

of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (d) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent uncommanded movement of the trimmable horizontal stabilizer (THS), which could result in reduced controllability of the airplane, accomplish the following:

Restatement of Requirements of AD 98-01-15

(a) Within 500 flight hours after January 28, 1998 (the effective date of AD 98-01-15, amendment 39-10277), perform an operational test of the THS override mechanism to determine if the override system functions correctly, in accordance with paragraph (a)(1) or (a)(2) of this AD, as applicable. Repeat the operational test thereafter at intervals not to exceed 500 flight hours.

(1) For Model A330 series airplanes: Perform the test in accordance with Airbus Service Bulletin A330-27-3051, dated February 13, 1997; and, prior to further flight, repair any discrepancy in accordance with this service bulletin.

(2) For Model A340 series airplanes: Perform the test in accordance with Airbus Service Bulletin A340-27-4058, dated February 13, 1997; and, prior to further flight, repair any discrepancy in accordance with this service bulletin.

New Requirements of This AD

(b) Within 15 months after the effective date of this AD, accomplish the actions specified by either paragraph (b)(1) or paragraph (b)(2) of this AD, in accordance with Airbus Service Bulletin A330-27-3056, Revision 01, dated May 5, 1998 (for Model A330 series airplanes), or Service Bulletin A340-27-4061, Revision 02, dated May 5, 1998 (for Model A340 series airplanes); as applicable.

(1) Replace three Flight Control Primary Computers (FCPC) (2CE1, 2CE2, and 2CE3), P/N LA2K01500190000, with new FCPCs, P/N LA2K01500210000; in accordance with the applicable service bulletin. Such replacement constitutes terminating action for the requirements of paragraph (a) of this AD.

(2) Replace the on-board replaceable module (OBRM) of the three FCPCs (2CE1, 2CE2, and 2CE3), P/N LA2K01500190000, with OBRMs that have been modified by converting FCPC P/N's to LA2K01500210000 in accordance with the applicable service bulletin. Such replacement constitutes terminating action for the requirements of paragraph (a) of this AD.

Spares

(c) As of the effective date of this AD, no person shall install on any airplane an FCPC, P/N LA2K01500190000.

Alternative Methods of Compliance

(d) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Operations Inspector or Principal Avionics Inspector or Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM-116.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM-116.

Special Flight Permits

(e) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Incorporation by Reference

(f) The actions shall be done in accordance with Airbus Service Bulletin A330-27-3051, dated February 13, 1997; Airbus Service Bulletin A340-27-4058, dated February 13, 1997; Airbus Service Bulletin A330-27-3056, Revision 01, dated May 5, 1998; or Airbus Service Bulletin A340-27-4061, Revision 02, dated May 5, 1998; as applicable.

(1) The incorporation by reference of Airbus Service Bulletin A330-27-3056, Revision 01, dated May 5, 1998; and Airbus Service Bulletin A340-27-4061, Revision 02, dated May 5, 1998; is approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51.

(2) The incorporation by reference of Airbus Service Bulletin A330-27-3051, dated February 13, 1997; and Airbus Service Bulletin A340-27-4058, dated February 13, 1997; was approved previously by the Director of the Federal Register as of January 28, 1998 (63 FR 1909, January 13, 1998).

(3) Copies may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Note 3: The subject of this AD is addressed in French airworthiness directives 98-124-069(B) (for Model A330 series airplanes) and 98-126-085(B) (for Model A340 series airplanes), both dated March 11, 1998.

(g) This amendment becomes effective on May 4, 2000.

Issued in Renton, Washington, on March 20, 2000.

Donald L. Riggins,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 00-7334 Filed 3-29-00; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 99-SW-75-AD; Amendment 39-11651; AD 2000-06-10]

RIN 2120-AA64

Airworthiness Directives; Bell Helicopter Textron Canada Model 407 Helicopters

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule; request for comments.

