

interval specified in paragraph 2.D. of the Accomplishment Instructions of Airbus Industrie Service Bulletin A300-54-6011, Revision 1, dated October 15, 1993.

New Requirements of This AD

Model A310 Series Airplanes

(c) For Model A310-221, -222, -322, -324, and -325 series airplanes: Perform an internal eddy current inspection to detect cracks in the lower spar axis of the pylon between ribs 9 and 10, in accordance with Airbus Industrie Service Bulletin A310-54-2016, dated November 12, 1991; or Revision 1, dated October 15, 1993; or Revision 02, dated June 11, 1999; at the time specified in paragraph (d) of this AD.

(1) If no crack is found, repeat the inspection thereafter at intervals not to exceed 2,500 landings.

(2) If any crack is found that is less than or equal to 30 mm: Perform subsequent inspections and repair in accordance with the methods and times specified in the service bulletin.

(3) If any crack is found that is greater than 30 mm, but less than 100 mm: Prior to the accumulation of 250 landings after crack discovery, repair in accordance with a method approved by the Manager, International Branch, ANM-116; or the DGAC (or its delegated agent).

(4) If any crack is found that is greater than or equal to 100 mm: Prior to further flight, repair in accordance with a method approved

by the Manager, International Branch, ANM-116; or the DGAC (or its delegated agent).

(5) Accomplishment of the modification specified in Airbus Industrie Service Bulletin A310-54-2022, dated October 15, 1993; or Revision 01, dated March 16, 1999; increases the threshold and repetitive interval of the inspections required by paragraph (c) of this AD to the threshold and interval specified in paragraph 2.D. of the Accomplishment Instructions of Airbus Industrie Service Bulletin A310-54-2016, Revision 02, dated June 11, 1999.

(d) Perform the initial inspection required by paragraph (c) of this AD at the earlier of the times specified by paragraphs (d)(1) and (d)(2) of this AD.

(1) Prior to the accumulation of 25,000 total landings, or within 500 landings after June 12, 1995, whichever occurs later.

(2) At the applicable time specified by paragraph (d)(2)(i), (d)(2)(ii), or (d)(2)(iii) of this AD.

(i) For airplanes that have accumulated fewer than 10,000 landings as of the effective date of this AD: Perform the inspection prior to the accumulation of 3,800 total landings, or within 1,500 landings after the effective date of this AD, whichever occurs later.

(ii) For airplanes that have accumulated 10,000 total landings or more, but fewer than 20,000 total landings, as of the effective date of this AD: Perform the inspection within 1,000 landings after the effective date of this AD.

(iii) For airplanes that have accumulated 20,000 total landings or more as of the

effective date of this AD: Perform the inspection within 500 landings after the effective date of this AD.

Alternative Methods of Compliance

(e) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM-116. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM-116.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM-116.

Special Flight Permits

(f) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Incorporation by Reference

(g) Except as provided by paragraphs (a)(3), (a)(4), (b)(3), (b)(4), (c)(3), and (c)(4) of this AD, the actions shall be done in accordance with the following Airbus Industrie service bulletins, as applicable.

Airbus Industrie Service Bulletin No.	Revision Level	Service Bulletin Date
A300-54-071	Original	November 12, 1991.
A300-54-071	1	October 15, 1993.
A300-54-0079	Original	October 15, 1993.
A300-54-6011	Original	November 12, 1991.
Change Notice O.A. A300-54-6011	Original	July 10, 1992.
A300-54-6011	1	October 15, 1993.
A300-54-6019	Original	October 15, 1993.
A310-54-2016	Original	November 12, 1991.
A310-54-2016	1	October 15, 1993.
A310-54-2022	Original	October 15, 1993.
A310-54-2022	01	March 16, 1999.
A310-54-2016	02	June 11, 1999.

(1) The incorporation by reference of Airbus Industrie Service Bulletin A310-54-2016, Revision 02, dated June 11, 1999; and A310-54-2022, Revision 1, dated March 16, 1999, is approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51.

(2) The incorporation by reference of the remaining Airbus Industrie publications was approved previously by the Director of the Federal Register as of June 12, 1995 (60 FR 25604, May 12, 1995).

(3) Copies may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Note 3: The subject of this AD is addressed in French airworthiness directive 1999-237-285(B), dated June 2, 1999.

(h) This amendment becomes effective on July 28, 2000.

Issued in Renton, Washington, on June 9, 2000.

Donald L. Riffin,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.
[FR Doc. 00-15185 Filed 6-22-00; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 99-SW-37-AD; Amendment 39-11787; AD 2000-12-09]

RIN 2120-AA64

Airworthiness Directives; Sikorsky Aircraft Corporation (Sikorsky) Model S-76A Helicopters

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD) for

Sikorsky Model S-76A helicopters. This AD requires inspecting the air conditioning system at specified intervals until installing a soft-start assembly retrofit kit to prevent a continuous flow of current through the soft-start resistor. This amendment is prompted by a report of overheating of the soft-start assembly. The actions specified by this AD are intended to prevent overheating of the air conditioning soft-start assembly, damage in the lower tailcone, an electrical fire, and subsequent loss of control of the helicopter.

DATES: Effective July 28, 2000.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of July 28, 2000.

ADDRESSES: The service information referenced in this AD may be obtained from Sikorsky Aircraft Corporation, Attn: Manager, Commercial Tech Support, 6900 Main Street, P. O. Box 9729, Stratford, Connecticut 06497-9129, phone (203) 386-7860, fax (203) 386-4703. This information may be examined at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT:

Terry Fahr, Aviation Safety Engineer, Boston Aircraft Certification Office, 12 New England Executive Park, Burlington, MA 01803, telephone (781) 238-7155, fax (781) 238-7199.

