallic vapor seals in the center fuel tank to detect holes, tears, or a change in shape; corrective action, if such damage is detected; and follow-up tests for leaks. This

amendment is prompted by reports of damaged metallic vapor seals observed during routine maintenance. This action is necessary to detect and correct damage to the metallic vapor seal in the center fuel tank, which could lead to leakage of fuel from the center tank into the air conditioning pack bay located below the center tank, providing a potential for fuel to be in contact with fuel ignition sources. This action is intended to address the identified unsafe condition.

DATES: Effective September 4, 2001.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of September 4, 2001.

ADDRESSES: The service information referenced in this AD may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Dan Rodina, Aerospace Engineer, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Ave. SW., Renton, Washington 98055-4056; telephone (425) 227-2125; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to all Airbus Model A310 series airplanes was published in the **Federal Register** on March 29, 2001 (66 FR 17127). That action proposed to require repetitive inspections of the metallic vapor seals in the center fuel tank to detect holes, tears, or a change in shape; corrective action, if such damage is detected; and follow-up tests for leaks.

Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due consideration has been given to the comments received from a single commenter.

Add Terminating Action

The commenter requests that the FAA revise the proposed rule to include a terminating action. The commenter notes that Airbus has issued Service Bulletin A310-28-2146, dated March 27, 2001. That service bulletin states

that, once the actions therein are accomplished, it cancels the inspection requirements of Airbus Service Bulletin A310-28-2138, dated June 28, 2000. (The proposed rule refers to that service bulletin as the appropriate source of service information.)

The FAA concurs. The Direction Générale de l'Aviation Civile (which is the airworthiness authority for France) has approved, and Airbus has recommended accomplishment of, Service Bulletin A310-28-2146, which describes procedures for replacement of metallic vapor seal panels with new, thicker metallic vapor seal panels. Such replacement raises the current fatigue life limitation on the metallic vapor seals and eliminates the need for the inspections required by this AD. Therefore, the FAA has revised this final rule to add a new paragraph (c) (and reorder subsequent paragraphs accordingly) to give operators the option to do the actions in that service bulletin as terminating action for the repetitive inspections required by this AD. Also, the FAA has added a new paragraph to the Cost Impact section in the preamble of this final rule to provide an estimate of the cost of this terminating action should an operator elect to do it.

Remove Reporting Requirement

The commenter requests that the FAA remove the reporting requirement that is specified in Airbus Service Bulletin A310-28-2138, dated June 28, 2000. The commenter states that the airplane manufacturer should already have adequate sampling data to understand the condition of the fleet, and, therefore, the reporting requirement is an unnecessary burden to the operator.

The FAA concurs with the intent of the commenter's request. However, the reporting requirement to which the commenter refers is not included in this AD, and the FAA cannot revise the referenced service bulletin. No change to the final rule is necessary in this regard.

Extend Repetitive Interval

The commenter requests that the FAA extend the repetitive interval for the repetitive inspections in paragraph (a) of the proposed AD from 600 to 750 flight hours. The commenter notes that its "B"-check interval is 350 flight hours, and the proposed 600-flight-hour interval would not allow for the proposed inspections to be done at a "2B"-check. Thus, it would not be able to do the inspections at a normal scheduled maintenance visit, which would negatively affect scheduling and increase the cost of the requirements of the proposed AD for the operator.