SUMMARY: This amendment adopts a new airworthiness directive (AD) applicable to Bell Helicopter Textron Canada (BHTC) Model 407 helicopters. This action requires preflight checking and repetitively inspecting the tail boom for a crack and replacing the tail boom if a crack is found. This amendment is prompted by four reports of cracks on the tail boom in the area of the horizontal stabilizer. The actions specified in this AD are intended to prevent separation of the tail boom and subsequent loss of control of the helicopter.

DATES: Effective April 14, 2000.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of April 14, 2000.

Comments for inclusion in the Rules Docket must be received on or before May 30, 2000.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Office of the Regional Counsel, Southwest Region, Attention: Rules Docket No. 99-SW-75-AD, 2601 Meacham Blvd., Room 663, Fort Worth, Texas 76137.

The service information referenced in this AD may be obtained from Bell Helicopter Textron Canada, 12,800 Rue de l'Avenir, Mirabel, Quebec JON1LO, telephone (800) 463-3036, fax (514) 433-0272. This information may be examined at the FAA, Office of the Regional Counsel, Southwest Region, Room 663, Fort Worth, Texas or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Sharon Miles, Aerospace Engineer, FAA, Rotorcraft Directorate, Regulations Group, 2601 Meacham Blvd., Fort Worth, Texas 76137, telephone (817) 222-5122, fax (817) 222-5961.

SUPPLEMENTARY INFORMATION: Transport Canada, the airworthiness authority for Canada, notified the FAA that an unsafe condition may exist on the BHTC Model 407 helicopters. Transport Canada advises that there have been several reports of cracks to the tail boom skin in the area of the horizontal stabilizer.

BHTC has issued Alert Service Bulletin 407-99-26, dated April 13, 1999 (ASB), which specifies a preflight check of the left-side of the tail boom before the next flight and before the first flight of every day thereafter. The ASB also specifies within the next 25 hours time-in-service (TIS) and thereafter every 50 hours inspecting any tail boom that has accumulated 600 or more hours TIS for a crack and replacing any cracked tail boom before further flight. Transport Canada classified this ASB as mandatory and issued AD CF-99-17, dated June 14, 1999, to ensure the continued airworthiness of these helicopters in Canada.

This helicopter model is manufactured in Canada and is type certificated for operation in the United States under the provisions of section 21.29 of the Federal Aviation Regulations (14 CFR 21.29) and the applicable bilateral airworthiness agreement. Pursuant to this bilateral airworthiness agreement, Transport Canada has kept the FAA informed of the situation described above. The FAA has examined the findings of Transport Canada, reviewed all available information, and determined that AD action is necessary for products of this type design that are certificated for operation in the United States.

Since an unsafe condition has been identified that is likely to exist or develop on other BHTC Model 407 helicopters of the same type design registered in the United States, this AD is being issued to prevent separation of the tail boom and subsequent loss of control of the helicopter. This AD requires a preflight check of the tail boom before further flight and thereafter before the first flight of each day. This AD also requires within 25 hours TIS and thereafter at intervals not to exceed 50 hours TIS, inspecting any tail boom that has accumulated 600 or more hours TIS for a crack with a 10X or higher magnifying glass and replacing any cracked tail boom with an airworthy tail boom before further flight. The actions are required to be accomplished in accordance with the ASB described previously. The short compliance time involved is required because the previously described critical unsafe condition can adversely affect the controllability and structural integrity of the helicopter. Therefore, checking the

tail boom for a crack is required prior to further flight and this AD must be issued immediately.

An owner/operator (pilot) may perform the visual check required by this AD but must enter compliance with this AD in the aircraft records in accordance with 14 CFR 43.11 and 91.417(a)(2)(v)). This AD allows a pilot to perform this check because it involves only a visual check for a crack in the tail boom and can be performed equally well by a pilot or a mechanic.

Since a situation exists that requires the immediate adoption of this regulation, it is found that notice and opportunity for prior public comment hereon are impracticable, and that good cause exists for making this amendment effective in less than 30 days.