SUPPLEMENTARY INFORMATION:

A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an AD for Sikorsky Model S-76A helicopters was published in the **Federal Register** on March 22, 2000 (65 FR 15280). That action proposed to require inspecting the soft-start assembly at intervals not to exceed 25 hours time-in-service until installing a soft-start assembly retrofit kit on the Aero Aire Air Conditioning System in 120 calendar days to prevent a continuous flow of current through the soft-start resistor.

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were received on the proposal or the FAA's determination of the cost to the public. The FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

The FAA estimates that 9 helicopters of U.S. registry will be affected by this AD, that it will take approximately 3

work hours per helicopter to accomplish the required actions, and that the average labor rate is \$60 per work hour. Aero Aire Service Bulletin No. 97002 states that the retrofit kit will be provided at no charge. Based on these figures, the total cost impact of the AD on U.S. operators is estimated to be \$1,620.

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding a new airworthiness directive to read as follows:

2000-12-09 Sikorsky Aircraft Corporation: Amendment 39-11787. Docket No. 99-SW-37-AD.

Applicability: Model S-76A helicopters with Aero Aire Air Conditioning System, part number (P/N) S-76A-1-2, modified in accordance with Supplemental Type Certificate SH4680SW, certificated in any category.

Note 1: This AD applies to each helicopter identified in the preceding applicability provision, regardless of whether it has been otherwise modified, altered, or repaired in the area subject to the requirements of this AD. For helicopters that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (c) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent overheating of the air conditioning soft-start control assembly, damage in the lower tailcone, a fire, and subsequent loss of control of the helicopter, accomplish the following:

(a) Within 25 hours time-in-service (TIS) and thereafter at intervals not to exceed 25 hours TIS, inspect the soft-start control assembly in accordance with the Accomplishment Instruction, Section III, of Aero Aire Corporation Service Bulletin No. 970001, Revision A, dated September 18, 1997, except neither contact nor return of the soft-start controller unit is required.

(b) Within 120 calendar days, install a soft start assembly retrofit kit (kit), P/N 76SB001, in accordance with the Accomplishment Instructions, Section III, of Aero Aire Corporation Service Bulletin 970002, dated December 18, 1997. Installing the kit is terminating action for the requirements of this AD.

(c) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Boston Aircraft Certification Office, FAA. Operators shall submit their requests through an FAA Principal Maintenance Inspector, who may concur or comment and then send it to the Manager, Boston Aircraft Certification Office.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Boston Aircraft Certification Office.

(d) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the helicopter to a location where the requirements of this AD can be accomplished.

(e) The inspection shall be done in accordance with the Accomplishment Instruction, Section III, of Aero Aire Corporation Service Bulletin No. 970001, Revision A, dated September 18, 1997. The modification shall be done in accordance with the Accomplishment Instructions, Section III, of Aero Aire Corporation Service Bulletin 970002, dated December 18, 1997. This incorporation by reference was approved by the Director of the **Federal Register** in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Sikorsky Aircraft Corporation, Attn: Manager, Commercial Tech Support, 6900

Main Street, P. O. Box 9729, Stratford, Connecticut 06497-9129, phone (203) 386-7860, fax (203) 386-4703. Copies may be inspected at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

(f) This amendment becomes effective on July 28, 2000.

Issued in Fort Worth, Texas, on June 8, 2000.

Larry M. Kelly,

Acting Manager, Rotorcraft Directorate,
Aircraft Certification Service.

[FR Doc. 00-15309 Filed 6-22-00; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 99-NM-330-AD; Amendment 39-11797; AD 2000-12-19]

RIN 2120-AA64

Airworthiness Directives; Boeing Model 747 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to certain Boeing Model 747 series airplanes, that requires repetitive inspections of the aft pressure bulkhead to detect cracking, and repair, if necessary. This amendment is prompted by a report of fatigue cracking found in the upper half of the aft pressure bulkhead. The actions specified by this AD are intended to detect and correct cracking in the aft pressure bulkhead, which could result in rapid decompression of the fuselage or overpressurization of the tail section.

DATES: Effective July 28, 2000.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of July 28, 2000.

ADDRESSES: The service information referenced in this AD may be obtained from Boeing Commercial Airplane Group, P.O. Box 3707, Seattle, Washington 98124-2207. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Rick Kawaguchi, Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Transport Airplane Directorate, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-1153; fax (425) 227-1181.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to certain Boeing Model 747 series airplanes was published in the **Federal Register** on February 2, 2000 (65 FR 4900). That action proposed to require repetitive inspections of the aft pressure bulkhead to detect cracking, and repair, if necessary.

Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due consideration has been given to the single comment received.

The commenter supports the proposed rule.

Conclusion

After careful review of the available data, including the comment noted above, the FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

Cost Impact

There are approximately 552 airplanes of the affected design in the worldwide fleet. The FAA estimates that 84 airplanes of U.S. registry will be affected by this AD.

It will take approximately 7 work hours per airplane to accomplish the required detailed visual inspection, at the average labor rate of \$60 per work hour. Based on these figures, the cost impact of the required detailed visual inspection on U.S. operators is estimated to be \$35,280, or \$420 per airplane, per inspection cycle.

It will take approximately 7 work hours per airplane to accomplish the required HFEC inspections, at the average labor rate of \$60 per work hour. Based on these figures, the cost impact of the required HFEC inspections on U.S. operators is estimated to be \$35,280, or \$420 per airplane, per inspection cycle.

The cost impact figures discussed above are based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted.

Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action: (1) Is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

2000-12-19 Boeing: Amendment 39-11797. Docket 99-NM-330-AD.

Applicability: Model 747 series airplanes, as listed in Boeing Alert Service Bulletin 747-53A2425, dated October 29, 1998; certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in