The FAA estimates that 200 helicopters will be affected by this AD, that it will take approximately 4 work hours to accomplish the inspections, and that the average labor rate is \$60 per work hour. Based on these figures, the total cost impact of the AD on U.S. operators is estimated to be \$48,000 assuming no tail boom will be replaced.

Comments Invited

Although this action is in the form of a final rule that involves requirements affecting flight safety and, thus, was not preceded by notice and an opportunity for public comment, comments are invited on this rule. Interested persons are invited to comment on this rule by submitting such written data, views, or arguments as they may desire. Communications should identify the Rules Docket number and be submitted in triplicate to the address specified under the caption **ADDRESSES**. All communications received on or before the closing date for comments will be considered, and this rule may be amended in light of the comments received. Factual information that supports the commenter's ideas and suggestions is extremely helpful in evaluating the effectiveness of the AD action and determining whether additional rulemaking action would be needed.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the rule that might suggest a need to modify the rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report that summarizes each FAA-public contact concerned with the substance of this AD will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments

submitted in response to this rule must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket No. 99-SW-75-AD." The postcard will be date stamped and returned to the commenter.

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

The FAA has determined that this regulation is an emergency regulation that must be issued immediately to correct an unsafe condition in aircraft, and that it is not a "significant regulatory action" under Executive Order 12866. It has been determined further that this action involves an emergency regulation under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979). If it is determined that this emergency regulation otherwise would be significant under DOT Regulatory Policies and Procedures, a final regulatory evaluation will be prepared and placed in the Rules Docket. A copy of it, if filed, may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding a new airworthiness directive to read as follows:

AD 2000-06-10 Bell Helicopter Textron

Canada: Amendment 39-11651. Docket No. 99-SW-75-AD.

Applicability: Model 407 helicopters, serial numbers 53000 through 53003, 53005 and higher, certificated in any category.

Note 1: This AD applies to each helicopter identified in the preceding applicability provision, regardless of whether it has been otherwise modified, altered, or repaired in the area subject to the requirements of this AD. For helicopters that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (d) of this AD.

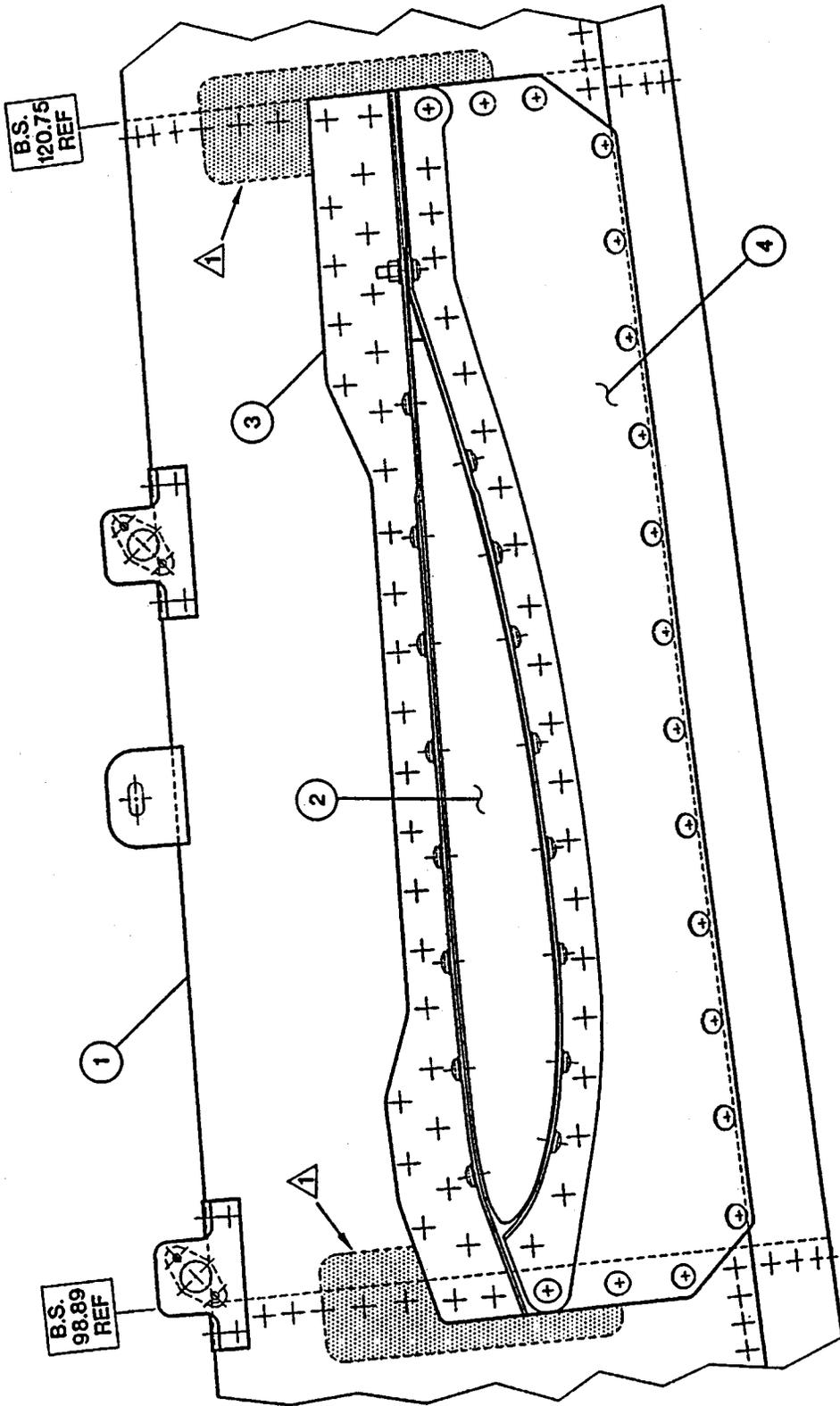
The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent separation of the tail boom and subsequent loss of control of the helicopter, accomplish the following:

(a) Before further flight and thereafter before the first flight of each day, check the left side of the tail boom for a crack in the areas shown in Figure 1. If a crack is found, replace the tail boom with an airworthy tail boom before further flight.

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LEGEND

- 1. Tailboom assembly (Ref.)
- 2. Horizontal stabilizer (Ref.)
- 3. Upper support (Ref.)
- 4. Lower support (407-023-800-121)

NOTES

- 1. Examine these areas for cracks on left side of tailboom only.
- 2. Horizontal stabilizer not shown for clarity.

Figure 1. Preflight Check of the Tailboom

(b) An owner/operator (pilot) holding at least a private pilot certificate may perform the visual check required by paragraph (a) but must enter compliance with paragraph (a) into the aircraft records in accordance with 14 CFR 43.11 and 91.417(a)(2)(v)).

(c) Within 25 hours time-in-service (TIS) and thereafter at intervals not to exceed 50 hours TIS, visually inspect any tail boom with 600 or more hours TIS for a crack using a 10X or higher magnifying glass, in accordance with the Accomplishment Instructions, Part II, of Bell Helicopter Textron Canada Alert Service Bulletin 407-99-26, dated April 13, 1999, except that you are not required to contact Bell Helicopter Product Support Engineering. If a crack is found, replace the tail boom with an airworthy tail boom before further flight.

(d) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Regulations Group, Rotorcraft Directorate, FAA. Operators shall submit their requests through an FAA Principal Maintenance Inspector, who may concur or comment and then send it to the Manager, Regulations Group.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Regulations Group.

(e) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the helicopter to a location where the requirements of this AD can be accomplished.

(f) The inspection of the tail boom shall be done in accordance with the Accomplishment Instructions, Part II, of Bell Helicopter Textron Canada Alert Service Bulletin 407-99-26, dated April 13, 1999. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Bell Helicopter Textron Canada, 12,800 Rue de l'Avenir, Mirabel, Quebec JON1LO, telephone (800) 463-3036, fax (514) 433-0272. Copies may be inspected at the FAA, Office of the Regional Counsel, Southwest Region, Room 663, Fort Worth, Texas, or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

(g) This amendment becomes effective on April 14, 2000.

Note 3: The subject of this AD is addressed in Transport Canada (Canada) AD CF-99-17, dated June 14, 1999.

Issued in Fort Worth, Texas, on March 21, 2000.

Henry A. Armstrong,

Manager, Rotorcraft Directorate, Aircraft Certification Service.

[FR Doc. 00-7552 Filed 3-29-00; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 71

[Airspace Docket No. 99-ACE-49]

Amendment to Class E Airspace; Cameron, MO

AGENCY: Federal Aviation Administration, DOT.

ACTION: Direct final rule; confirmation of effective date.

SUMMARY: This document confirms the effective date of a direct final rule which revises Class E airspace at Cameron, MO.

DATES: The direct final rule published at 64 FR 72925 is effective on 0901 UTC, April 20, 2000.

FOR FURTHER INFORMATION CONTACT: Kathy Randolph, Air Traffic Division, Airspace Branch, ACE-520C, DOT Regional Headquarters Building, Federal Aviation Administration, 901 Locust, Kansas City, MO 64106; telephone: (816) 329-2525.

SUPPLEMENTARY INFORMATION: The FAA published this direct final rule with a request for comments in the **Federal Register** on December 29, 1999 (64 FR 72925). The FAA uses the direct final rulemaking procedure for a non-controversial rule where the FAA believes that there will be no adverse public comment. This direct final rule advised the public that no adverse comments were anticipated, and that unless a written adverse comment, or a written notice of intent to submit such an adverse comment, were received within the comment period, the regulation would become effective on April 20, 2000. No adverse comments were received, and thus this notice confirms that this direct final rule will become effective on that date.

Issued in Kansas City, MO on March 24, 2000.

Herman J. Lyons, Jr.,

Manager, Air Traffic Division, Central Region.

[FR Doc. 00-7856 Filed 3-29-00; 8:45 am]

BILLING CODE 4910-13-M

DEPARTMENT OF TRANSPORTATION

Office of the Secretary

14 CFR Part 255

[Docket No. OST-2000-6984]

RIN 2105-AC75

Third Extension of Computer Reservations Systems (CRS) Regulations

AGENCY: Office of the Secretary, Department of Transportation.

ACTION: Final rule.

SUMMARY: The Department is revising its rules governing airline computer reservations systems (CRSs), 14 CFR part 255, to change the rules' expiration date for a third time. This revision changes the date from March 31, 2000, to March 31, 2001, to keep the rules from terminating on March 31, 2000. The rules will thus remain in effect while the Department continues its reexamination of the need for CRS regulations. The Department finds that the current rules should be maintained because they are necessary for promoting airline competition and helping to ensure that consumers and their travel agents can obtain complete and accurate information on airline services. The Department previously extended the rules from December 31, 1997, to March 31, 1999, and from March 31, 1999, to March 31, 2000.

DATES: This rule is effective on March 31, 2000.

FOR FURTHER INFORMATION CONTACT: Thomas Ray, Office of the General Counsel, Department of Transportation, 400 Seventh St. SW., Washington, DC 20590, (202) 366-4731.

SUPPLEMENTARY INFORMATION: To ensure that we periodically review the need for our CRS rules and their effectiveness, section 255.12 of the rules establishes a sunset date. The original sunset date was December 31, 1997. We have changed the rules' expiration date twice before, once to March 31, 1999, 62 FR 66272 (December 18, 1997), and then to March 31, 2000, 64 FR 15127 (March 30, 1999).

We are now changing the sunset date to March 31, 2001, because we have been unable to complete our reexamination of the current rules by March 31, 2000. Given our view that the current rules should be maintained pending our reexamination of the need for rules, we proposed to change the rules' expiration date to March 31, 2001, and gave interested persons an opportunity to comment on that proposal. 65 FR 11009 (March 1, 2000).